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Volume 1 of 2

RESULTS OF THE AFRSI DETAILED-ENVIRONMENT TEST OF THE 0.035-SCALE SSV PRESSURE-LOADS MODEL 84-0 IN THE AMES 11x11 FT. TWT AND THE LEWIS 8x6 FT. AND 10x10 FT. SWT (OA-310A, B, C)

SPACE SHUTTLE AEROTHERMODYNAMIC DATA REPORT

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Data Management Services



DMS-DR-2459 NASA-CR 167,685

Volume 1 of 2

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TEST OF THE 0.035-SCALE SSV PRESSURE-LOADS
MODEL 84-0 IN THE AMES 11x11 FT. TWT AND THE
LEWIS 8x6 FT. AND 10x10 FT. SWT
(OA-310A, B, C)

Ъy

B. A. Marshall and J. Marroquin Rockwell International Space Transportation Systems Division

Prepared under NASA Contract Number NAS9-16283

bу

Data Management Services
Chrysler Military-Public Electronic Systems
Michoud Engineering Office
New Orleans, Louisiana 70189

for

Systems Engineering Division

Johnson Space Center National Aeronautics and Space Administration Houston, Texas

N84-31261#

WIND TUNNEL TEST SPECIFICS:

Test Number: ARC 587-1-11 LeRC 046 LeRC 074 Tunnel: llxll-foot 8x6-foot 10x10-foot NASA Series Number: OA-310A OA-310B OA-310C Model Number: 84-0 84-0 84-0

Test Dates: 8-8-83 through 9-12-83 through 11-7-83 through 8-18-83 11-15-83 9-22-83

Occupancy Hours: 144 56 96

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Concurrence

Chrysler Military-Public Electronic Systems/Michoud Engineering Office assumes no responsibility for the data presented other than display characteristics.

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ABSTRACT

Detailed orbiter aerodynamic and aeroacoustic pressure data were obtained in a three-part experimental investigation (OA-310A, B and C) which was conducted during the period from August to November, 1983. Test OA-310A, B and C was conducted in three NASA facilities: OA-310A in the Ames 11x11-foot Transonic Wind Tunnel; OA-310B in the Lewis 8x6-foot Supersonic Wind Tunnel; and OA-310C in the Lewis 10x10-foot Supersonic Wind Tunnel. Test data were obtained to support analysis of the Space Transportation System (STS) -6 Advanced Flexible Reusable Surface Insulation (AFRSI) anomaly using the 0.035-scale Space Shuttle vehicle pressure-loads Model 84-0.

During Test OA-310A, B and C, data were obtained for detailed orbiter aerodynamic and aeroacoustic environments in the areas of the orbiter where AFRSI is to be applied to OV-099 and OV-103. Emphasis was placed on acquiring detailed aeroacoustic data and time-averaged pressure distributions on five affected areas: (1) canopy; (2) side of fuselage; (3) upper surface of wing; (4) OMS pods; and (5) vertical tail. Data were obtained at nominal ascent and entry atmospheric flight trajectory conditions between M=0.6 through M=3.5.

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SCHEDULE	COEFFICIENTS PLOTTED
A	Cp VS x/IB
8	Cp VS x/cv
С	Cp VS x/cW

INTRODUCTION

Advanced Flexible Reusable Surface Insulation (AFRSI) is presently being used as a replacement for most of the Low-Temperature Reusable Surface Insulation (LRSI) tiles on the Space Shuttle Orbiter Vehicle. The AFRSI is a quilted blanket consisting of silica fiber felt insulation material with a quartz fabric ONL cover and a glass fabric IML lining. The quilting is done with quartz thread stitched through the three layers of material. The blanket IML is bonded to the skin of the vehicle while the OML face is exposed to the high pressure gradients, the fluctuating acoustic pressures, and the wind shear stresses attendant during entry into the atmosphere. The blankets are very flexible and susceptible to damage due to the hardness and brittleness of the individual fibrous elements.

The purpose of Test OA-310A, B, and C was to obtain data to support analysis of the STS-6 AFRSI anomaly using the 0.035-scale Space Shuttle Vehicle pressure-loads Model 84-0. Data were obtained for detailed orbiter aerodynamic and aero-acoustic environments in the areas of the orbiter where AFRSI is to be applied to OV-099 and OV-103. Emphasis was placed on acquiring detailed aeroacoustic data and time-averaged pressure distributions on five affected areas:

(1) canopy; (2) side of fuselage; (3) upper surface of wing; (4) OMS pods; and (5) vertical tail.

Data were obtained at nominal ascent and entry atmospheric flight trajectory conditions between M=0.6 through M=3.5. Also, model angles of attack, sideslip angles, rudder, speedbrake, and elevon deflections were varied. No internal balance was used during Test OA-310A, B, and C; however, the sting was gaged for deflection data during Test OA-310A.

INTRODUCTION (Concluded)

Test OA-310A was conducted in the NASA/Ames Research Center (ARC) 11x11-foot Transonic Wind Tunnel. Test OA-310B was conducted in the NASA/Lewis Research Center (LeRC) 8x6-foot Supersonic Wind Tunnel and Test OA-310C was conducted in the LeRC 10x10-foot Supersonic Wind Tunnel.

This report contains information on the conduct of Test OA-310A, B, and C and descriptions of the test facilities and instrumentation. Photographs of the 0.035-scale Space Shuttle Vehicle pressure-loads Model 84-O are included. In addition, static pressure data are tabulated and sample plotted data are presented.

NOMENCLATURE

Symbol	Mnemonic	Definition
Ср	СР	Pressure Coefficient
dB		Volume of Sound (decibel)
٥F		Degrees Fahrenheit
ft		Feet
in.	INCHES	Inches
M	MACH	Freestream Mach Number
N	ETA	Percent Span
P	P	Freestream static pressure, psia
PHI	PHI	Angular location measured clockwise from bottom of fuse lage, degrees
PL		Local static pressure, psia
PRMS	-	Root Mean Square (RMS) pressure in psia
psf		Pounds per square foot
psia		Absolute pressure in pounds per square inch
Pt	PT	Freestream total pressure, psf
Q, q	Q(PSF)	Freestream dynamic pressure, psf
°R		Degrees Rankine
	RN/L	Unit Reynolds number, million per ft.
sq ft	SQ.FT.	Square feet
Х		Model-scale station
X/C		Percent chord (local)
X/Cy	X/CV	Chordwise location on vertical tail, fraction of local chord
X/ _{CW}	X/CW	Chordwise location on wing surface, fraction of local chord
X/ _{LB}	X/LB	Longitudinal location of orbiter body surface, fraction of body length

NOMENCLATURE (Continued)

Symbo1	Mnemonic	<u>Definition</u>
XO		Full-scale station
Υ		Model-scale buttplane
YO		Full-scale buttplane
Z		Model-scale waterplane
ZO		Full-scale waterplane
α	ALPHA	Model angle of attack, degrees
β	BETA	Model sideslip angle, decrees .
$^{\delta}$ BF	BDFLAP	Model body flap deflection angle, degrees
$\delta_{f e}$	ELEVON	Model elevon deflection angle, degrees
$^{\delta}$ e $_{ m I}$	IB-ELV	Model inboard elevon deflection angle, degrees
$^{\delta}$ e $_{0}$	OB-ELV	Model outboard elevon deflection angle, degrees
δ _R	RUDDER	Model rudder deflection angle, degrees
δSB	SPDBRK	Model speed brake deflection angle, degrees
%		Percent
	SREF	Wing reference area, ft ²
	LREF	Reference length, inches
	BREF	Wing reference span, inches
	XMRP,YMRP, ZMRP	Location of the moment reference point in the Orbiter coordinate system, inches

NOMENCLATURE (Concluded)

Other Symbology Includes:

STS

Symbol	Definition
AFRSI	Advanced Flexible Reusable Surface Insulation
ARC	Ames Research Center
ESP	Electro Scan Pressure
IML	Inner Mold Line
KULI	Kulite
LeRC	Lewis Research Center
LRSI	Low-Temperature Reusable Surface Insulation
NA	Not Applicable
NASA	National Aeronautics and Space Administration
No., NUMB, #	Number
OMIL	· Outer Mold Line
OML OMS	······································
	· Outer Mold Line
OMS	· Outer Mold Line Orbiter Maneuvering System
OMS ORIF	Orbiter Maneuvering System Orifice
OMS ORIF OV	Orbiter Maneuvering System Orifice Orbiter Vehicle

Space Transportation System

REMARKS

Prior to run 4 of Test OA-310A, static pressure orifices numbered 345 and 369 were determined to be plugged. It should also be noted that prior to run 36, static pressure orifices numbered 119, 120, 142, 143, 144, 170, 171, 172, 221, 222, 240, 345, 488, and 512 were deleted from the data printout because they were not producing good data.

Kulites numbered K14, K24, K54, K65, K98, K103, K104, K105, K106, and K108 did not produce usable data during Test OA-310A. No Kulite data were obtained during run 6.

During Test OA-310B, not all test objectives were met. Airloads and aeronoise data from Mach numbers 1.4 through 2.0 were expected for this test. However, data were acquired for only Mach 1.4 and 1.6. Due to a malfunction of the Lewis 8x6-ft tunnel's number 2 drive motor, no data were obtained at Mach 1.8 or 2.0.

The following pressure taps were omitted from Model 84-O during Test OA-310B: 119, 120, 142, 143, 144, 170, 171, 172, 488, and 512. Tap No. 406 was plugged during Test OA-310B and Tap Nos. 426 and 513 were considered unusable. These three pressure taps were deleted from the data reduction output.

One hundred eight Kulites were mounted in the orbiter Model 84-O. However, only 100 Kulites were able to be recorded during Test OA-310B and C due to channel availability. During Test OA-310B, Kulites numbered K14, K19, K74, K80, K87, K92, K95, K102, K103, and K108 were not recorded. It should also be noted that Kulite K5 responded only intermittently throughout Test OA-310B.

REMARKS (Concluded)

The following pressure taps were omitted from Model 84-O during OA-310C: 119, 120, 142, 143, 144, 170, 171, 172, 488, and 512. The following pressure taps were plugged during Test OA-310C: 147, 406, and 506. Pressure taps numbered 210, 426, and 431 leaked during this test. It should also be noted that pressure tap 306 was found to have a bad leak prior to run 12; therefore, data obtained from tap 306 after run 11 should be considered questionable.

Kulites numbered K12, K16, K19, K21, K24, K27, K31, and K39 were the eight Kulites that were not recorded during OA-310C due to channel availability. However, Kulite K92 was giving bad data during runs 9, 10, and 11 and was replaced with Kulite K16 prior to run 12.

CONFIGURATIONS INVESTIGATED

Model Description

The model tested during Test OA-310A, B and C was a 0.035-scale model of the Space Shuttle Orbiter Vebicle, designated 84-0 (see Figure 1). The model was designed to the OV102 outer moldline specifications.

All major model components are constructed of aluminum alloy. All stings and supporting hardware are constructed of stainless steel. All load-carrying components are designed to meet the ARC and LeRC maximum facility specified safety factors.

INSTRUMENTATION

The orbiter Model 84-O was supported on sting support hardware compatible with tunnel sting and strut assemblies. During Test OA-310A in the ARC llx11-foot wind tunnel, the W-1144-S-3 sting was attached to the A9758D-125-2 Ames straight sting. It should also be noted that a clinometer was mounted inside Model 84-O only during Test OA-310A. Also, no balance was used during any portion of Test OA-310A, B and C.

During Test OA-310A, the orbiter Model 84-O was instrumented with 337 static pressure orifices. The locations of these orifices are shown in Table III. These steady-state pressures were measured utilizing eight of twelve S-type Scanivalve modules on two drive assemblies. Rockwell provided the Scanivalves, the Scanivalve drives, and the pressure transducers required. The drive assemblies were mounted in the model.

One hundred and ten high-frequency low-temperature (250°F) differential pressure transducers (Kulites) were mounted in selected locations as shown in Table IV. Rockwell Laboratory and Test representatives supported the Kulite measurements with signal conditioning, preamplification, frequency analysis, and recording equipment.

Prior to testing at LeRC (OA-310B and C), same modifications were made to the model instrumentation. First, the low-temperature (250°F) Kulites were replaced with high-temperature Kulites compensated to 350°F to accommodate the higher testing temperatures. Also, the model was modified from having an internal Scanivalve system to an external system utilizing steel tubing routed from the model to outside the test section.

INSTRUMENTATION (Continued)

During testing at LeRC, Model 84-O was instrumented with 335 static pressure orifices of which 331 were utilized for data acquisition. These time-averaged pressures were measured using 12 electro scan pressure (ESP) modules. LeRC provided these modules and all electrical installation items necessary for their operation. Rockwell supplied the stainless steel tubing and connections to the pressure taps on the model.

All instrumentation leads and static pressure hardlines were routed externally along the main sting fixture and connected to LeRC's patchboard. The basic static pressure tap locations are as follows:

 Vertical Tail
 =
 35

 Upper Wing
 =
 53

 Elevons
 =
 23

 Forward Fuselage
 =
 21

 Mid-Fuselage
 =
 20

 Canopy
 =
 69

 OMS
 =
 110

 331
 =
 331

Of the one hundred and eight high-frequency high-temperature differential pressure transducers (Kulites) mounted on Model 84-O, only 100 Kulites were able to be recorded due to channel availability during testing at LeRC.

The basic Kulite locations on the model were as follows:

Conopy = 24

Forward Fuselage = 7

Aft Fuselage = 12

Body = 29

Vertical Tail = 10

Wing/Elevon = 26

Thermocouples were used to determine Kulite transducer environmental temperatures for calibration and correction purposes. The six chromel/alumel

INSTRUMENTATION (Concluded)

thermocouples were installed in the vicinities of Kulite numbers 7, 29, 42, 64, 77, and 104.

TEST FACILITY DESCRIPTION

The NASA/Ames 11-foot Transonic Wind Tunnel is the transonic leg of the Ames Unitary facility. It is a closed circuit, single return, continuous flow, variable-density tunnel. The 11x11x22-foot test section is slotted to permit transonic testing. The nozzle has adjustable sidewalls. The tunnel air is driven by a 3-stage axial flow compressor powered by four wound-rotor induction motors. The speed of the motors is varied as necessary to provide the desired Mach number. The motors have a combined output of 180,000 horse-power for continuous operation or 216,000 horsepower for one hour. Tunnel temperature is controlled by aftercoolers and a cooling tower. Four 30,000 cubic-foot storage tanks provide dry air for tunnel pressurization.

The tunnel can be operated at nominal Mach numbers of 0.5 to 1.4, unit Reynolds numbers of 1.7 to 9.4 \times 10⁺⁶ per foot, dynamic pressures of 150 to 2000 (psf), and a total temperature of 540 to 610 ($^{\circ}$ R), respectively. This tunnel is used for force and moment, pressure, internal air flow/inlet, and dynamic-stability tests.

The NASA/Lewis Research Center &x6-ft Supersonic Wind Tunnel is capable of attaining test section flow in the Mach number range from 0.36 to 2.0. The change in Mach number is continuous up to 1.3 and in increments of 0.1 between 1.3 and 2.0. The tunnel may be operated in either of two modes; aerodynamics cycle, or propulsion cycle. During the aerodynamic cycle, the tunnel is operated as a closed system with dry air added only as required to maintain the desired tunnel conditions. This cycle is used primarily for aerodynamic flow studies where contaminants are not introduced into the airstream.

TEST FACILITY DESCRIPTION (Continued)

The test section is 8 ft high and 6 ft wide with parallel side walls for a total length of 23 feet, 6 in. The test section is perforated on four sides. Perforations start 9 ft 1 in. from the upstream end of the test section and extend 14 ft 5 in. downstream. This perforation provides approximately 6 percent porosity; however, this can be reduced and varied along the length of the test section.

Models are installed through an access door in the bottom of the tunnel diffuser downstream of the test section. The opening is 16 ft long by 6 ft wide. Two overhead cranes are provided in the ceiling of the diffuser section. Models on special dollies are lifted into the diffuser section and rolled to the test section for installation.

Sting-mounted models are mounted to the strut which extends through the tunnel floor when supporting a model and retracted below the tunnel floor when not in use. The angle of attack can be remotely varied from 0 degrees to +15 degrees.

Two pair of Schlieren windows are located in the side walls. The 26.5-inch diameter windows are located eight inches off center in a 42.5-inch steel disc which, when rotated, allows the window to cover any portion of the 42.5-inch diameter circle.

The NASA/Lewis Research Center 10x10-foot Unitary Supersonic Wind Tunnel is a closed loop continuous flow facility with a Mach number capability from 2.0 to 3.5 in either an aerodynamic or propulsion circuit. The aerodynamic circuit, used for these investigations, has a stagnation pressure capability

TEST FACILITY DESCRIPTION (Concluded)

of 0.1 to 2.36 atmospheres at a stagnation temperature of $1160^{\circ}F$ giving a Reynolds number capability from 0.2 to 2.6 x $10^{6}/ft$. The dynamic pressure varies from 20 to 720 psf. The propulsion circuit of the tunnel has a stagnation pressure capability of 0.62 to 2.36 atmospheres at a stagnation temperature of $1160^{\circ}F$ for a Reynolds number variation of 2.1 to 2.8 x $10^{6}/ft$ and a dynamic pressure variation of 500 to 600 psf. This circuit can accept either air breathing or rocket engines for testing.

TEST PROCEDURES

During the course of Test OA-310A, B and C, data were recorded at nominal Mach numbers from 0.60 to 3.50. Data were also recorded for an angle of attack range of -6 degrees to 40 degrees and sideslip angles of -4 to +4 degrees.

Nominal entry and ascent pitch and yaw attitudes from previous flights were duplicated during the course of Test OA-310A, B and C. A summary of test conditions and runs completed during Test OA-310A, B and C is shown in Tables I and II, respectively.

DATA REDUCTION

Standard tunnel equations were used for computing all tunnel conditions. Local static-pressure coefficient data were calculated using the following equation.

$$C_{p} = \frac{P_{L} - PX144}{q}$$

Fluctuating pressure data were recorded on magnetic tape and reduced during and after the test.

Local sound pressure levels were calculated as follows:

$$dB = 20 \log \frac{P_{RMS}}{2.94 \times 10^{-9}}$$

REFERENCES

- 1. R. B. Kingsland and M. E. Nichols, STS23-0467, "Pretest Information for AFRSI Detailed-Environments Tests of the 0.035-Scale SSV Pressure-Loads Model 84-0 in the Ames 11-foot Transonic Wind Tunnel and the Lewis 8x6-foot and 10x10-foot Supersonic Wind Tunnels (OA-310)" (July 1983)
- 2. NASA TM X-71542, "NASA/Lewis 8x6-ft Supersonic Wind Tunnel". (May 1974)

TEST : 0A-310A	•		DATE: 8-18-83
	TEST CON	DITIONS	
	·		· · · · · · · · · · · · · · · · · · ·
MACH NÚMBER	Total Pressure	Dynamic Pressure	
MACH NUMBER	(pounds/sq. f+.)	(pounds/sq. ft)	
0.60	2075-4025	410 - 795	
0.80	1395 + 2705		
0.90	1225 - 2370		
0.95	1160 -> 2250		
1.05	1070 72070	*	•
1.10	1360 71665	540 → 660	
1.15	1325 -> 1620		
1.25	1280 + 1565	+	
1.40	1255 -> 2120	540 -> 915	
BALANCE UTILIZED	NA		
	CADACITY	ACCHDACY	COEFFICIENT
	CAPACITY	ACCURACY	TOLERANCE.
NF			
SF			
AF			··
PN			
RM			
YM	·		
COMMENTS			
			}

TABLE I (CONTINUED)

TEST : 0A-310E	3 ·		DATE: 11-15-83
•	TEST CON	IDITIONS	
MACH NÚMBER	Total Pressure (pounds/sq. f+)	Dynamic Pressure (pounds/5q. ft)	
1.40	2505	1080	
1.60	2800	1180	
			•
<u> </u>			
	<u> </u>		
BALANCE UTILIZED	NA	· · · · · · · · · · · · · · · · · · ·	
	CAPACITY.	ACCURACY	COEFFICIENT TOLERANCE.
NF			
SF			
AF			
PM RM			
YM			
COMMENTS			
			1

TABLE I (CONCLUDED)

TEST : 0A-310C	·		DATE: 9-22-83
	TEST CON	IDITIONS	
	I		
MACH NÚMBER	(pounds/sq f+)	Dynamic Pressure (pounds/5q. ft)	
2.00	1120	400	
2.20	1260		
2.50 3.50	1560		
3.50	3560	4	
			,
	<u> </u>		
BALANCE UTILIZED	NA		·
	CADACITY	100110107	COEFFICIENT
	CAPACITY	ACCURACY	TOLERANCE.
NF			
SF			
AF			
PM			
RM			
YM			
COMMENTS.			

\$ SEE TABLE II (CONCLUDED) FOR COMPONENT IDENTIFICATION

ins (CL

4

TABLE II (Cont'd)

ARC TEST : O		DAT	A SE	T/RU	N NU	MBER	DATE 30 Aug 1982									
DATA SET IDENTIFIER	CONFIGURATION	α	M	9	<u> </u>	Sen	S 2	Se	1			7	85	TA	7.	_
RA2 \$19	ovice orbiter	Aı	0,8		_	5	87.2		-				72	73	74	
20	,		0,9										76	78	79	
21			0,95				1	1					80	81	82]
• 22			0.6				25	5					239	240	241	
23			0.8									2	42	243	244]
24			0,9		\bot							2	245	246	247] ;
25			0.95		1		, ,					2	48	249	250	
24			0.6	-	1	 - - - - - - - - - 	55					1	92	194		15
27		╌╂┼┼	<u> </u>	- -		┨-}-			_			_	89	190	191	13
	· · · · · · · · · · · · · · · · · · ·	$-\downarrow \downarrow \downarrow$	0,9		\bot	╂-╁-	 						86	187	188	43
		-	0.95	 -	╂┼	+	 			ļ			183	184	185	┨
30			1.05	1-		╂-┼-	 	┟╧╁╌			 		180	181	182	┨
31	ļ	-HH	11.10	├ -├-	1	╂╌┼╾	╂-┼		-		 		177	178	179	┨
32			1.25	-	╂┼		 			 	ļ		74	175	176	1
33			1.40	1	\bot	+-	$\bot \bot$	1	-		ļ <u> </u>		171	172	173	┨
34			0,6	-	+	 -	} -}-	-5		ļ	 - -			219	220	┨
35		-111	0,8	\perp	╂┼	1	╁╁							216	1	4
36			0,9				1 1	1	<u> </u>		<u> </u>	2	212	213	214	上
1	7 13 19	<u> </u>	25		31		37		43	49	5.5	<u>t 1</u>		۲,) : '
		4 - 4 - 4 - 4 -	-1-1-				LENTS		سسل	 		ISVAR	1 (1)	ISVA	H (S)	N D
a OR SCHED					Ç	OFFFIC	TENTS					IDVAK	(1)	ISVAI	-	,

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	- //	J	DAT	A SE	T/RU	שא או	MBER	COLI	. A T 101	1 SUMM	ARY	DATE				3 of 2
(ONFIL	URATION	OK	M	1	See	San	\\ \\ \\ \\ \\ \\ \\ \\ \\ \\ \\ \\ \\	S -			————		γ	_A	ETA.	4
04101	Orbiter	A,		600	_	5	55	-5						209	210	211
	`		1.05											206	207	208
			1.10											203	204	205
			1,25											200	201	202
			140				1							197	198	199
			0.6				25							227	228	229
			0,8											230	231	232
			0,9											233	234	235
			0,95		1		1	1						236	237	238
			06		-5	-5	55	0						83	84	85
1			0.8											87	108	109
			0.9											104	105	106
			0.95											101	102	103
			1.05	\prod			\prod							98	99	100
		1	1.10						1					95	96	97
				1	1									92	93	94
		1			17	11.			1							90
,		-1+		11	0	0	1		†							135
β	3 19		.35	A.J.	31 1 4	OE FFIC	37	L.A.A.A	43	49	-	:5	61 4 4 4 5 1 7 7		<u>.,</u>	
	CONFIG	CONFIGURATION OVIO2 Orbiter 1 1 13 19	CONFIGURATION CONFIG	CONFIGURATION (CONFIGURATION (CONFIGURATION	CONFIGURATION	CONFIGURATION S	ASIOA ONFIGURATION S M 3 S S S S S S S S S S S S S S S S S	ASIOA DATA SET/RUN NUMBER (ONFIGURATION (C) M 3 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	CONFIGURATION S M 3 S S S S S S S S S	CONFIGURATION	CONFIGURATION CONFIGURATIO	DATA SET/RUN NUMBER COLLATION SUMMARY CONFILURATION (M 3 St	ASIOA DATA SET/RUN NUMBER COLLATION SUMMARY DATE CONFIGURATION CONFIGURATION	ASIOA DATA SET/RUN NUMBER COLLATION SUMMARY CONFIGURATION S. M. 2 Set So Set	CONFIGURATION DATA SET/RUN NUMBER COLLATION SUMMARY DATE SO AUB CONFIGURATION SCHOOL SET SET	DATA SET/RUN NUMBER COLLATION SUMMARY DATE 20 Aug 198 Sec Se

TABLE II (Cont'd)

ARC TEST : OA	587-1-11		0.47					0.66		TION	dana dan	UAT	30	Aug		4 0
· · · · · · · · · · · · · · · · · · ·	ISIOA	<u>_</u>	DATA	A SE	I/RU	N NU	MBE	K ((JL t. A	1104 21	YSAMML		. 30			
DENTIFIER	CONFIGURATION	a	M	4	Ser	Sec	18-	S			7		-γ	-4	ETA O	7
RA2#55	ovioz Orbiter	Ail	0,8		0	0	55							130	131	132
56	`		0.9	1	1		1							1126		127
57		111	0,95											123	124	125
· 58			105				1				<u> </u>			120	121	122
59			1.10			\prod	\prod							117	1/8	119
60			1.25											114	115	116
61		1	1,40											111	112	113
62		A2	09	1											128	
63		Ai	1,40	915	Ţ	1	1								110	
64		A3	0.6	600	10	9	0							168	169	170
65	,		89		\ \ \									165	166	167
66			0.9				T							162	163	164
67			0,95			\prod	Π							159	160	161
68			1,05			1	11							156	157	158
69		111	1.10		1		11	1				1	1	153	154	155
70		111	1.15		1	1-1-	11							150	151	152
71		111	1,25			11.	11	+				1-1-		147	148	149
72		-1;+	1,40		1	1	11	+-,	,			1	1	144	145	7
		1, ', 1		<u> </u>	<u> </u>		<u> </u>		- ا			<u>.</u>		1.7.7		
, , , , ,			25		31 	OE I FIG	ILL L	 5	43 4.4.1-	<u></u>	49	<u> </u>	150	AH (II	ISVA	7 H 12) N
a OR SCHEDUL									•							

NASA-MSFC MAP

	587-1-11	···	·		,								RH		5 of	5
EST : O	1310A		DAT	A SET	[∕RUI	טא א	MBER	COLL	OITA_	N SUMMARY	DATE	30	Aug	198	2	
DATA SET	CONFIGURATION			·			·		·				B	EIA	787	
DENTIFIER		Ø	_M_	Pr	Sec			Se					-4	0	4	ĺ
	ov 1020 rbiter	6	MI	216	5	5	55	9		 	- -		60	61	62	l
74		444	M ₂	29.2			↓ ↓	1					63	64	65	İ
75		444	M	21,6		\sqcup	-	-5					221	222	223	l
76		$\bot \!\!\! \downarrow \!\!\! \downarrow$	M2	29.2			 			<u> </u>			224		226	į
77		111	M	21.6				5					25]	252	253	İ
78		11	Mz	292	1	1	Ш.	1	ļ	<u> </u>			254	255	256	
79		1-4-	MI	216	10	9		0					138	139	140	37
1 80	<u> </u>	111	M _≥	292	1	1	1	1	ļ	 		_	141	142	143	Š
		+		-	-		-		-	 		-	A	LPHA		
	1	B	M	9	SeI	<u>ک</u> وہ	SSB	SR						60		
A2 #81	ovioz Orbiter	В	09	600	0	0	55	0						129		
									-			-				1
				-		•										1
	13 19		 25_	ــــــــــــــــــــــــــــــــــــــ	31	<u> </u>	37	L	43	49	:5	. 1		٤,	1	<u>.</u>
<u> </u>			.1	4-4-4	1		11		1					بمبا		
a OR Schedu	40011 44	-3 -2	-15 -	15	0,0	5.1.	1.5.2	3,4	1.4	MACH M2	= .6,.7,.8	3,9	95,1.0	المراجع	H (2)	4 6

HASA-MSFC-MAI

\$ SEE TABLE II (CONCLUDED) FOR COMPONENT IDENTIFICATION

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DATA SET IDENTIFIER					BETA							
		ø	M	\$	Sez	See	See	Sa	POINT #	-2	0	2
A4801	OV102 ORBITER	θ_{I}	2.0	400	5	5	55	0	72 - 1/3	101	201	301
02			2.0				7	\mathcal{I}	129-170	401	501	601
03		\coprod	2.2	\coprod					171- 212	102	202	302
04		\coprod	2.5			\sqcup _			213->254	103	203	303
05			3.5		1	Y			255-296	104	204	304
06			2.0		-5	-5			306→347	105	205	305
07			2.2						348-389	106	206	306
08		Ш	2.5						390-431	107	207	307
09		Υ	3.5		*	*			432-473	108	208	306
10		15	Ma		0	0		Y	488 - 505 509 - 517	109	209	309
11		A	2.0		7	T		5	519-560	110	210	310
12		$ \Upsilon $	2.5					T	561-602	111	211	311
13			3.5		Y	Y		Y	732→773	115	215	3/5
14			2.0		-5	-5		-5	690-731	114	214	314
15			2.5		7	\bigcap		71	648-689	113	2/3	313
16		Y	3.5		Y	Y	Y	4	606-648	112	212	
17		A ₂	2.0		15	5	87.2	0	1155-71166	129	229	329
1 18	Y	A	2.5	1	F	Ŧ	Ŧ	Ŧ	1063 - 1105	127	227	327
	7 13 19		25		31		37	43	#1074 SKIPPI L	1	67	
1111					1		1				1	

M. 20, 2, 4, 15, 2.6, 2.8, 3.0, 3.2, 3.4, 3.5 \$ SEE TABLE I (CONCLUDED) FOR COMPONENT IDENTIFICATION

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TABLE II (Cont'd)

<u> </u>	0 A 310 C		DAT	A SE.	T/RU	N NU	MBER	COL	ATION	SUMMARY	DAT	E /	0/1	1/8.	3
DATA SET	CONFIGURATION				1		V		······					BET	
		الحد	M	8				SR		1106	+ //33		-2	0	2
	OVIDE ORBITE	1		400		5	87.2	0			1/33		128	228	
20		A31	Ŧ	┨-}-	T	T	T				1139			-	628
21		Ro	2.0		0	-5	0				→852		120	220	320
22		-	2.2	- -							₹837		119		319
23		-144	2.5							804	₹818		118	218	318
24		Y	3.5	11							788		116	216	3/6
25		2	Me		1	1	Y			\mathcal{Q}			117	217	3/7
26		As	2.0		خ	5	55	i		853	-714*		121	221	32/
27		17	2.5		$ \uparrow $	7				915	→974		122	222	322
Y 28	Y	1	3.5	Y	*	¥	V	Y		975	-1034		123	223	323
		1	-	-	<u> </u>		-							LPH	
		B	M	9-	Sez	Seo	Sse	SR					12.7	24.0	23,6
A4# 29	OVIO2 ORBITER	В.	2.5	400	5	5	55	0					126		
¥ 30		- B,		100		5	55							224	325
		+										-			
		廿	1												
	7 13 19	9	25		31	<u> </u>	37		43	49	55_	61		67	
1111	ليبسلسين						LENTS	-			2->+2(4)	حبيلت		ببيا	R (2)

^{1 776, 781, 784, 789 +803, 804, 811, 816, 820 → 822, 820, 850 630, 840 344, 800 +} SKIPP V 874 +875

NASA-MSFC-MAF

TABLE II (Concluded)

DATASET/RUN NUMBER COLLATION SUMMARY INDEX TO DATA TABULATIONS

Dataset		ØA31	0A	ØA31	OB	ØA310	C
Fourth Character	Component Description	Tab Pg.No.	Fiche Pg.No.	Tab Pg.No.	Fiche Pg.No.	Tab Pg.No.	Fiche Pg.No.
С	Canopy and For- ward Fuselage	1-1054	1-18	1-109	1- 2	1- 804	1-13
В	Fwd. Side Fuselage	1055-1584	18-26	110-163	2- 3	805-1185	13-19
M	Mid-Side Fuselage	1585-2186	26-35	164-225	3- 4	1186-1620	19-26
Ø	OMS and Aft Fuselage	2187-2869	35-46	226-296	4- 5	1621-2150	26-35
L	Left Surface of Vertical Tail	2870-3552	46-57	297-367	5- 6	2151-2680	35-43
R	Right Surface of Vertical Tail	3553-4165	57-67	368-430	6- 7	2681-3168	43-51
U	Upper Wing (Left)	4166-5982	67-96	431-616	7-10	3169-4574	51-73
Z	Upper Wing (Right)	N/T	N/T	617-654	10-11	4575-4854	73-78

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TABLE III. STATIC PRESSURE ORIFICE LOCATIONS

Vertical Tail

OPIF	FULL	%	MODEL	MODEL
NUMB	SCALE	CHOPD	SCALE	SCALE
#	ZO	(X/E)	N	Z
2345678981234561-8981234567898123456	60000000000000000000000000000000000000	0.000000000000000000000000000000000000	48.45393004436286439933444362864395643956439564395643956439564395555555555	21.000 21.000 21.000 21.000 21.000 21.000 21.000 21.000 21.000 21.000 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 23.800 24.600 26.600 26.600 26.600

Upper Wing

ORIF NUMB #	FULL SCALE YO	SP ЯН N	% CHOPD (X/C)	MODEL SCALE X	MODEL SCALE Y
	YO 0.00000000000000000000000000000000000	N 42277777777777777777777777777777777777	(X/C) 0.010000000000000000000000000000000000	X 605004999337771009993377710099933777100993377710099337771009993377710099933777100999337771009993388888888888888888888888888888888	Y -7.000 -7.000 -7.000 -7.000 -7.000 -7.000 -7.000 -7.000 -7.0
138 139 140 141	-365.3 -365.3 -365.3	0.780 0.780 0.780 0.780	0.450 0.500 0.550 0.612	46.127 46.581 47.034 47.597	-12.786 -12.786 -12.786 -12.786

Upper Wing

OPIF NUMB # .	FULL SCALE YO	SPAN N -	% CHORD - (X/C)	MODEL SCALE X	MODEL SCALE Y
2344567890-234567890-23456789 11144567890-234567890-23456789 177777777777777777777777777777777777	-3655.3333333333333333333333333333333333	03.00000000000000000000000000000000000	0.6885320087532700000000000000000000000000000000000	48.28637.04815929420449.58677899946149.488.932694420449.58686778899946149.449.658693693694444.444.444.444.444.444.444.444.444.4	-12.7866 -12.78666666666666666666666666666666666666
180 181	-420.1 420.1	0.897 0.897	0.967 0.997	50.551 50.756	-14.704 -14.704

Forward Side Fuselage

ORIF	FULL	FULL	MODEL	MODEL
NUMB	SCALE	SCALE	SCALE	Scale
#	XO	ZO	X	Z
203 204 205 207 208 207 208 207 208 207 208 207 208 207 208 201 201 201 201 201 201 201 201 201 201	9595559955995595959 52935599559955959 566665666666666665656	350 350 350 360 365 360 400 400 400 400 415 460 460 460	20.650 21.875 24.150 21.875 21.875 21.875 21.650 21.525 21.525 22.400 23.275 23.625 24.150 21.875 21.875 21.875 21.875 21.875 21.875 21.875 21.875	12.250 12.250 12.250 12.600 13.125 13.475 14.000 14.000 14.000 14.000 14.000 14.000 14.525 15.575 16.100 16.100

TABLE III. STATIC PRESSURE ORIFICE LOCATIONS (Continued)

Mid Side Fuselage

	,,,,,,			•
ORIF NUMB #	FULL SCALE XO	FULL SCALE ZO	MODEL SCALE X	MODEL SCALE Z
2222222334567890 22222223334567890 222222222222222222222222222222222222	928 10070 10070 10070 10070 10070 10070 10070 10070 10070	360 360 360 360 400 400 400 400 400 400 460 460 460	32.480 35.450 35.450 35.210 35.310 35.310 35.480 33.530 33.530 35.210 35.210 35.210 35.210 35.210 35.210 35.210 35.210 35.210	12.600 12.600 12.600 13.300 13.685 14.000 14.000 14.000 14.000 14.000 14.000 14.000 14.000 14.000 14.000

Canopy and Forward Fuselage

ORIF NUMB #	FULL SCALE XU	FULL SCALE YO	STA ANGLE PHI	MODEL SCALE X	MODEL SCALE Y
				SCALE	SCALE Y 00000000000000000000000000000000000
3478 3449 350 350	480 490 500 510 520	-49.5 -52.0 -52.5 -53.0 -53.5	150 150 150 150 1 50	16.800 17.150 17.500 17.850 18.200	-1.733 -1.820 -1.838 -1.855 -1.873

TABLE III. STATIC PRESSURE ORIFICE LOCATIONS

(Continued)

Canopy and Forward Fuselage

OPIF	FULL	FULL	STA	MODEL	MODEL
NUMB	SCALE	SCALE	ANGLE	SCALE	SCALE
#	XO	YO	PHI	X	Y
1234567.000.4234 5555555555664334 600000000000000	530 540 550 550 550 550 550 641 4430 4470	-54.0 -54.0 -54.0 -54.0 -52.0 -517.0 -517.0 -6504.0 -77.0 -77.0 -77.0 -77.0 -77.0 -77.0 -77.0 -77.0	150 150 150 150 150 120 120 120 120 120	18.550 18.900 19.250 19.600 19.950 20.650 12.250 12.650 15.650 16.100 16.450	-1.890 -1.890 -1.895 -1.830 -1.830 -1.810 -1.813 -1.813 -2.459 -2.459 -2.770 -2.870
365	490	-86.0	120	17.150	-3.010
366	510	-89.5	120	17.850	-3.133
367	530	-91.0	120	18.550	-3.185
368	550	-92.5	120	19.250	-3.238
369	570	-95.0	120	19.950	-3.325
370	590	-97.5	120	20.650	-3.413

OMS Pods and Aft Fuselage

OR IF	FULL SCALE XO	FULL SCALE YO	FULL SCALE ZO	STA ANCLE PHI	MODEL Scale- X	MODEL SCALE Y	MODEL Scale Z
402	1215		400.0	98	42.525		14.000
402	1245		400.0	90	43.575		14.000
404	1265		400.0	ទុក្ខ	44.275		14.000
405	1285		400.0	<u> 50</u>	44.975		14.000
406	1300		400.0	90 22	45.500		14.000
407	1306		400.0	90	45.710		14.000
408	1312		400.0	90	45.920		14.000
409	1318	**********	400.0	90 22	46.130		14.000
410	1325		400.0	90	46.375		14.000
411	1330		400.0	90	46.550		14.000
412	1350		400.0	90	47.250		14.000
413	1375		400.0	90 22	48.125		14.000
414	1430		400.0	98 405	50.050		14.000
415	1215	**********	429.7	105	42.525		15.040
416	1245	**** (0.00 true true	429.7	105	43.575		15.040
417	1265		429.7 429.7	105	44.275		15.040 15.040
418	1285		429.7 429.7	105	44.975 45.500		15.040
419 420	1300		429.7	105 165	45.710		15.040
420 421	1306 1312		429.7	105	45.920		15.040
421 422	1312		427.1	105	46.130		15.145
423 423	1318		4 4.1	105.	46.130 46.375		15.194
424	1330		425.2	105	46.550		15.232
425	1350		435.6	105	47.250		15.246
426	1375		435.2	105	48.125		15.267
427	1430		439.2	105	50.050		15.372
428	1215		439.6	110	42.525		15.38£
429	1245	***	439.6	110	43.575		15.386
43C	1265		439.6	110	44.275		15.386
431	1285		439.6	110	44.975		15.386
432	1300		439.6	110	45.500		15.386
433	1306		439.6	110	45.710		15.386
434	1312		439.6	110	45.920 .		15.386
435	1318		441.6	110	46.130		15.456
436	1325		444.2	110	46.375		15.547
437	1330		445.5	110	46.550		15.593
438	1350		448.5	110	47.250		15.698
439	1375		451.9	110	48.125		15.817
440	1430		455.4	110	50.050		15.939
441	1215		459.4	120	42.525		16.079
442	1245		459.4	120	43.575		16.079
*443	1265		459.4	120	44.275		16.079
444	1285		459.4	120	44.975		16.079
445	1300		459.4	120	45.500		16.079
446	1306		459.4	120	45.710		16.079
447	1312		460.0	120	45.920		16.100
448	1318		463.4	120	46.130		16.219
449 450	1325 1330		467.3 469.3	1 20 1 20	46.375 46. 5 50		16.356 16.426

ORIF NUMB #	FULL SCALE XO	FULL SCALE YO	FULL SCALE ZO	STA ANGLE PHI	MODEL SCALE X	MODEL SCALE Y	MODEL ' SCALE Z
- 1234567890-1234567890-1234567890123456789 - 1234567890-12345677777777888888899999999999999999999999	73730 5550 6 2 6 5 0 0 0 5 0 5 0 5 0 5 0 5 0 5 0 5 0 5		474 9 479.2 482 8		25050555500 21052775500 21052777500 21052777500 210527770 210527770 210527770 210527770 210527770 210527770 210527770 210527770 210527770 210527770 210527770 210527770 2105270 210527770 210527770 210527770 210527770 210527770 21052770 21052770 21052770		16.62?
500	1306	-15.8		174	45 710	-0.553	

OMS Pods and Aft Fuselage

ORIF NUMB #	FULL SCALE XO	FULL SCALE YO	FULL SCALE ZO	STA ANGLE PHI	MODEL SCALE X	MODEL SCALE Y	MODEL SCALE Z
501	1312	-15.8		174	45,920	0.553	
502	1318	-158		174	46.130	-0.553	
503	1325	-15.8		174	46.375	-0.553	
504	1330	-15.8		174	46.550	-0.553	
505	1350	-15.8		174	47.250	-0.553	
506	1375	-15.8		174	48.125	-0.553	
507	1430	-14.9		174	50.050	-0.522	
508	1215	0.0		180	42.525	0.000	
509	1245	0.0		180	43.575	0.000	
510	1265	0.0		180	44.275	0.000	
511	1285	9,0		180	44.975	0.000	
512	1300	0.0	-	183	45 500	0.000	
1 513	130€	0.0		180	45,710	0.000	

Table IX Kulite Locations

	UL I	FULL SCALE XO	FULL' SCALE YO	FULL SCALE ZO	MODEL SCALE ·X	MODEL SCALE Y.	MODEL SCALE Z
ι	1	370	9		12.950	0.000	
	2	370	-20		12.950	-0.700	
	3	415			14.525	0.000	
	4	415 415	-20 -40		14.525 14.525	-0.700 -1.400	
	2	415	-40 -55		14.525	-1.925	
	7	440	-JJ 8		15.400	0.000	
	ģ	440	-20		15.400	-0.700	
	4567mm	440	-40		15.400	-1.400	
	10	440	-55		15.400	-1.925	
	11	450	9		16.100	0.000	
	12	460	-20		16.100	-0.700	
	13	480	9		16.800	0.000	
	14	480 480	-20 -40		16.800	-0.700 -1.400	
	15 16	400 500	-40 0		16.800 17.500	0.000	
	17	500	-20		17.500	-0.700	
	18	500	-40		17.500	-1.400	
	19	500	-55		17.500	-1.925	
	20	520	Ō		18.200	0.000	
	21	520	-20		18.200	-0.700	
	22	520	-40		18.200	-1.400	
	23	560	Ø		19.600	0.000	
	24	560	-40		19.600	-1.400	
	25	580		400	20.300		14.000
	26	688		420	21.000 21.000		14.700
	27	699 629		380 460	21.000		13.300
	28 29	620 620		400	21.700		16.100 14.000
	30	620		350	21.700	<u> </u>	12.250
	-31	640		420	22.400		14.700
	32	640		380	22.400		13.300
	33	690		400	24.150	~	14 00C
	37	920		400	32.200		14 000
	38	920		350	32.200		12 250
	39	950		400	33.250		14 000
	40	1000		460	.35.000		16.100
	41	1000		420	35.000		14,700
	42 43	1000 1000		400 380	35.000 35.000		14.000 13.300
	44'	1000		350 350	35.000		12.250
7	45	1035		400	36.225		14.000
ì	46	1070		400	37.450		14.000
	47	1070		350	37.450		12.250
	48	1140		400	39.900		14.000
	49	1200		460	42.000		16.100
	50	1200	,	400 .	42.000	,	14.000

Table I▼ Kulite Locations (Continued)

KUL I	FULL SČALE XŪ	FULL SCALE YO	FULL SCALE ZO	MODEL SCALE X.	MODEL SCALE Y	MODEL SCALE Z
51	1260	0		44.100	0.000	
52	1260	-75		44.100	-2.625	
53	1280	-40		44.800	-1.400	
54 55	1280 1280	-75 	460	44.800 44.800	-2.625	16.100
56	1300	9	400	45.500	0.000	16.100
57	1300	-20		45.500	-0.700	
58	1300	-40		45.500	-1.400	
59	1300	-75		45.500	-2.625	
60	1300		460	45.500		16.100
51	1320	-20		46.200	-0.700	
62	1320	- 40		46.230	-1.400	
63	1320	-75		46 200	-2.625	
64	1320		460	46 200		16 100
65 66	132C 1340	-20	400	46 200 46,900	-0 700	14.000
67	1340	-40 -40		46.900	-1.400	
68	1340	-75		46.900	-2.625	
69	1340		460	46.900		16.100
70	1340		400	46.900		14.000
71	1380	-75		48.300	-2.625	
72	1420	-20		49.700	-0.700	
73	1420	-75	~	49.700	-2.625	
74	1420		400	49.700		14.000
75	1480	-20		51.800	-0.700	
76 7 7	1480	-75 -140		51.800	-2.625 -4.900	
78	1000 1035	-140 -140		35.000 36.225	-4.900	
79	1935	-190		36.225	-6.650	
60	1070	-140		37.450	-4.900	
81	1070	-190		37.450	-6.650	
82	1070	-220		37.450	-7.700	
83	1090	-140		38.150	-4.900	• •
84	1090	-190		38.150	-6.650	
85	1090	-220		38.150	-7.700	
86	1140	-140 -100		39.900 39.900	-4.900 -6.650	
87 ጽ ዮ	1140 1140	-190 -220		39.900	-7.700	
89	1200	-140		42.000	-4.900	
90	1280	-380		44.800	-13.300	
91	1300	-380		45.500	-13.300	
92	1320	-380		46.200	-13.300	
93	1340	-340		46.900	-11.900	
94	1348	-360		46.900	-12.600	~~~~
95	1340	-380 480		46.900	-13.300	
96 47	1340	-400 -420		46.900 46.900	-14.000 -14.700	
98	1340 1400	-380		49.000	-13.300	
99	1420	-360		49.700	-12.600	
100	1420	-380		49.700	-13.300	~~~~

Table I▼ Kulite Locations (Concluded)

KUL I NUMB	FULL SCALE XO	FULL SCALE YO	FULL SCALE ZO	MODEL SCRLE X	MODEL SCALE Y	MODEL SCALE
101	1420	-420		49,700	-14.700	
102	1440	-380		50.400	~13.300	
103	1520		680	53.200		23.800
194	1380		560	48.300		19.600
105	1540		680	53.900		23.800
106	1630		780	57.050		27.300
107	1580		720	55.300		25 20c
108	1550		700	54.250		24 500
109	1560		680	54.600	~	23.800
110	1530		650	53.550		22.750
111	1490		56A	52.150		19.600
112	1595		660	55.825		23.800
113	1620		680	56,700		23 800

TABLE V
LIST OF BAD DATA POINTS

ØA310A

COMPONENT	IDENTIFIER	<u>. M</u>	<u>β</u>	<u>α</u>	TAP NUMBERS
Canopy and	RA2C04	0.95	4	6	333 through 370
Forward Fuselage	RA2C70	1.15	4	-2	ALL
	RA2C73	1.155 +1.217	4	6	ALL
Forward Side Fuselage	RA2B70	1.15	4	-2	ALL
ruserage	RA2B73	1.155 +1.217	4	6	ALL
Mid-Side	RA2M04	0.95	4	6	228 through 239
Fuselage	RA2M70	1.15	4	-2	ALL
OMS and Aft Fuselage	RA2Ø04	0.95	4	6	428 through 453 467 through 494
	RA2Ø70	1.15	4	-2	ALL
	RA2Ø73	1.155 +1.217	4	6	ALL
Left Surface of	RA2L18-21	ALL	ALL	ALL	20
Vertical Tail	RA2L70	1.15	4	-2	ALL
	RA2L73	1.155 +1.217	4	6	ALL
Right Surface of Vertical Tail	RA2R70	1.15	4	-2	ALL
vertical latt	RA2R73	1.155 +1.217	4	6	ALL
Upper Wing (Left)	RA2U04	0.95	4	6	132 through 145 176 through 180
	RA2U70	1.15	4	-2	ALL
	RA2U73	1.15 +1.217	4	6	ALL

TABLE V. (Cont'd)

ØA310B

COMPONENT	IDENTIFIER	<u>M</u>	<u>β</u>	<u>a</u>	TAP NUMBERS				
OMS and Aft Fuselage	ALL	ALL	ALL	ALL	406,426,488,512				
Upper Wing (Left)	ALL	ALL	ALL	ALL	119,120,142 through 144, 170 through 172				
ØA31 <u>0C</u>									
Canopy and	RA4C06-08	ALL	ALL	ALL	341				
Forward Fuselag	RA4C10	1.99 +2.17 +2.37	ALL	15	352,355 through 357 F				
Forward Side Fuselage	RA4B10	2.17 +2.37	ALL	15	210				
OMS and	ALL	ALL	ALL	ALL	406,426,488,506,512				
Aft Fuselage	RA4Ø17-20	ALL	ALL	ALL	412,482				
	RA4Ø21-25	ALL	ALL	ALL	482				
	RA4Ø26-28	ALL	ALL	ALL	412,482				
Left Surface of	RA4L01	2.0	ALL	ALL	22				
Vertical Tail	RA4L01	2.0	-2	ALL	18				
	RA4L01	2.0	0	24 to 40	18				
	RA4L01	2.0	2	26 to 40	18				
	RA4L05 & 09	3.5	-2	25 to 38	22				
	RA4L21-25	ALL	ALL	ALL	22				
Right Surface of	f				,				
Vertical Tail	RA4R05 & 09		ALL	25 to 38	35				
	RA4R13	3.5	ALL	24 to 38	35				
	RA4R21-25	ALL	ALL	ALL	35				

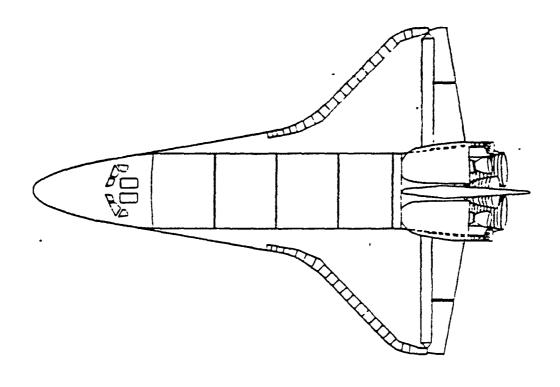
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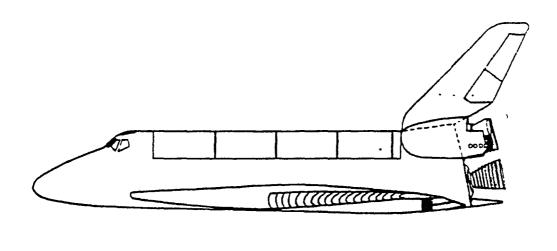
ØA310C (Cont'd)

COMPONENT	IDENTIFIER	M	<u>β</u>	<u>a</u>	TAP NUMBERS
Upper Wing (Left)	ALL	ALL	ALL	ALL	119,120,142 through 144, 170 through 172
	RA4U01-09	ALL	ALL	ALL	130 through 133,135, 137 through 141,147, 148, 150 through 153, 159
	RA4U06-09	ALL	ALL	ALL	145
		1.99 +2.17 +2.37	ALL	15	ALL
	RA4U11	2.0	ALL	10 to 24	147,150
	RA4U13	3.5	-2,0	ALL	147
	RA4U17	2.0	ALL	ALL	147,150
	RA4U17	2.0	-2	ALL	134
	RA4U18	2.5	-2	ALL	147
	RA4U19	3.5	ALL	ALL	147,150
	RA4U19	3.5	-2	ALL	134,136
	RA4U21	2.0	-2	ALL	147,150
	RA4U21	2.0	0	2,4,6	147,150
	RA4U21	2.0	2	-2,0,4,6	147,150
	RA4U22	2.2	ALL	2,4,6	147,150
	RA4U24	3.5	ALL	ALL	147
	RA4U26	2.0	-2	15 to 22.4	147
	RA4U27	2.5	2	ALL	147
	RA4U28	3.5	ALL	ALL	134,136,147
	RA4U29	2.5	-2 to 0	12.7	134,136

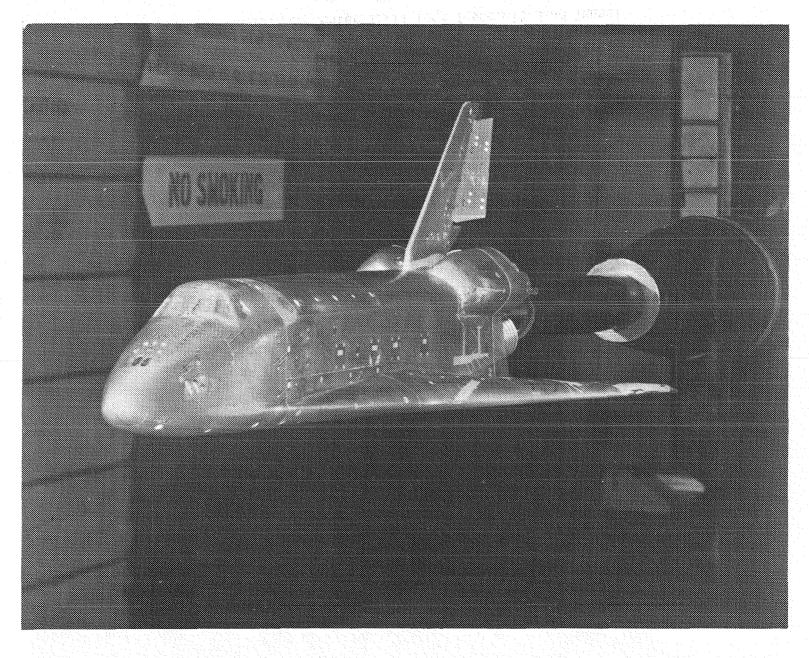
TABLE V. (Concluded)

COMPONENT	IDENTIFIER	<u>M</u>	<u>β</u>	<u>a</u>	TAP NUMBERS	
	RA4U29	2.5	ALL	12.7	147,150	
	RA4U30	3.5	ALL	ALL	136	,
	RA4U30	3.5	ALL	21	147,150	
	RA41130	3.5	-2 to 0	23	147	



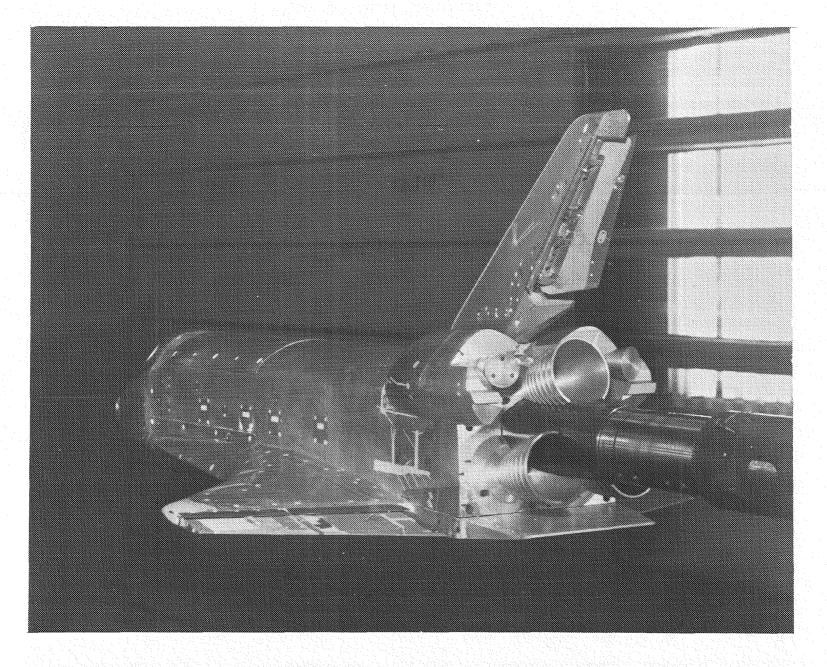


a. Sketch of Space Shuttle Orbiter Model 84-0 Figure 1. Model Sketches

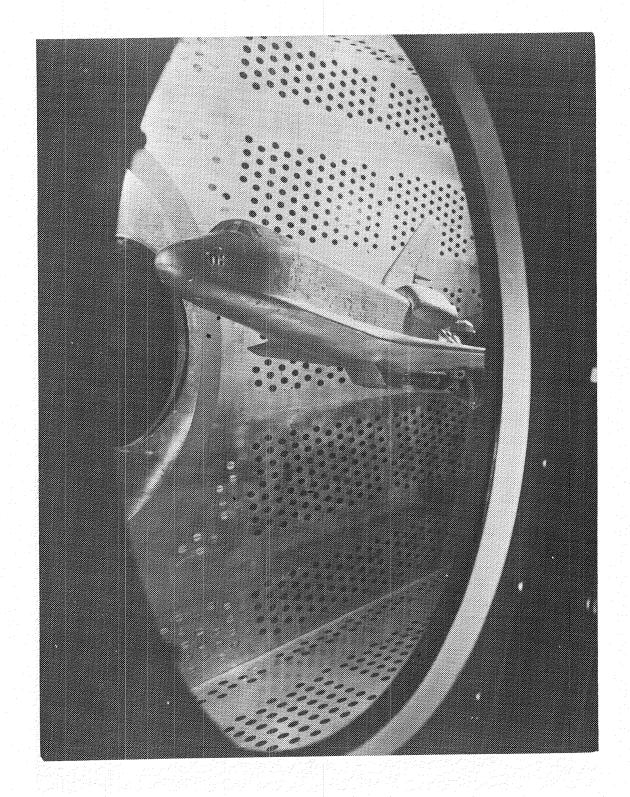


a. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Ames Research Center 11x11 foot Transonic Wind Tunnel

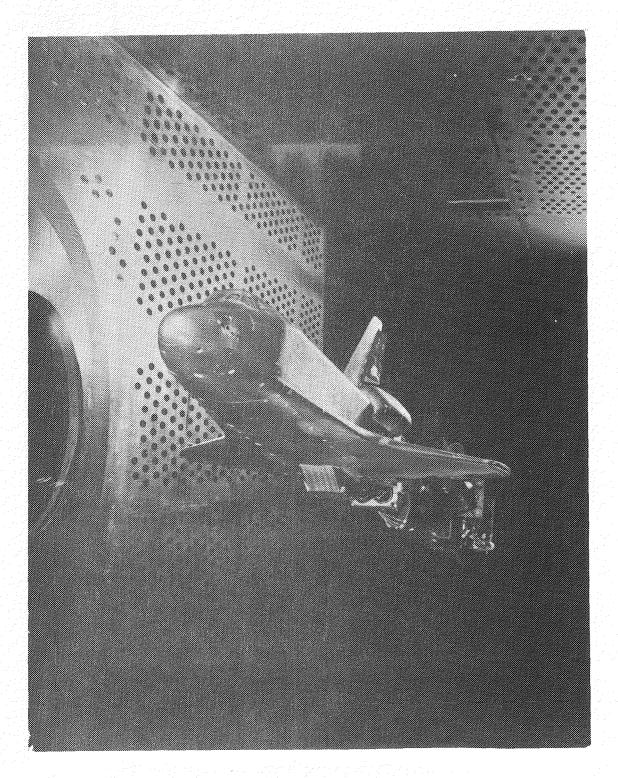
Figure 2 - Model Photographs



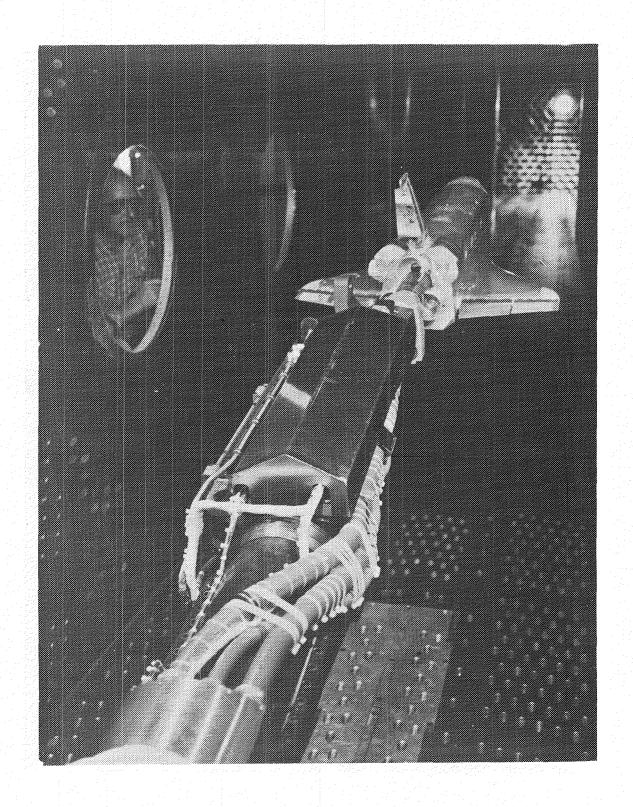
b. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Ames Research Center 11x11 foot Transonit Wind Tunnel



c. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Lewis Research Center 8x6 foot Supersonic Wind Tunnel



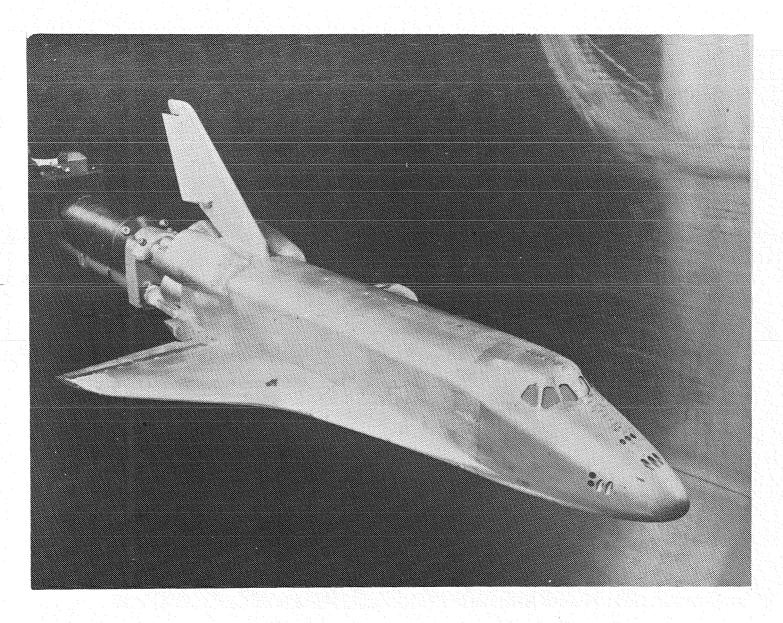
d. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Lews Research Center 8x6 foot Supersonic Wind Tunnel



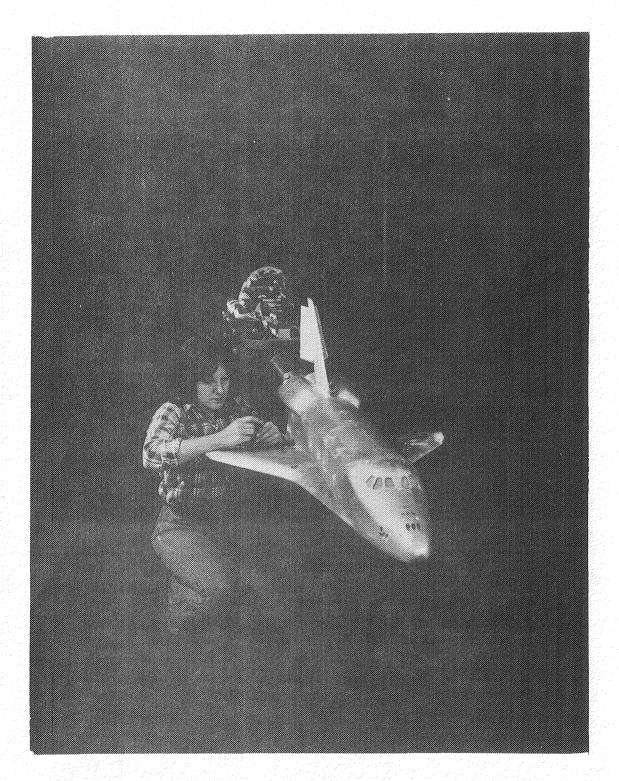
e. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Lewis Research Center 8x6-foot Supersonic Wind Tunnel Figure 2 (Cont'd)



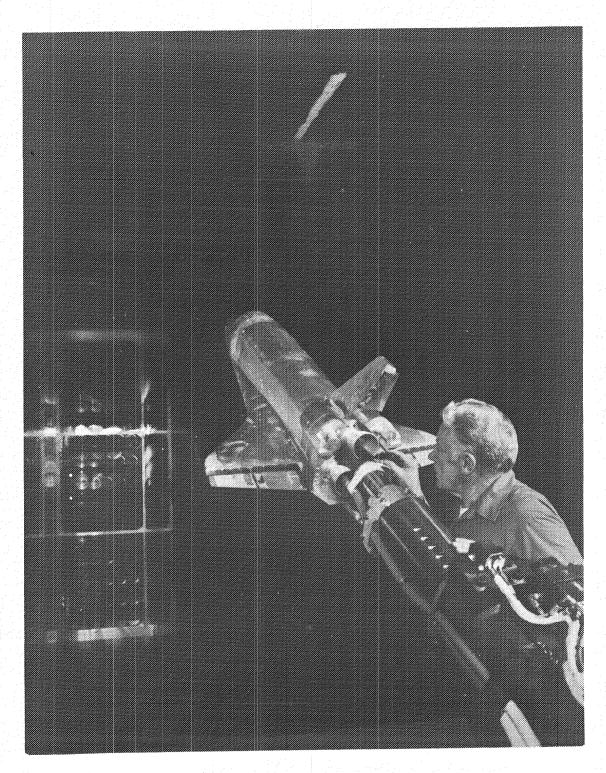
f. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Lewis Research Center 10x10 foot Supersonic Wind Tunnel



g. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Lewis Research Center 10x10-foot Supersonic Wind Tunnel Figure 2 (Cont'd)



Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/Lewis Research Center 10x10-foot Supersonic Wind Tunnel

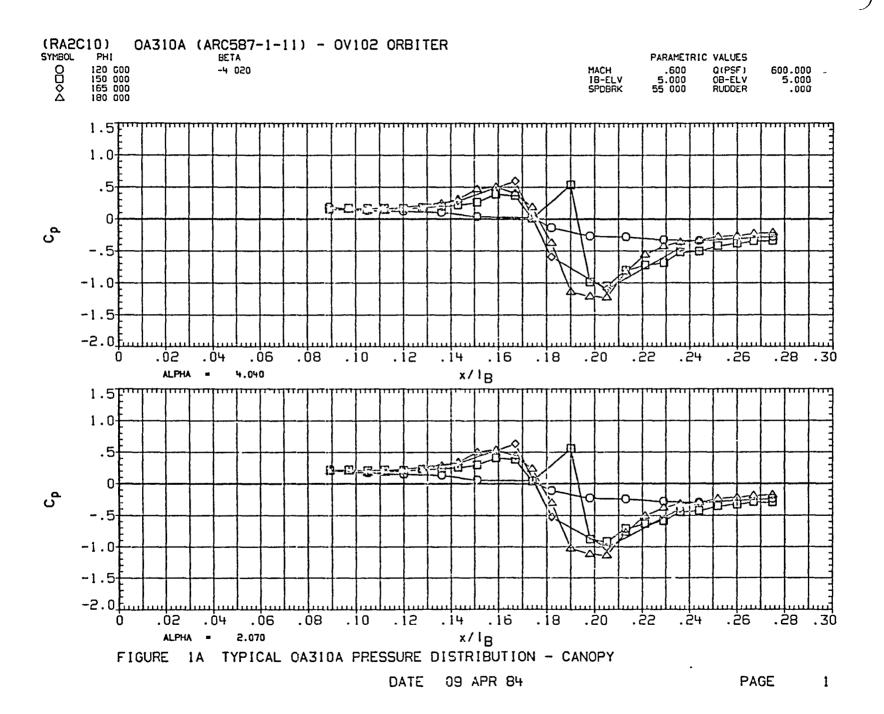


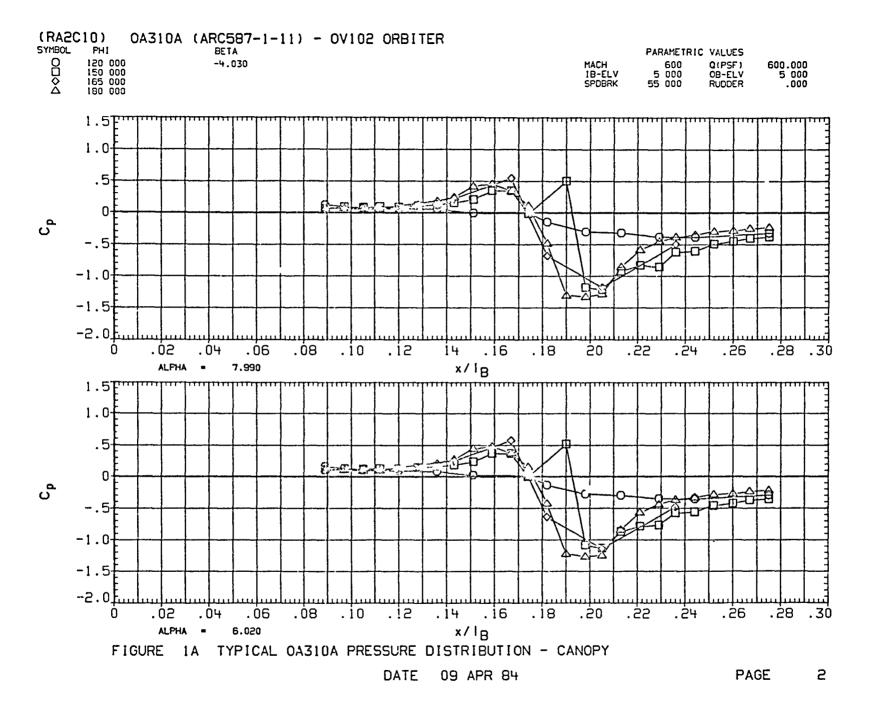
i. Installation Photograph of the 0.035-scale Space Shuttle Vehicle Pressure-Loads Model 84-0 in the NASA/AFWAS Research Center 10x10 foot Supersonic Wind Tunnel

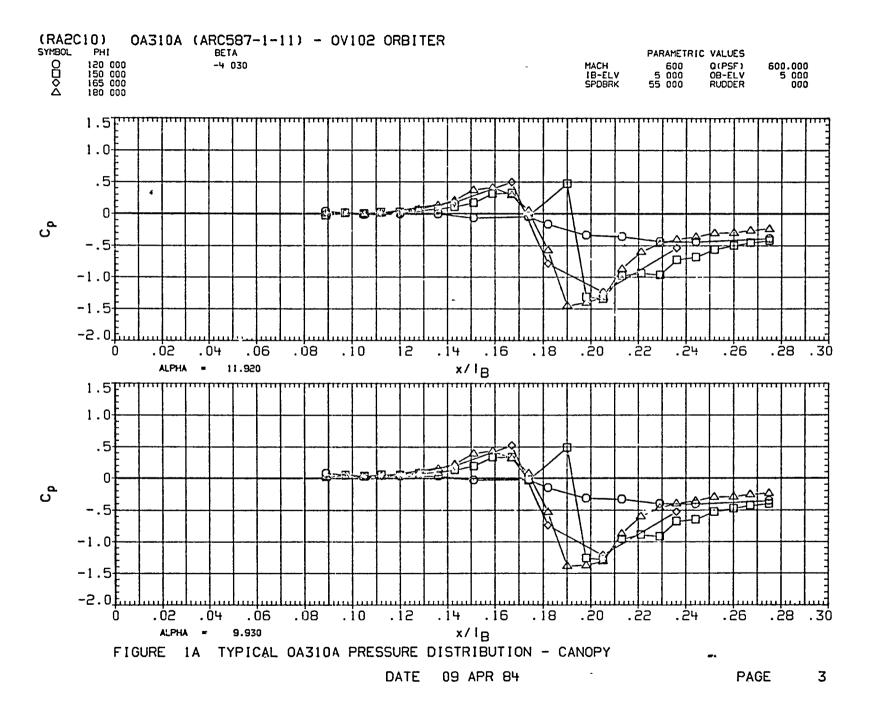
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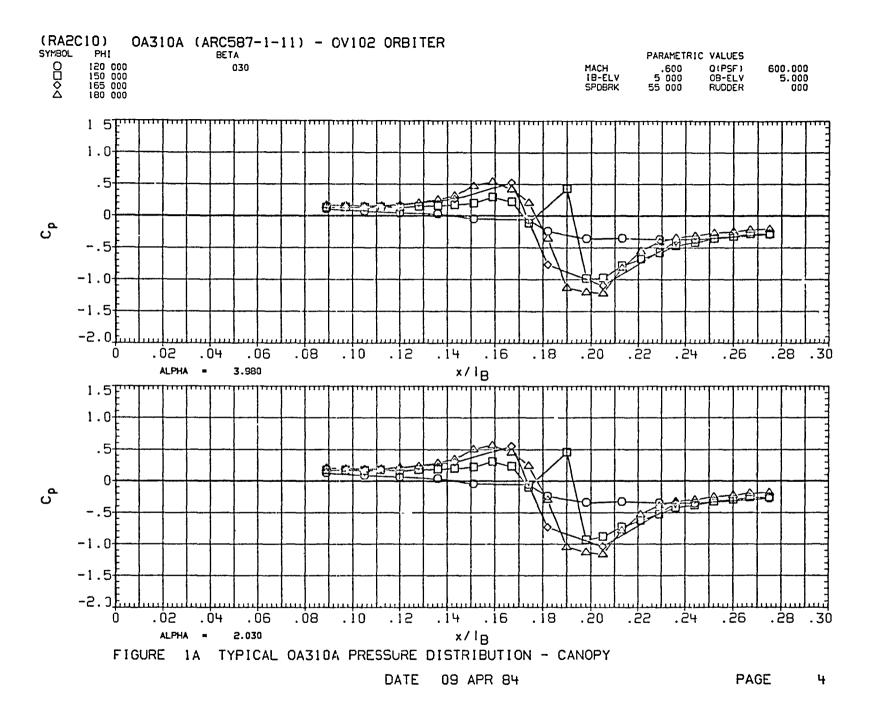
Data Figures

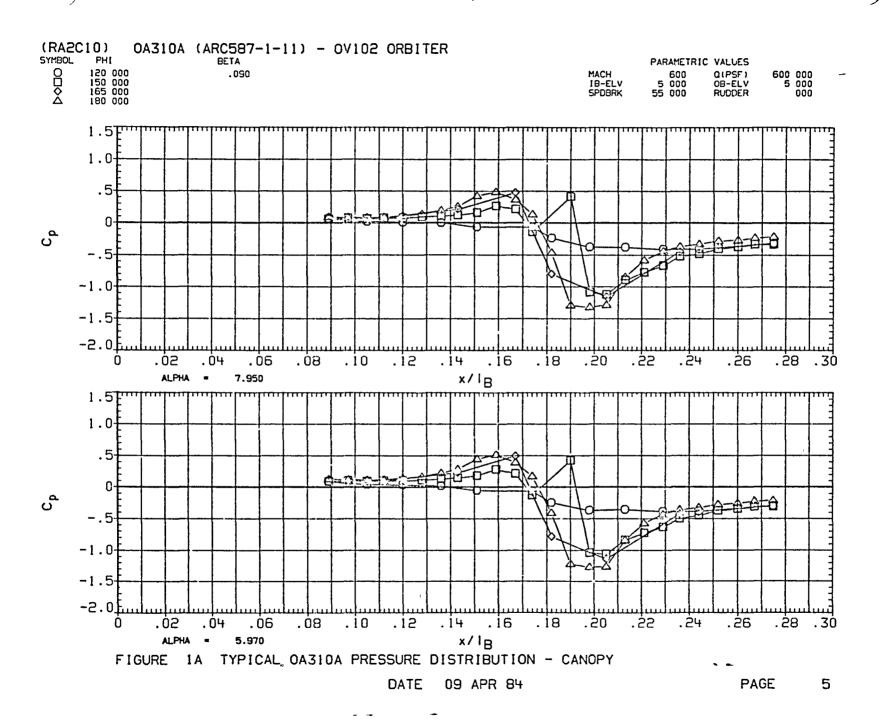
Data tabulations may be found in Volume 2 (microfiche only) or obtained from DMS on request.

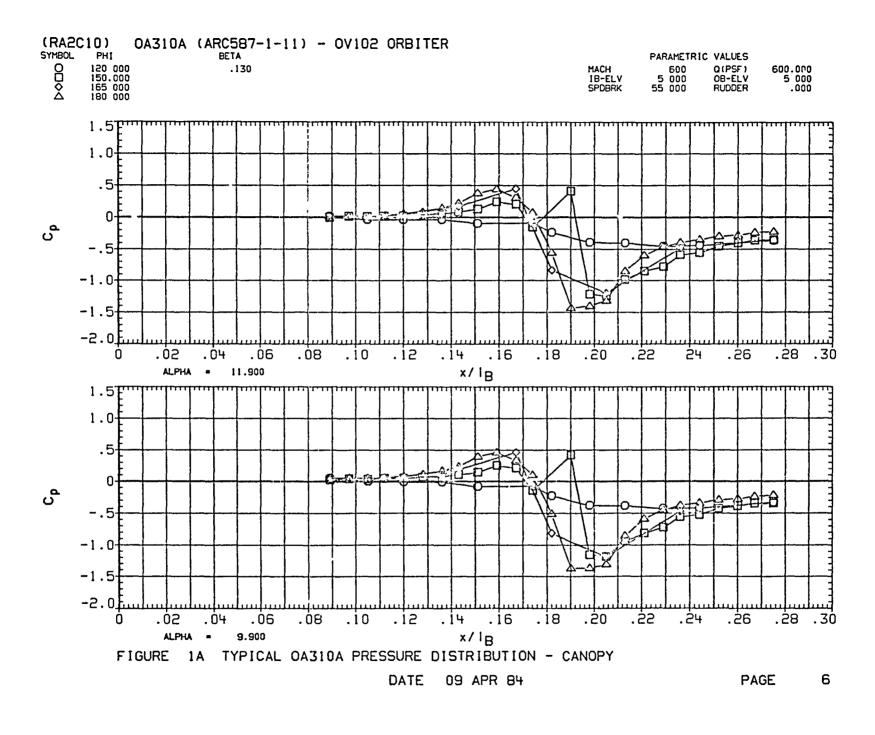




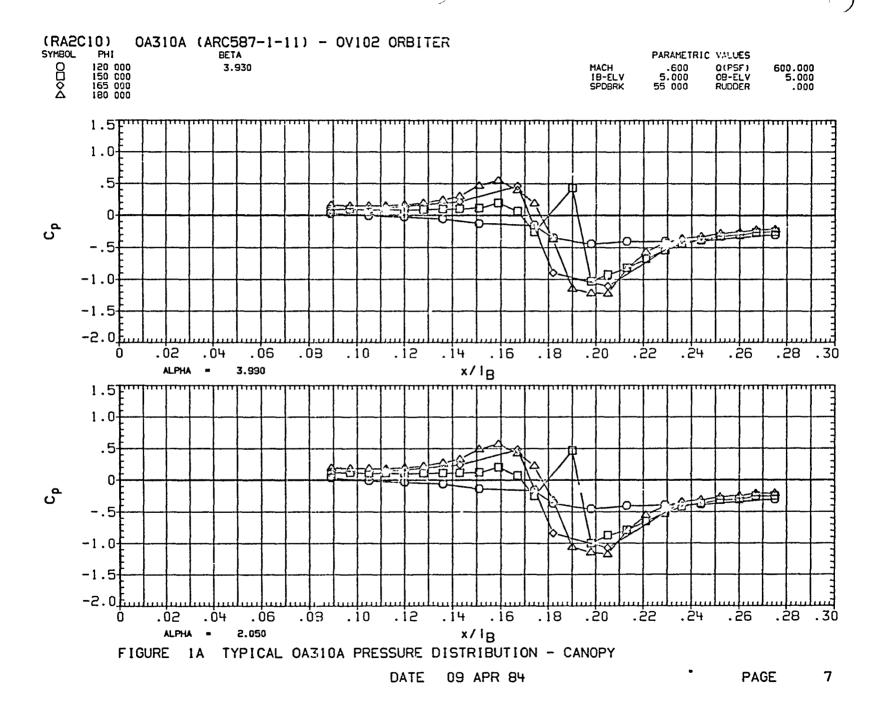


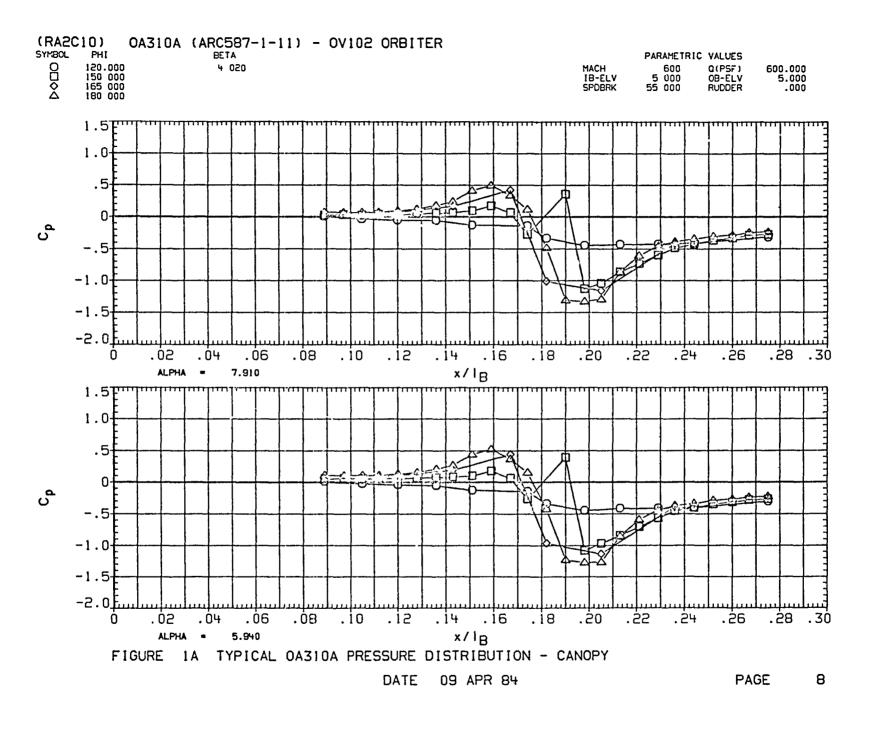


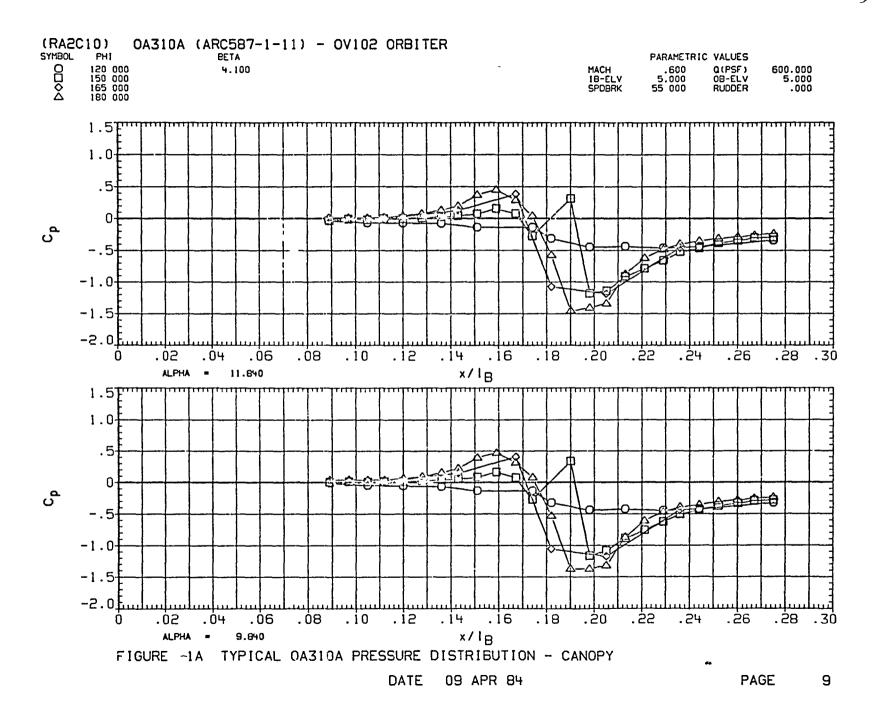


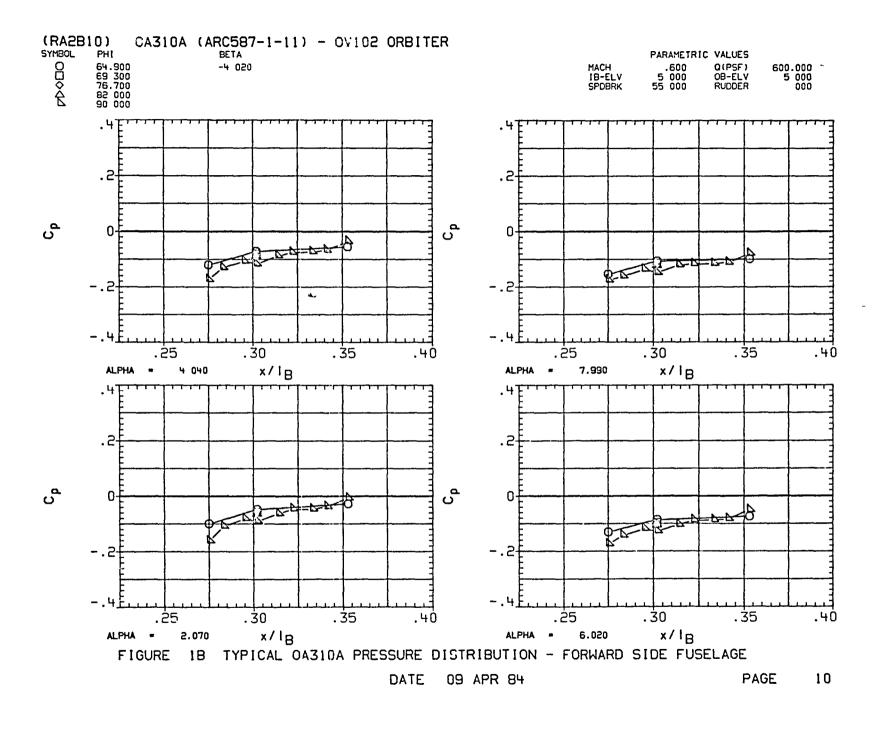


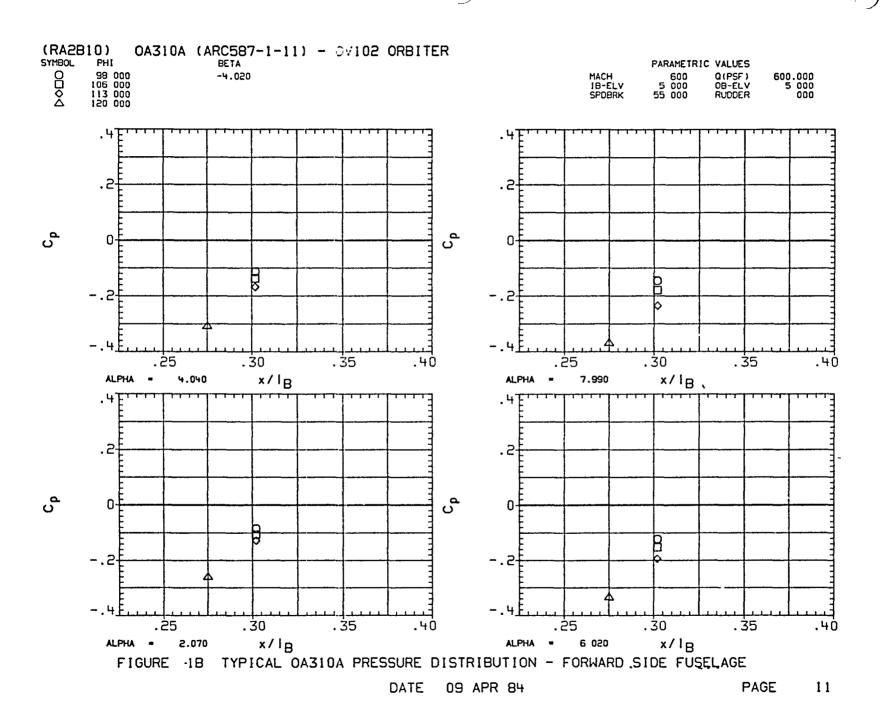
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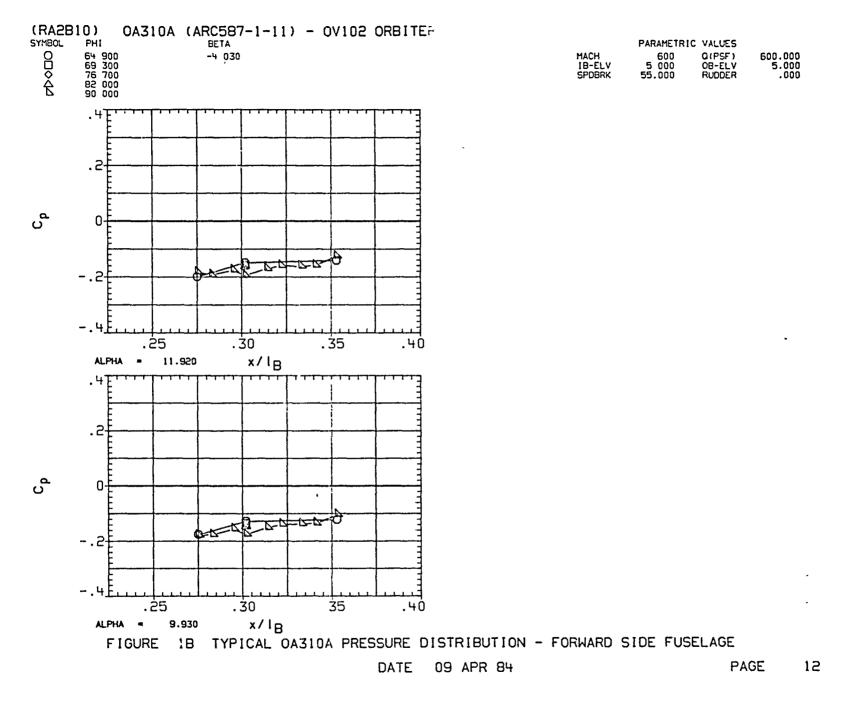


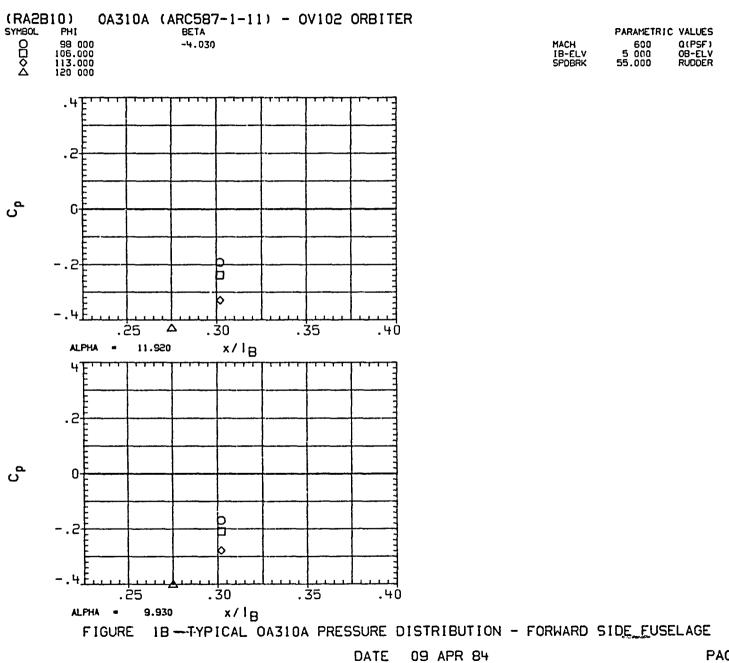








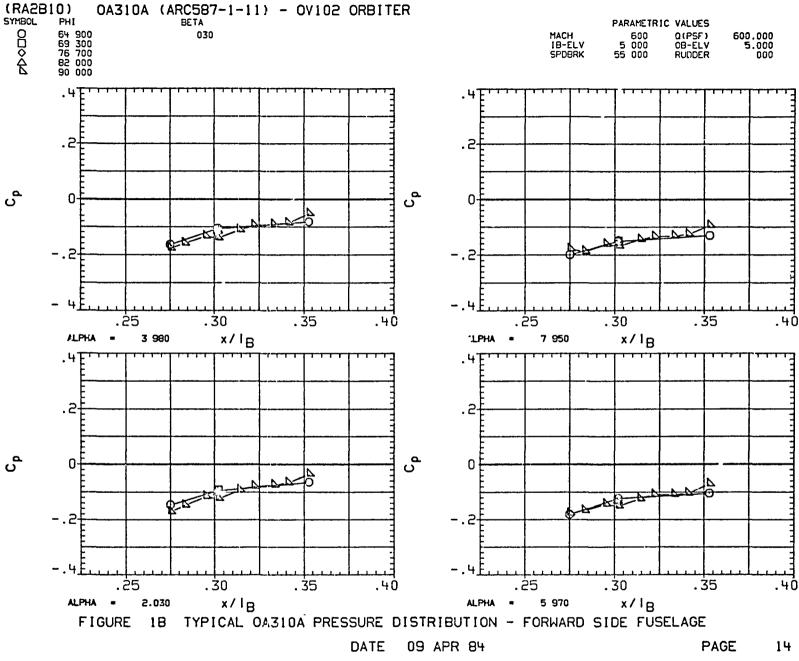


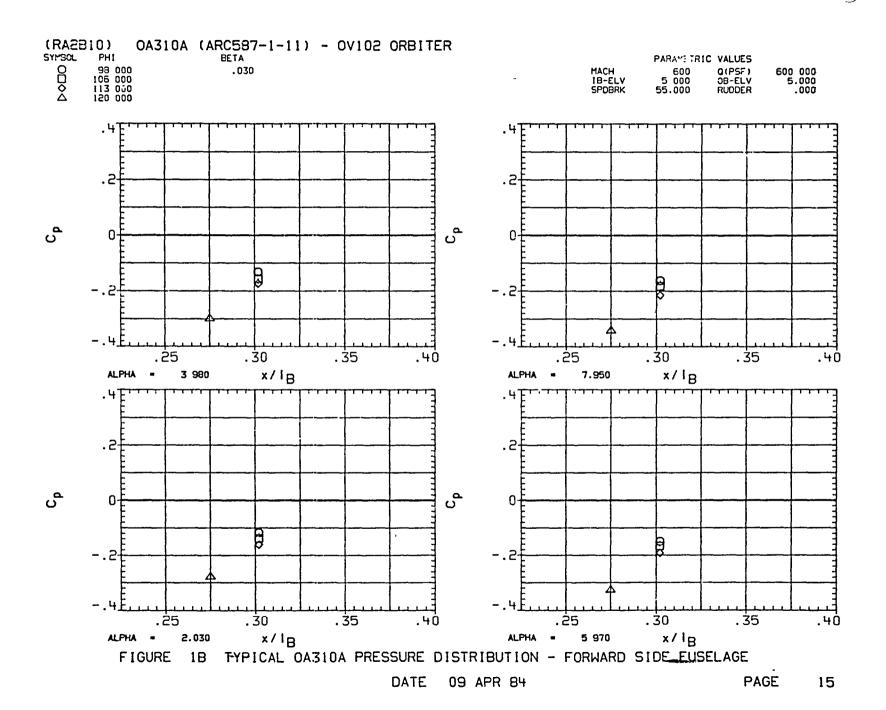


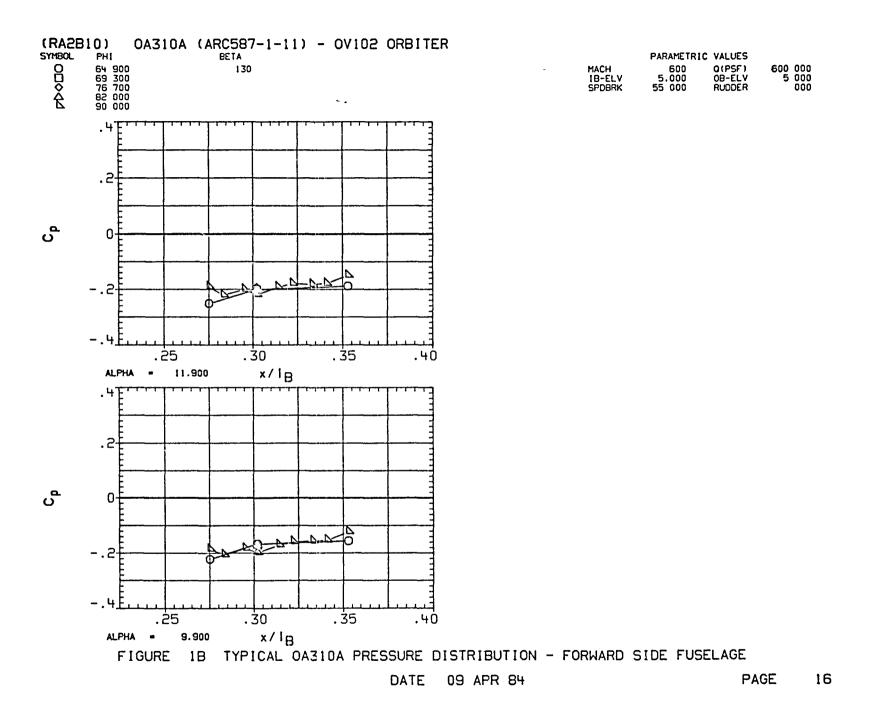
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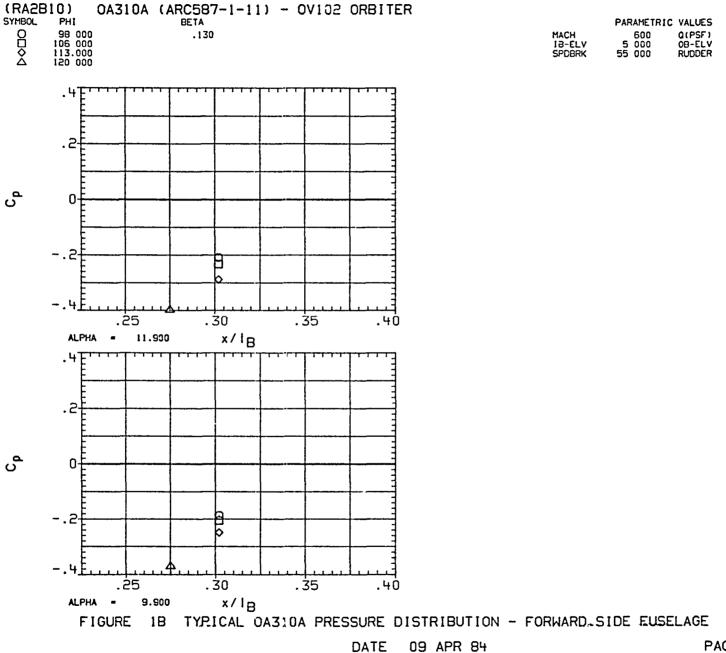
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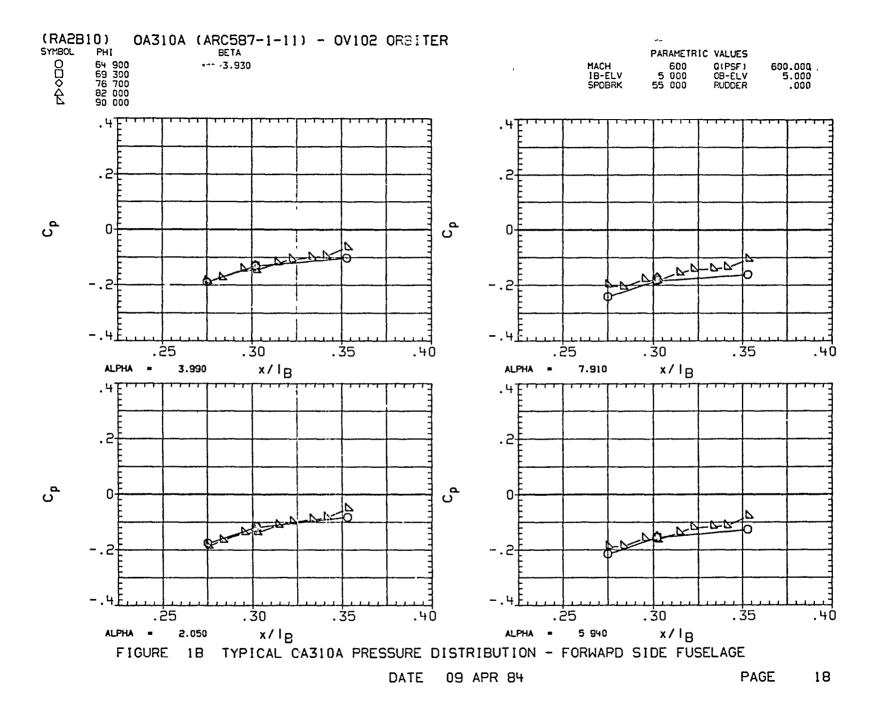


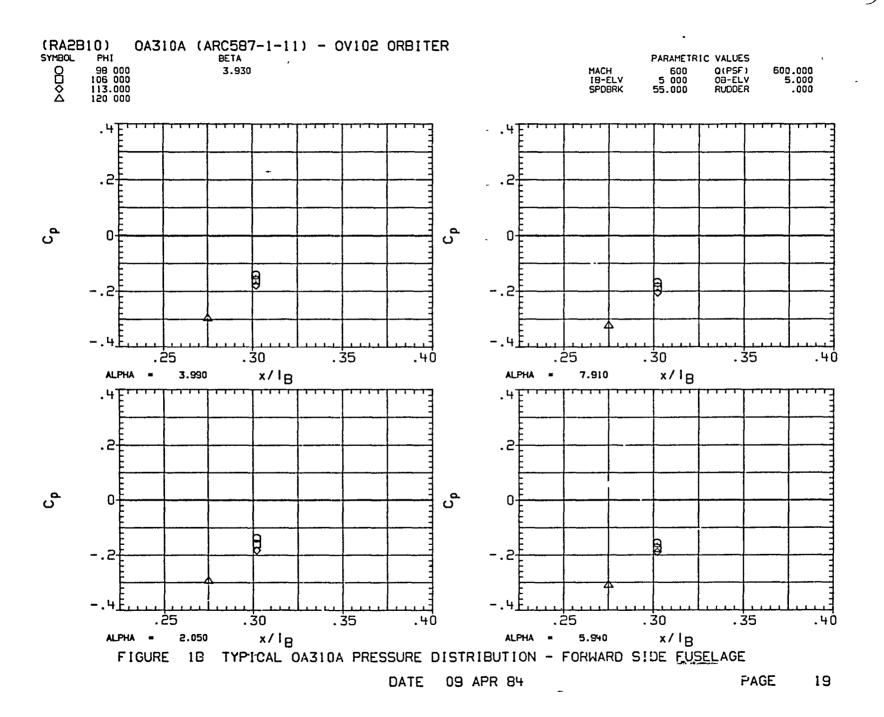


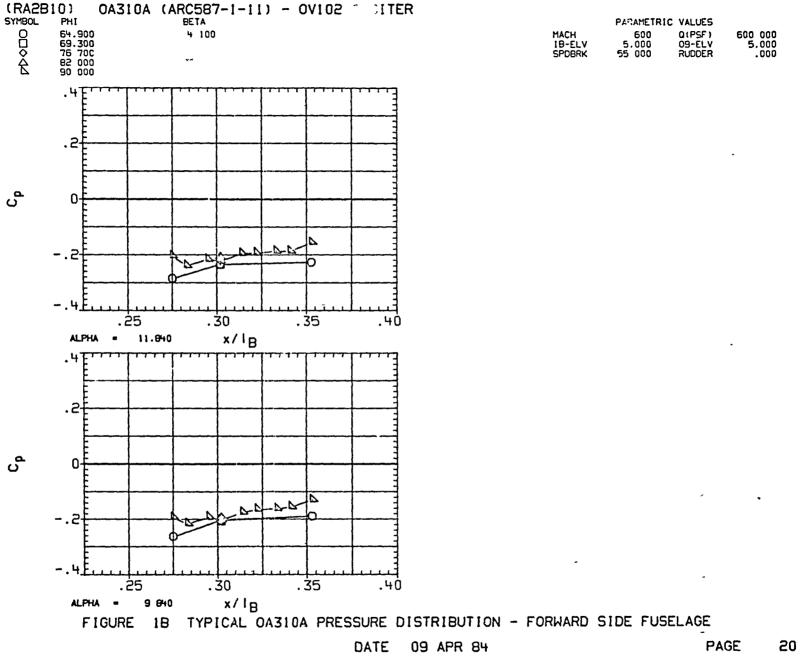
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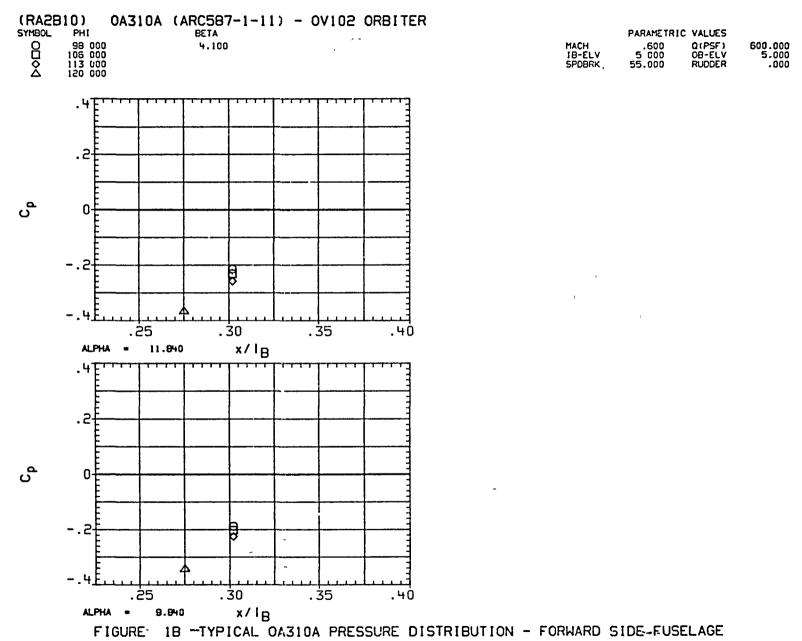
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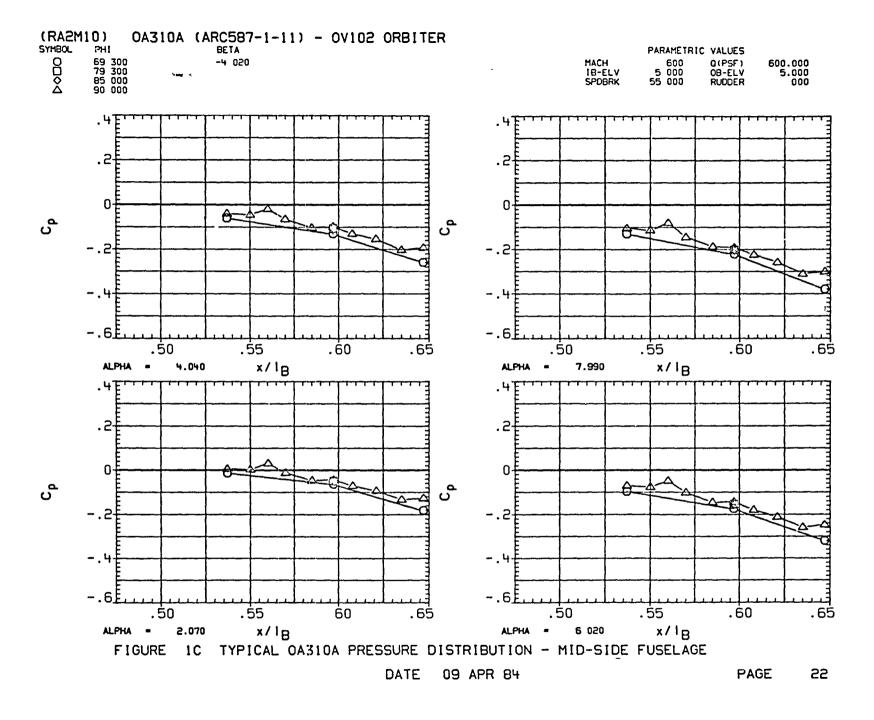




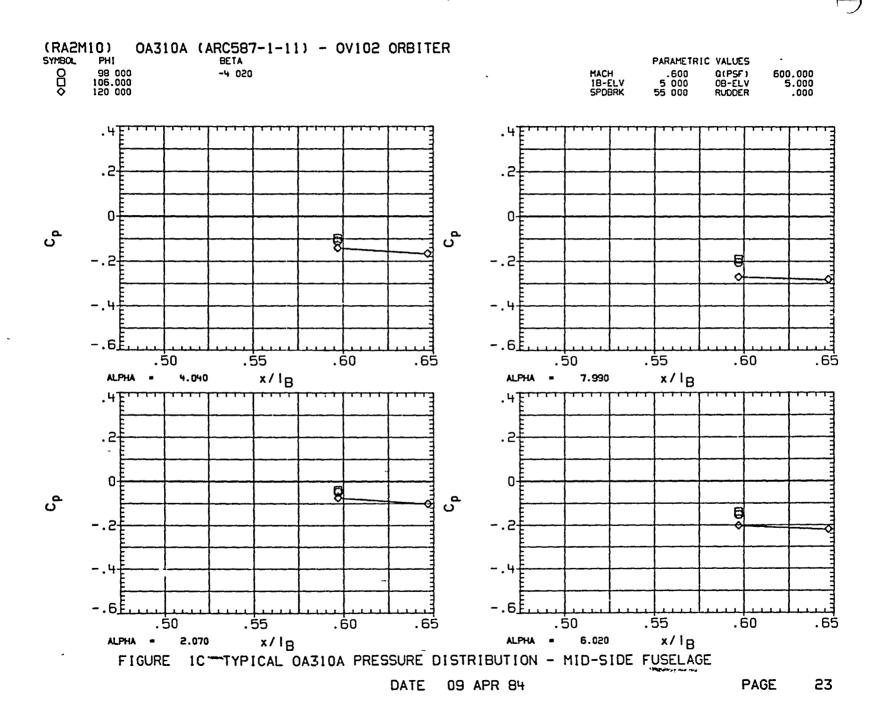


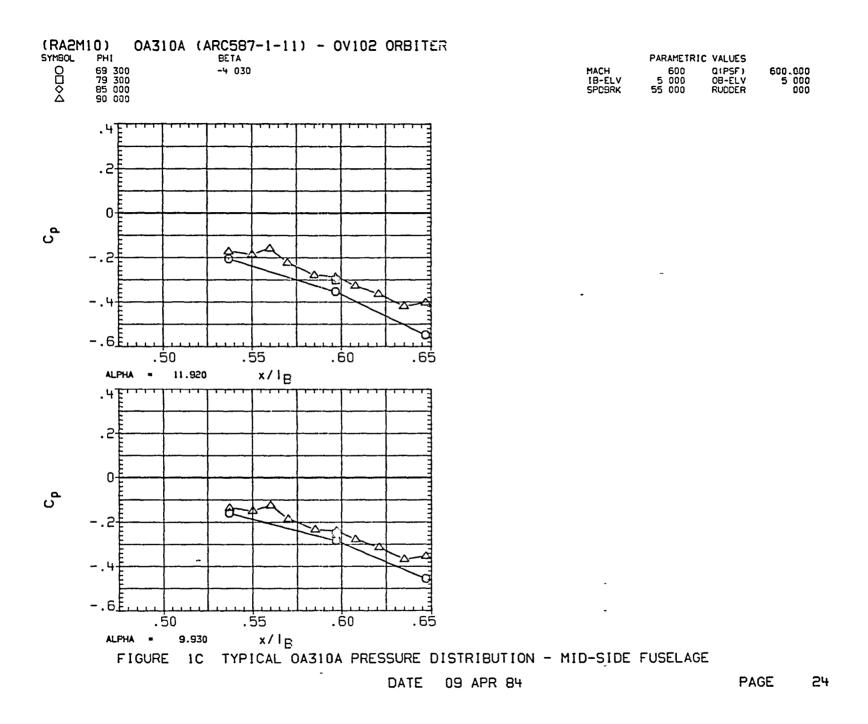


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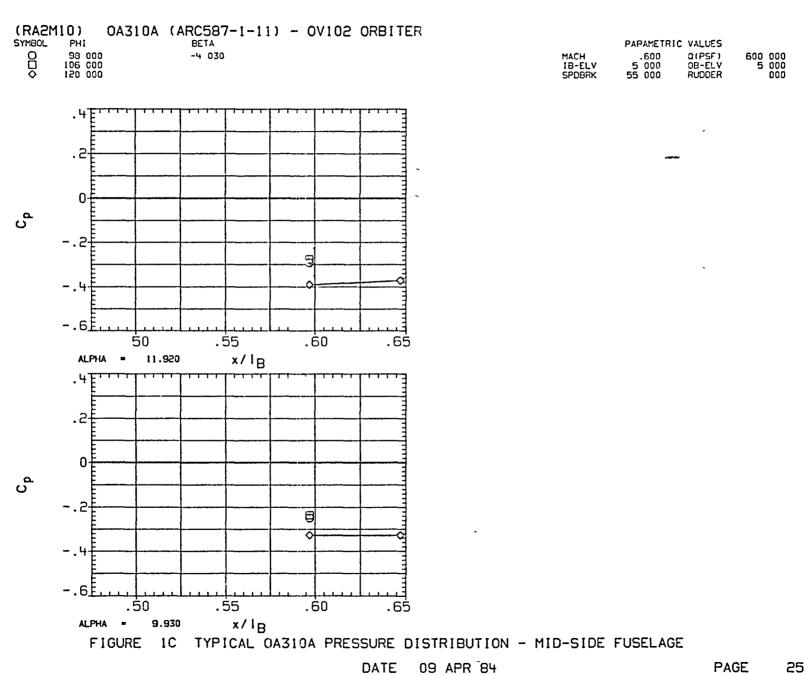


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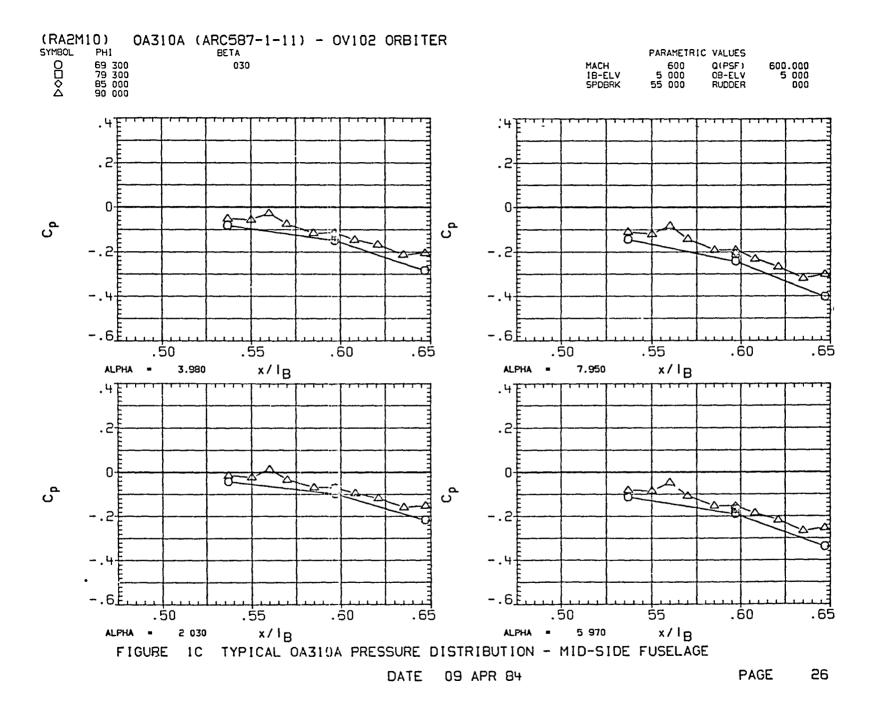


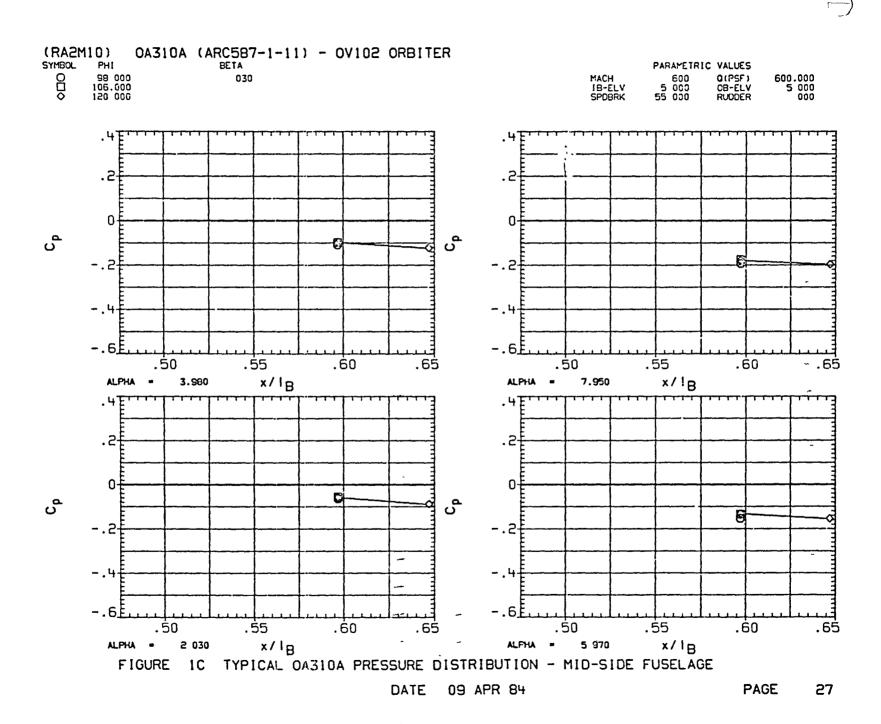


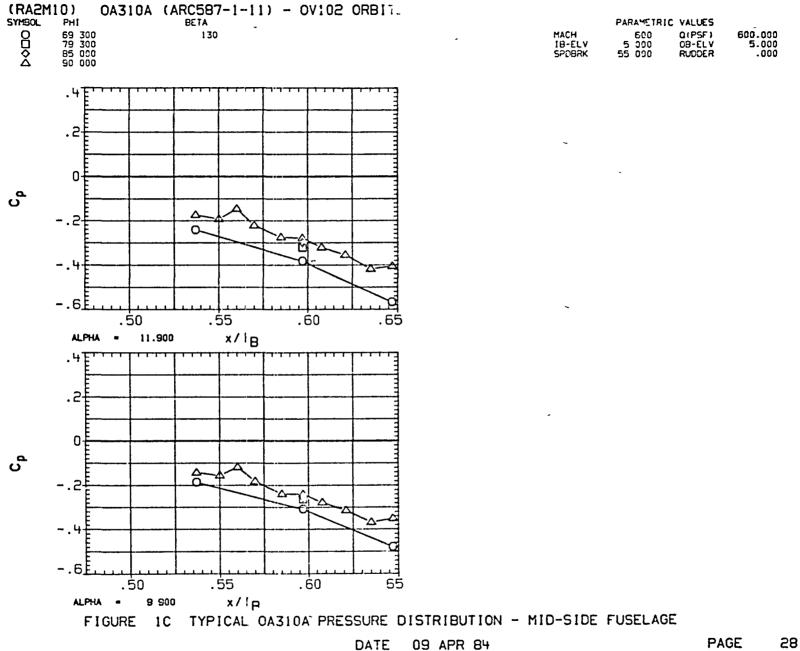
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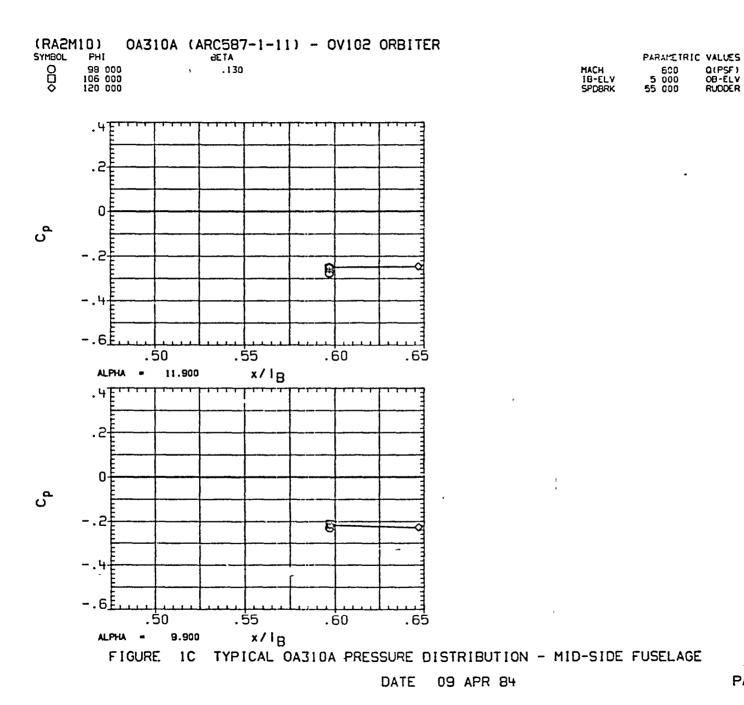
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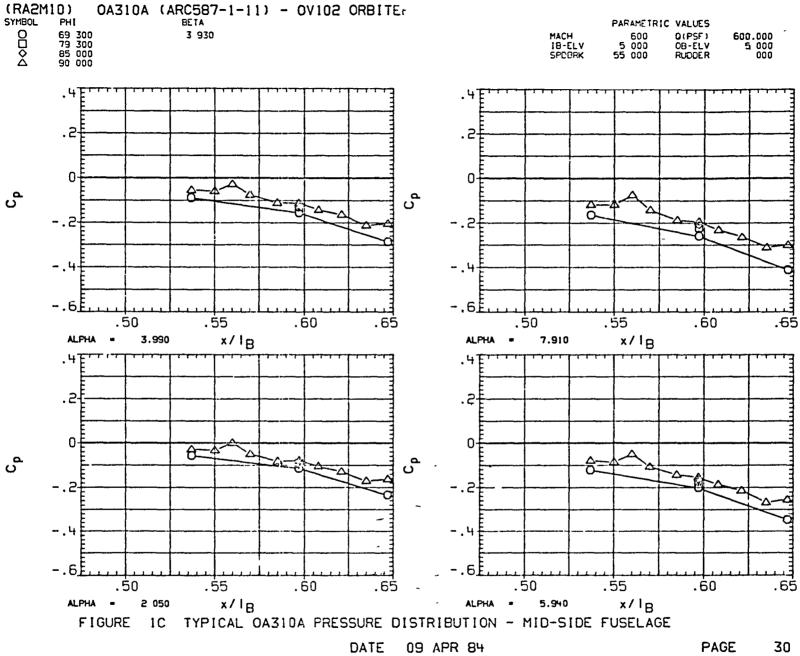


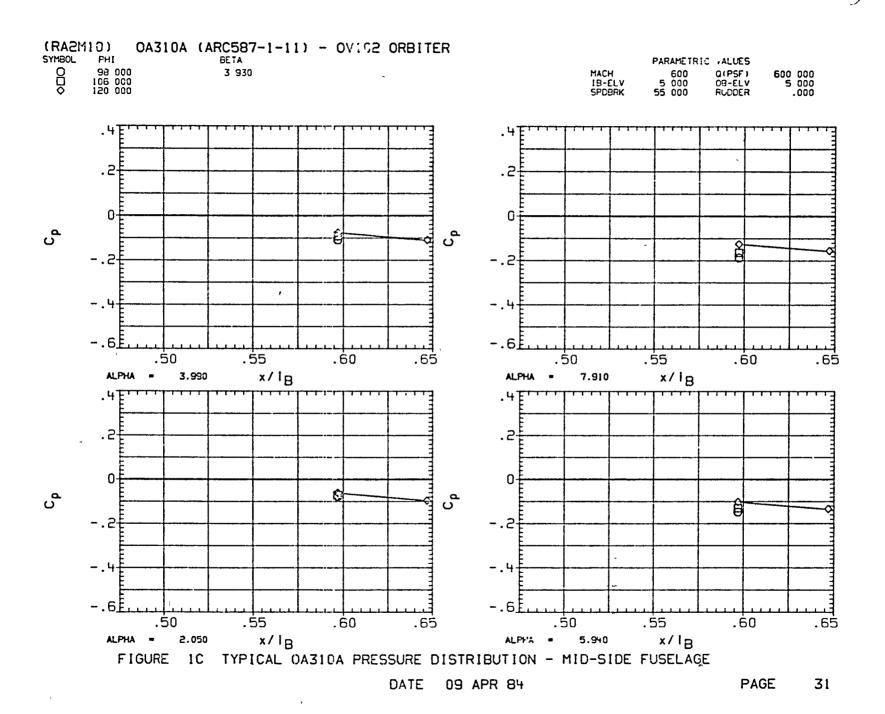


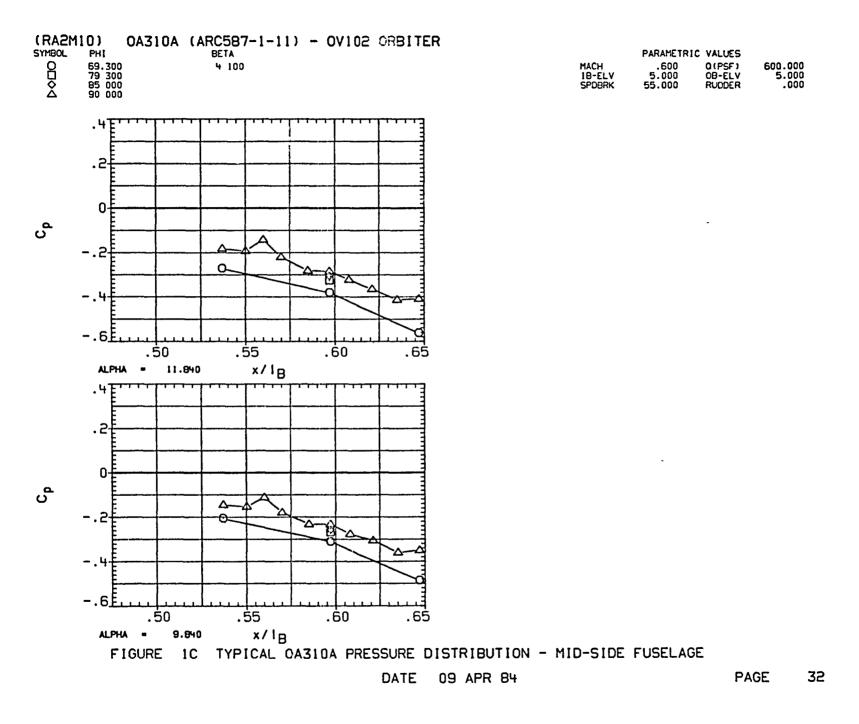
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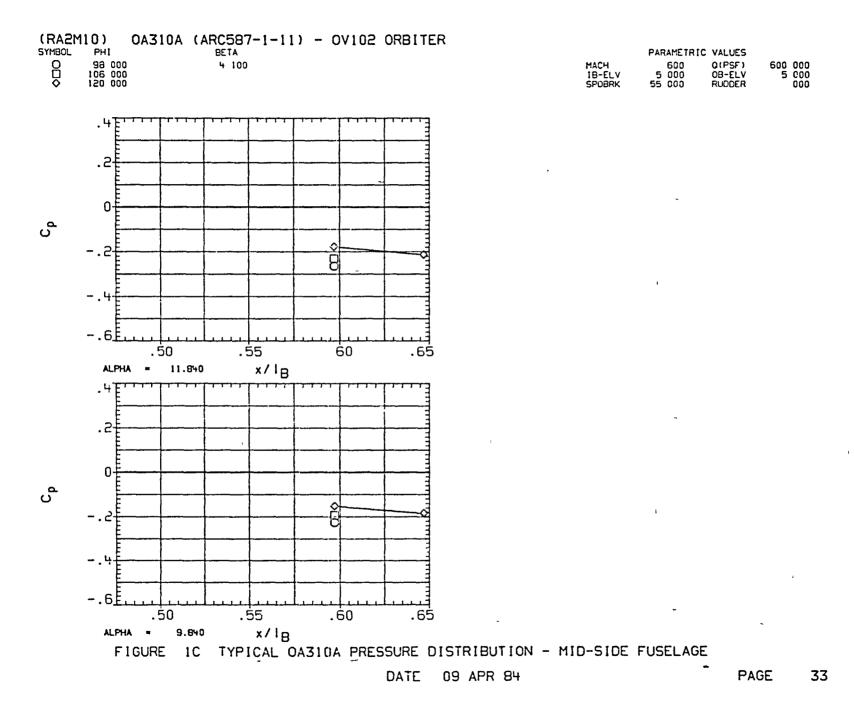
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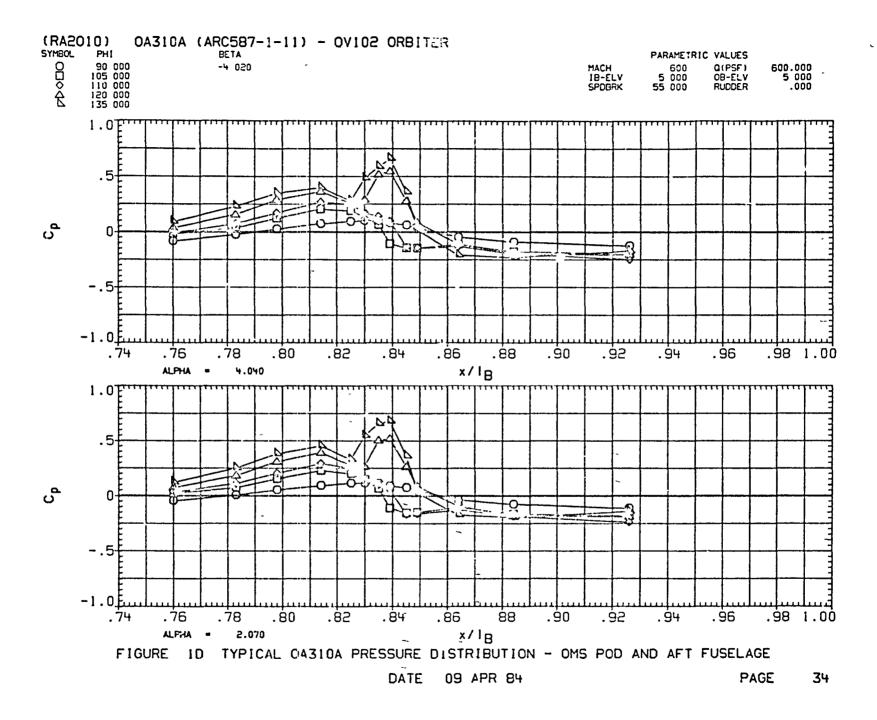






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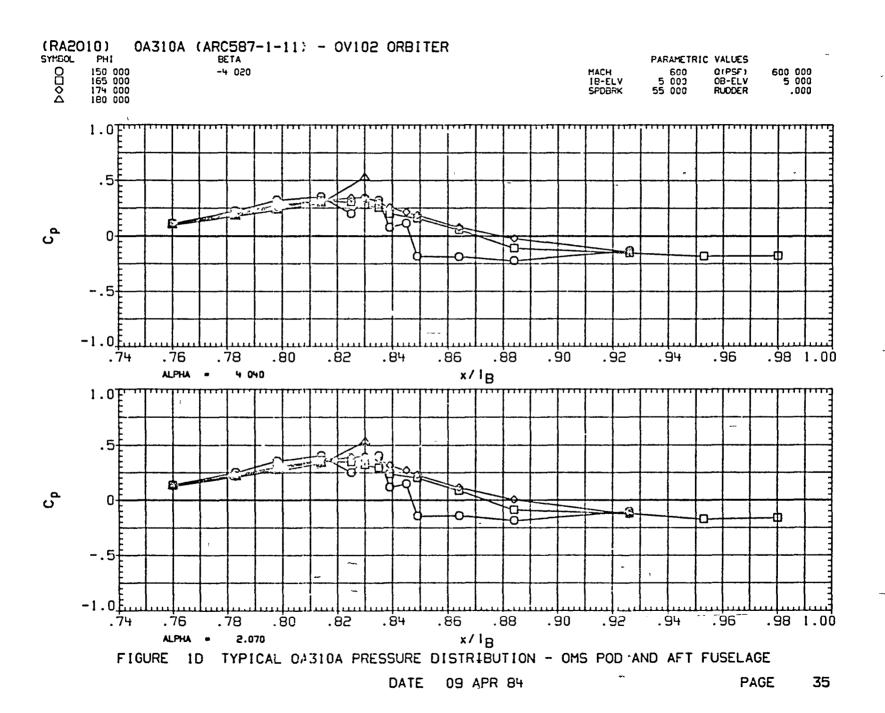


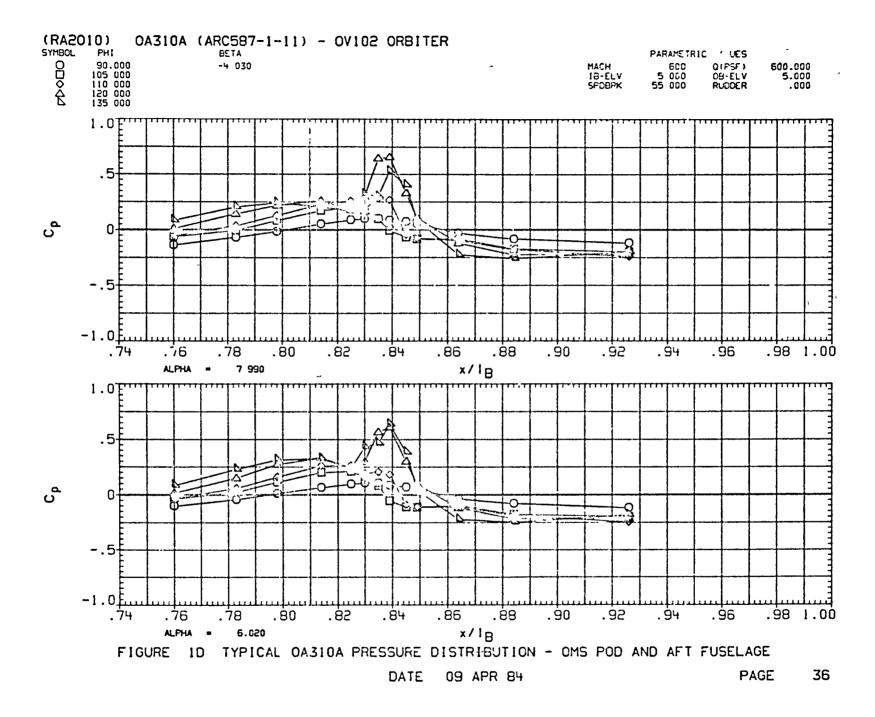


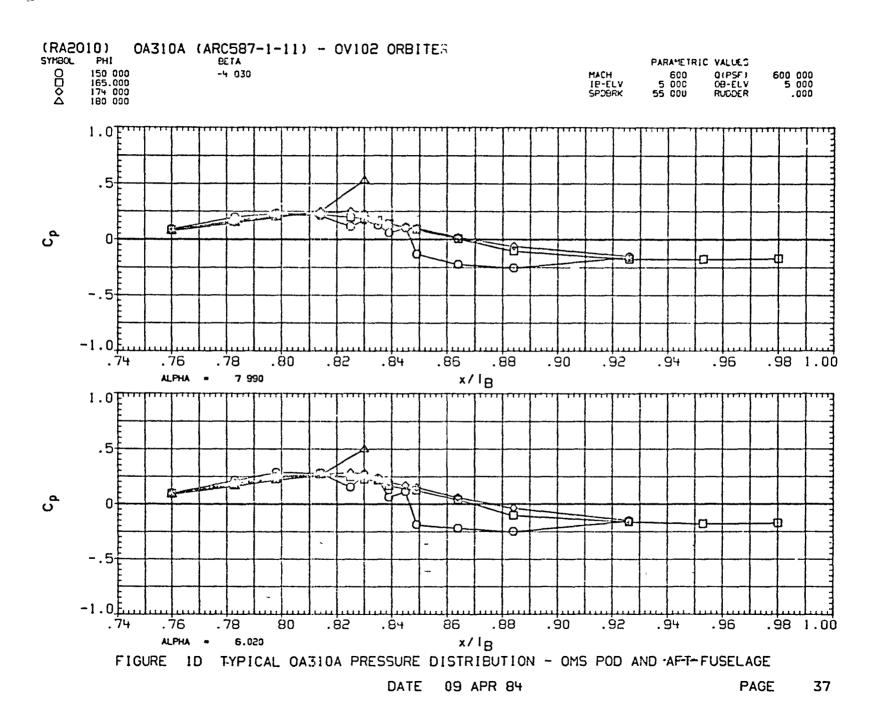
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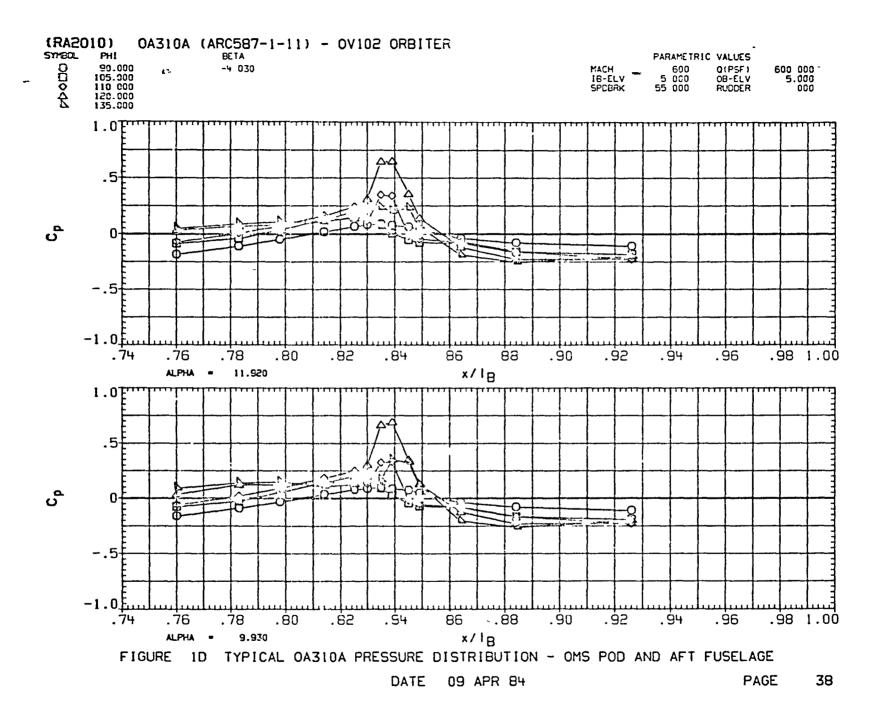
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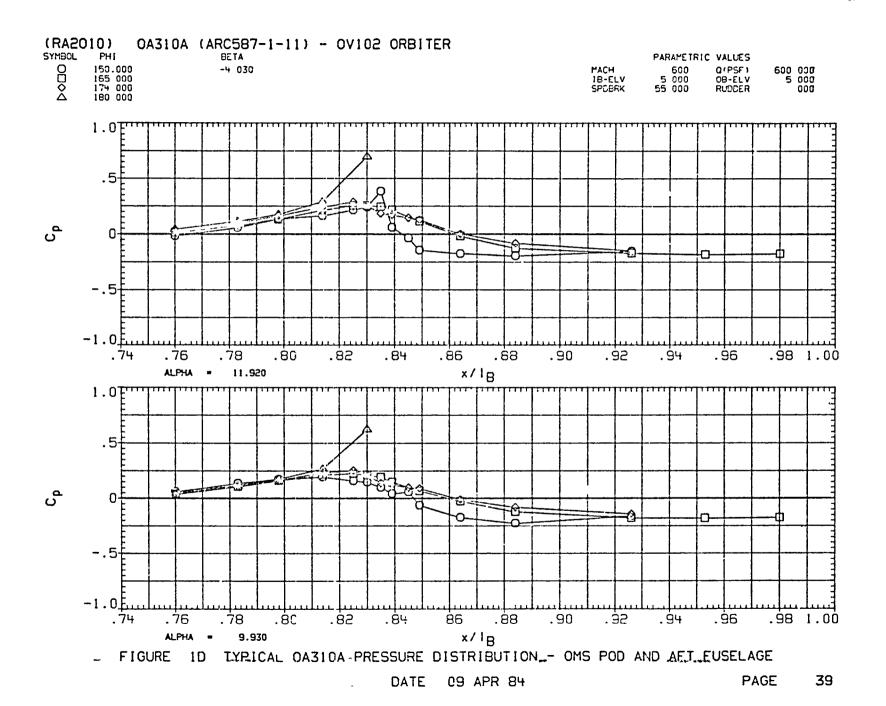
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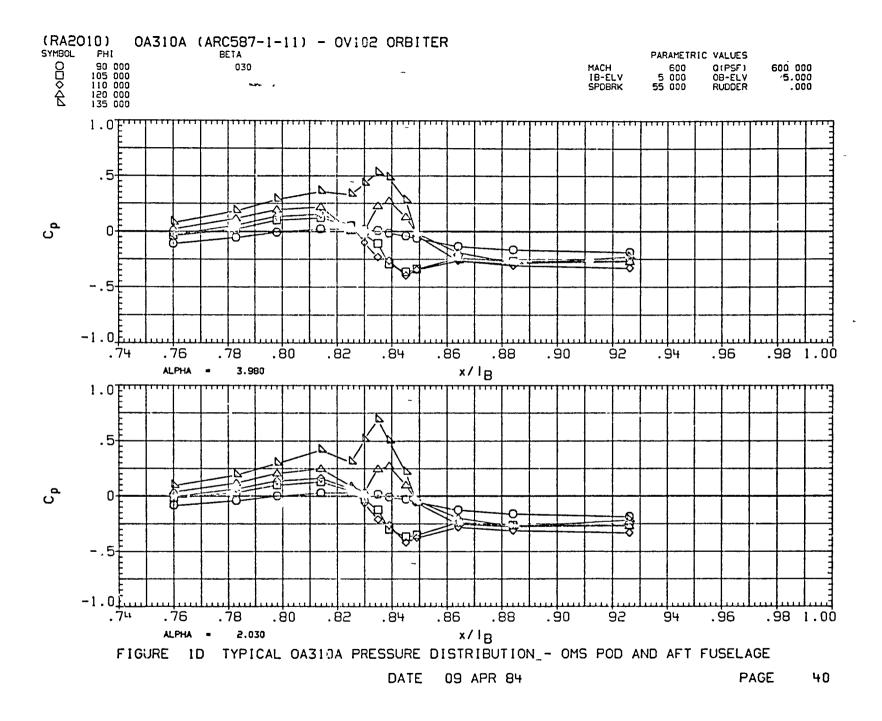


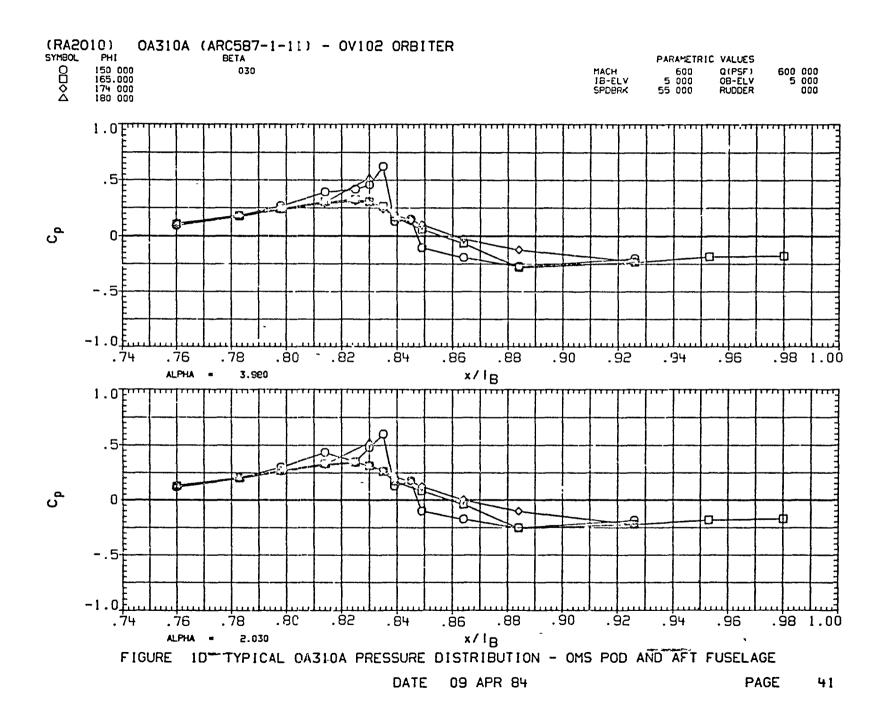


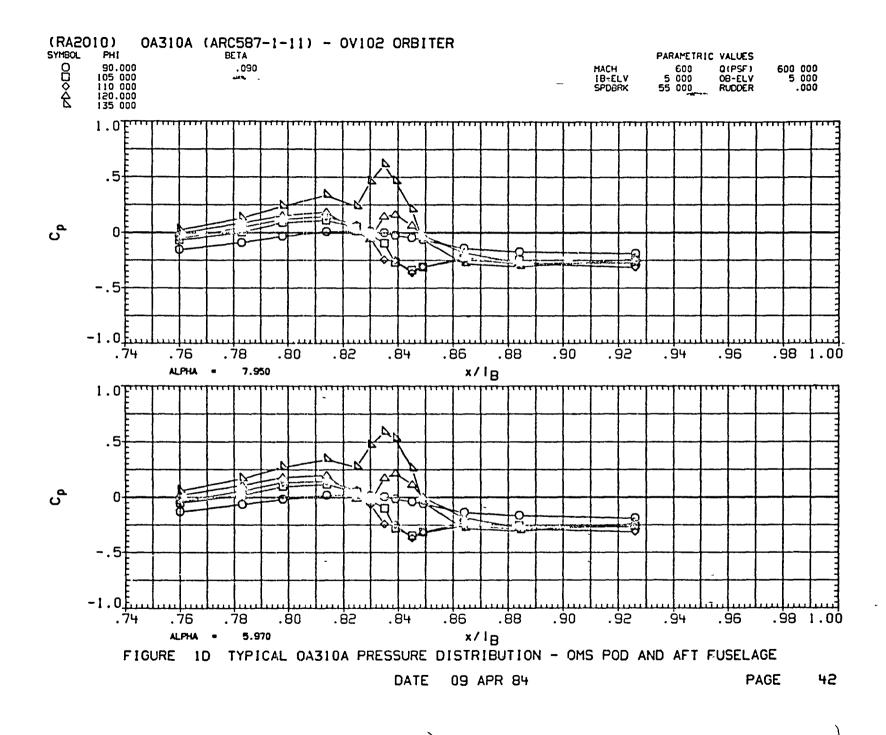


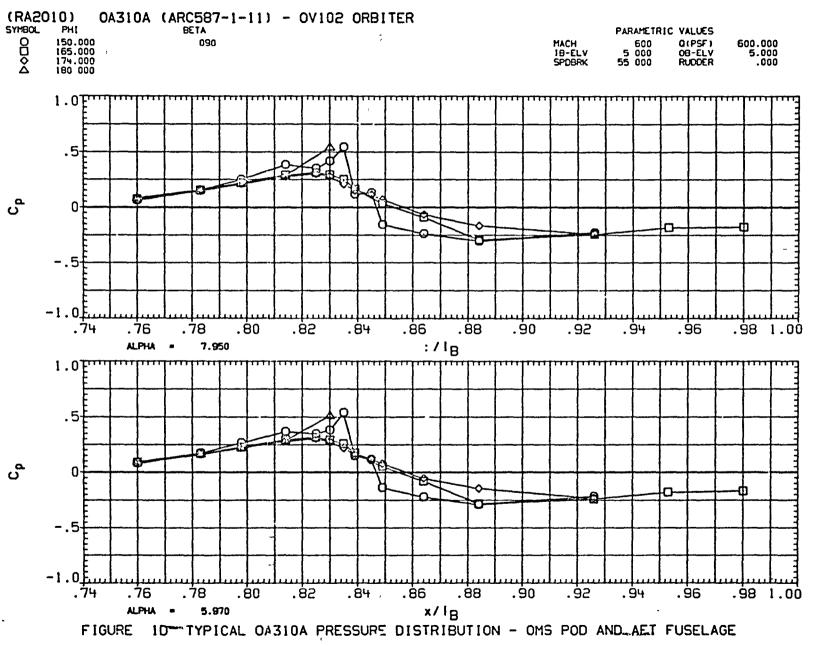






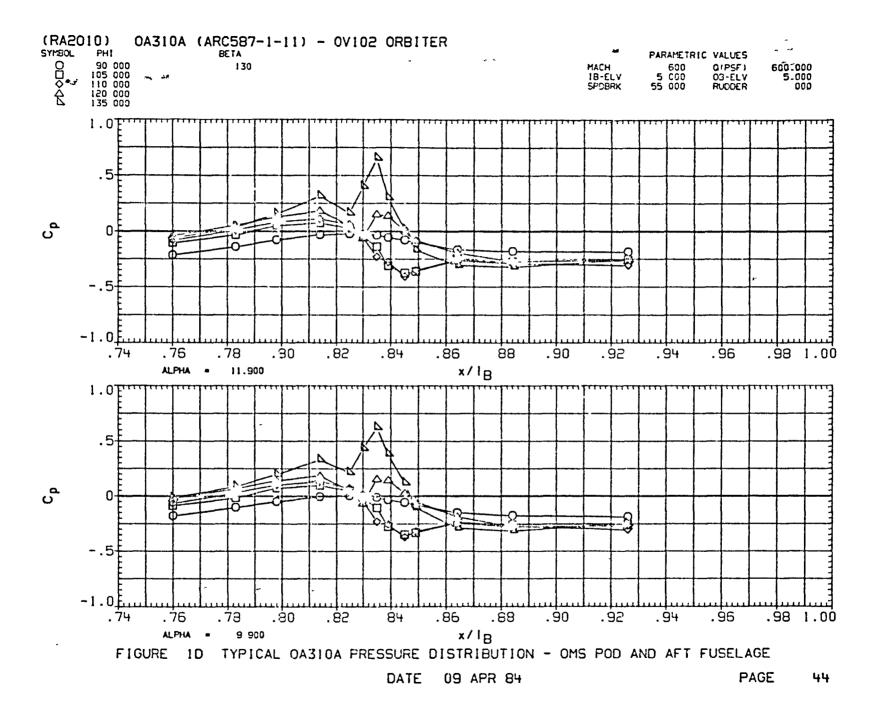


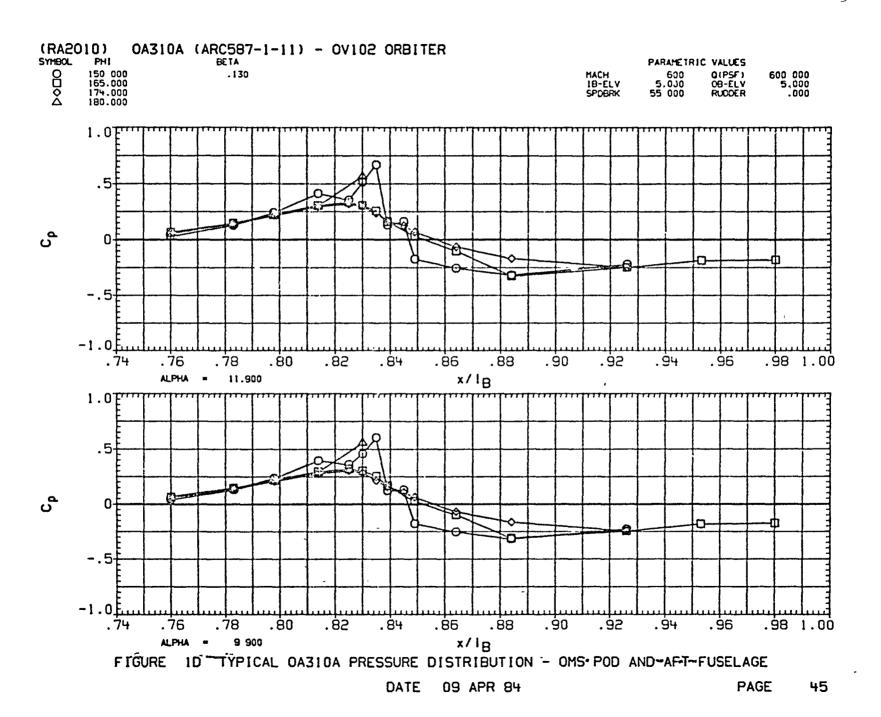


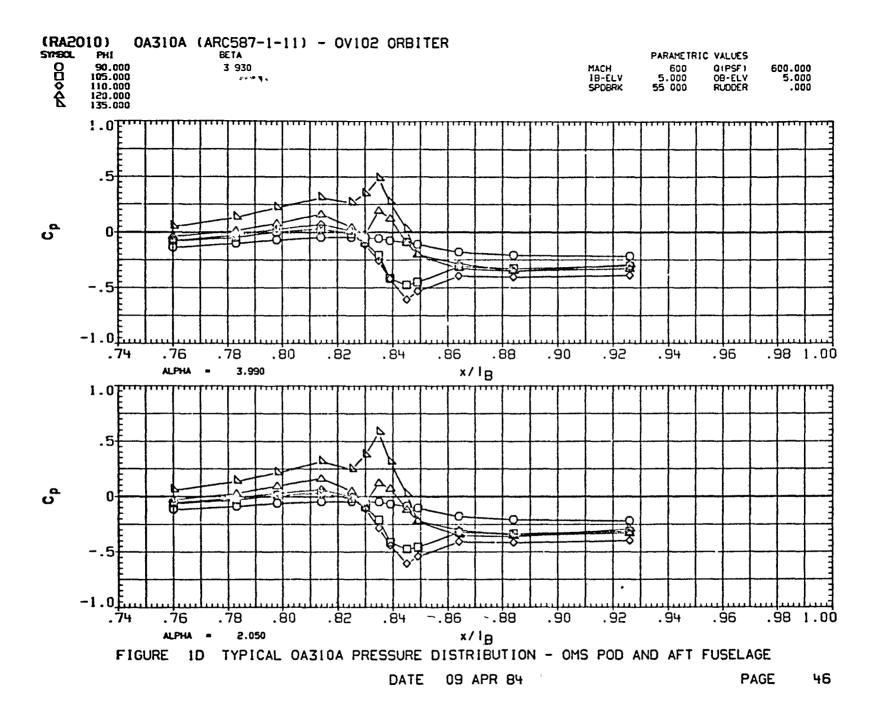


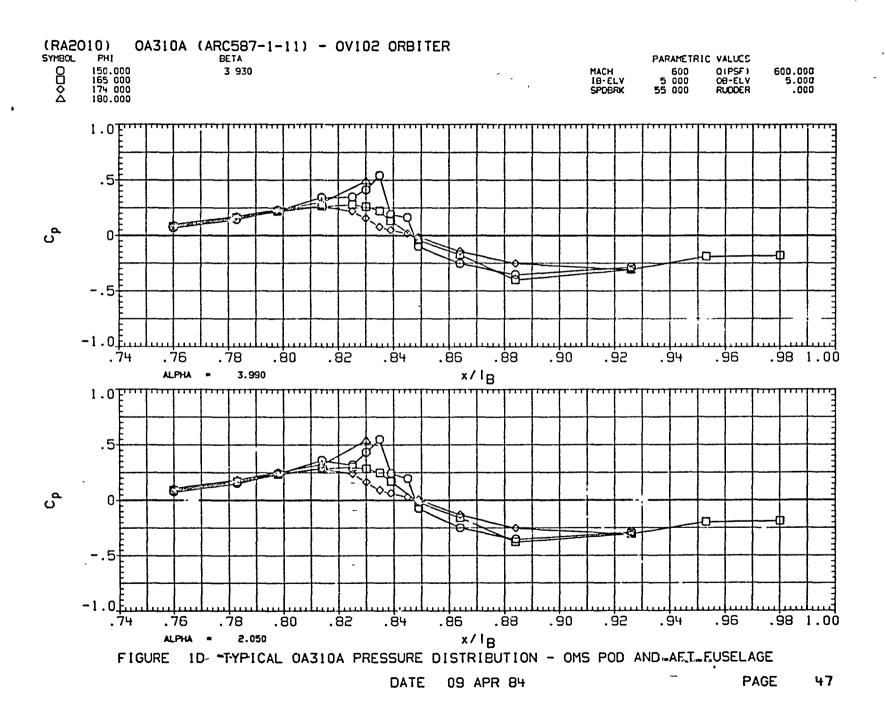
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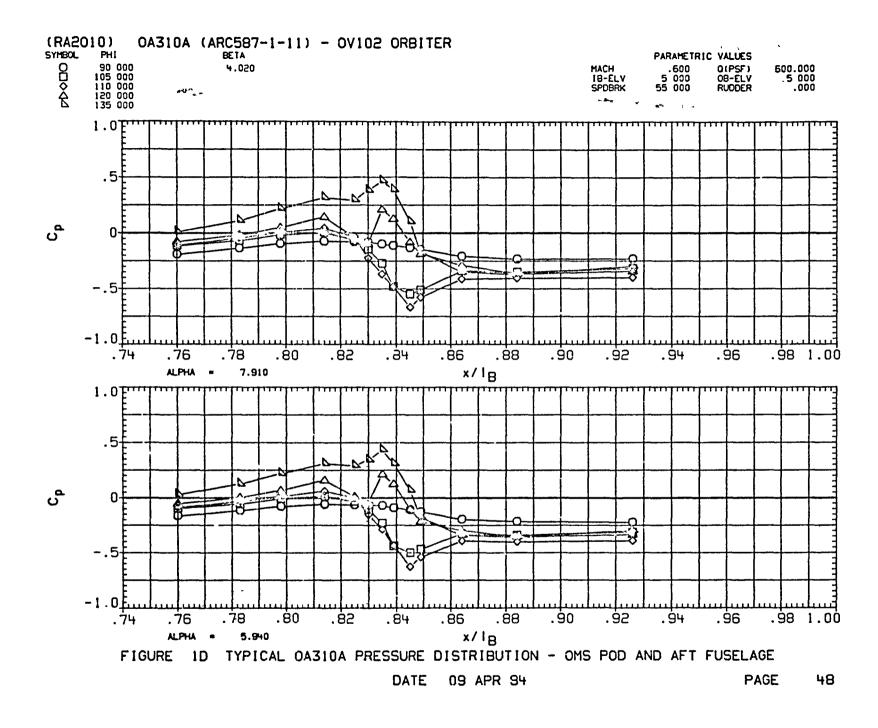
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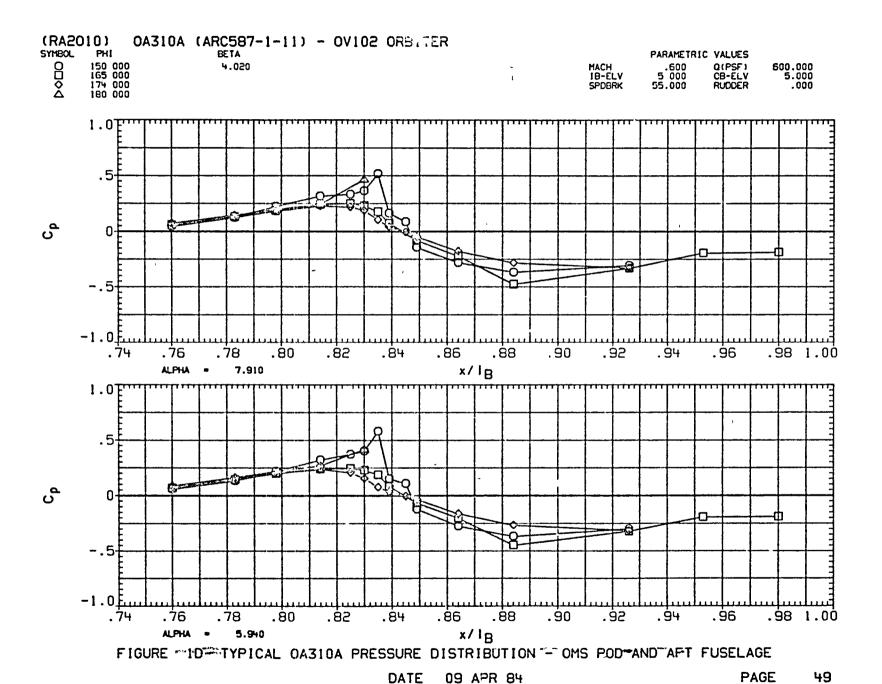


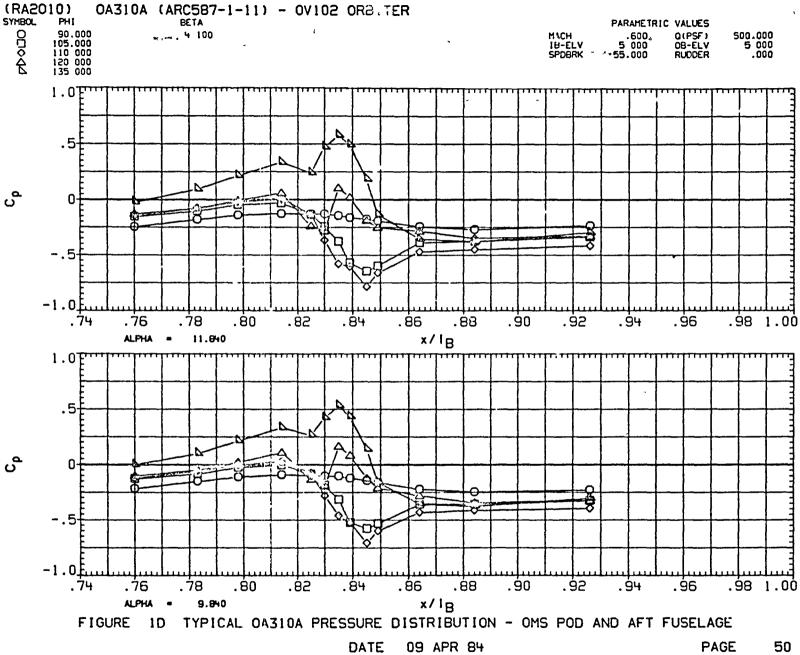


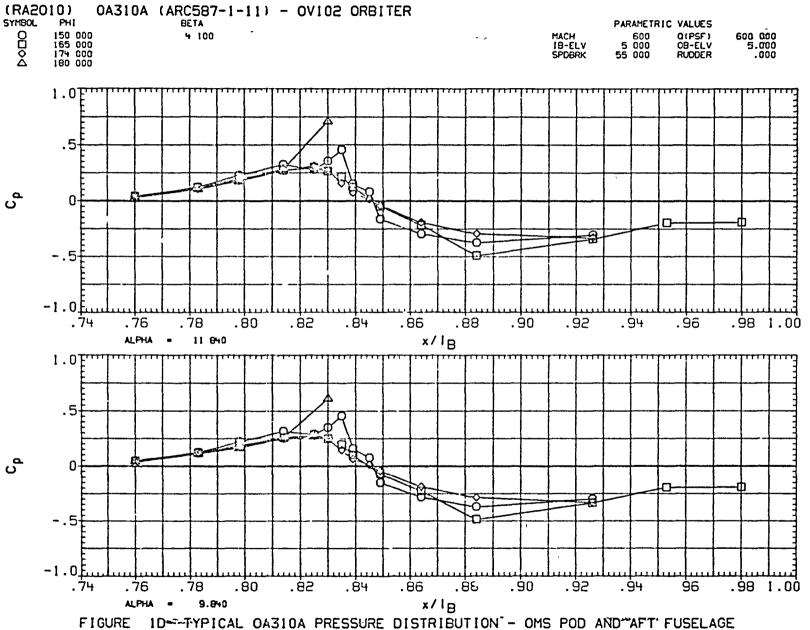






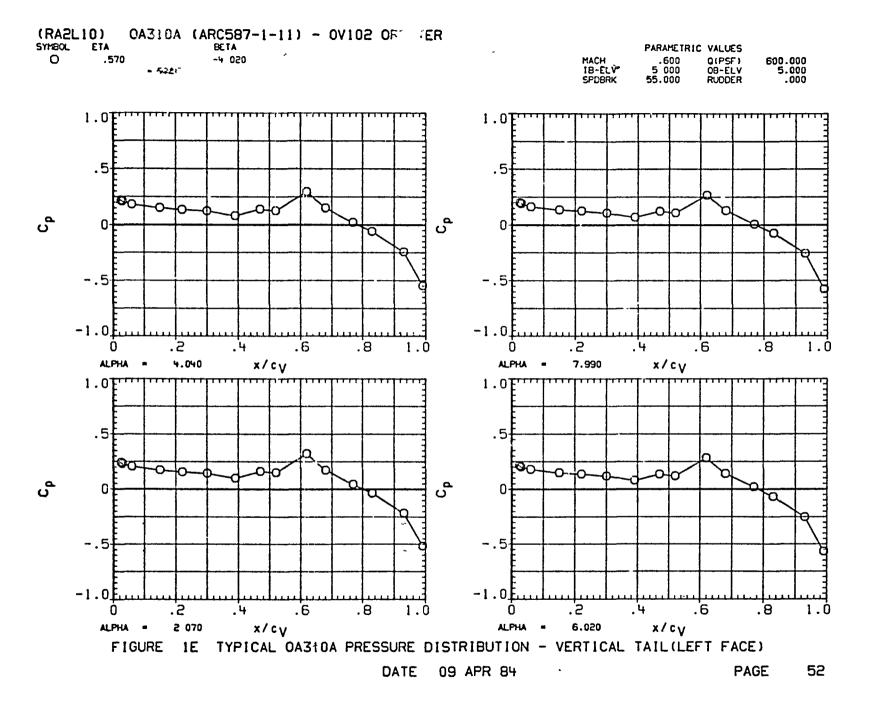






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(RA2L10) OA310A (ARC587-1-11) - OV102 ORBITER
SYMBOL ETA
O .570 -4.030

PARAMETRIC VALUES

MACH .600 Q(PSF) 600.000
1B-ELV 5 000 08-ELV 5 000
SPDBRK 55.000 RUDDER .000

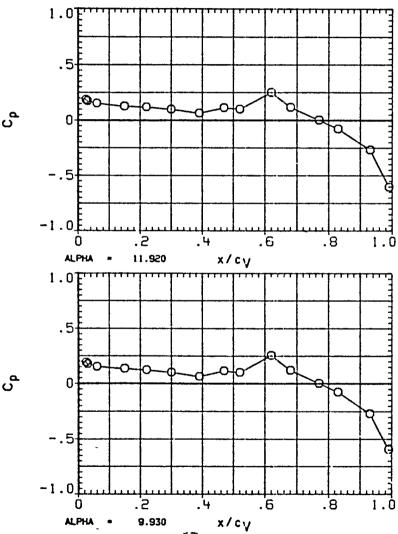
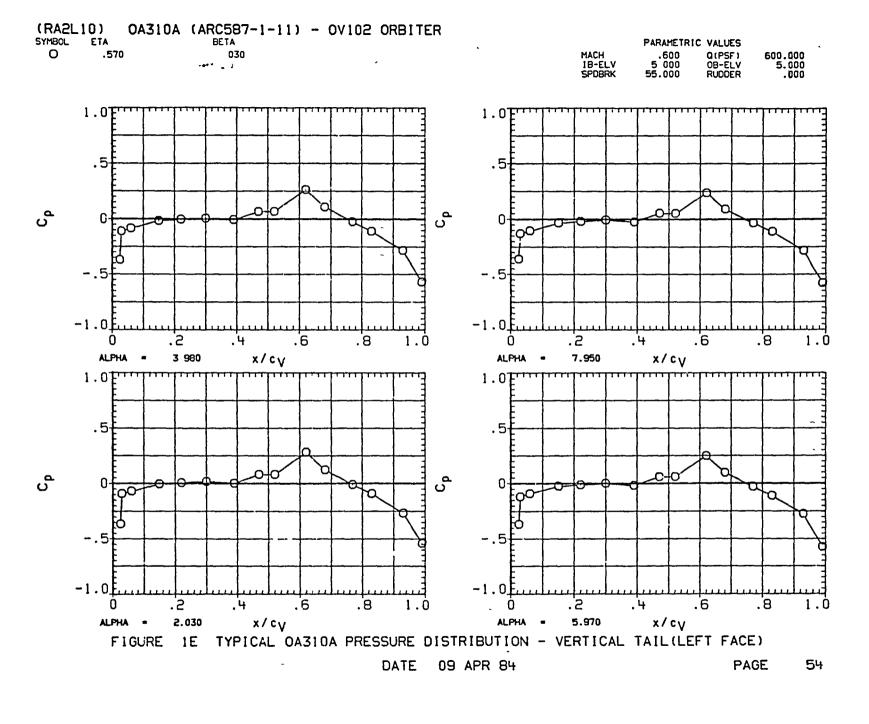


FIGURE 1E TYPICAL CA310A PRESSURE DISTRIBUTION - VERTICAL TATE (LEFT FACE)

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(RA2L10) OA310A (ARC587-1-11) - OV102 OTBITER
SYMBOL ETA
O .570 .130

PARAMETRIC VALUES

HACH .600 0(PSF) 600 000

18-ELV 5.000 08-ELV 5.000

SPDBRK 55.000 RUDDER .000

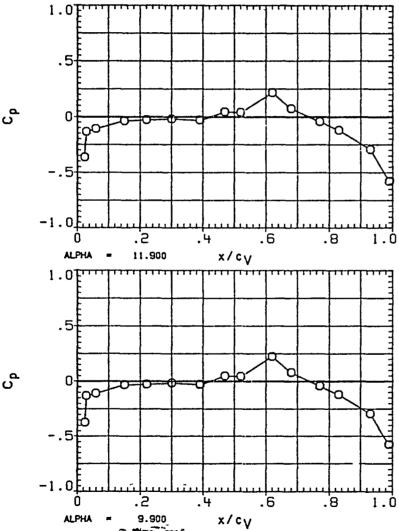
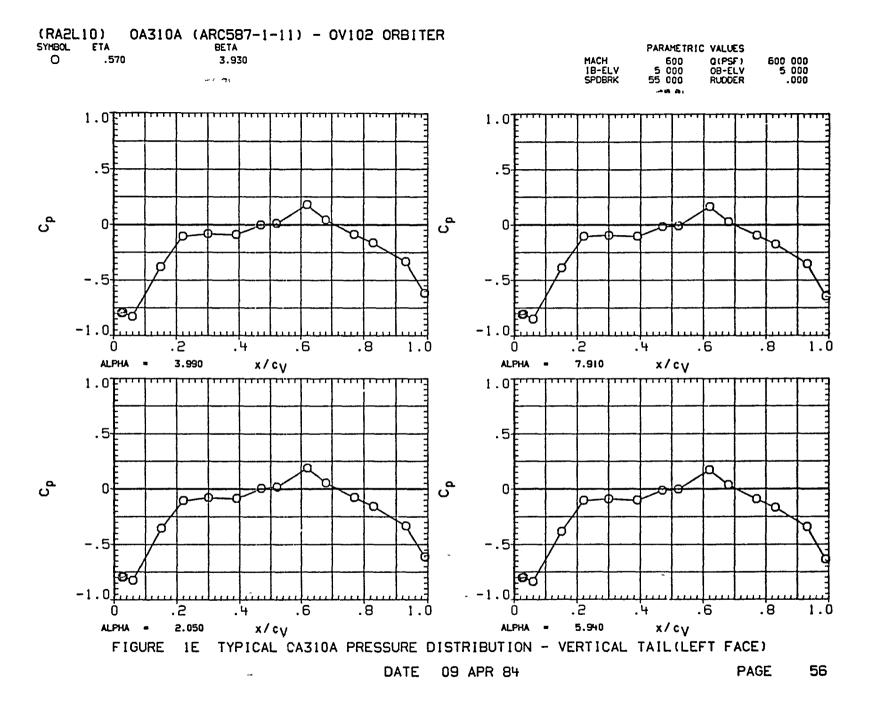
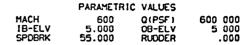


FIGURE TYPICAL 043104 PRESSURE DISTRIBUTION - VERTICAL TAIL (LEFT FACE)







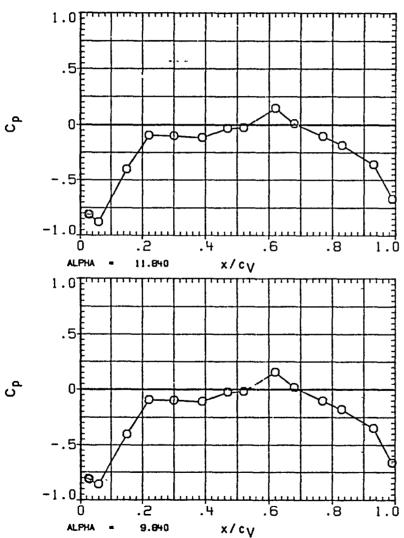
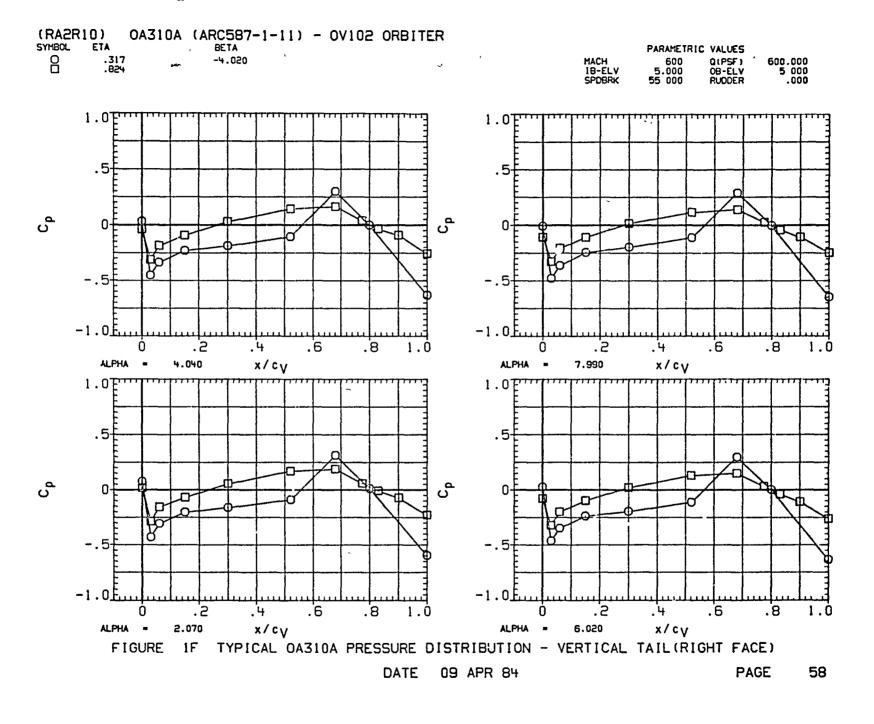
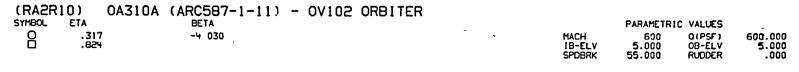


FIGURE TETTYPICAL OASIOA PRESSURE DISTRIBUTION - VERTICAL TAIL (LEFT FACE)





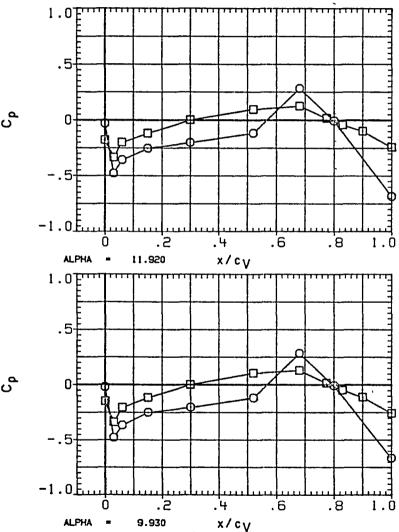
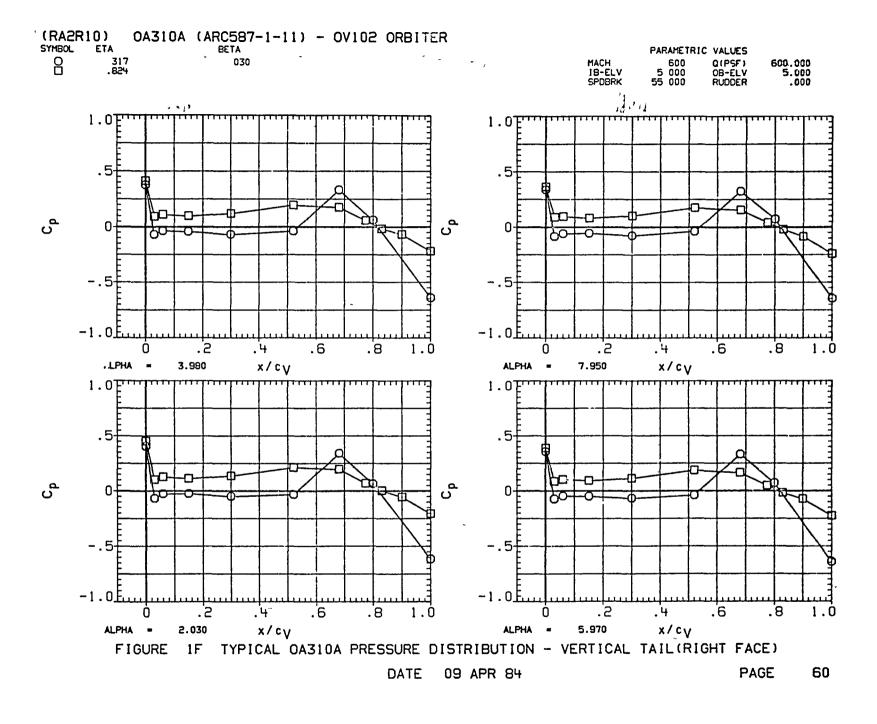
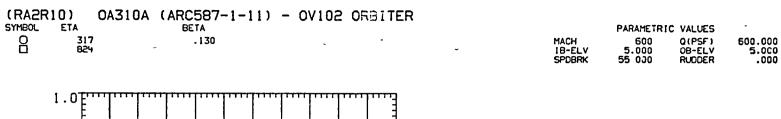


FIGURE 1F-TYPICAL 0A310A PRESSURE DISTRIBUTION - VERTICAL TAIL (RIGHT FACE)





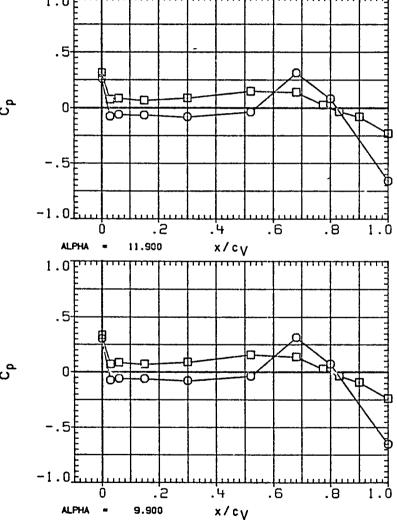
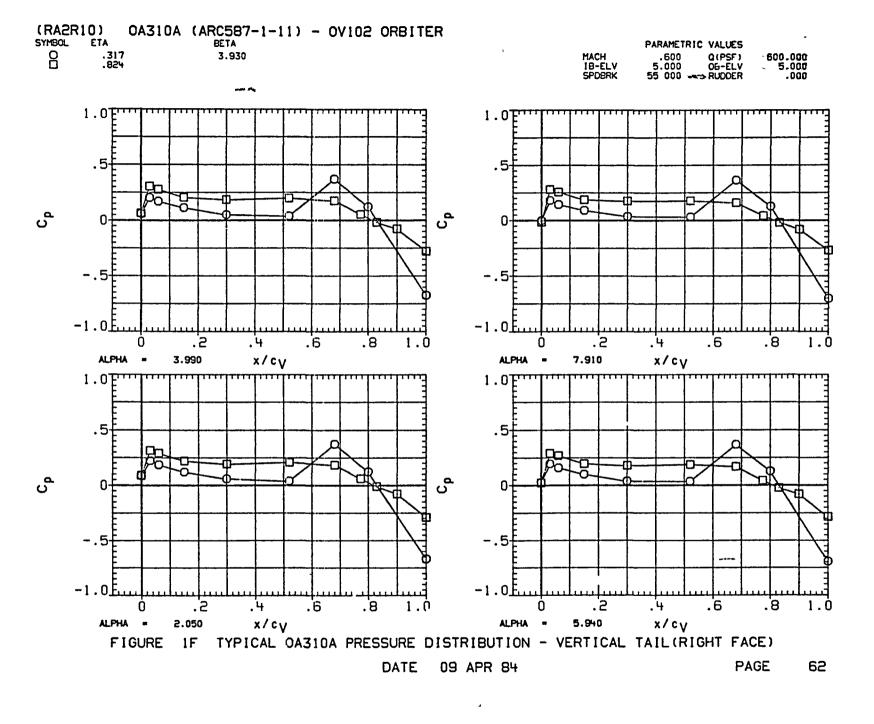


FIGURE 1F TYPICAL 0A310A PRESSURE DISTRIBUTION - VERTICAL TAIL (RIGHT FACE)

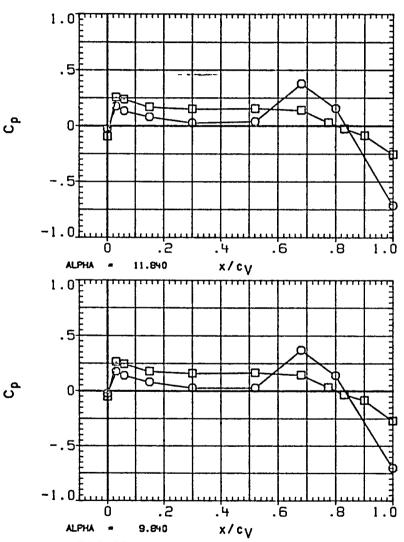
DATE 09 APR 84



(RA2R10) OA310A (ARC587-1-11) - OV102 ORBITER
SYMBOL ETA
O .317
BETA
4.100

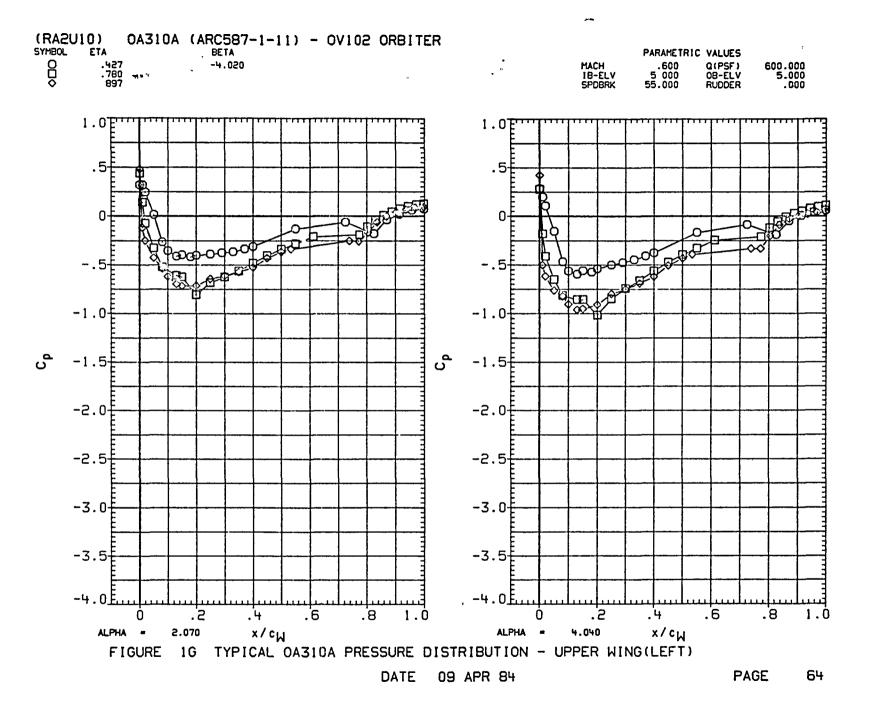
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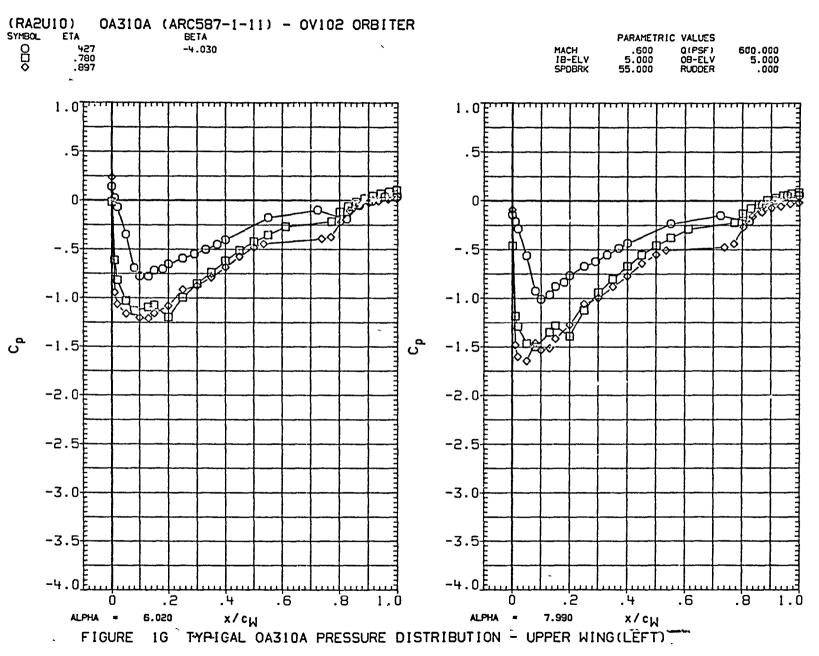
MACH .600 Q(PSF) 600.000
1B-ELV 5.000 0B-ELV 5.000
SPDBRK 55.000 RUDDER .000



IGURE IF TYPICAL 0A310A PRESSURE DISTRIBUTION - VERTICAL TAIL (RIGHT FACE)

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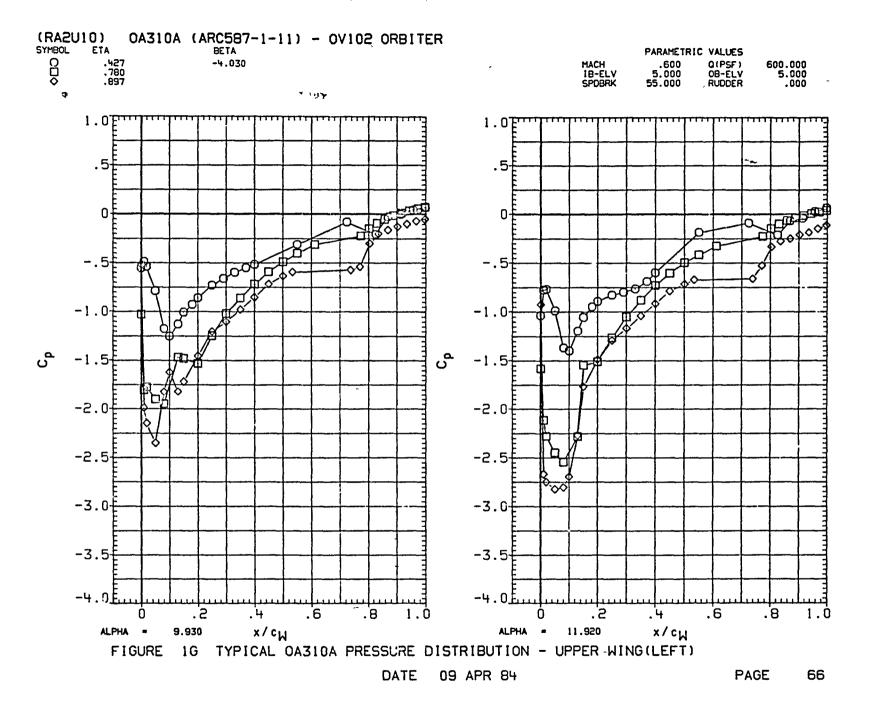


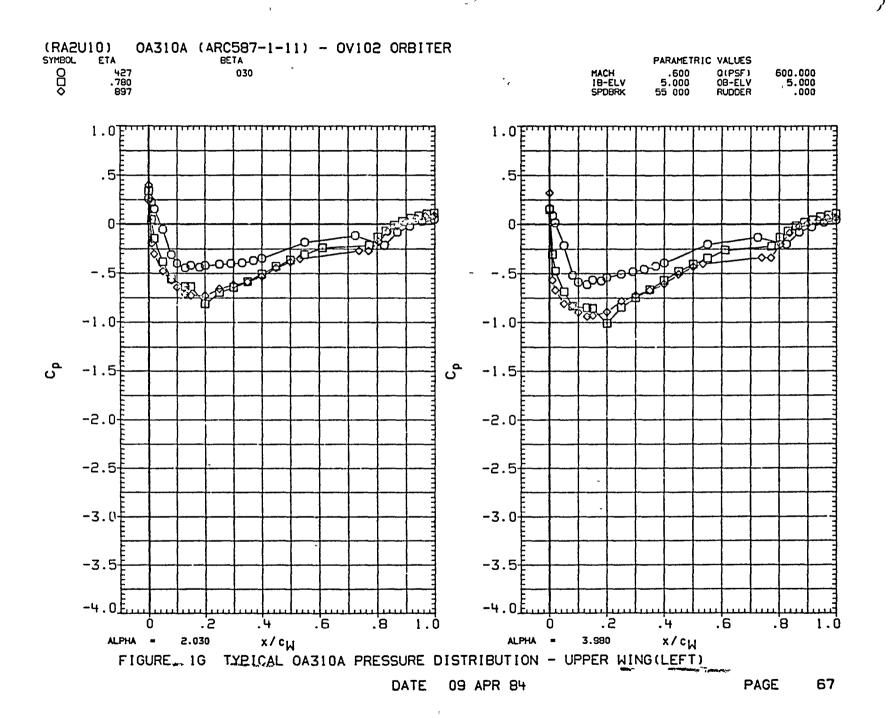


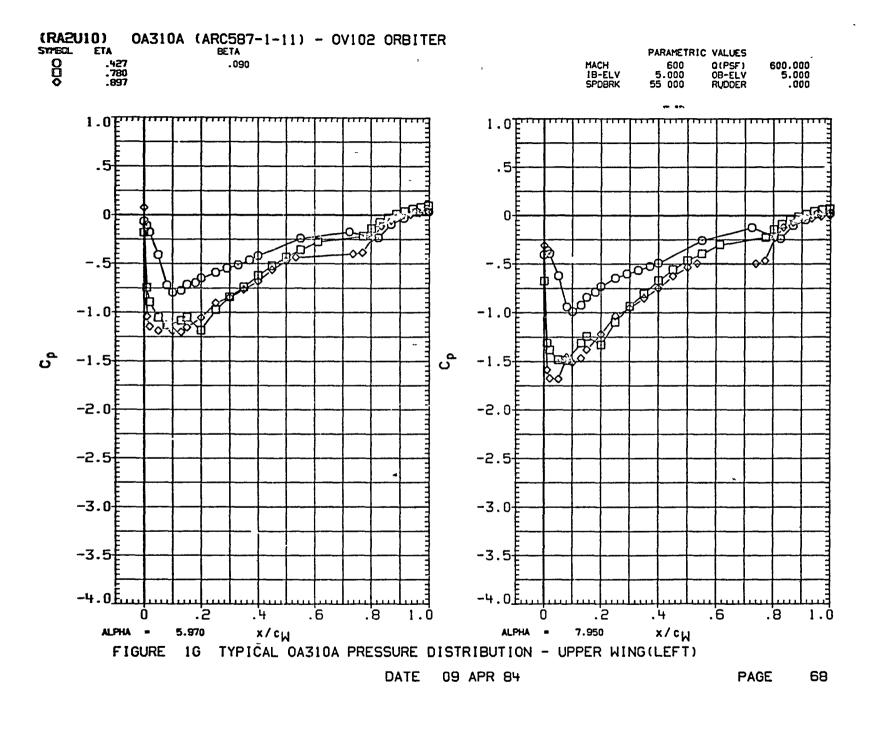
DATE 09 APR 84

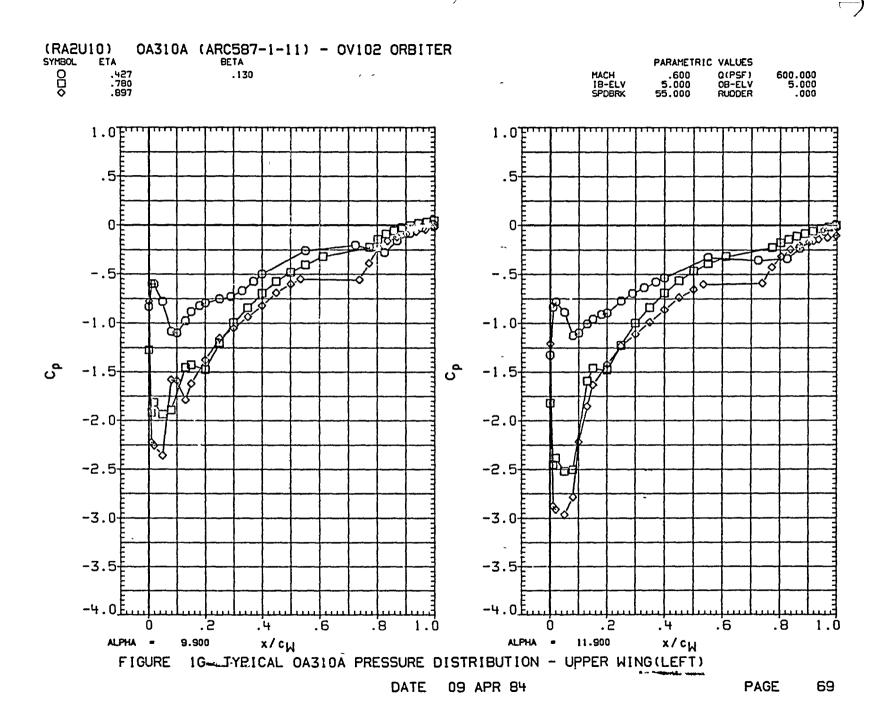
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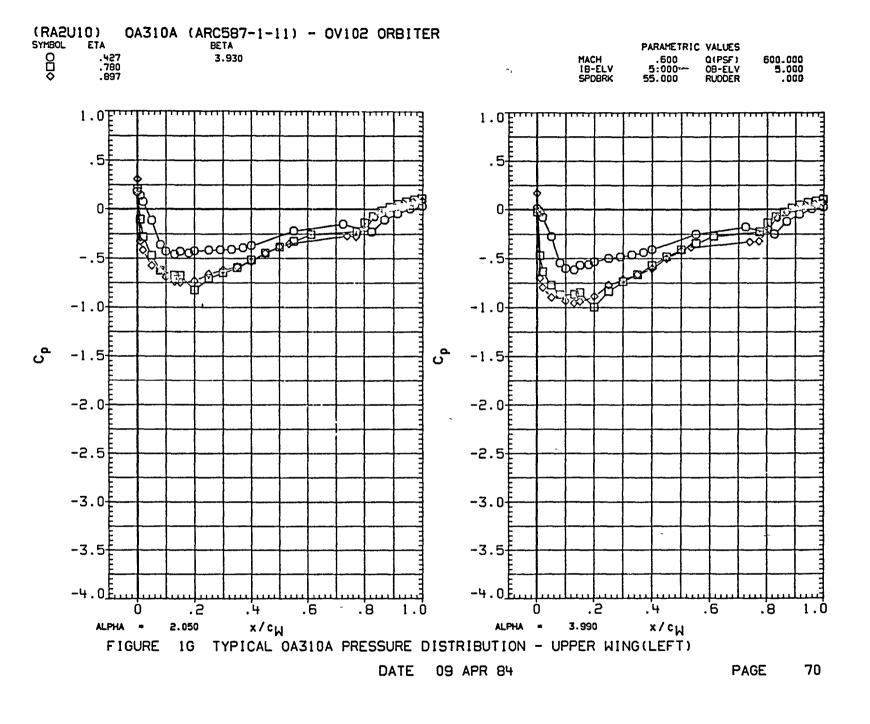
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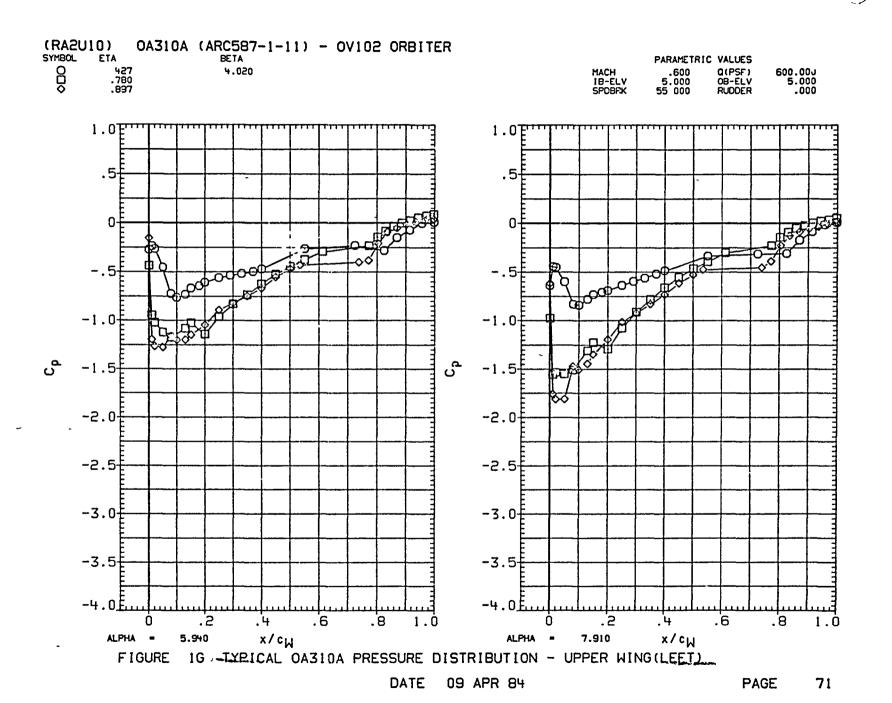


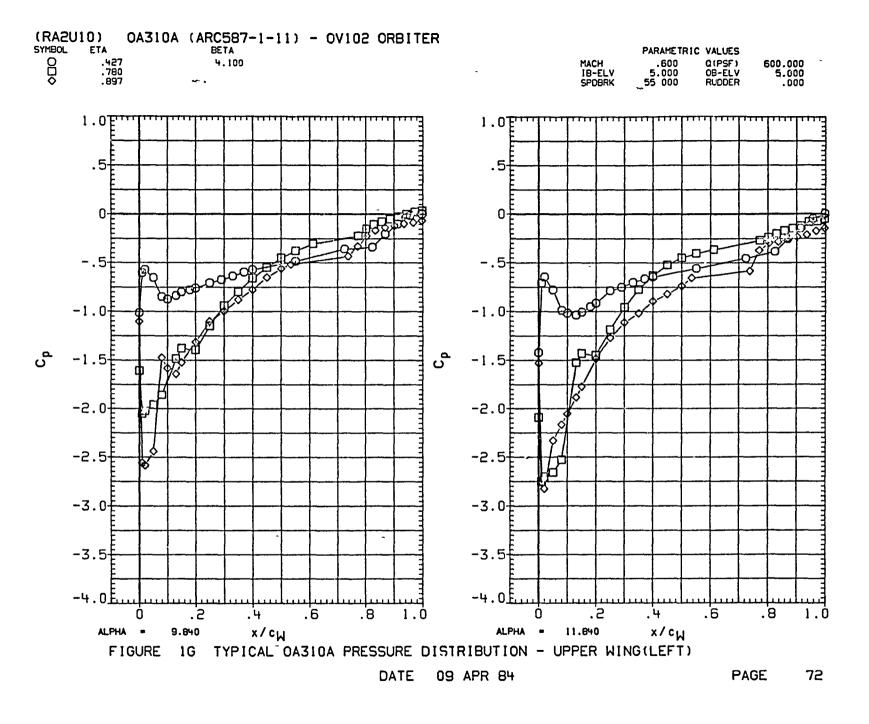


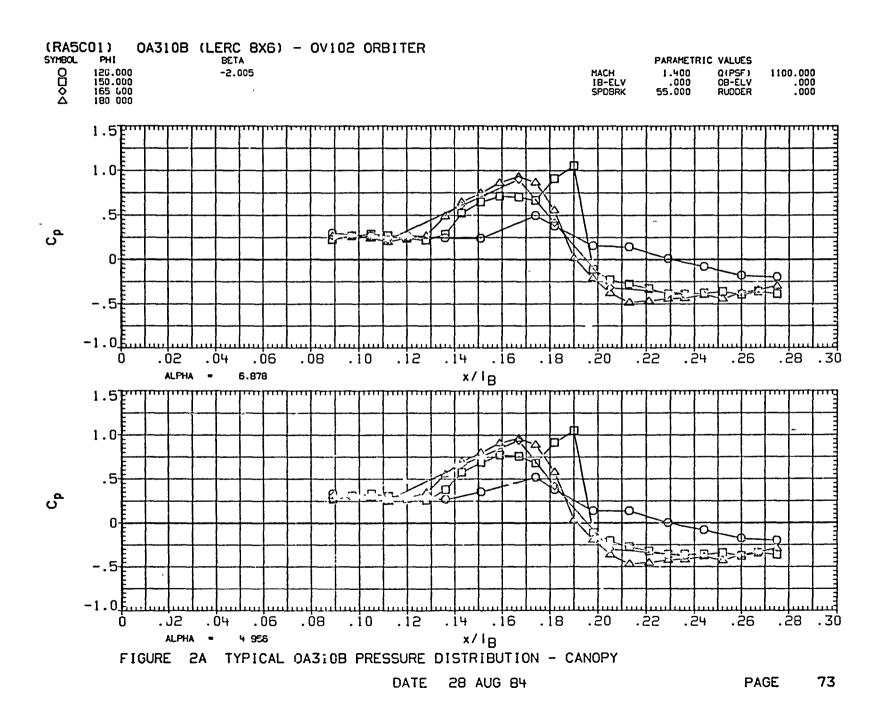


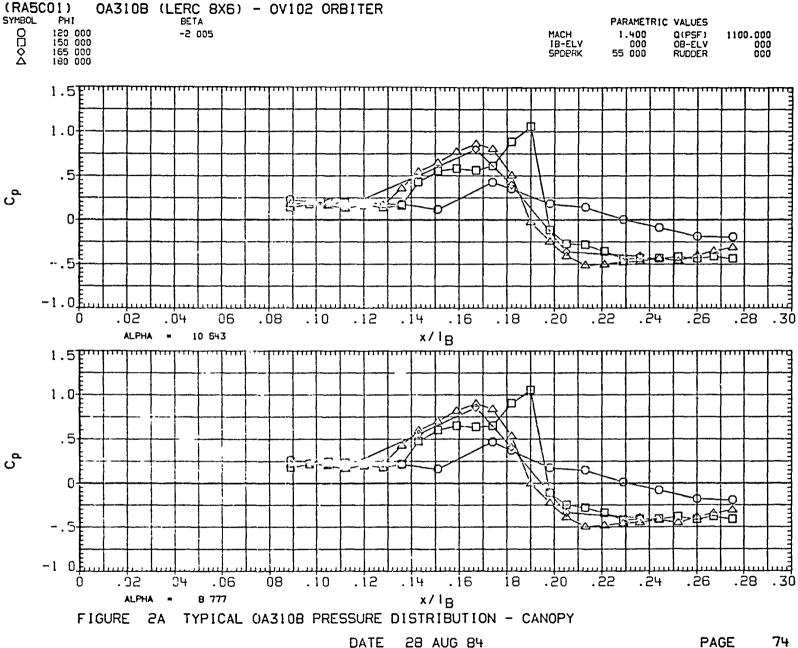


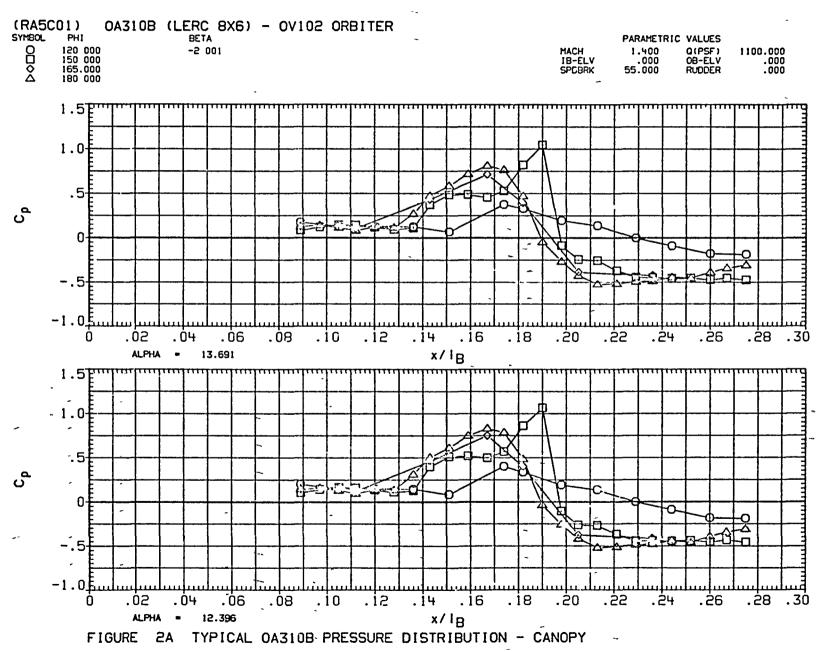






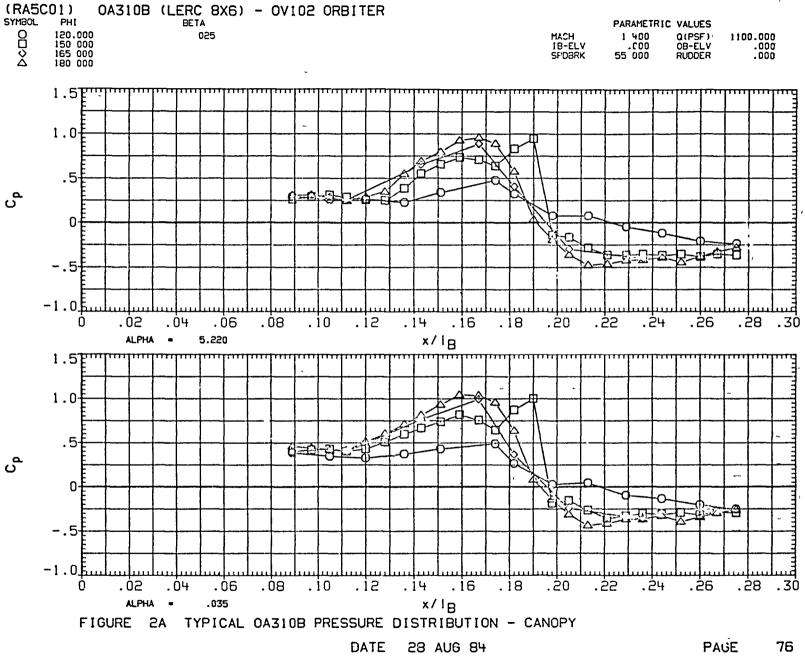


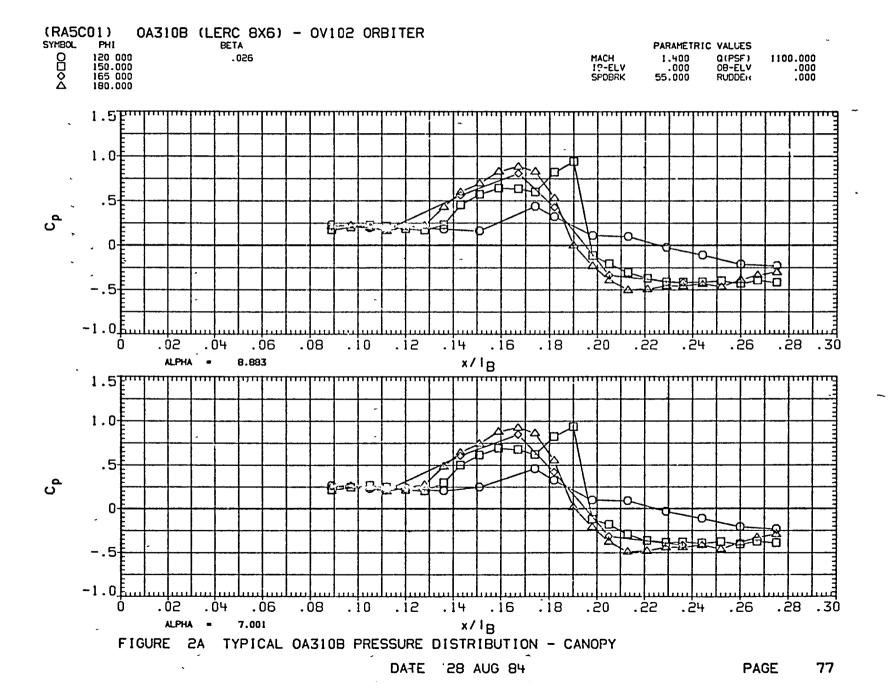


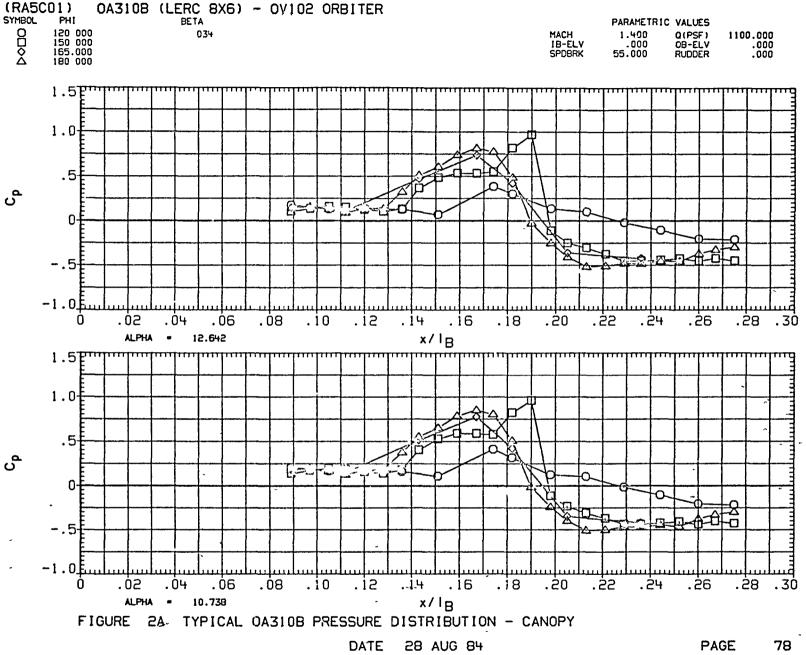


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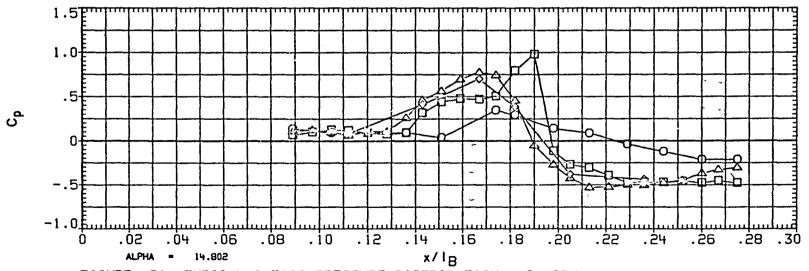
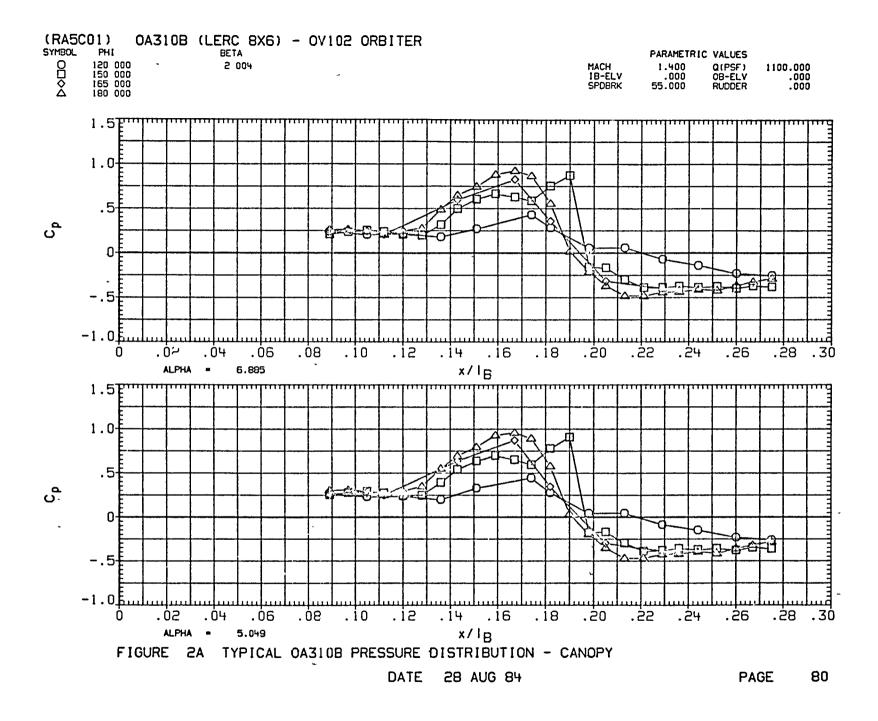
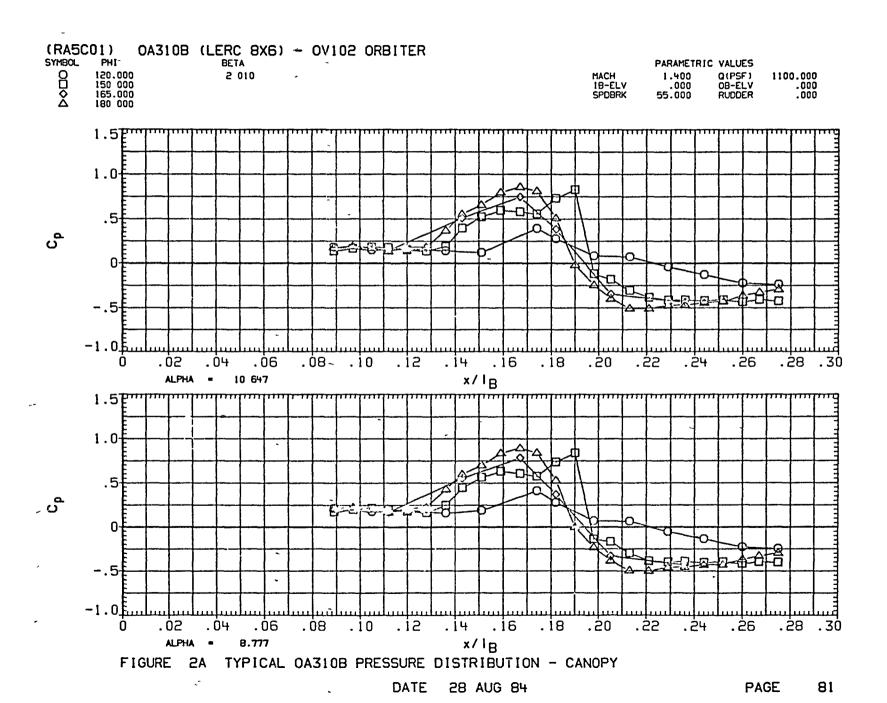


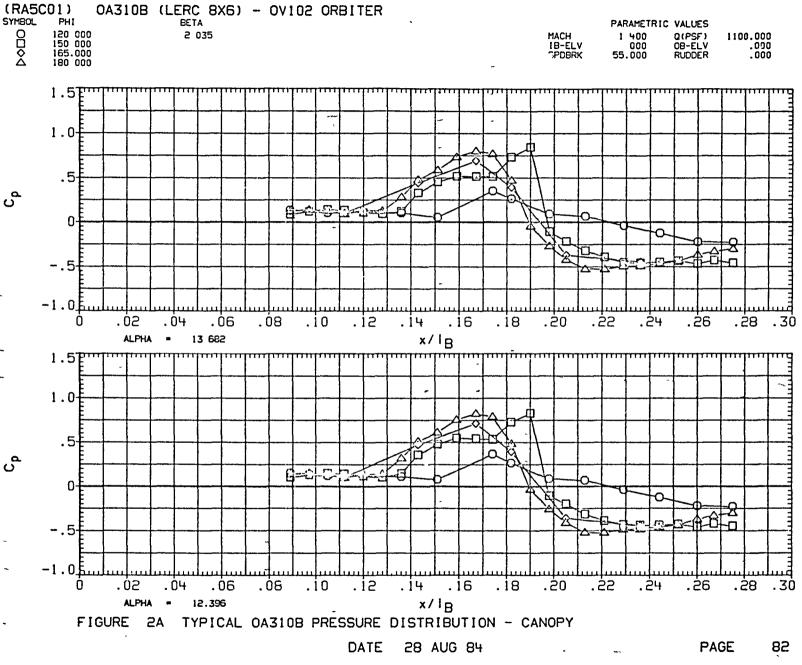
FIGURE 2A TYPICAL 0A310B_PRESSURE DISTRIBUTION - CANOPY

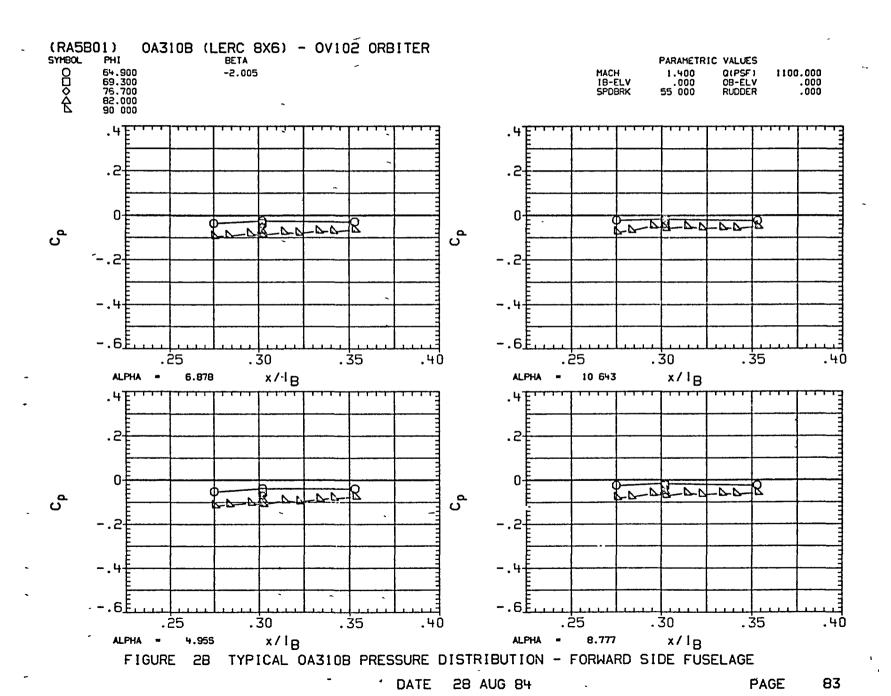
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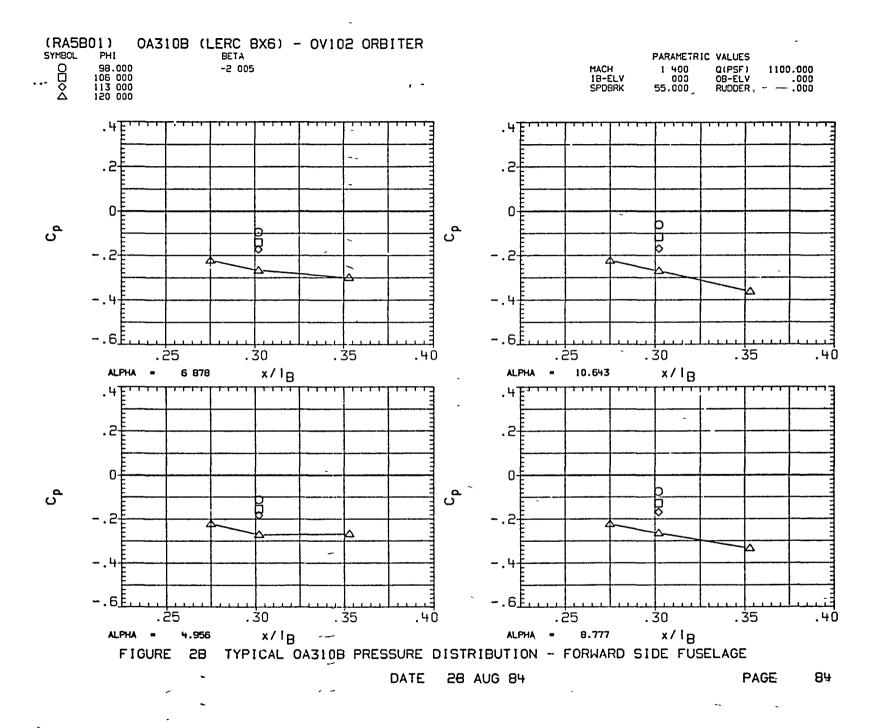
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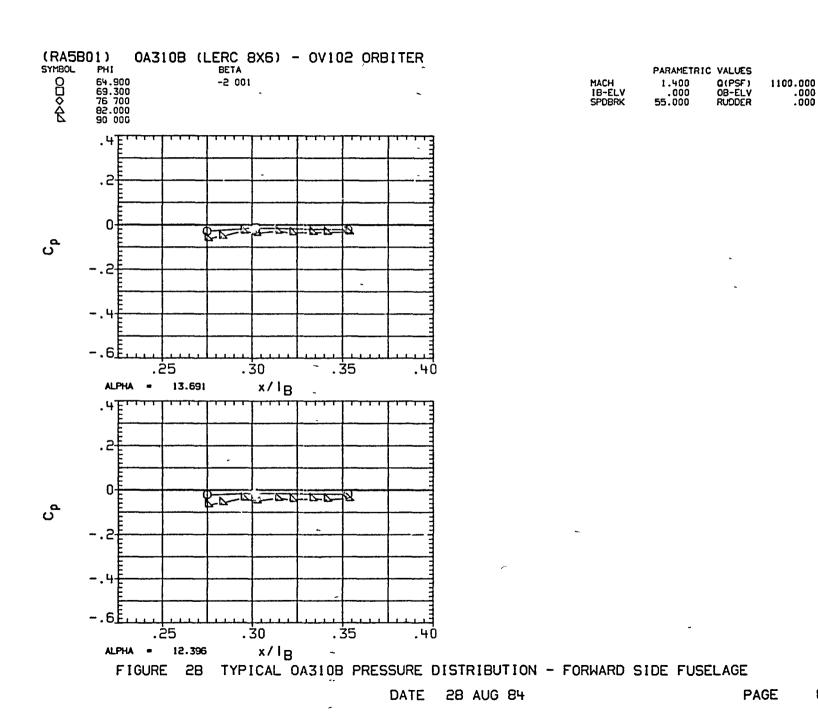


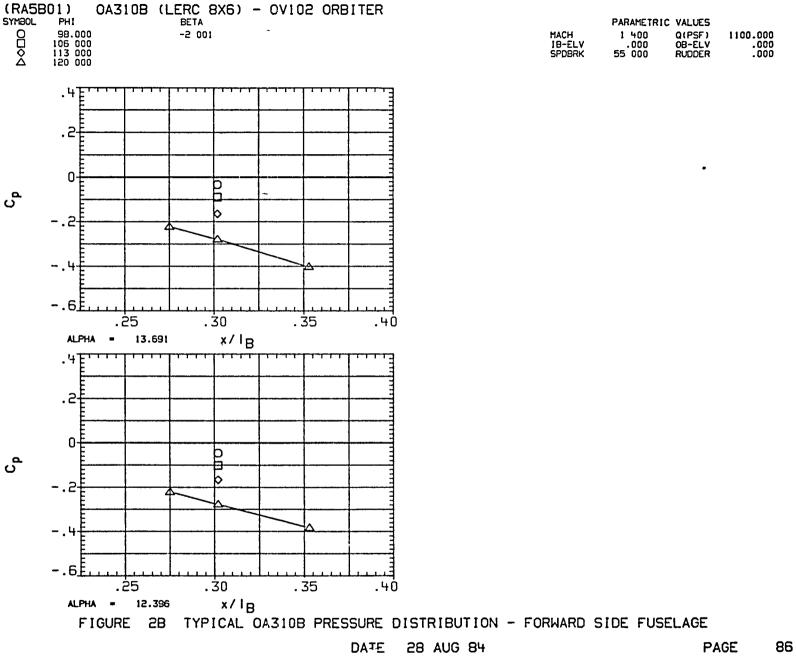


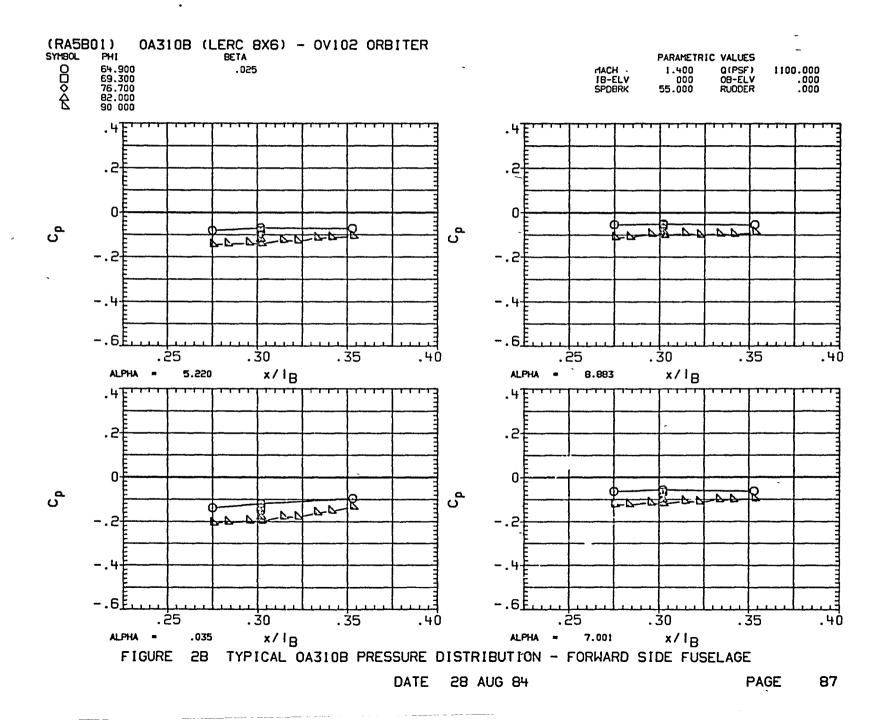


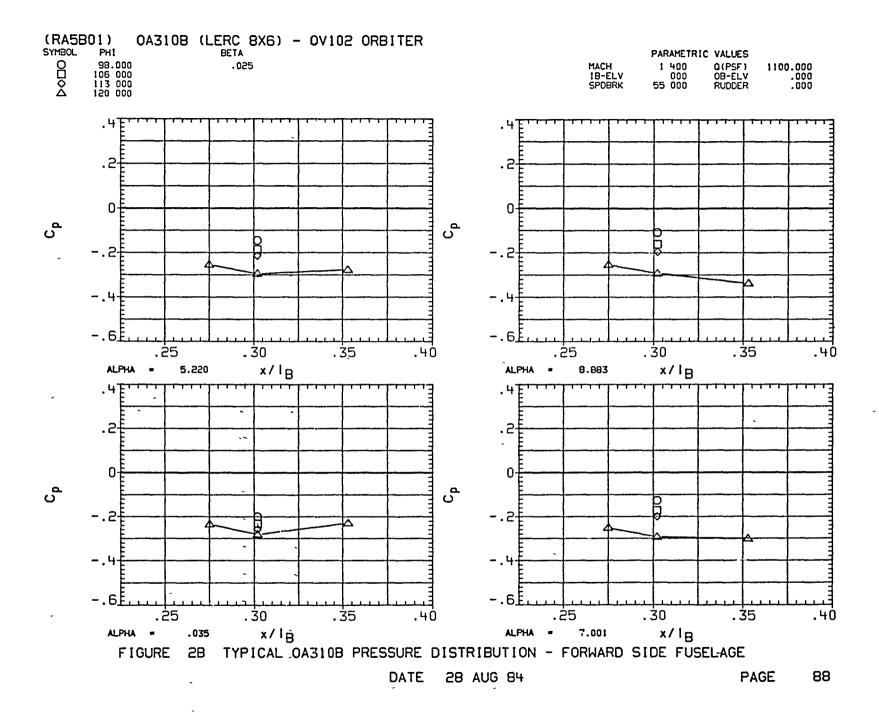


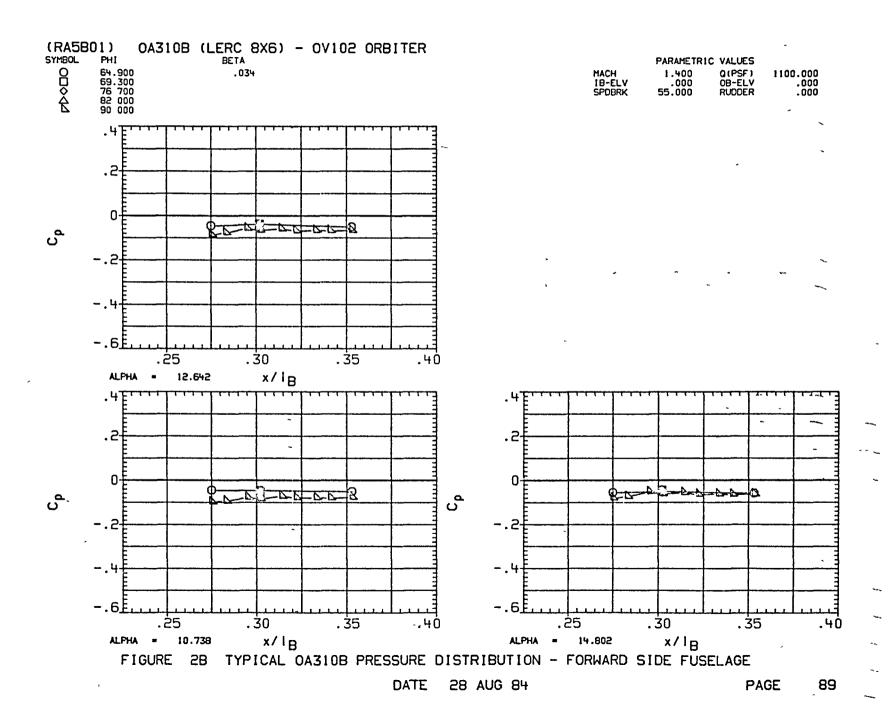
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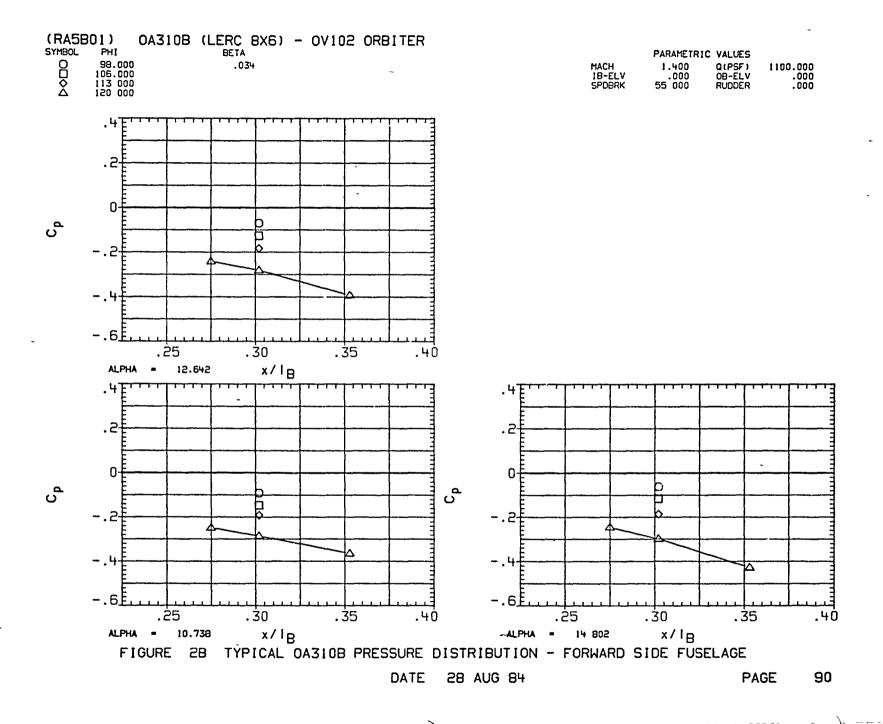


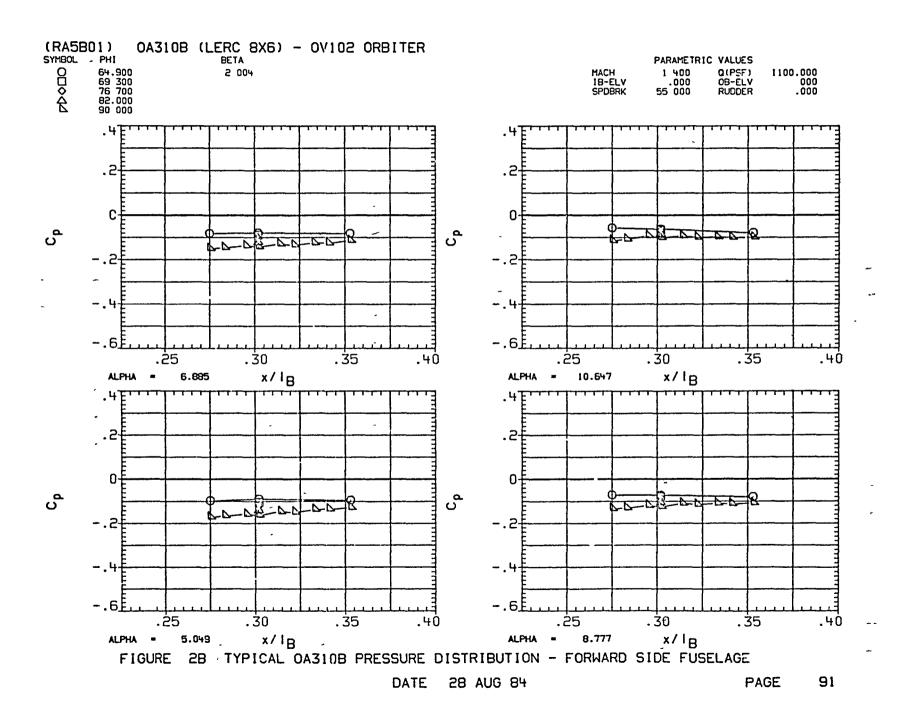


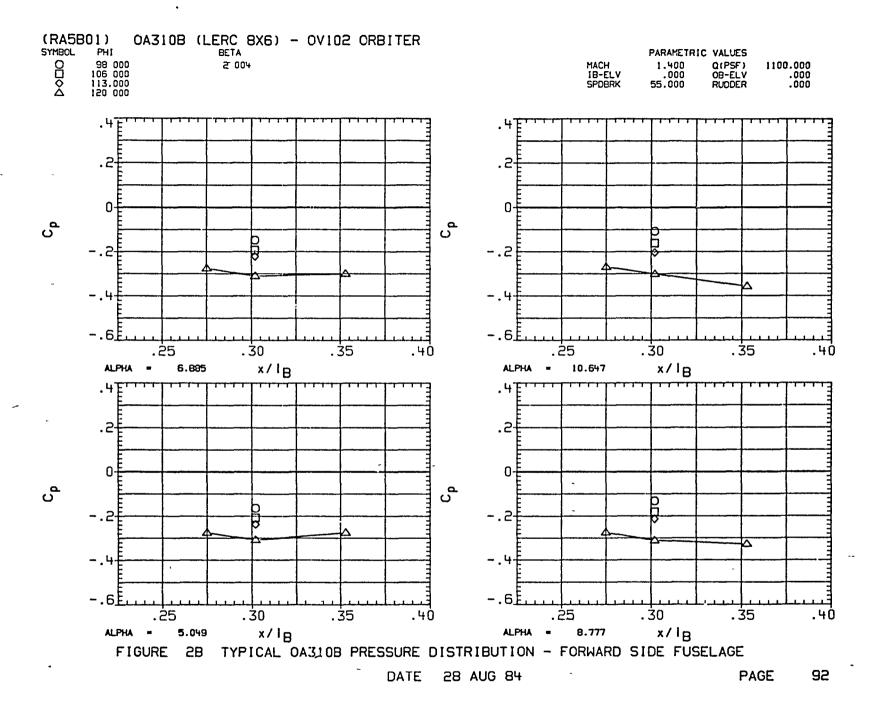


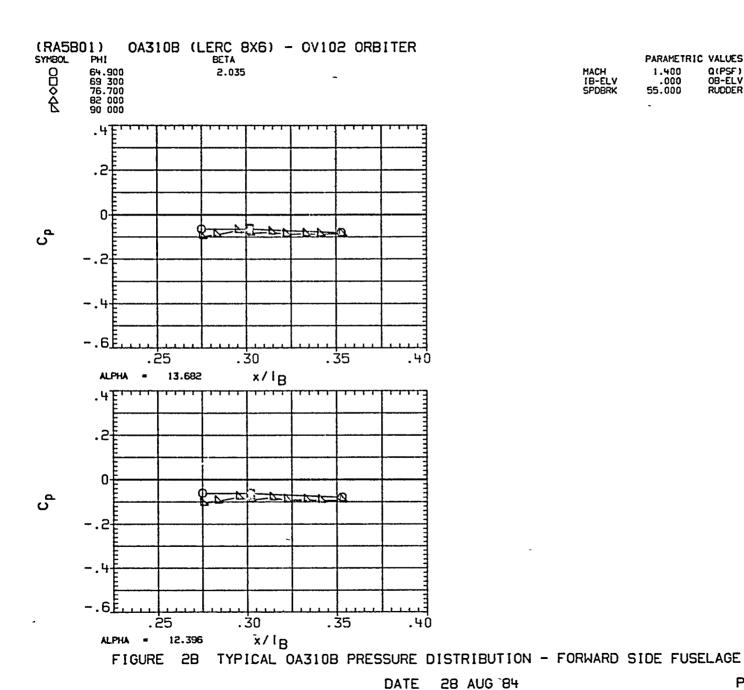






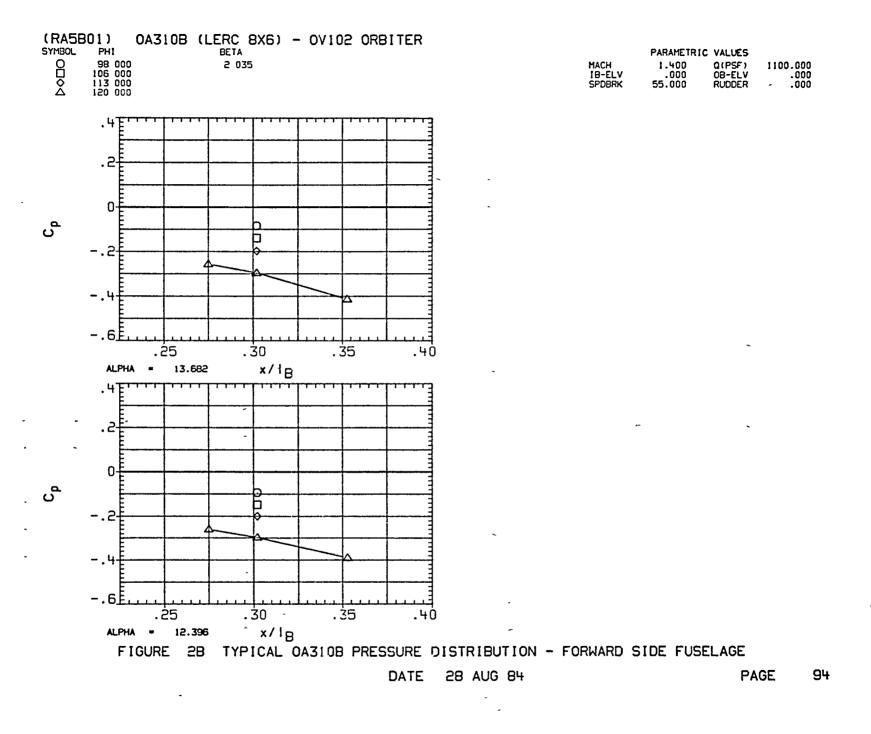


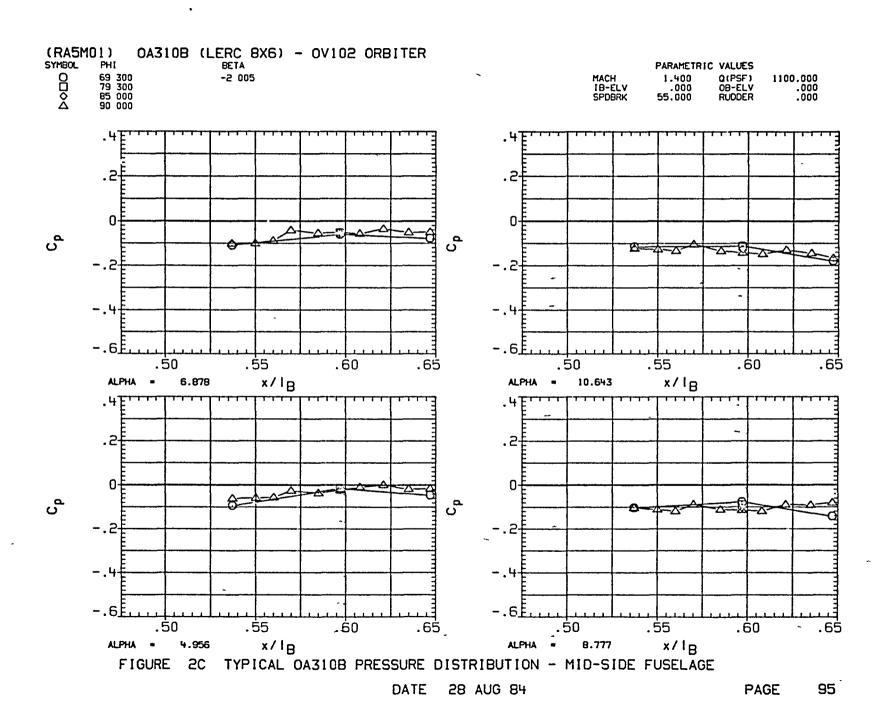


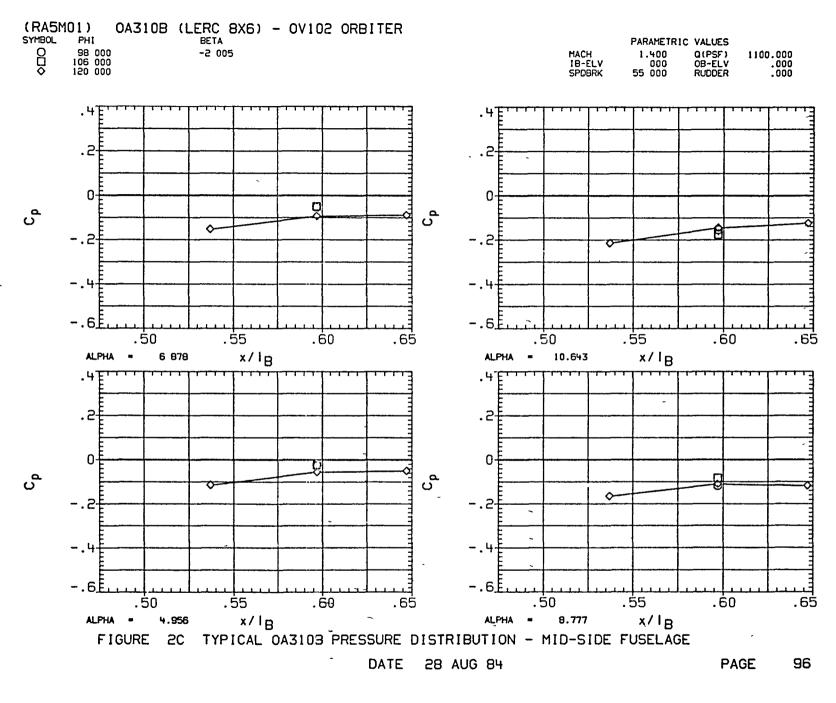


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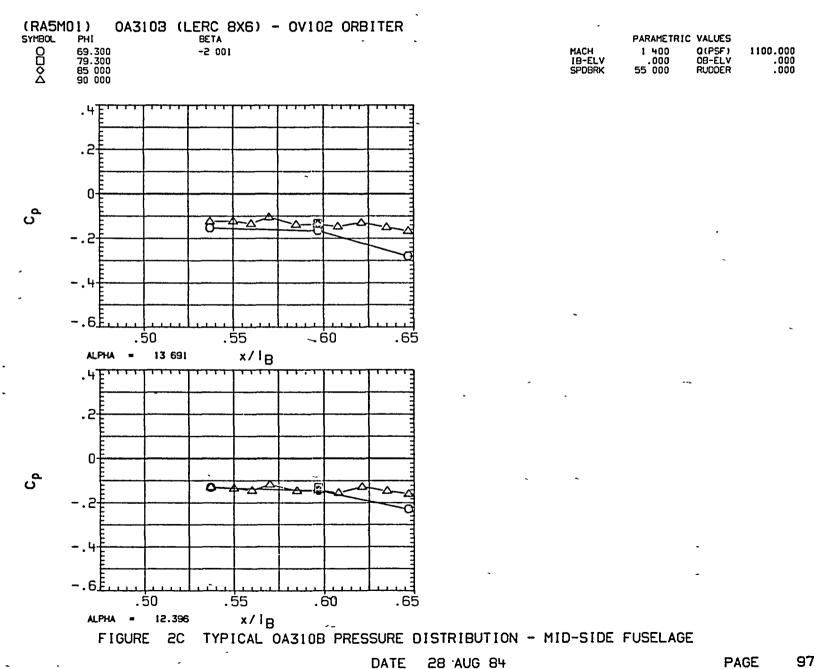
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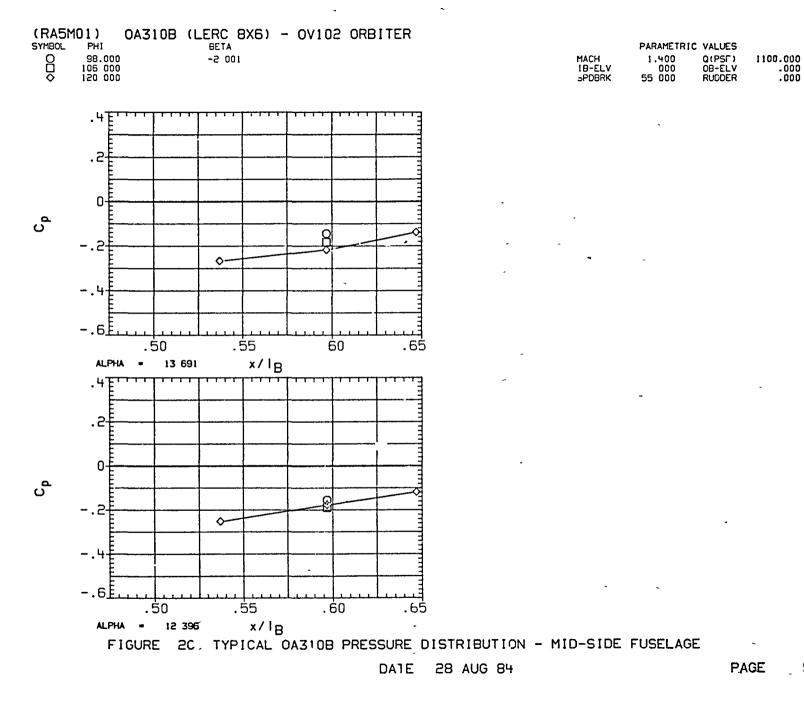


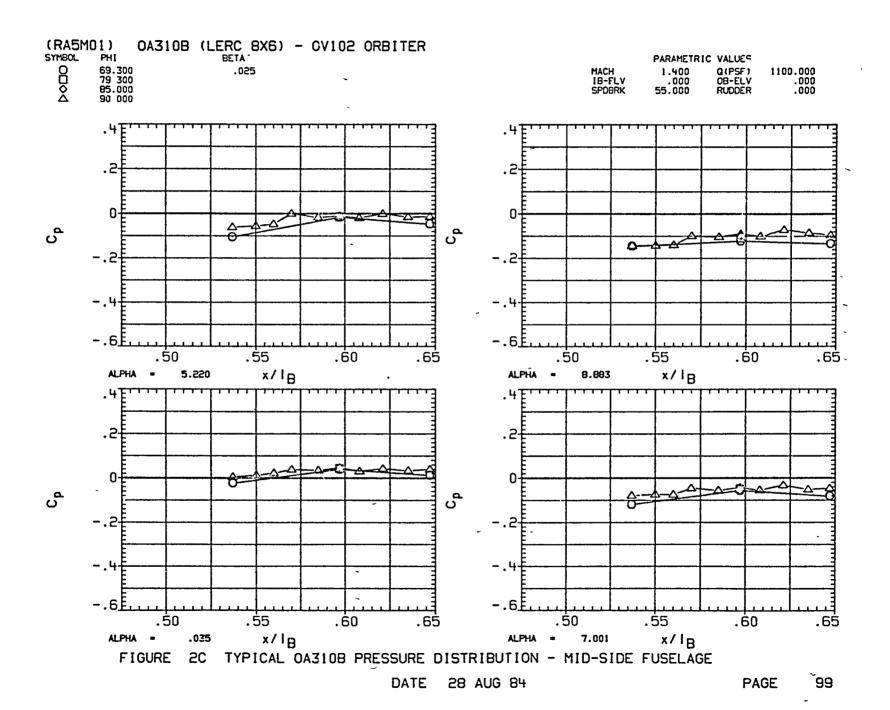


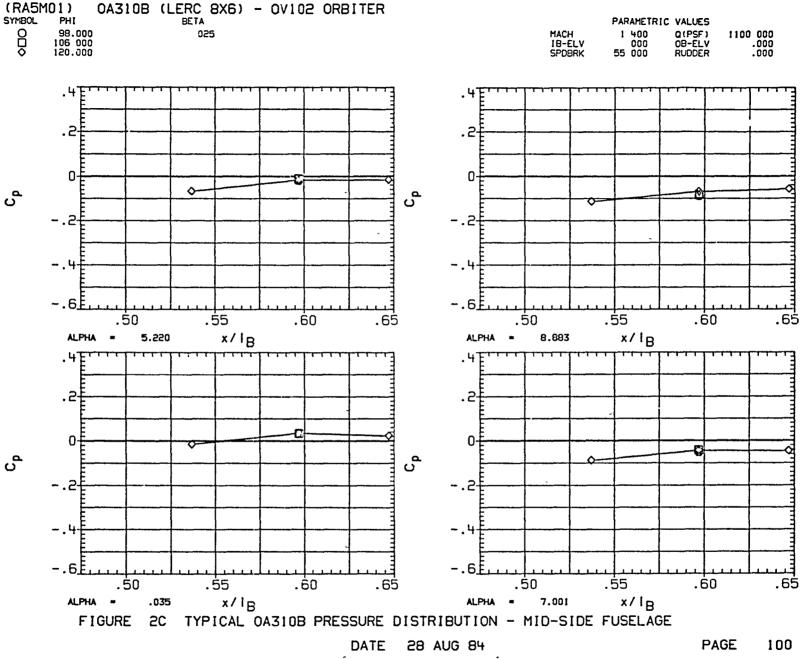
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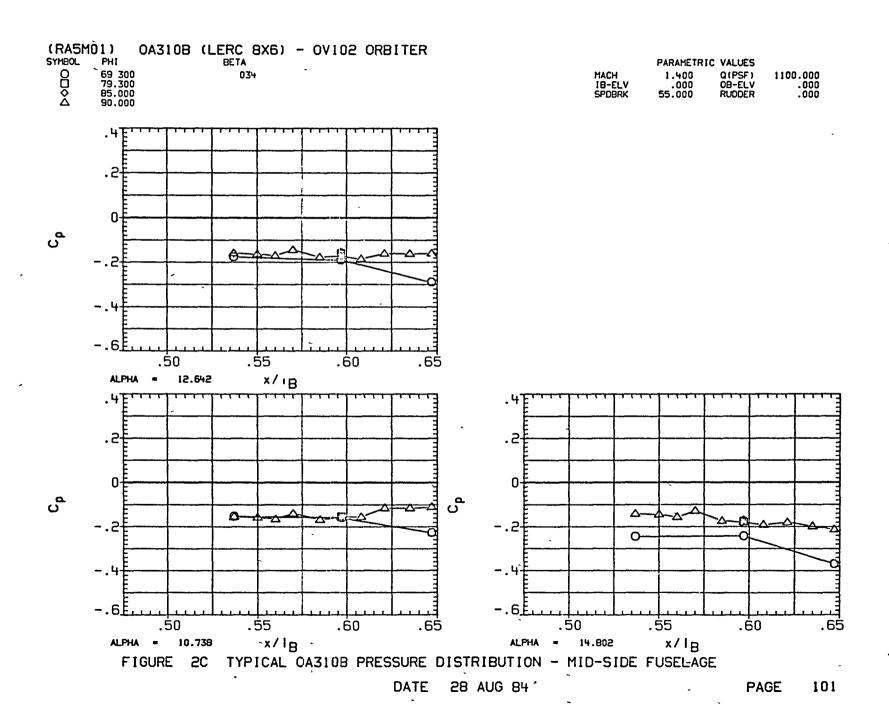


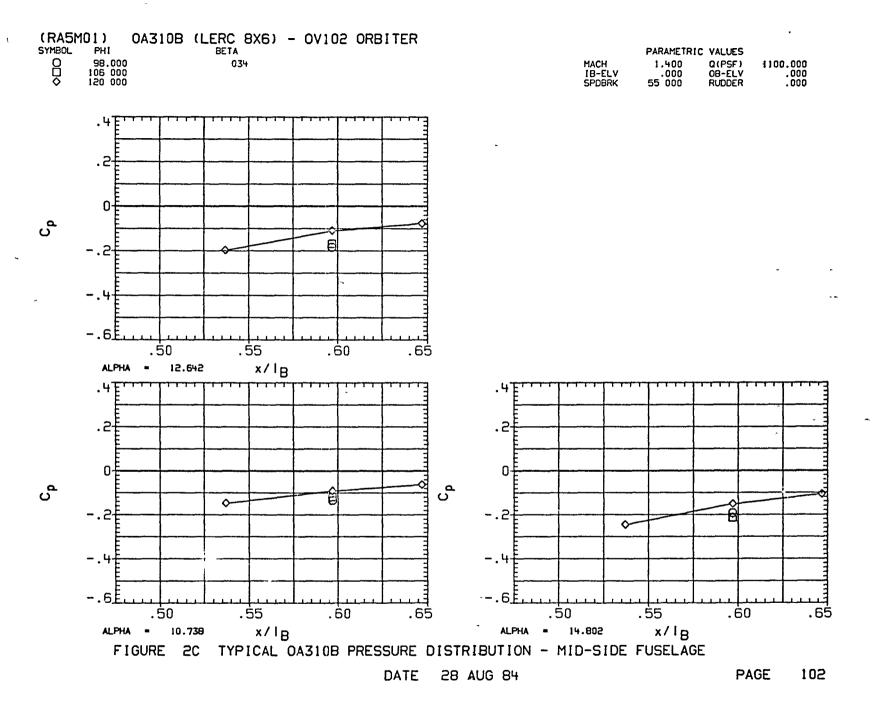
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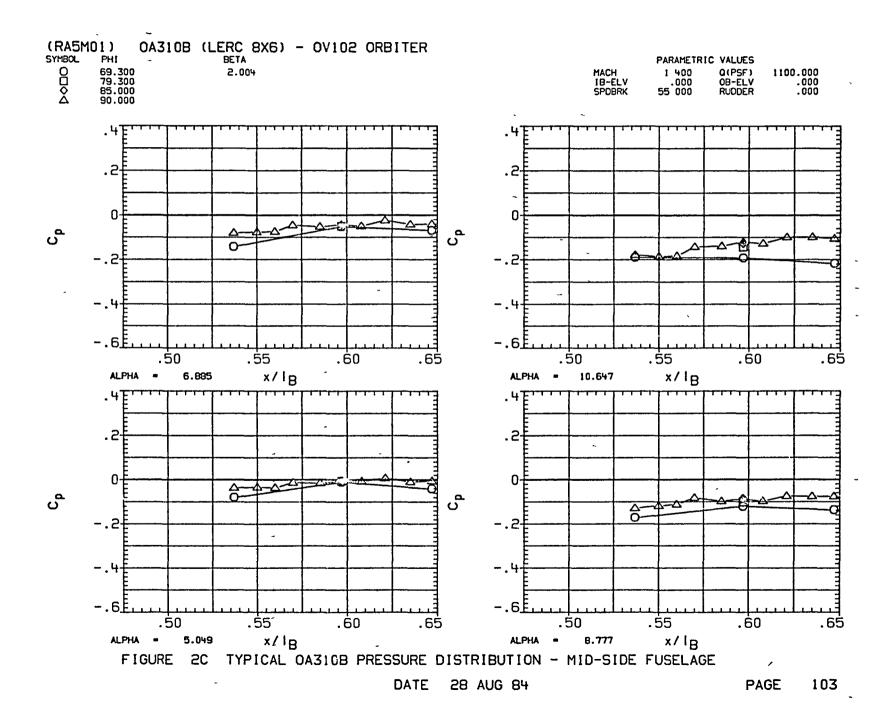


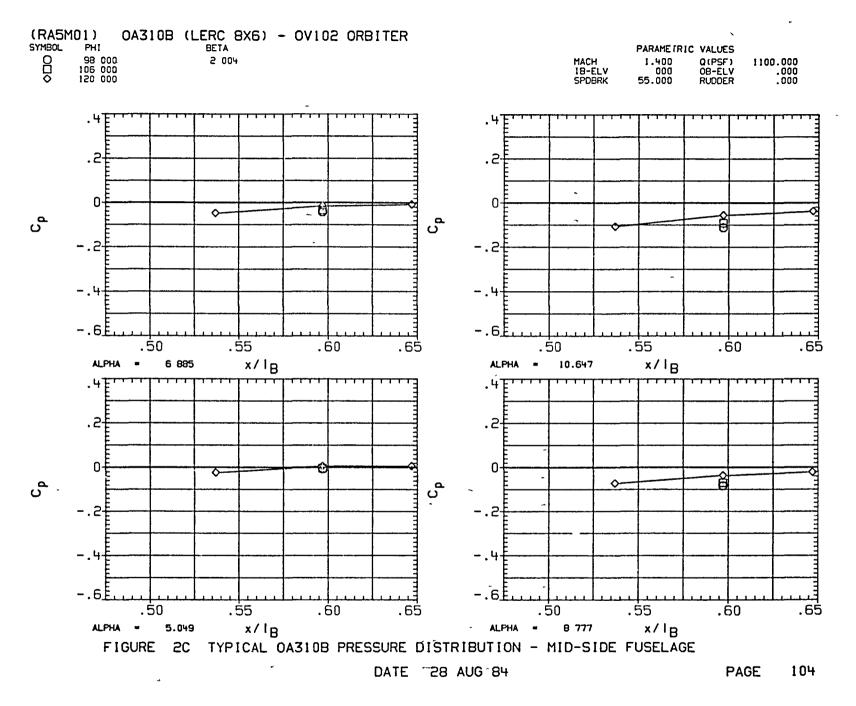


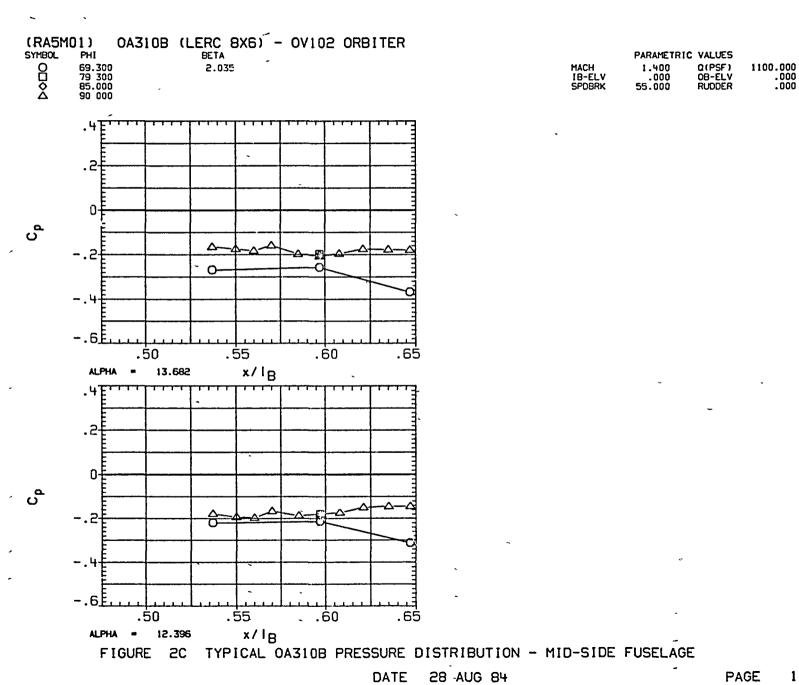




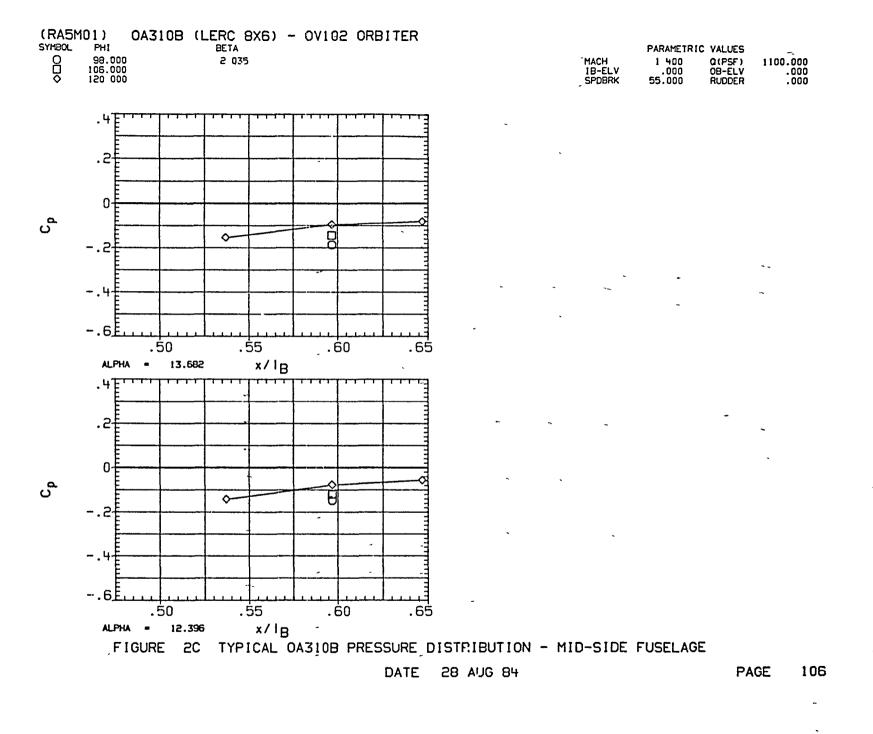


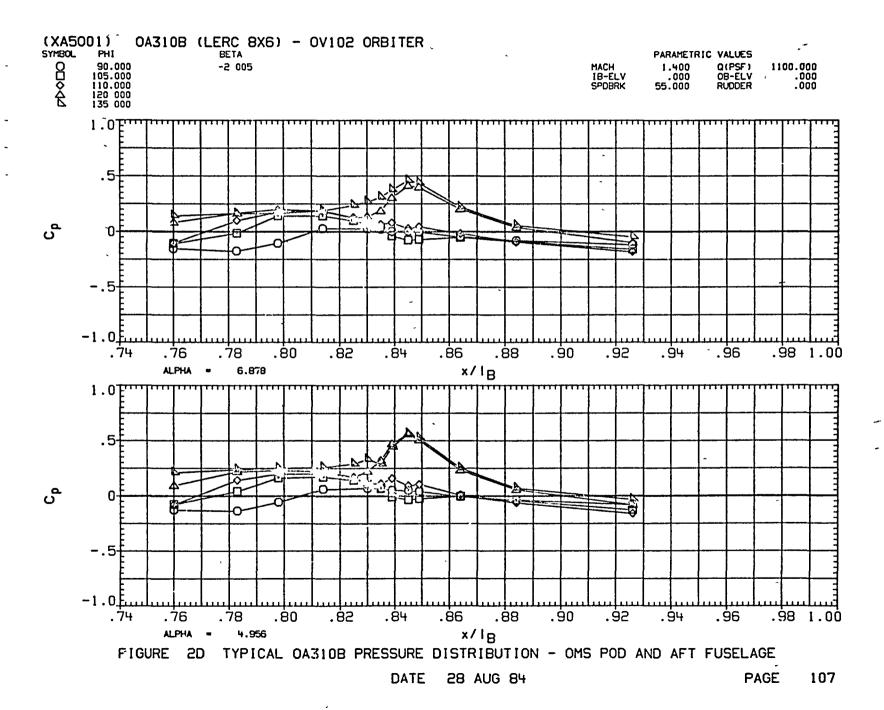


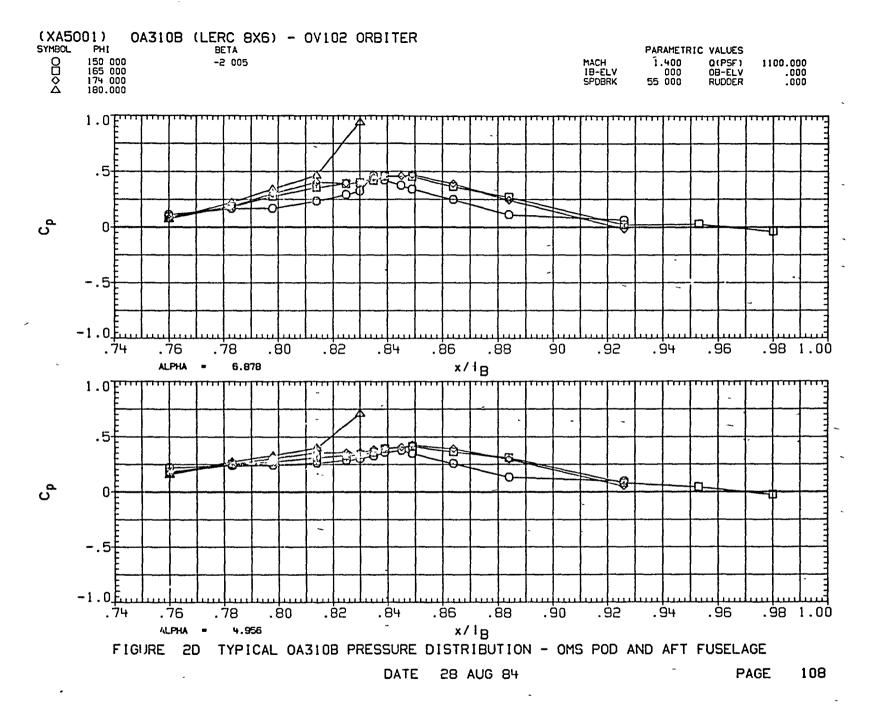


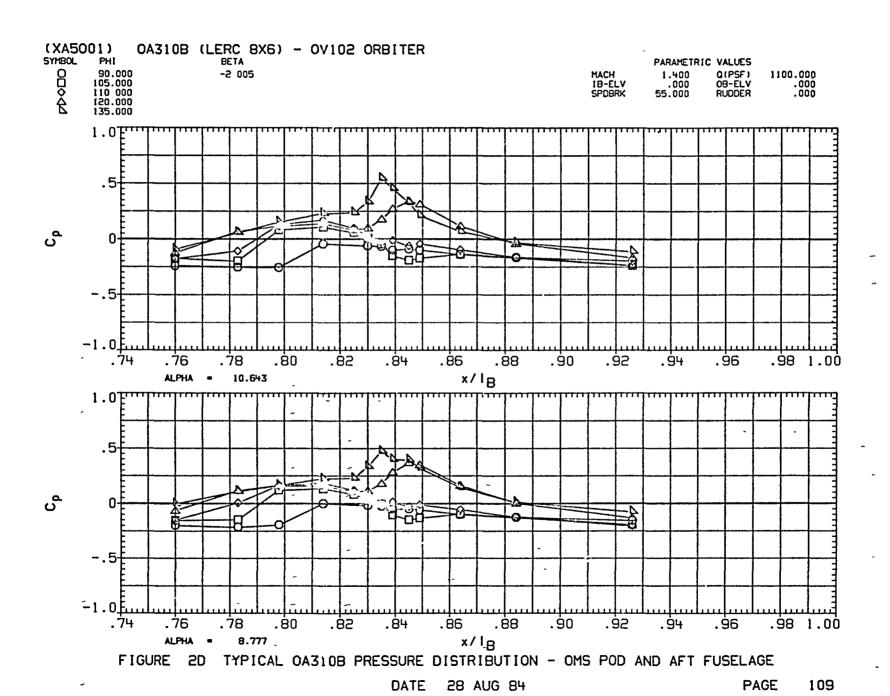


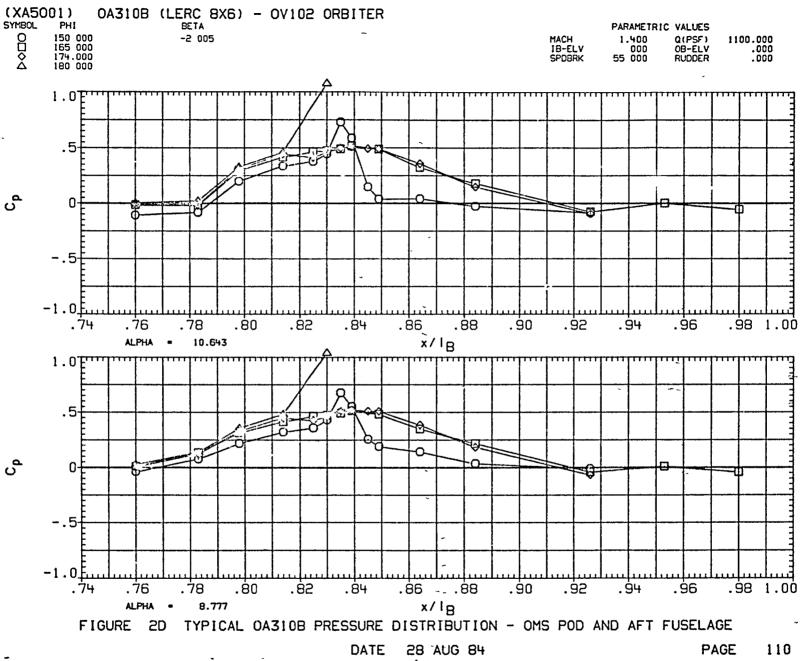
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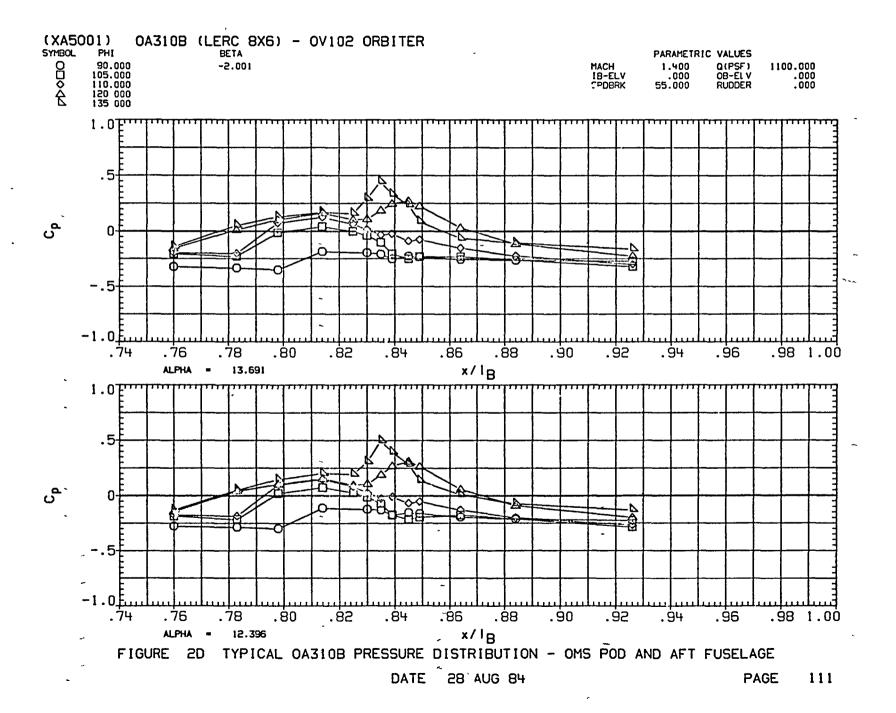


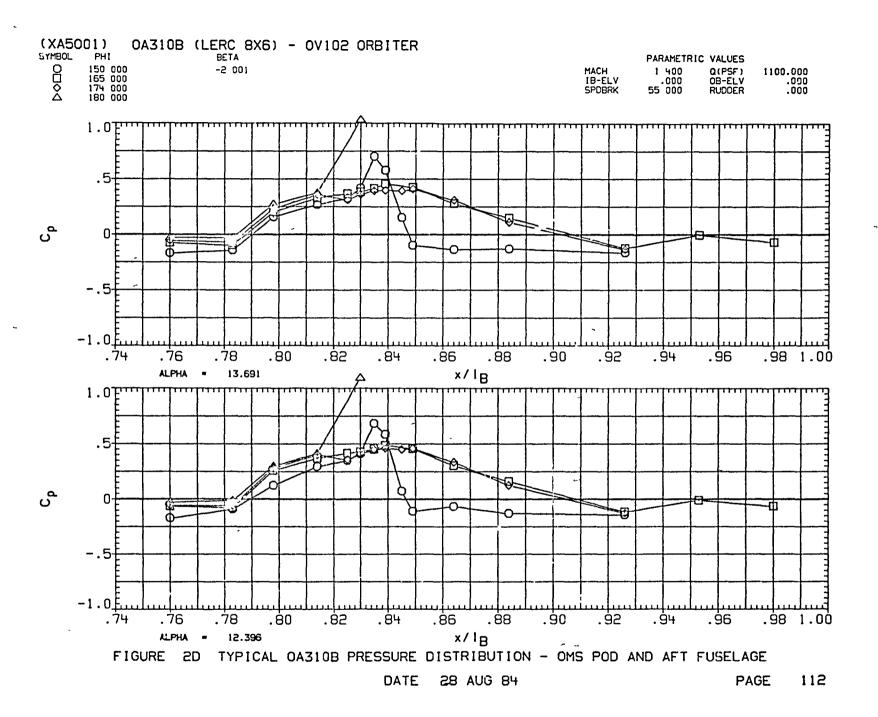


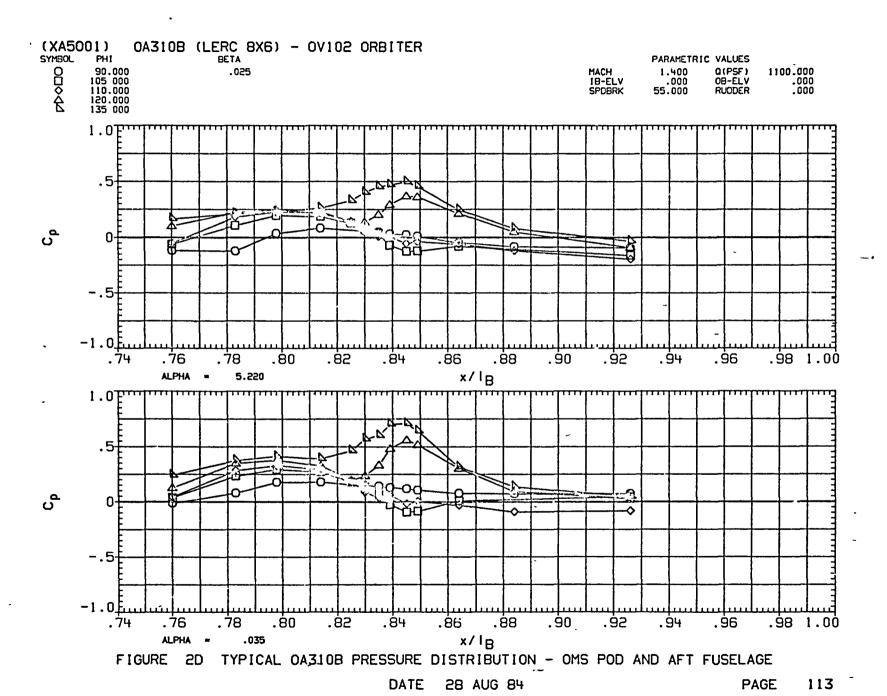


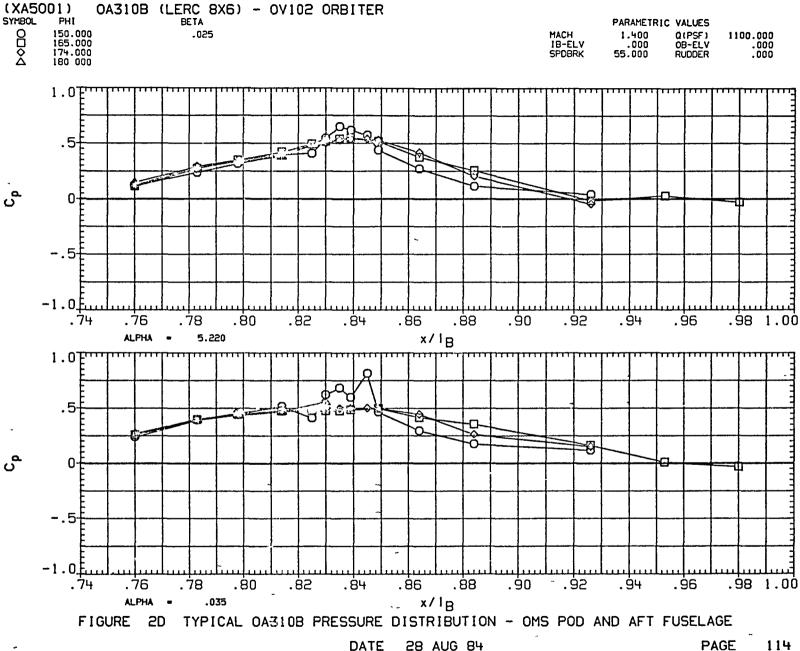




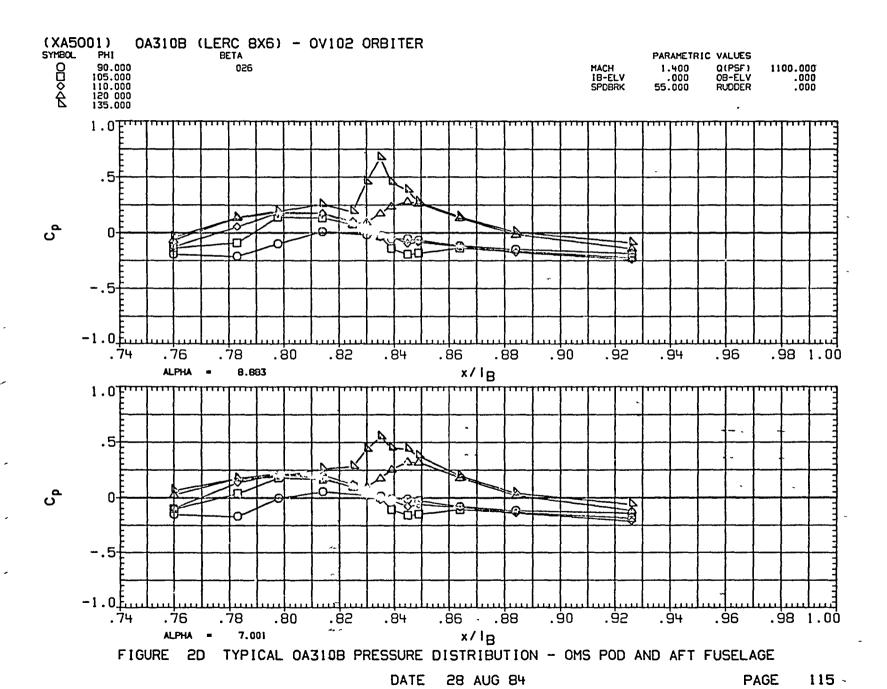


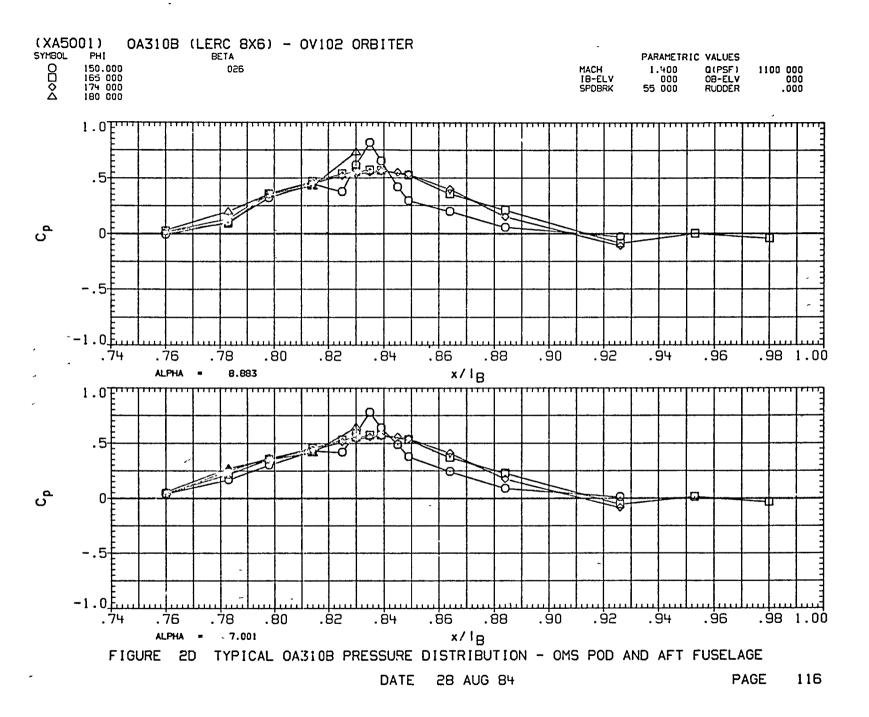


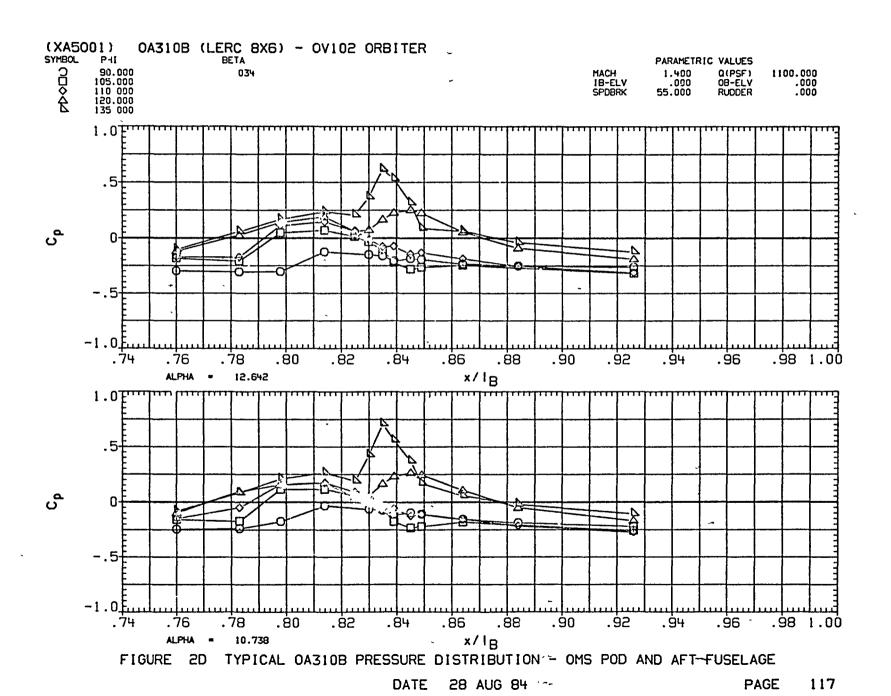


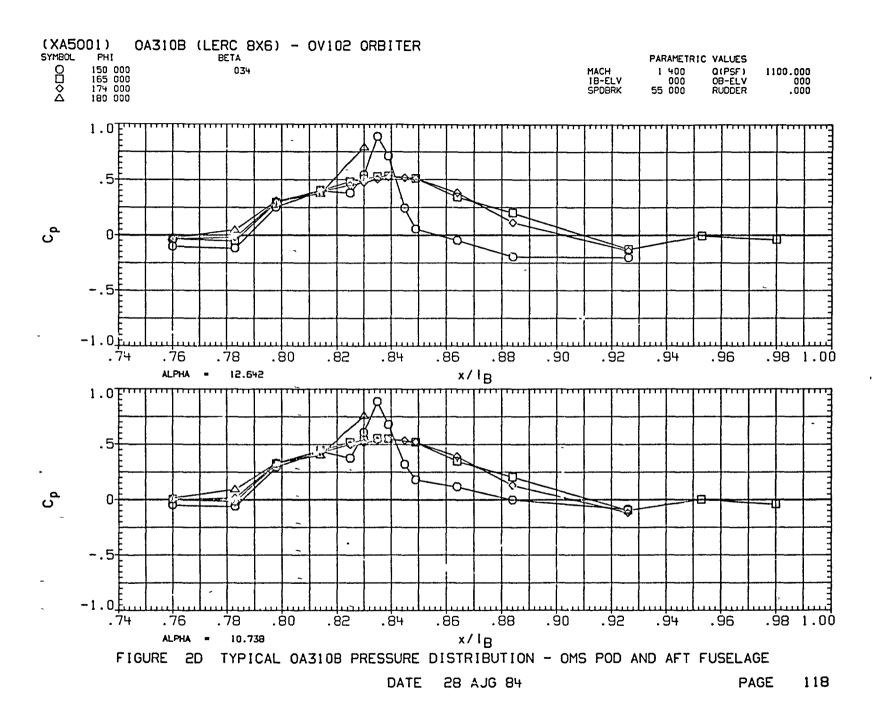


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(XA5001) OA310B (LERC 8X6) - OV1(D2 ORBITER
SYMBOL PH! BETA	PARAMFTRIC VALUES
O 90.000 .034	- MACH 1 400 Q(PSF) 1100.000
105.000	000. V3-ELV .000 0B-ELV .000.
♦ 110.000	SPOBRK 55 000 RUDDER .000
↑ 120.000 ↑ 135.000	

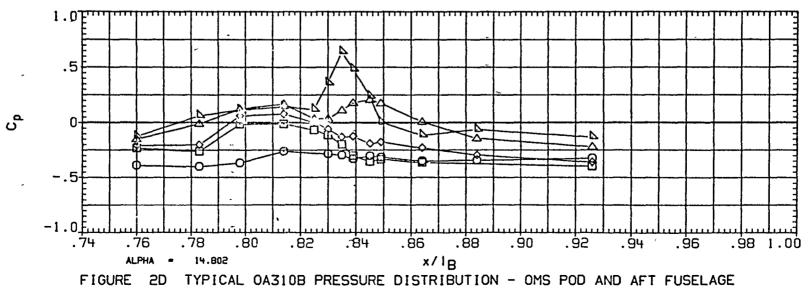
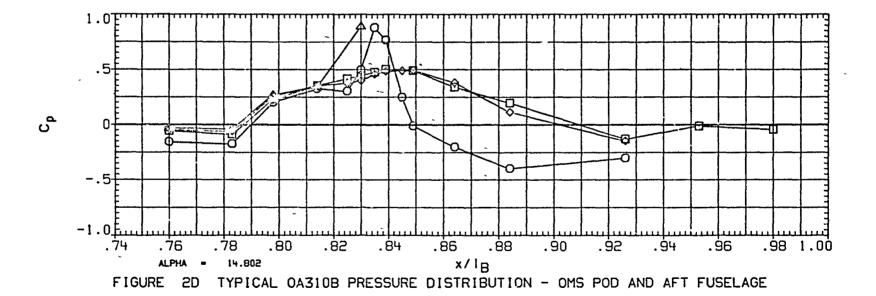


FIGURE 2D TYPICAL 0A310B PRESSURE DISTRIBUTION - 0MS POD AND AFT FUSELAGE

DATE 28 AUG 84 PAGE

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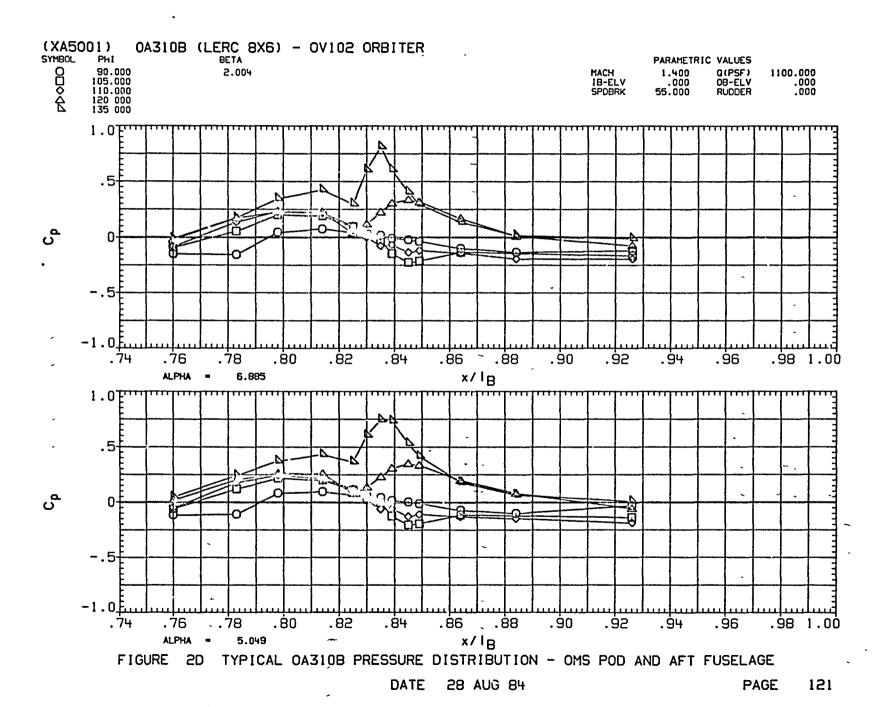
(XA50	01)	0A310B	(LERC 8X6)	_	OV102	ORBITER					
SYMBOL	PHI		BETA						PARAMETRI	IC VALUES	
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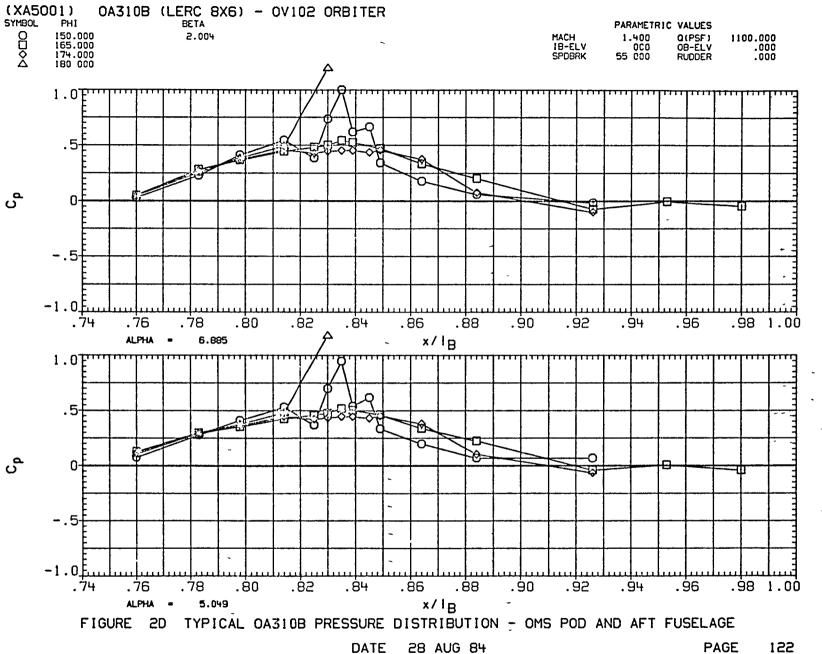


DATE 28 AUG 84

PAGE

120





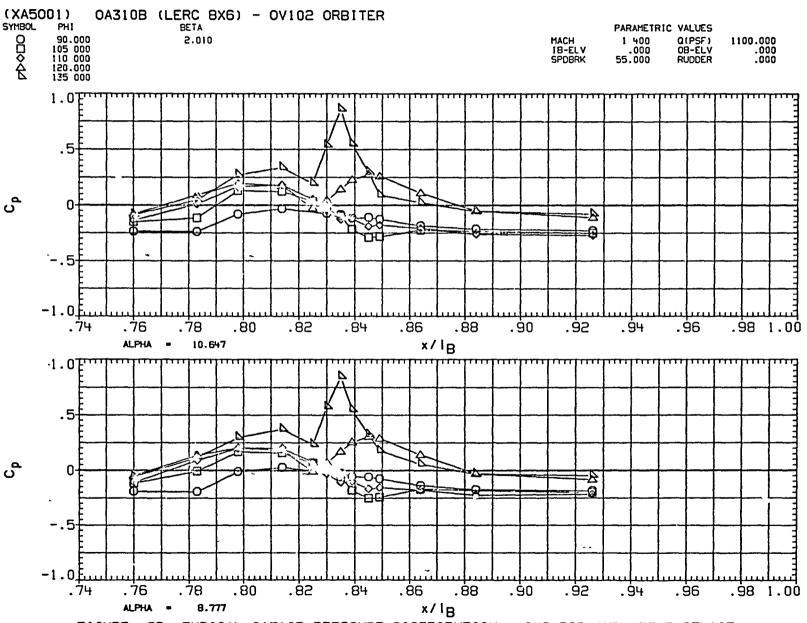
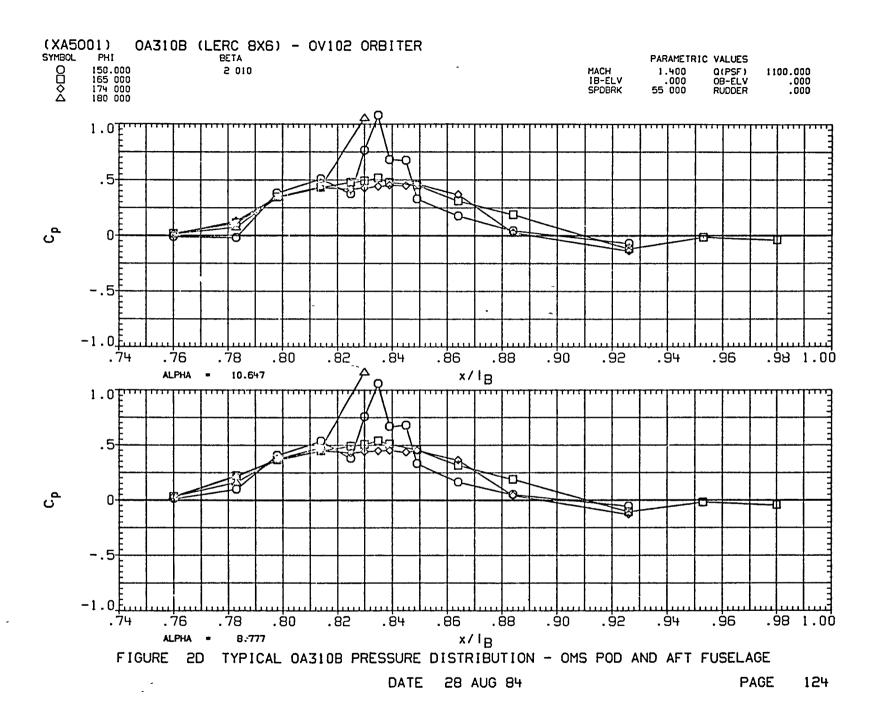
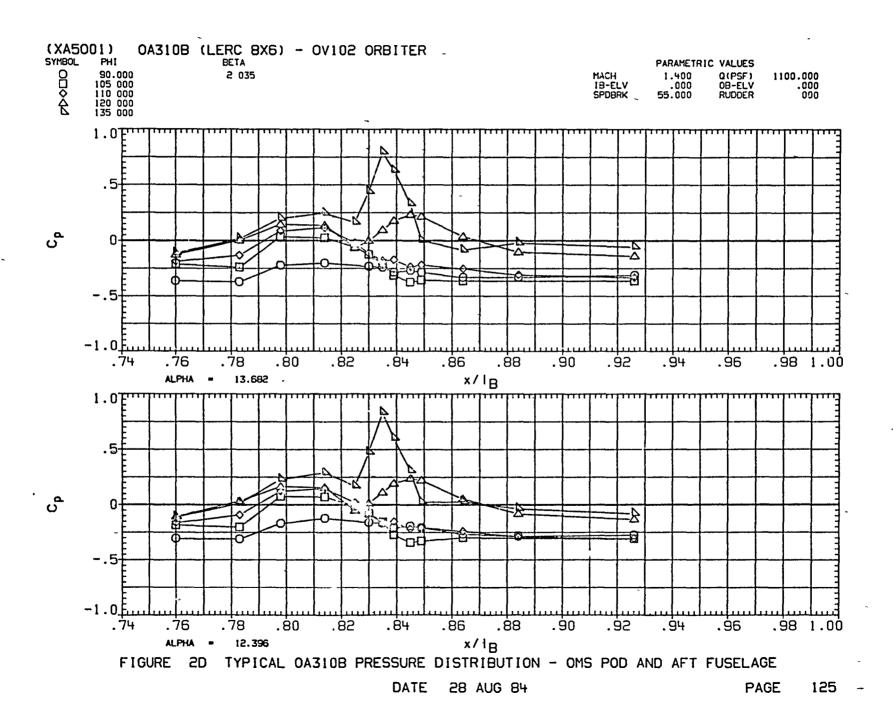


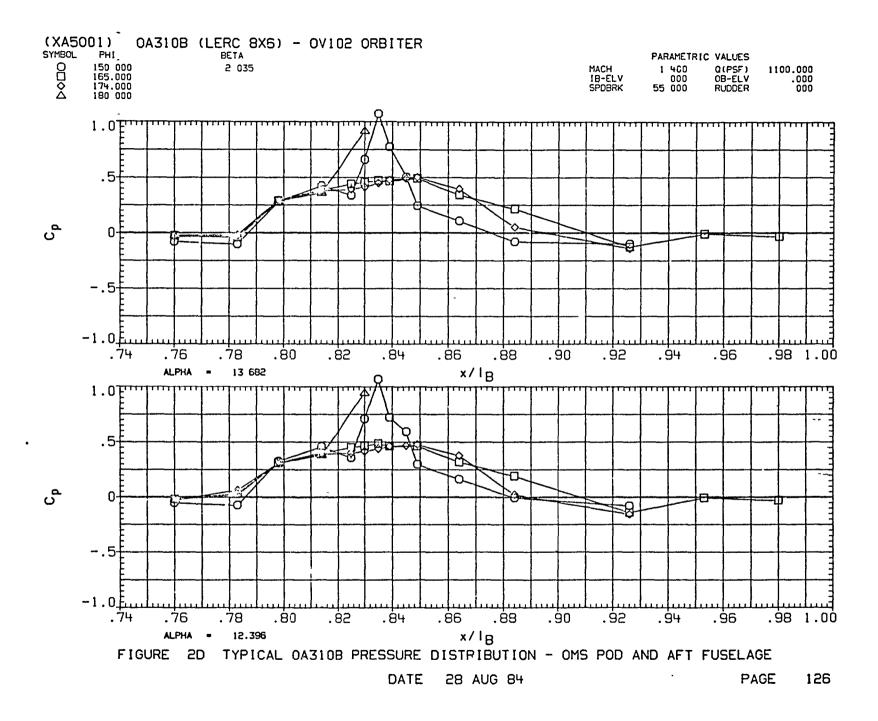
FIGURE 2D TYPICAL 0A310B PRESSURE DISTRIBUTION - OMS POD AND AFT FUSELAGE

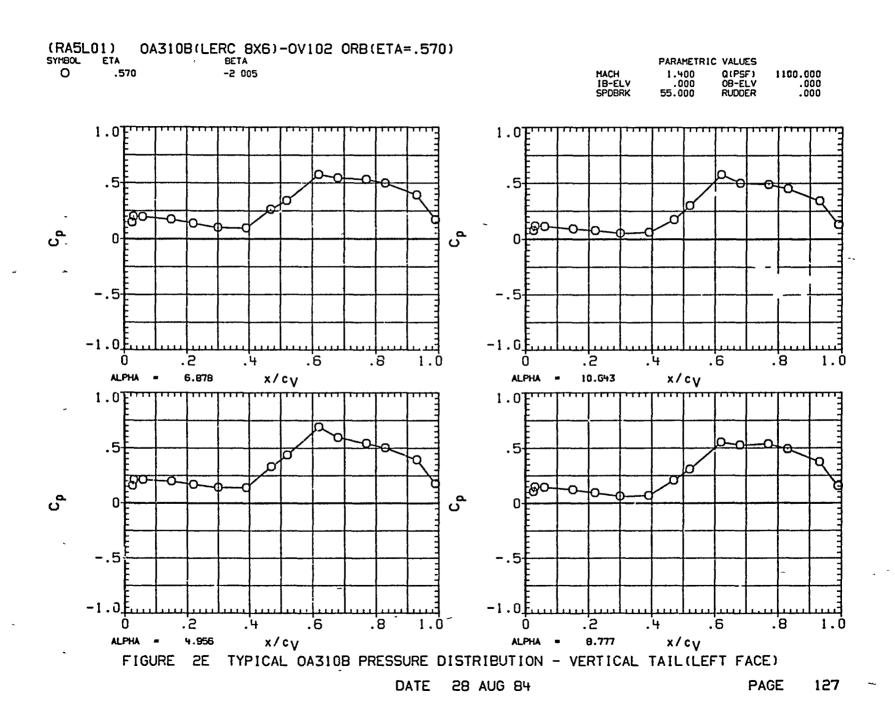
DATE 28 AUG 84

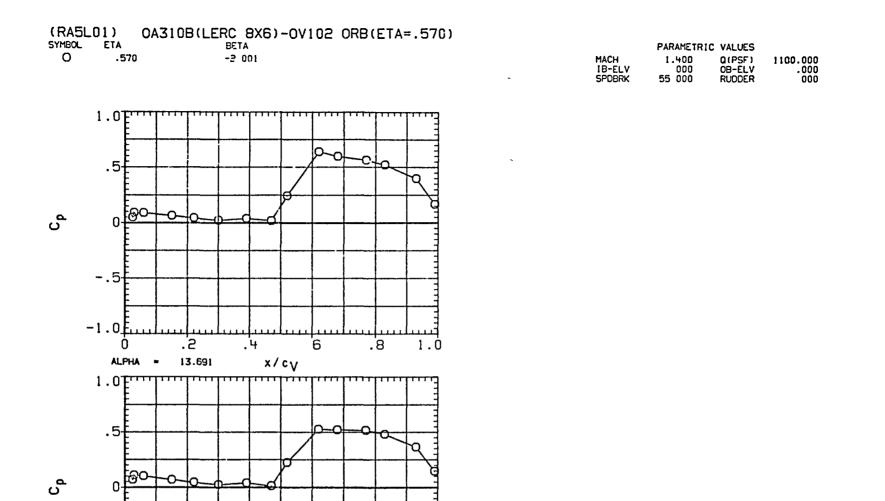
PAGE 123











ALPHA = 12.396 x/cV FIGURE 2E TYPICAL 0A310B PRESSURE DISTRIBUTION - VERTICAL TAIL (LEFT FACE) **DATE 28 AUG 84**

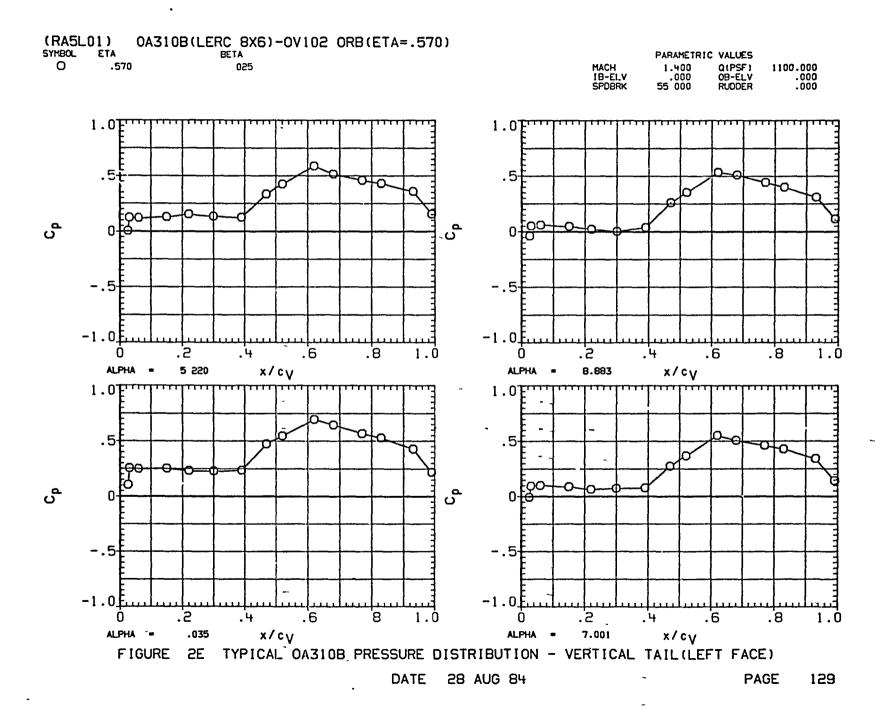
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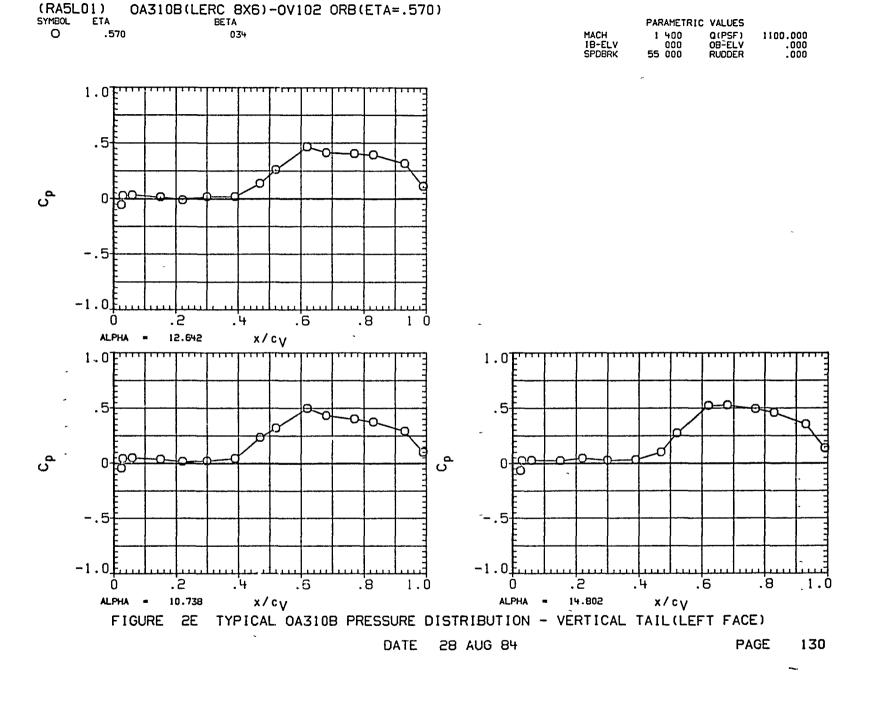
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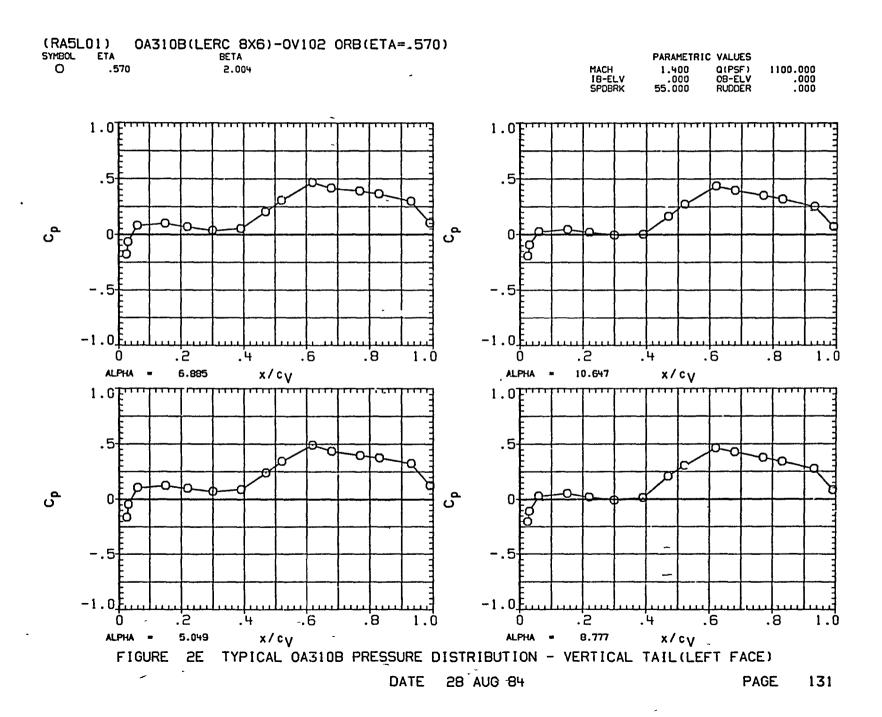
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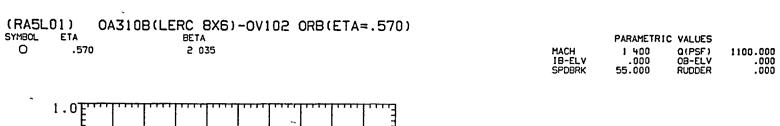
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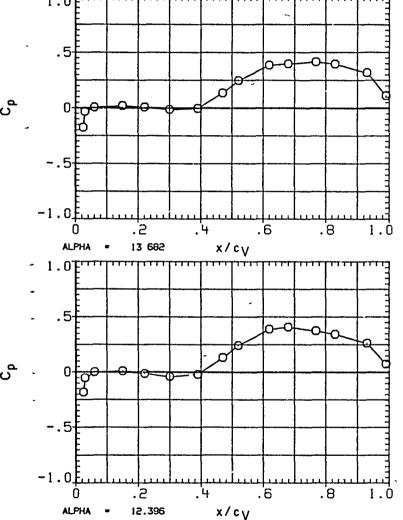
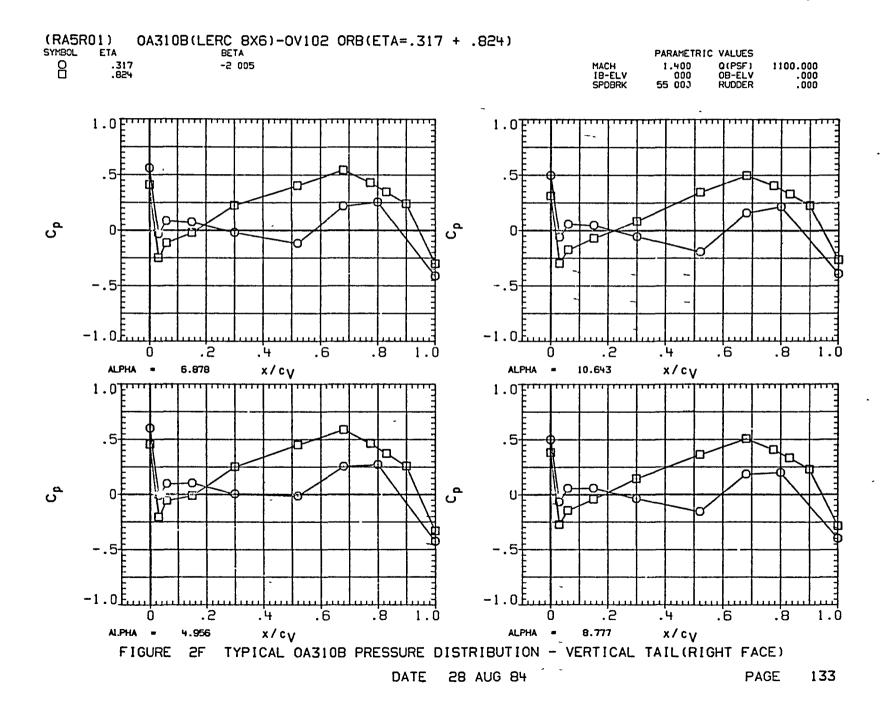
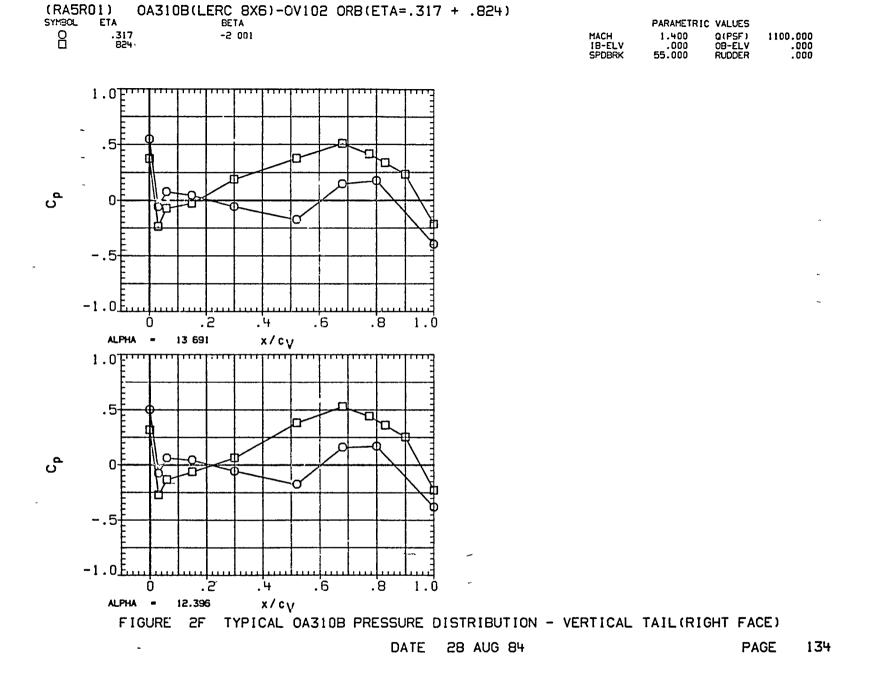
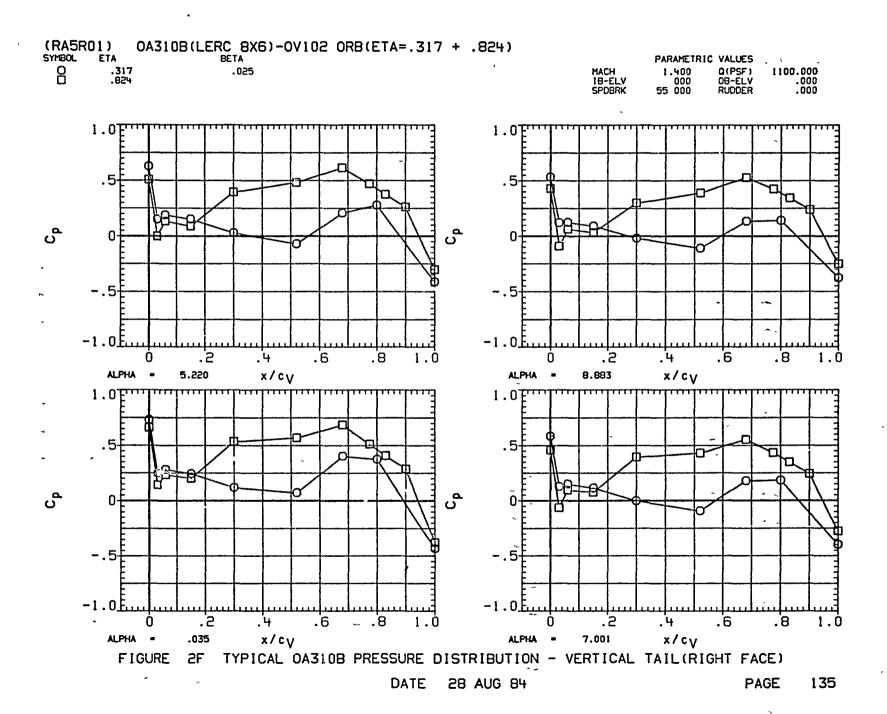


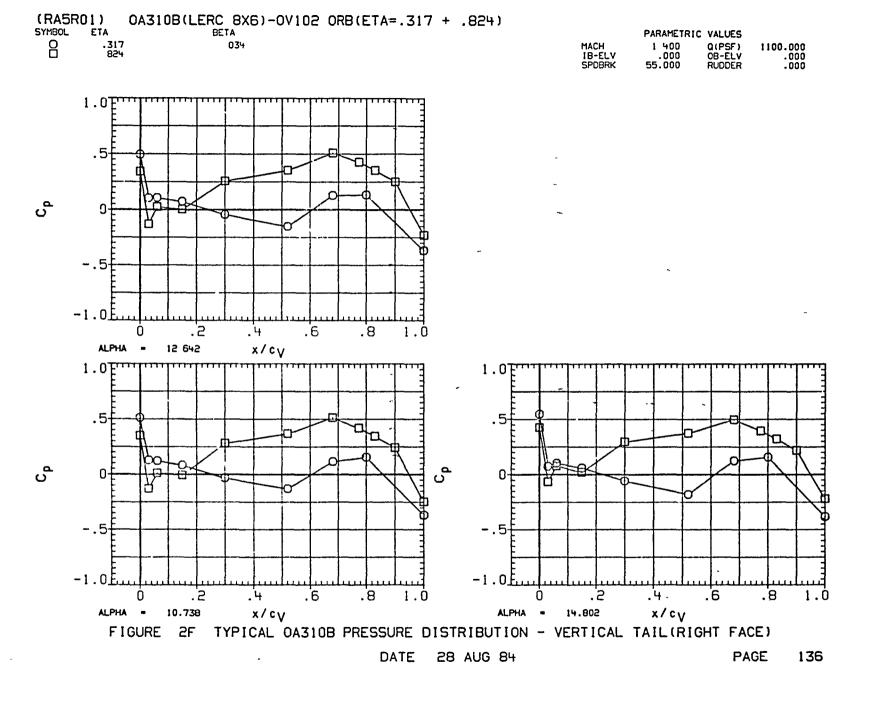
FIGURE 2E TYPICAL 0A310B PRESSURE DISTRIBUTION - VERTICAL TAIL(LEFT FACE)

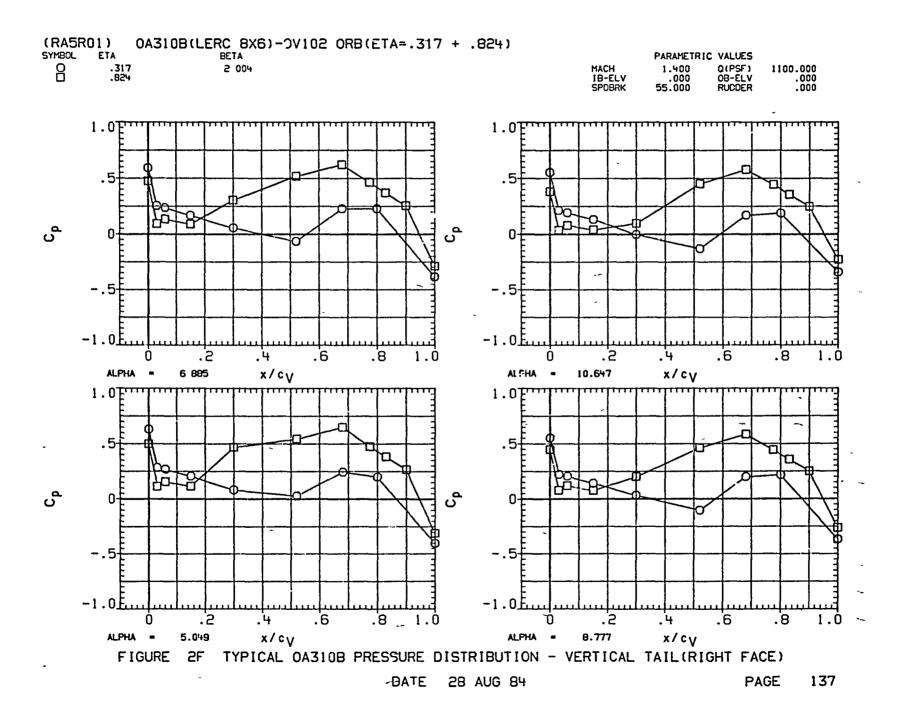
DATE 28 AUG 84 PAGE 1-32

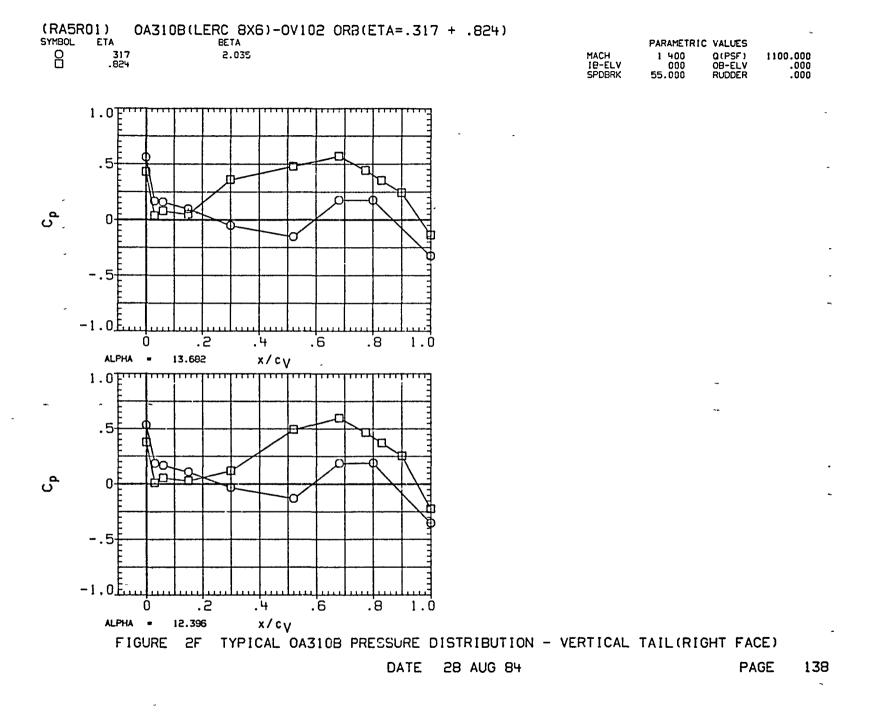


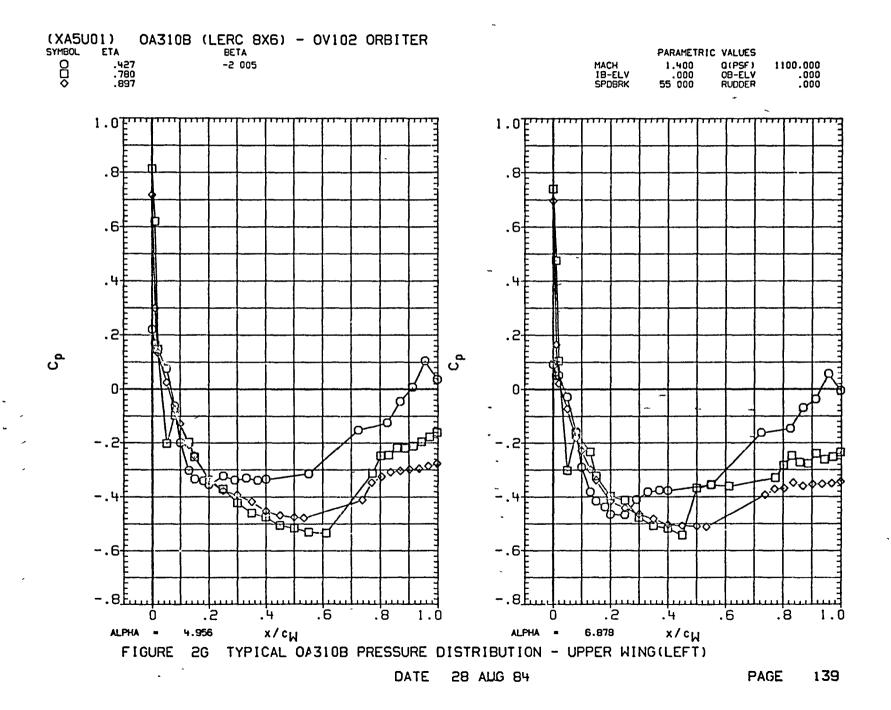


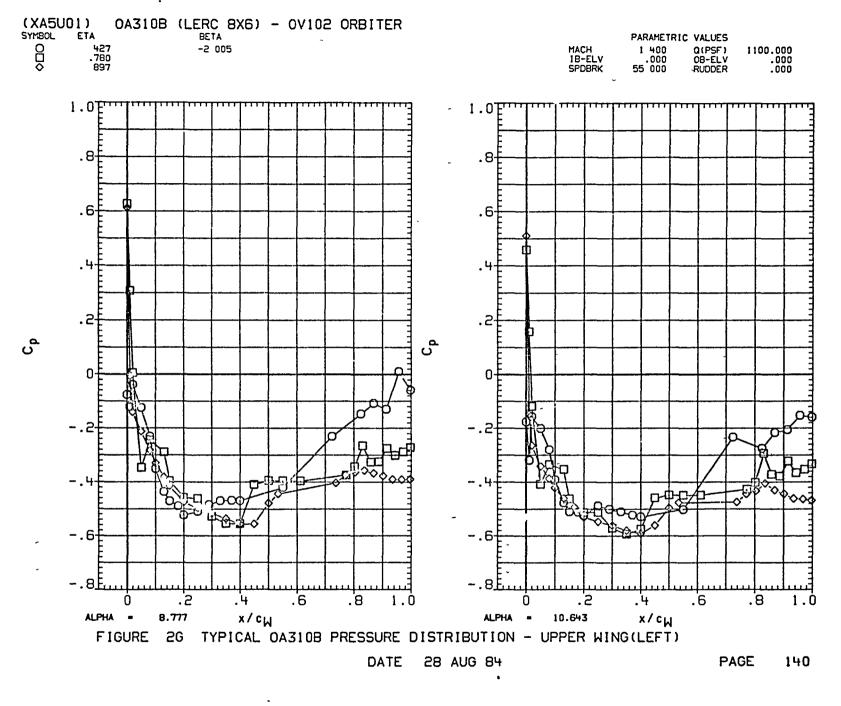


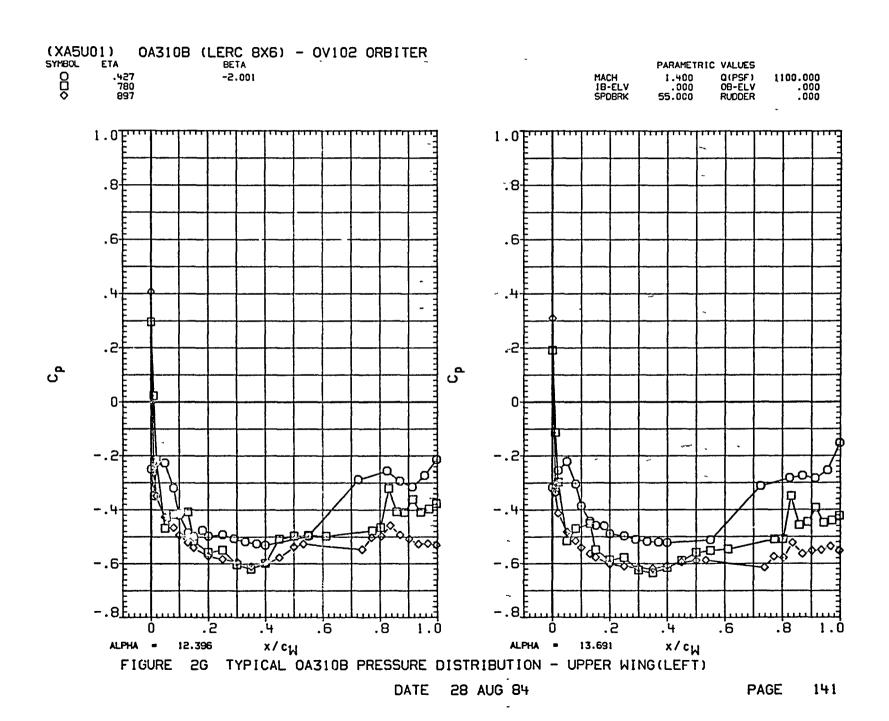


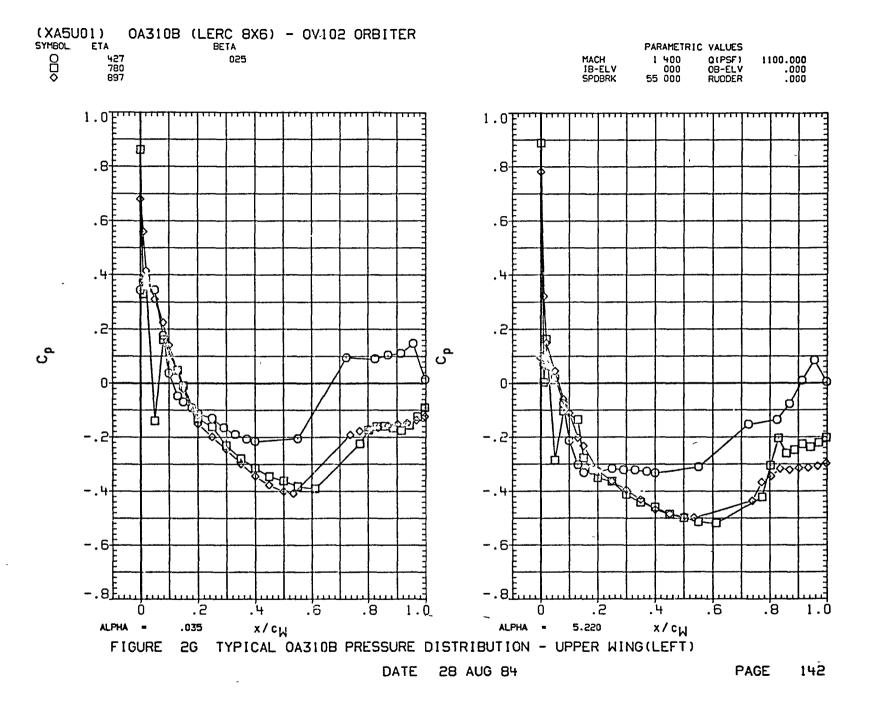


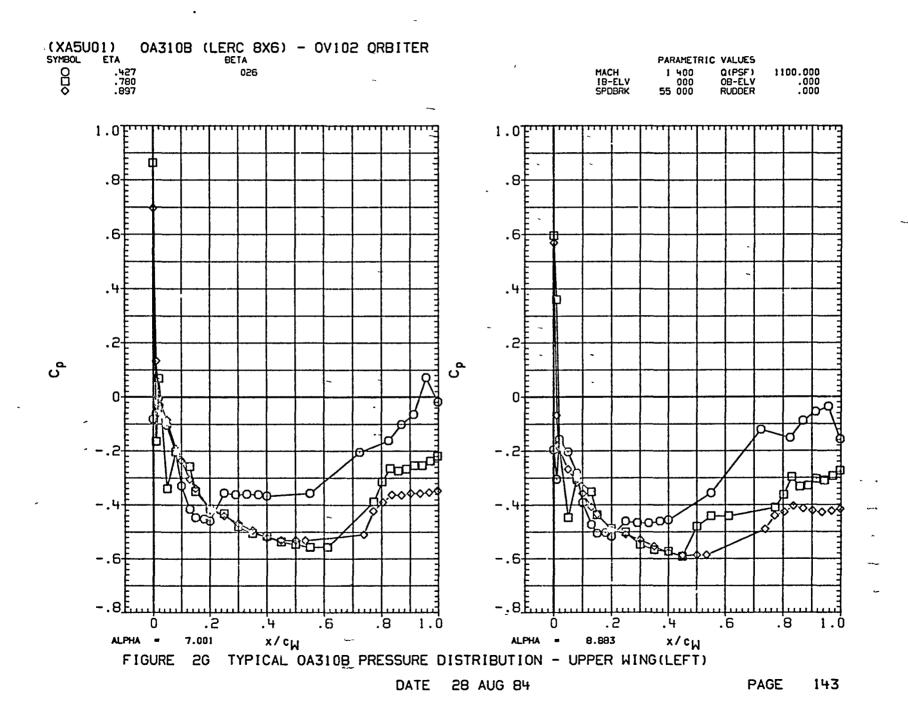


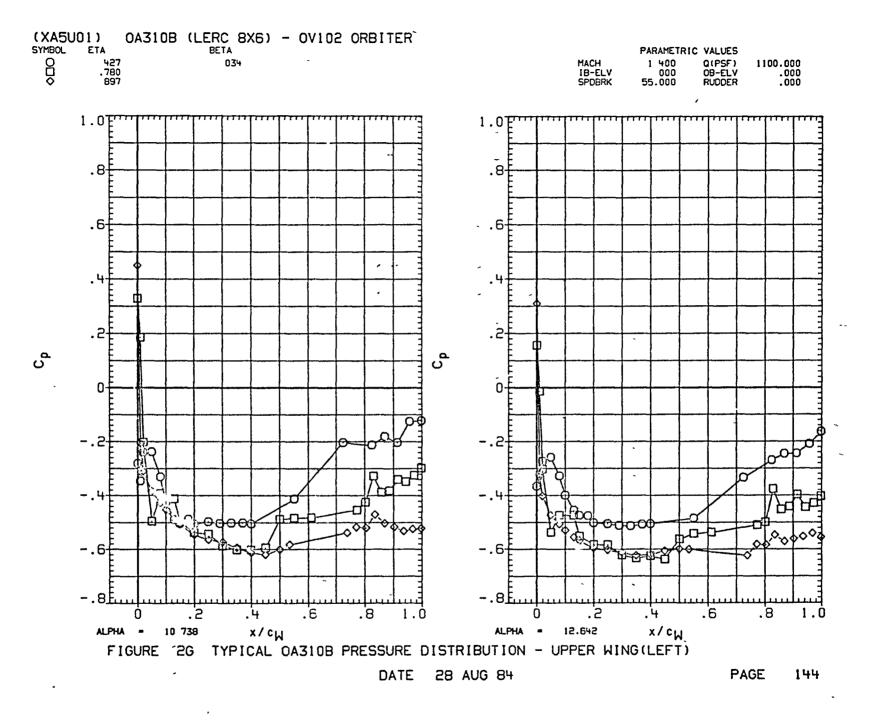












(XA5U01) SYMBOL ETA OA310B (LERC 8X6) - OV102 ORBITER ETA PARAMETRIC VALUES 000 .427 .780 .897 MACH IB-ELV SPDBRK 1.400 .000 55.000 .034 1.0F .8 .6 .2 0--.6 Э. .'z .'4 .8 1.Ò ALPHA = 14.802 x/cM FIGURE 2G TYPICAL 0A310B PRESSURE DISTRIBUTION - UPPER WING(LEFT)

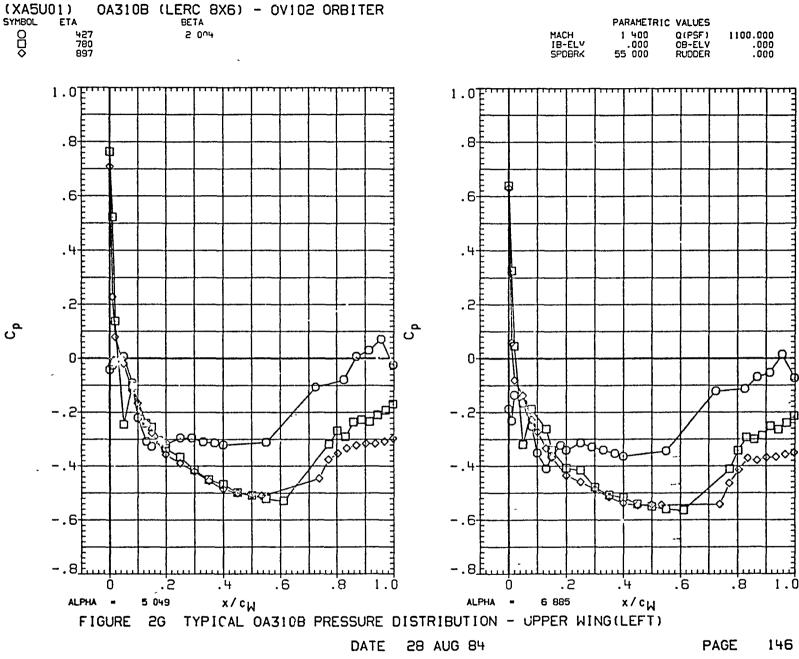
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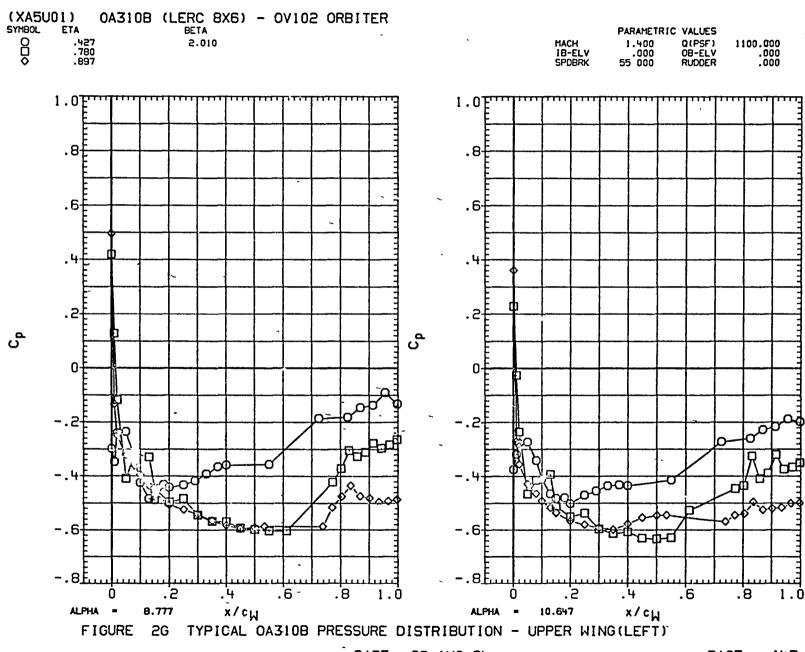
28 AUG 84

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Q(PSF) 08-ELV RUDDER

1100.000 .000 .000

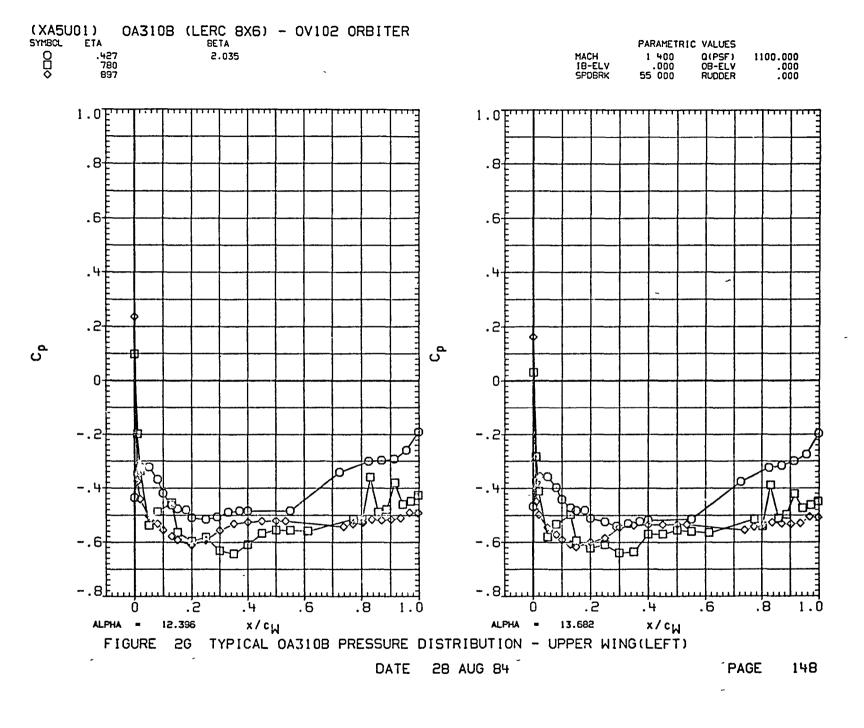


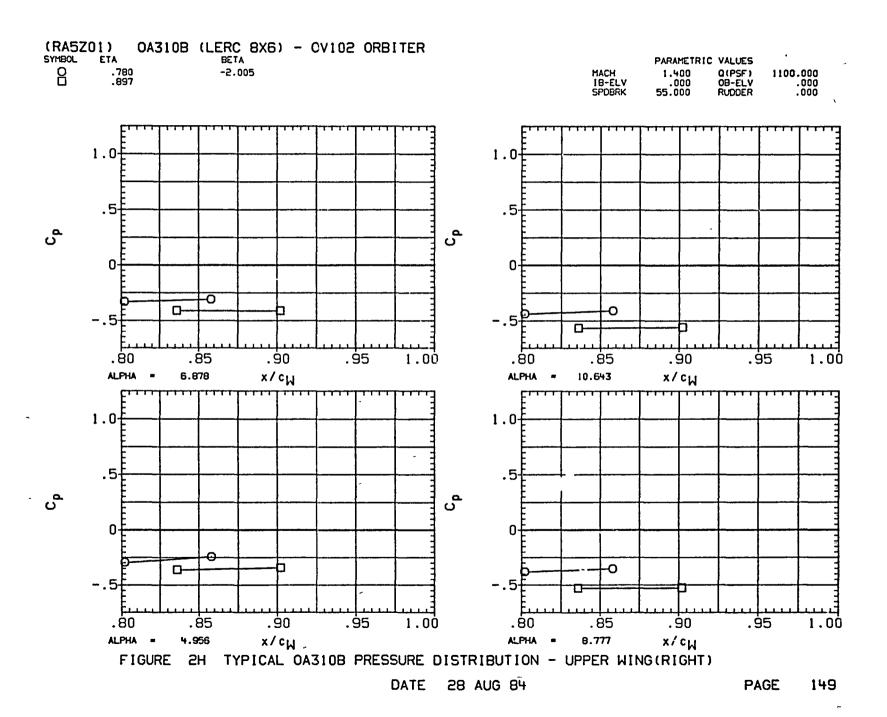


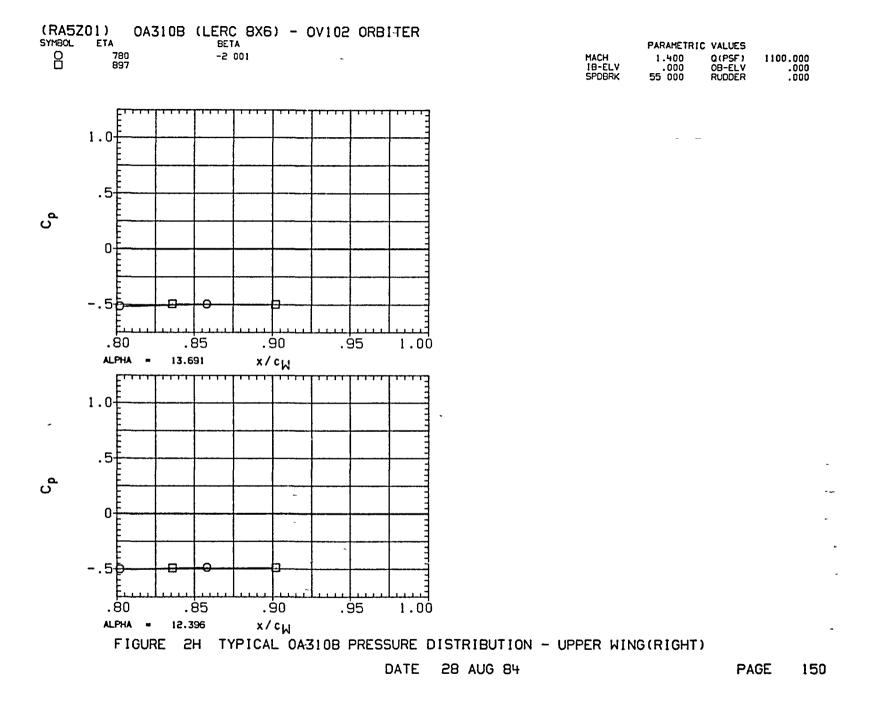
DATE 28, AUG 84

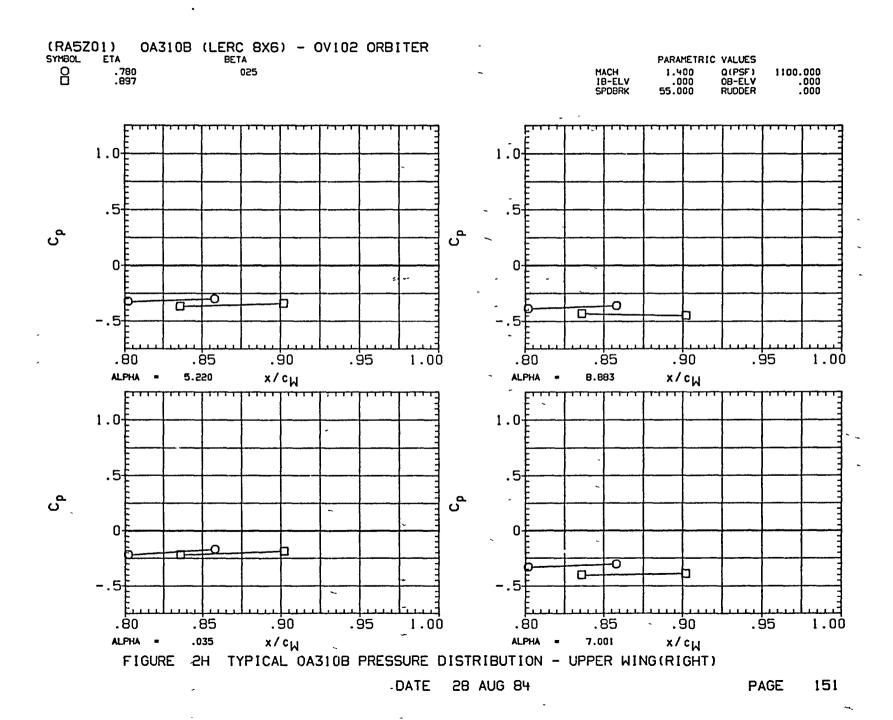
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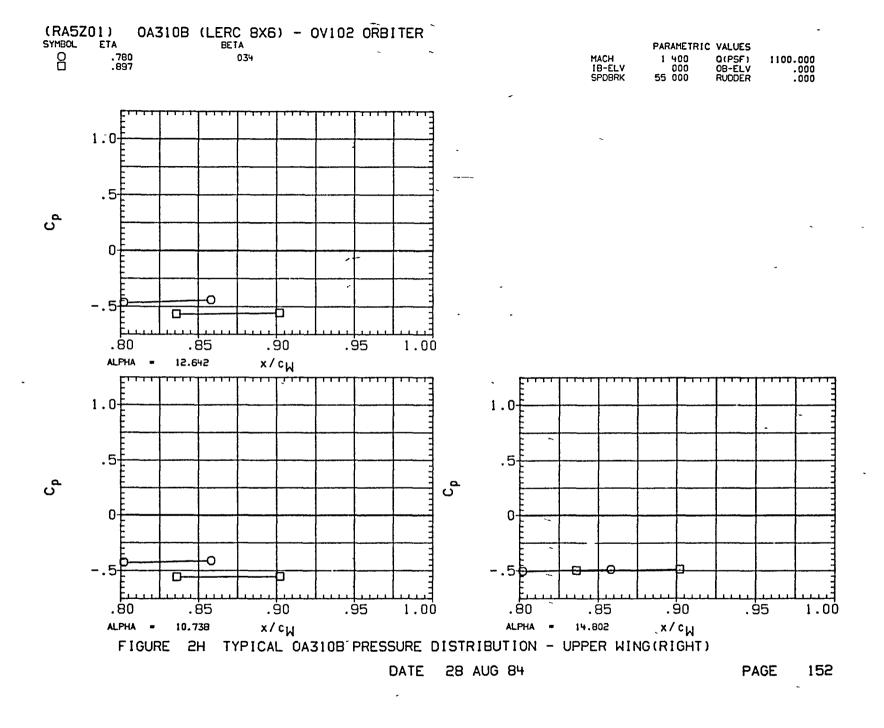
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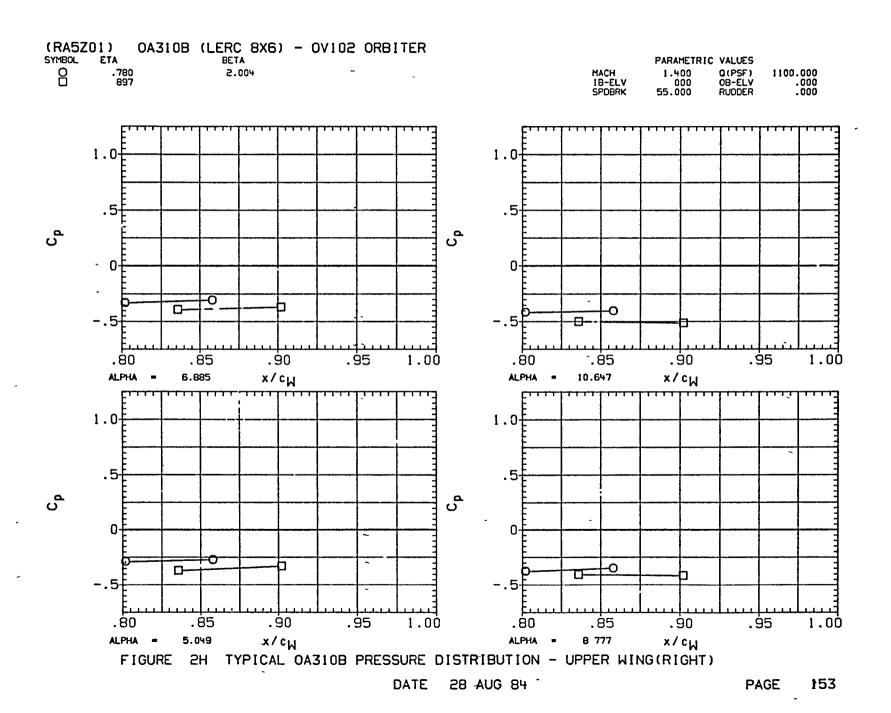


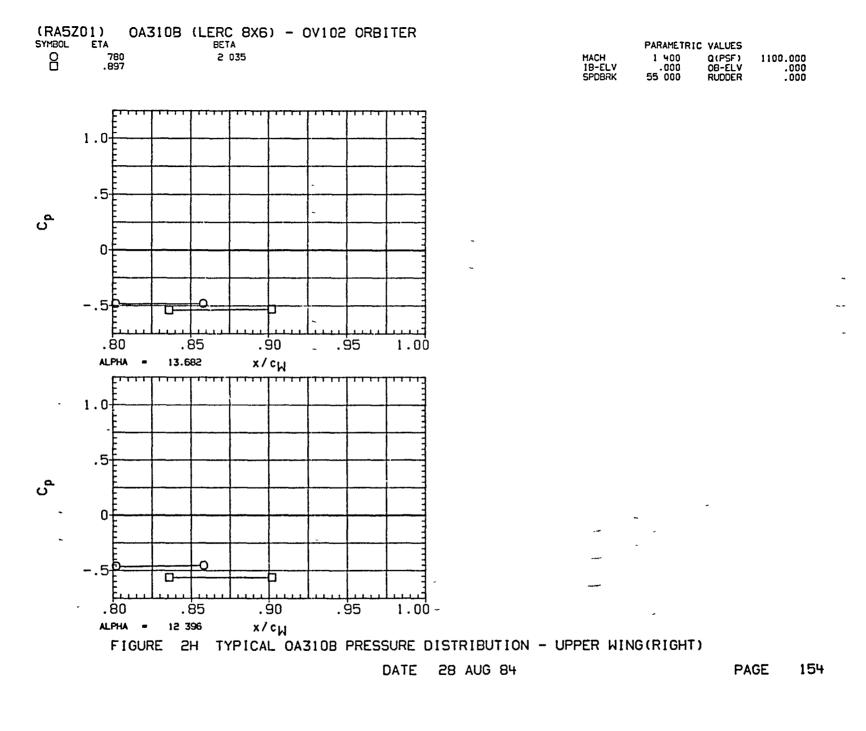


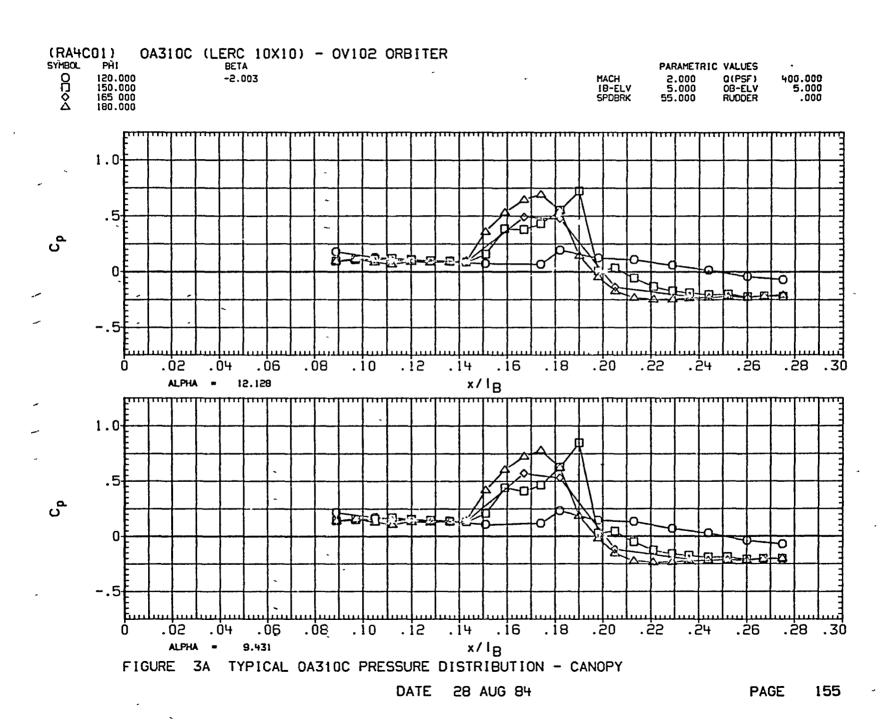


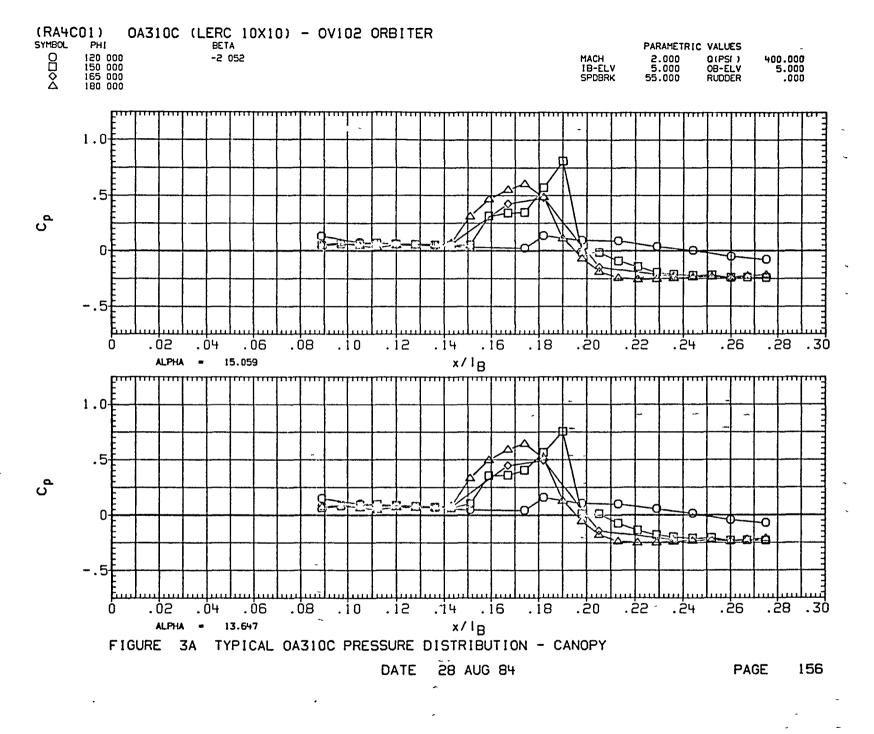


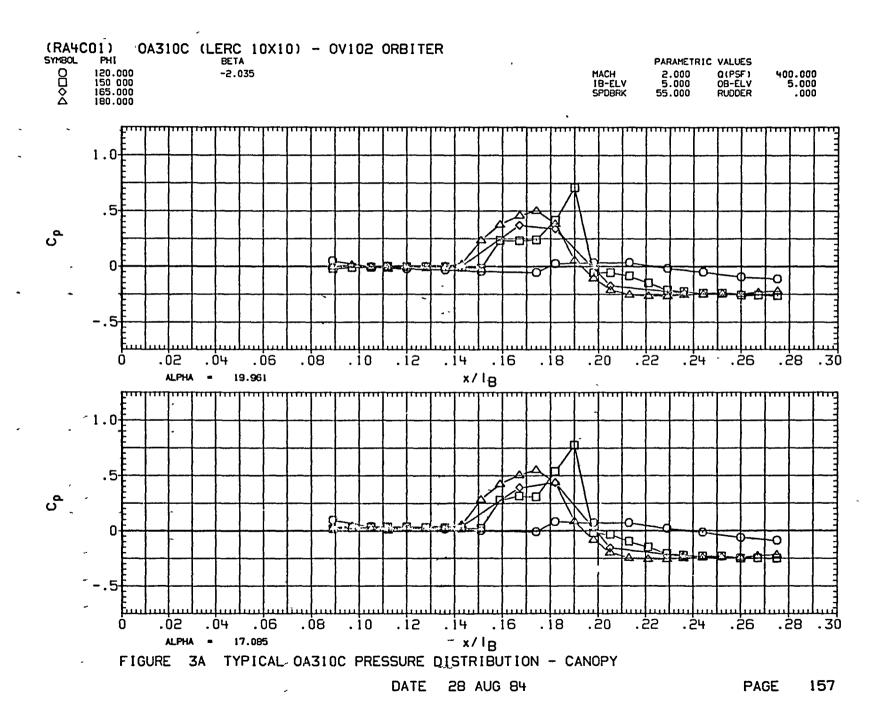


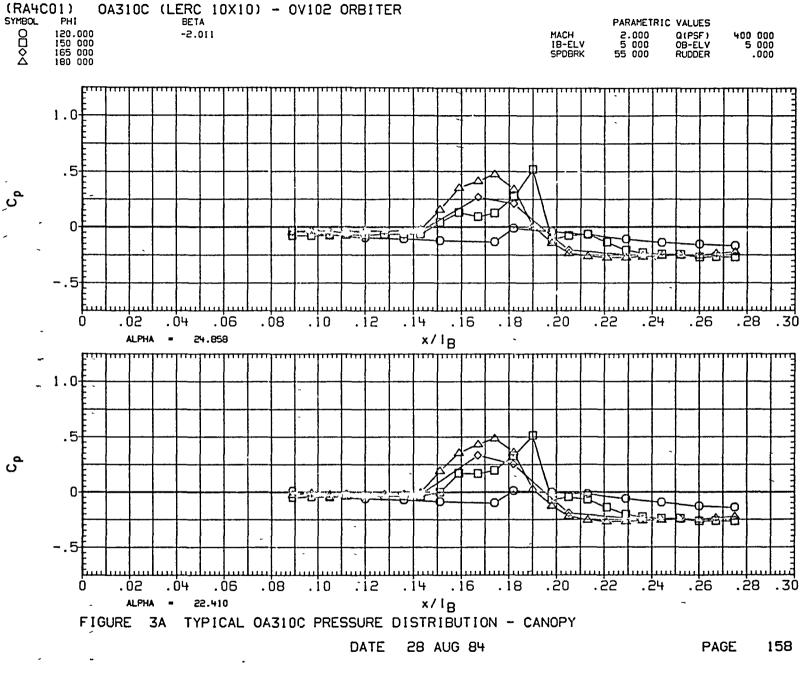


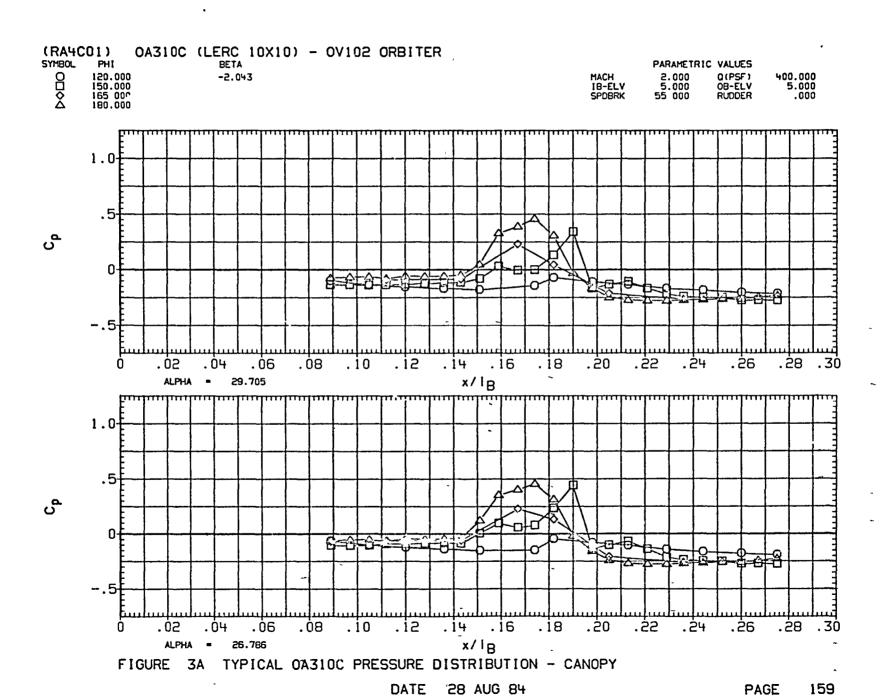


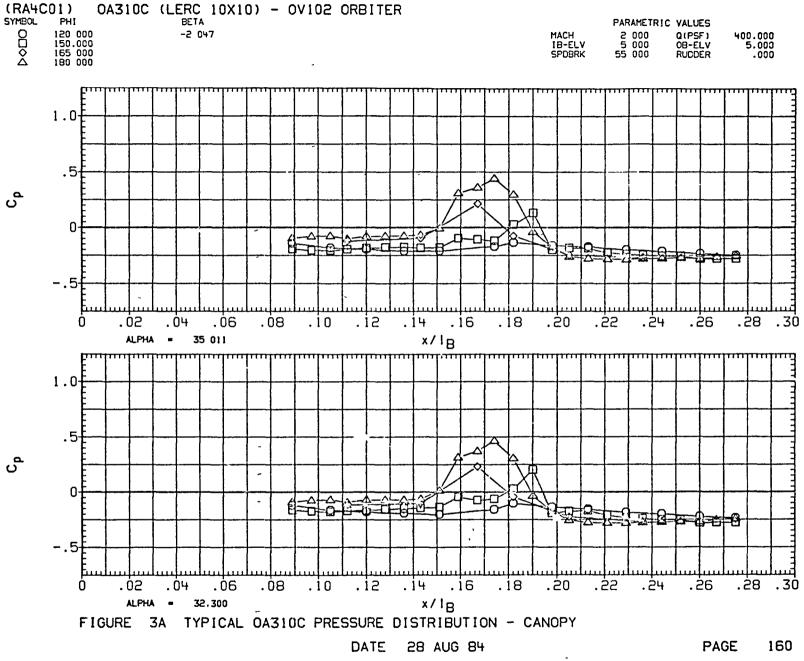


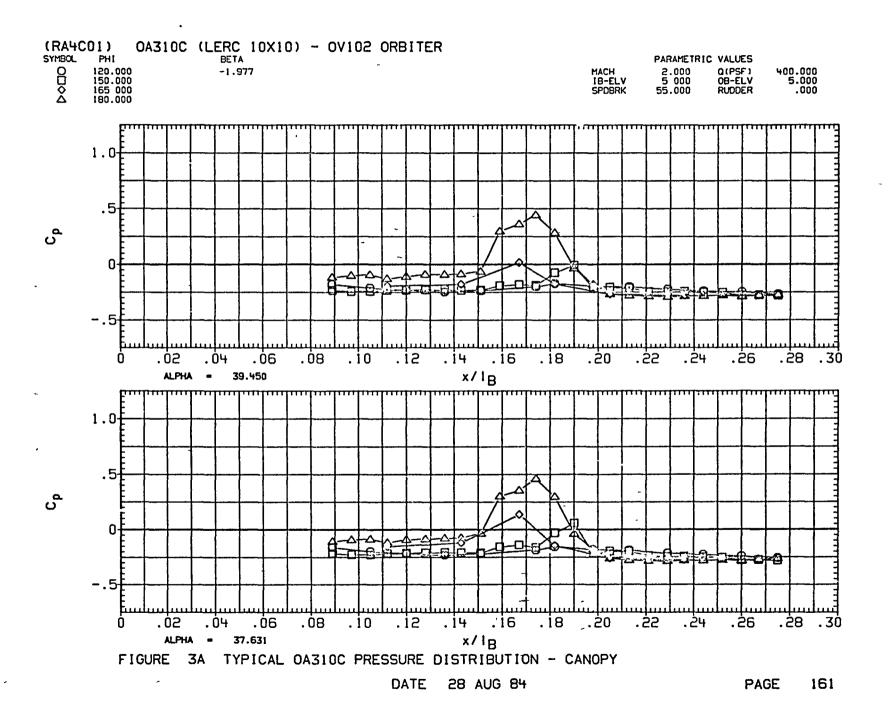


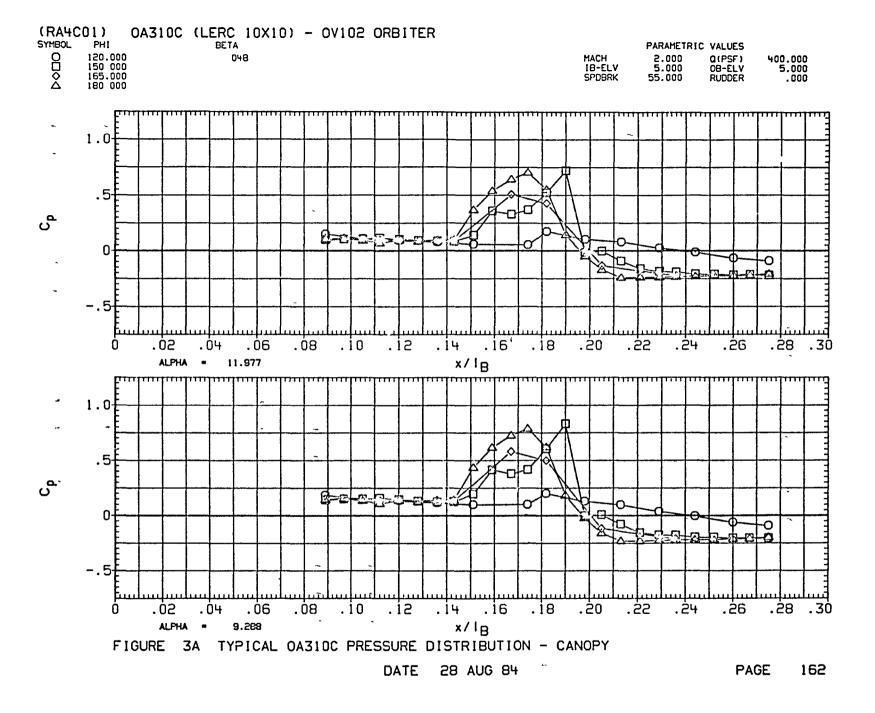


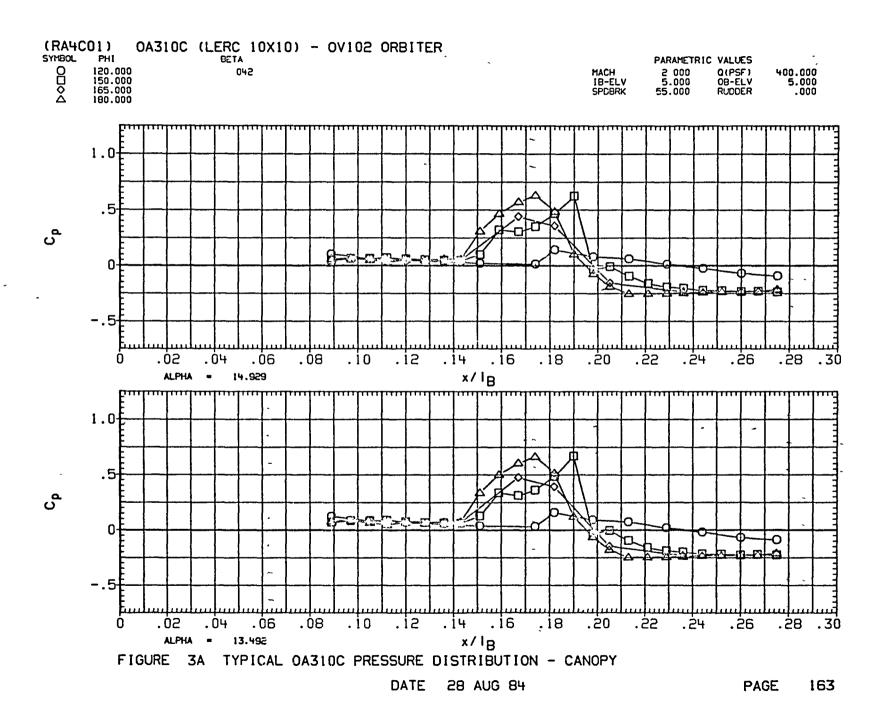


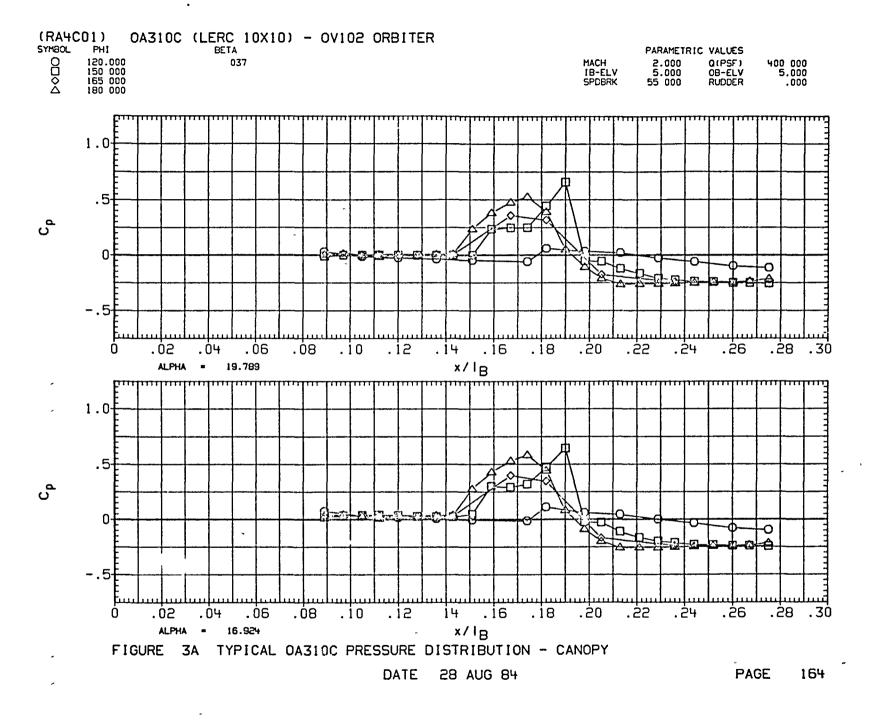


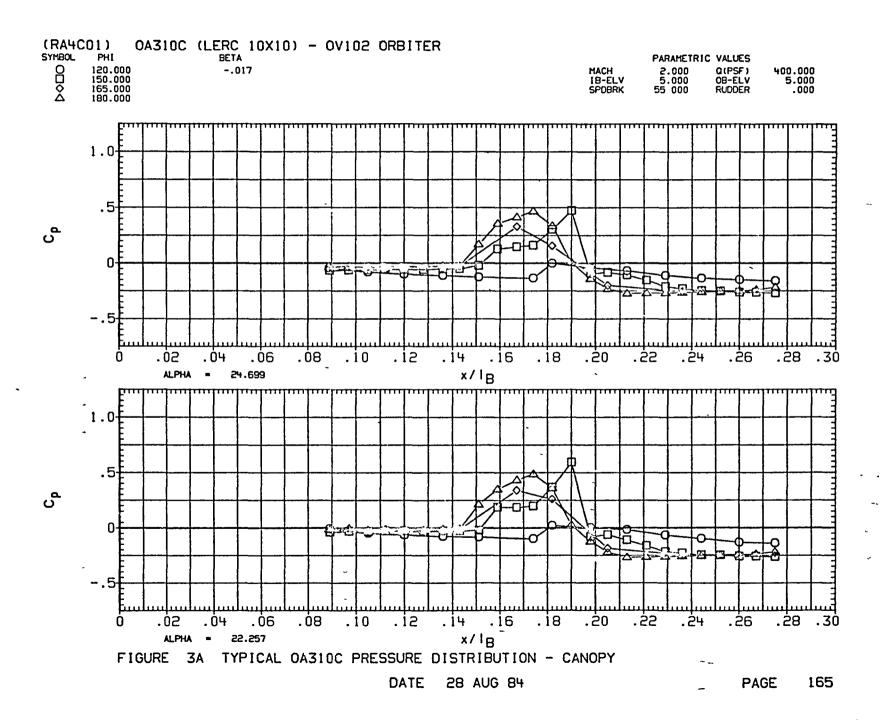


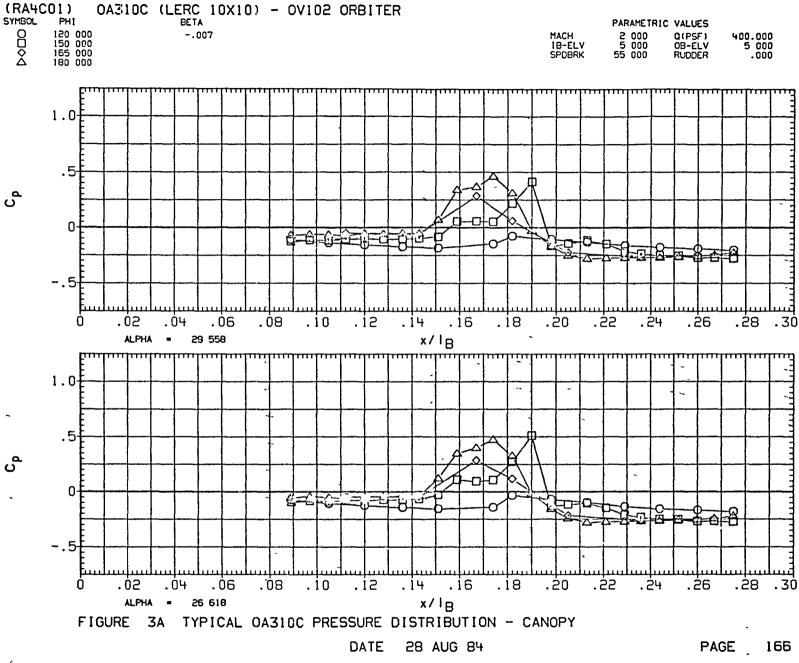


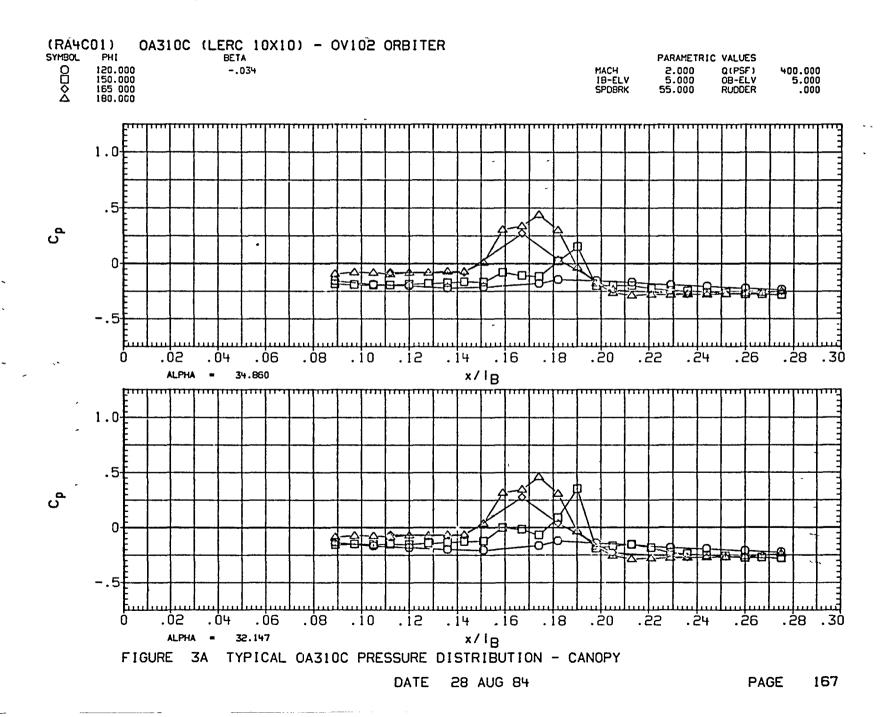


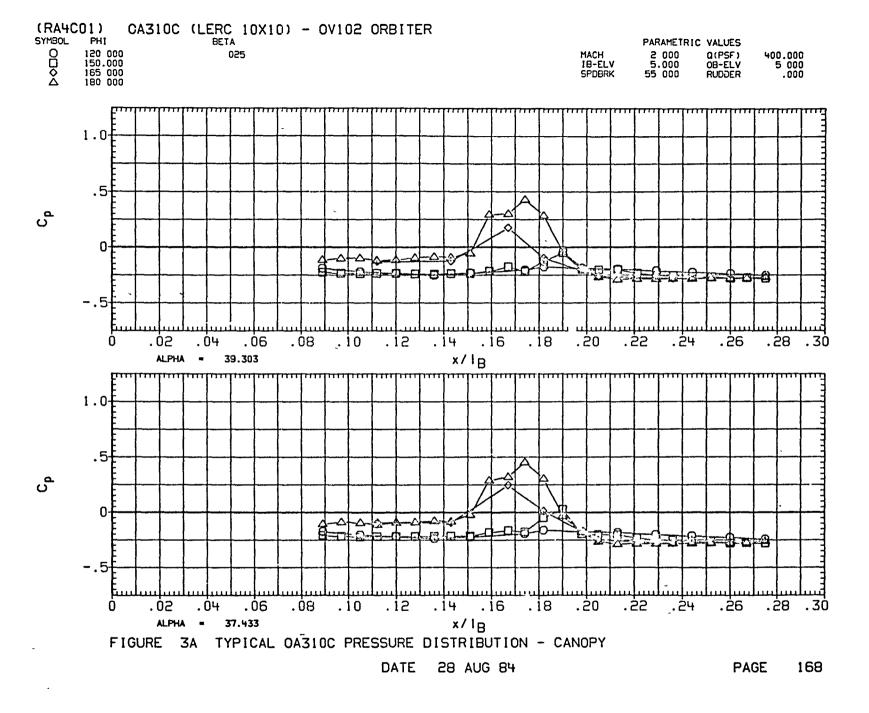


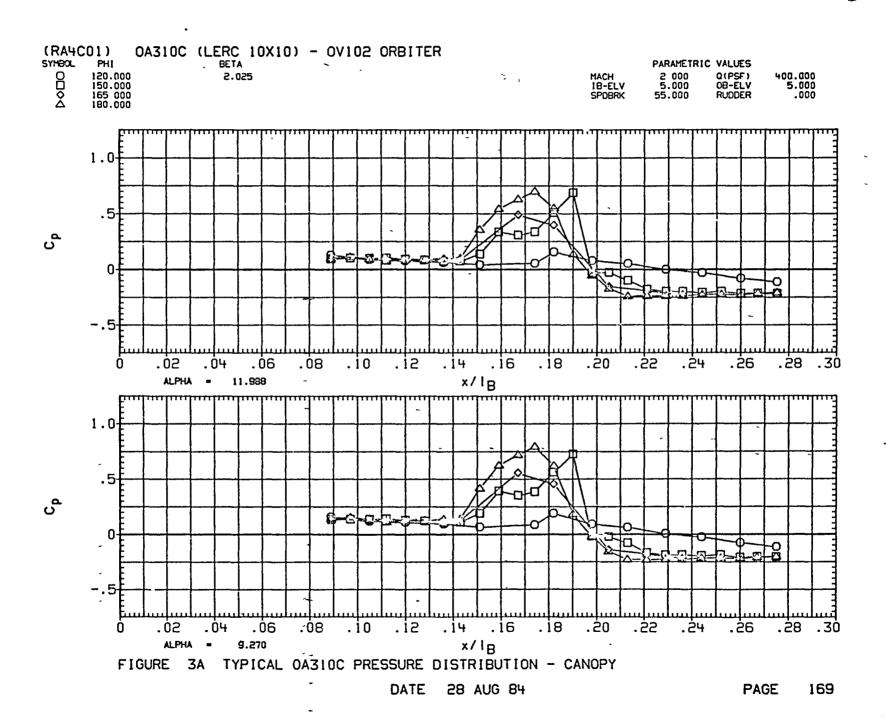


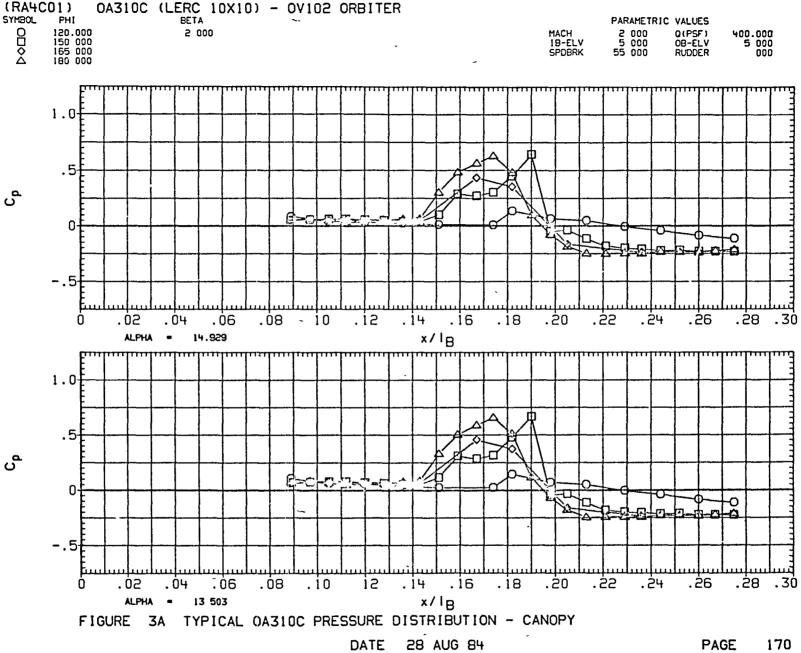


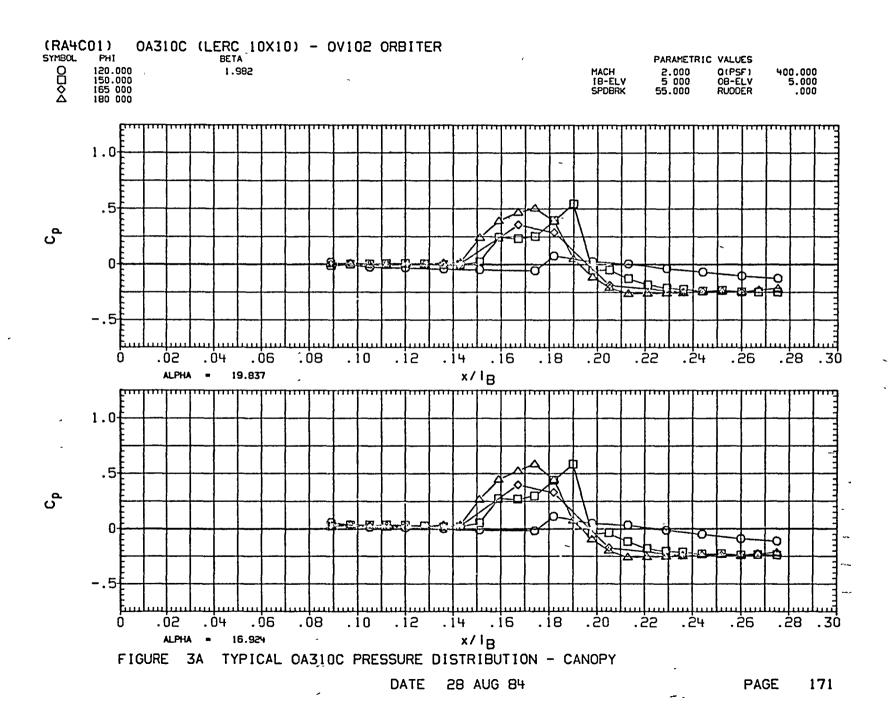


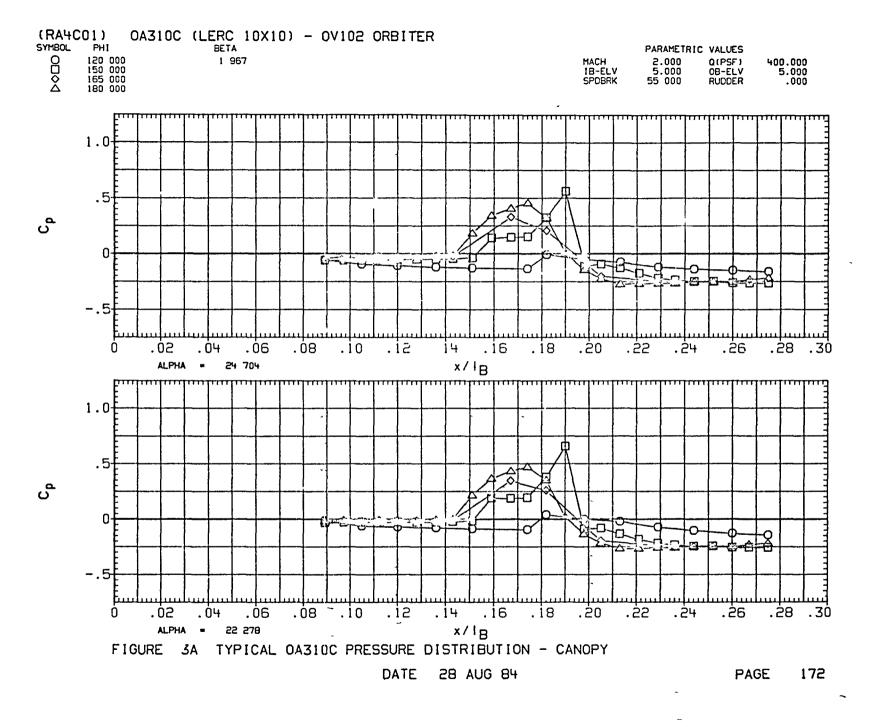


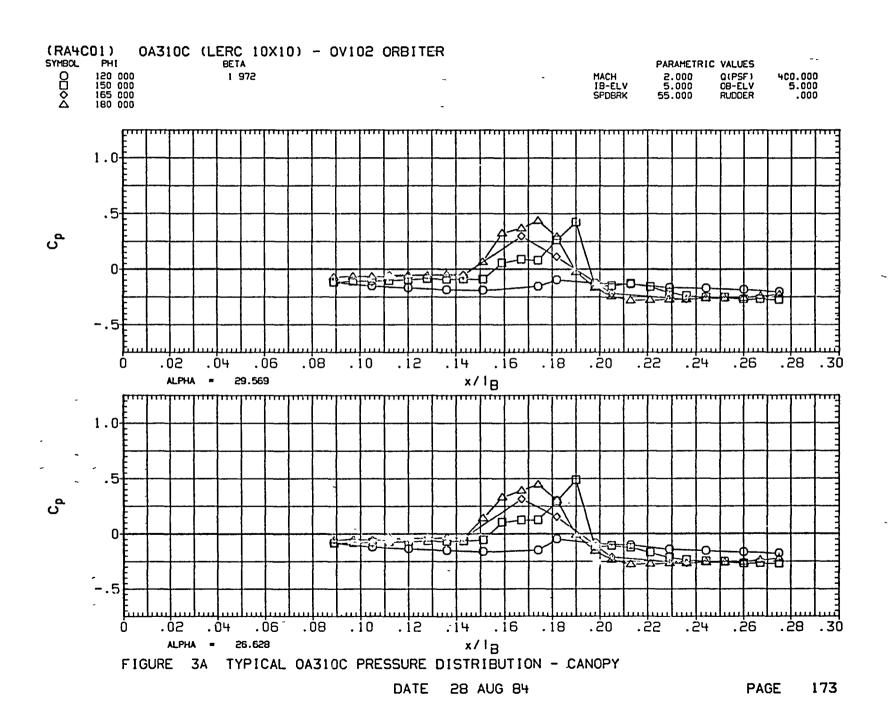


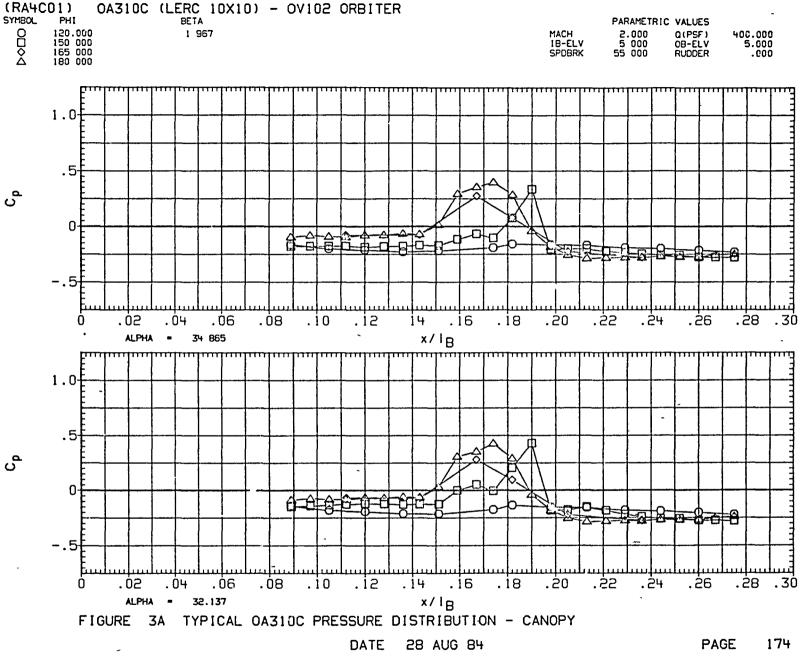


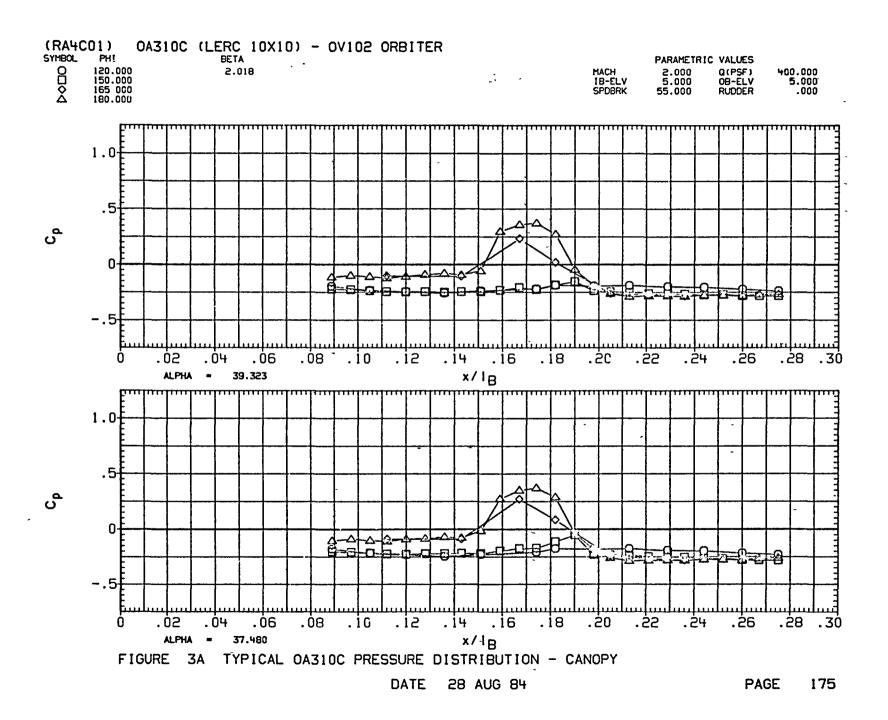


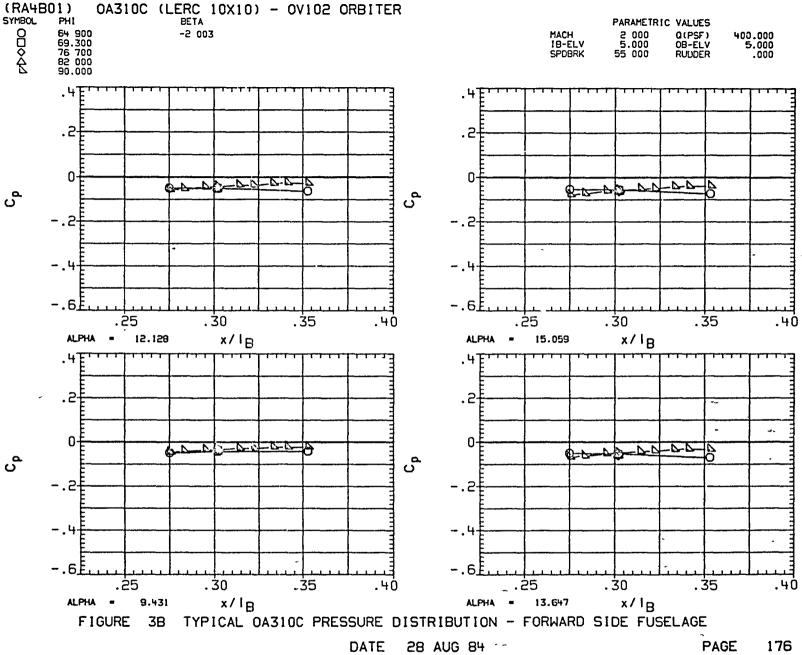


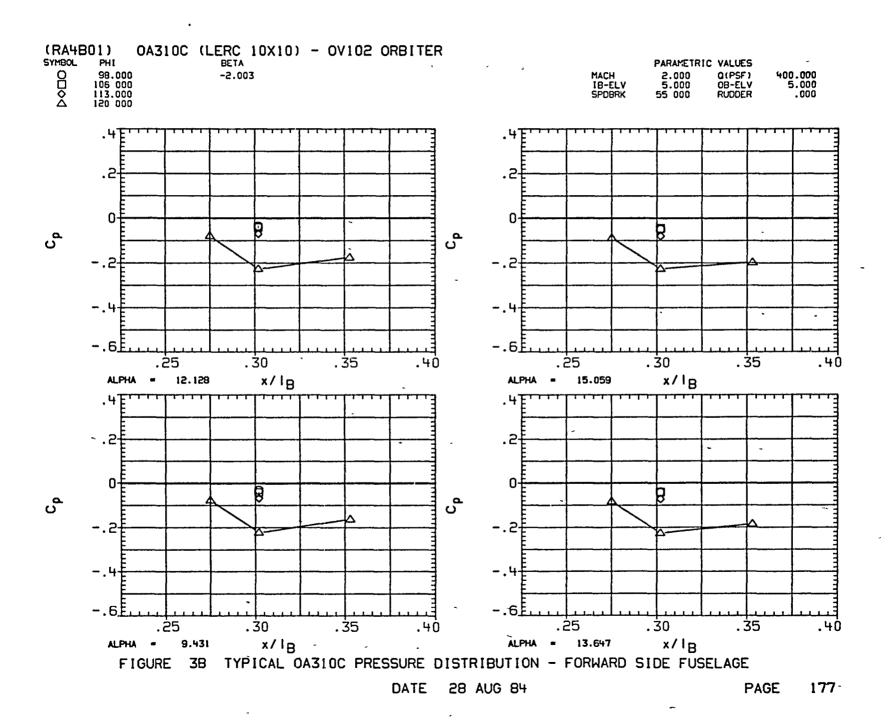


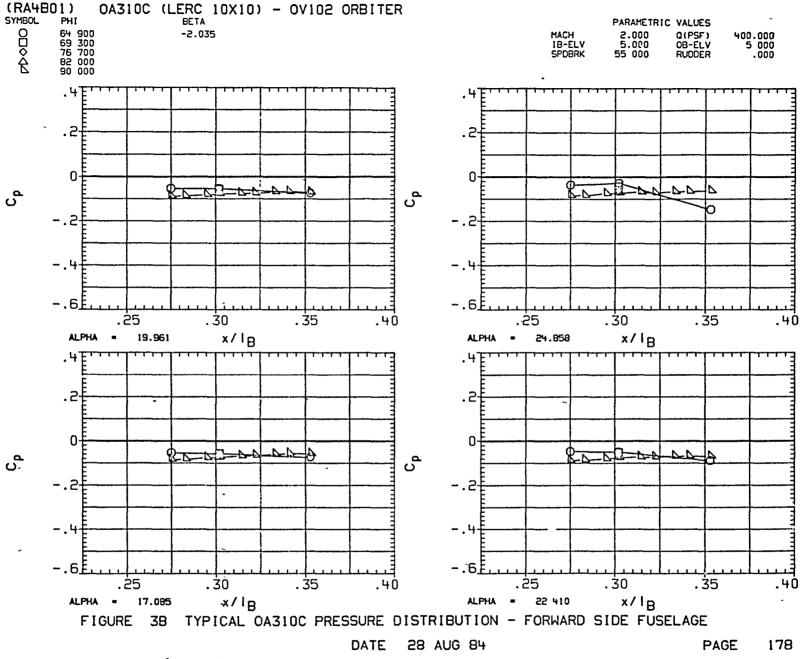


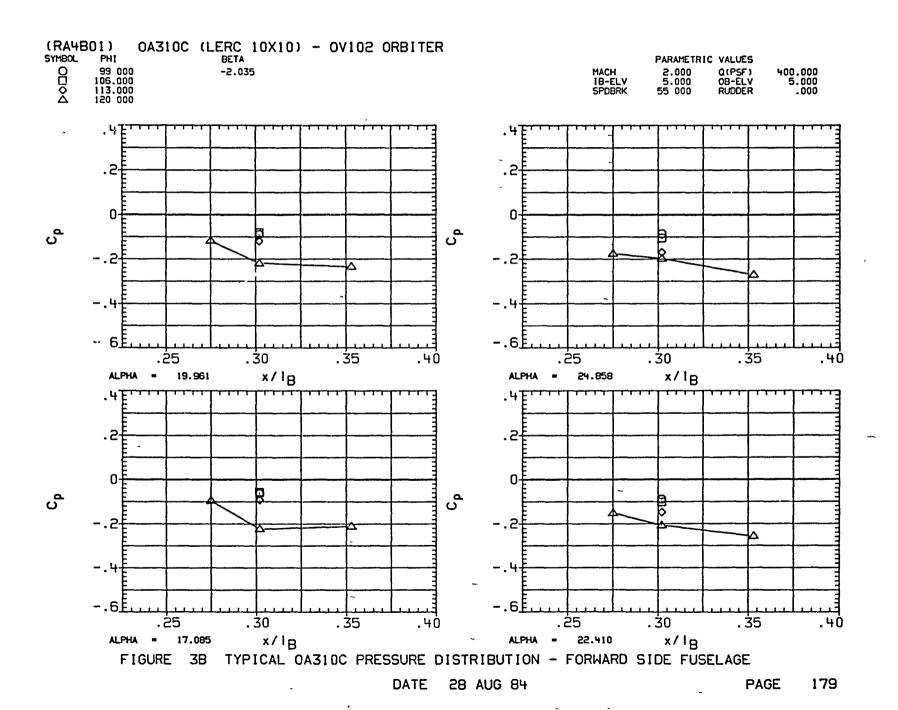


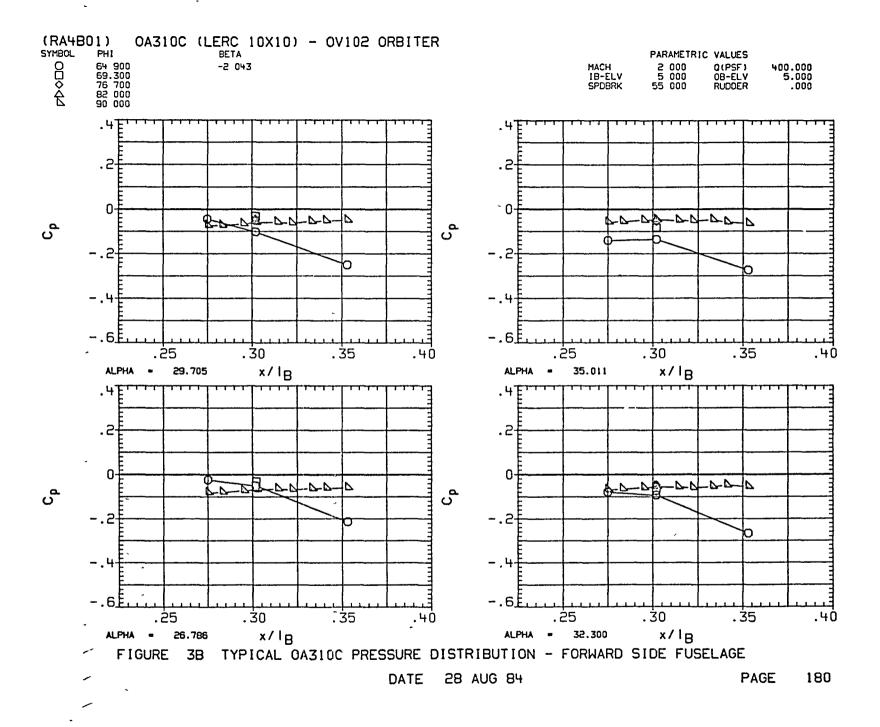


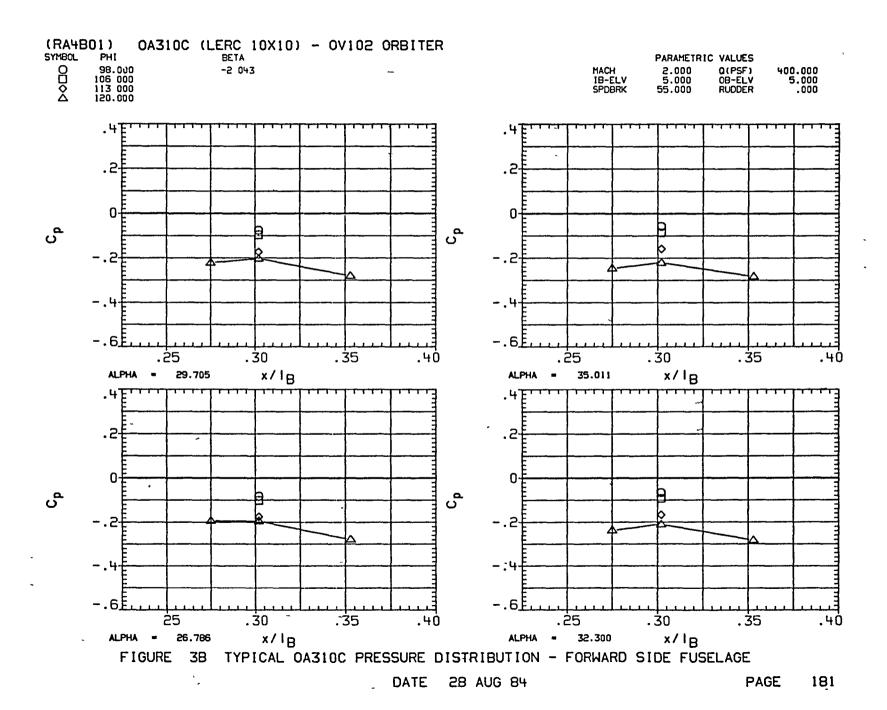


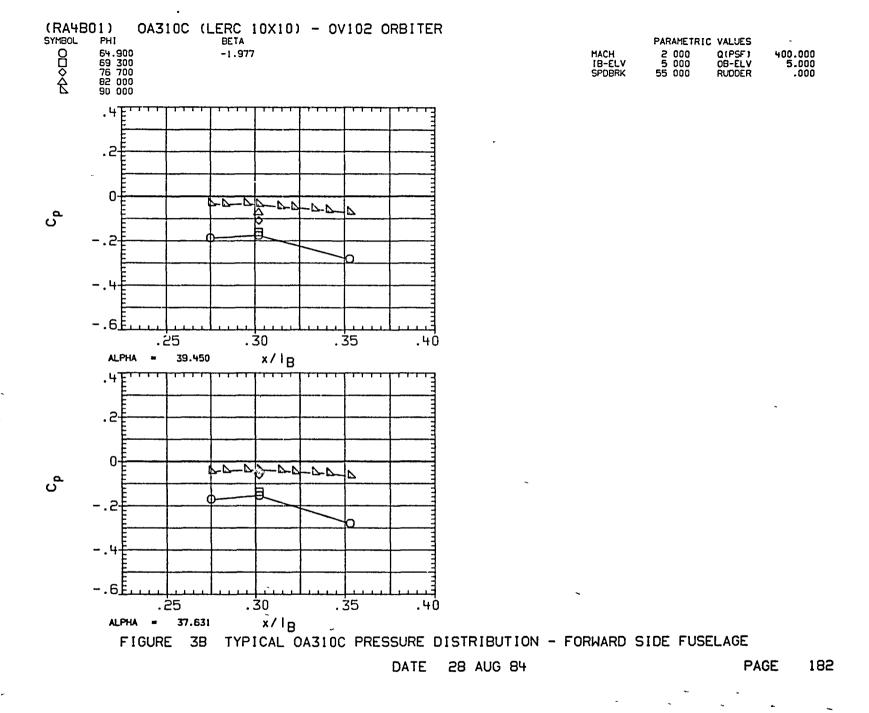


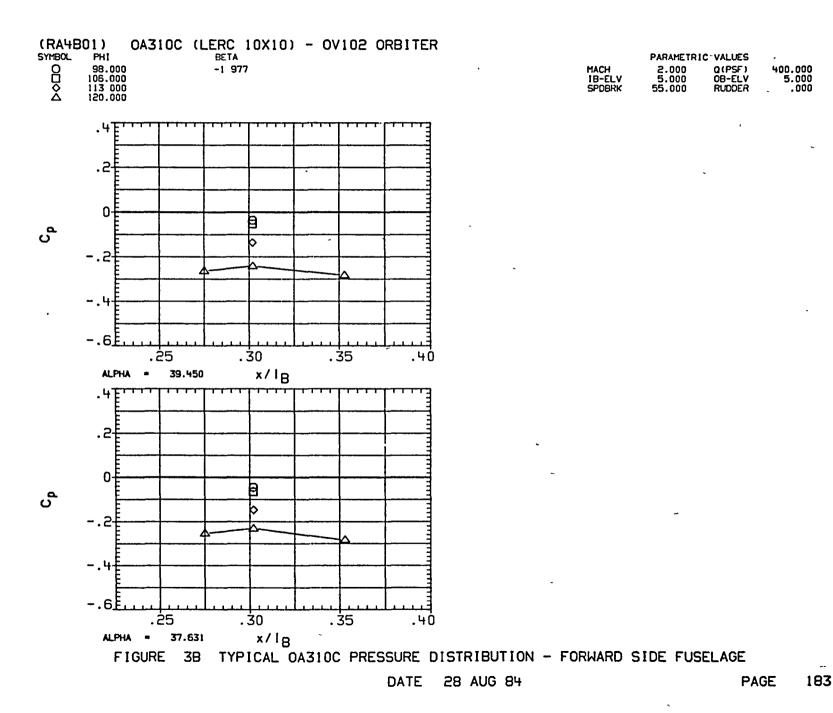


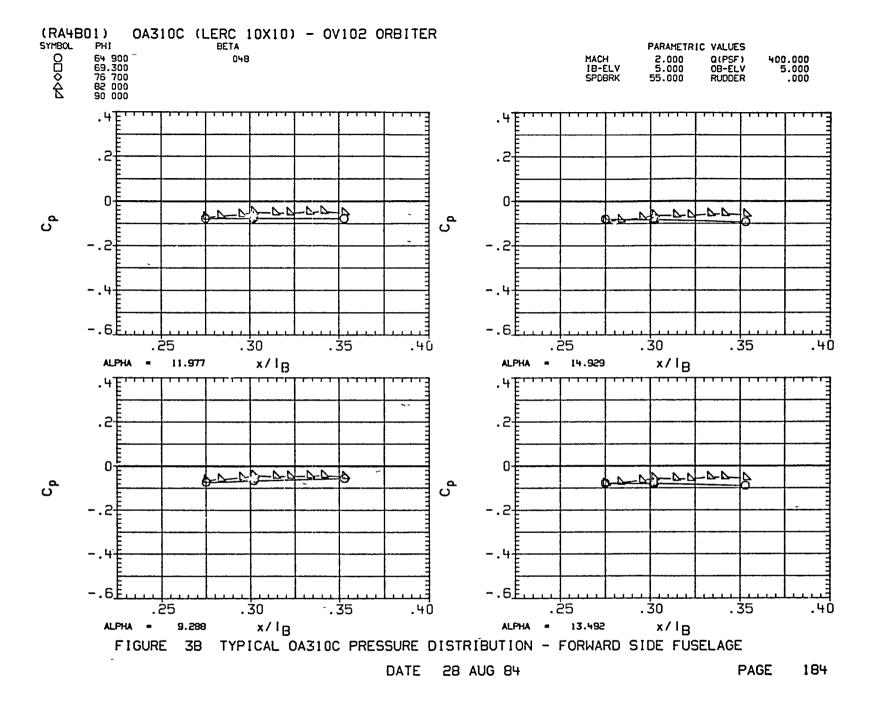


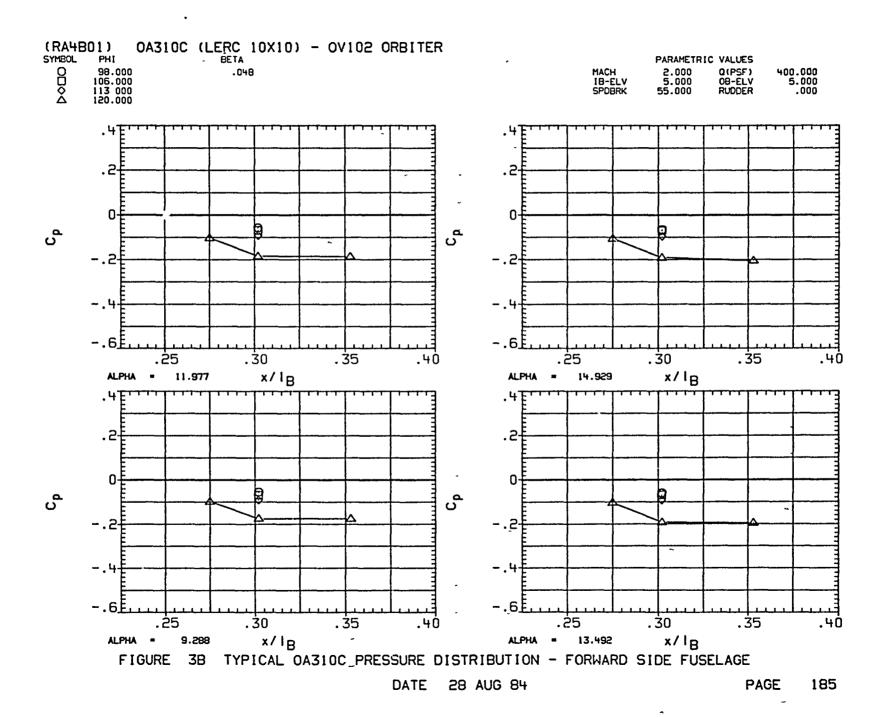


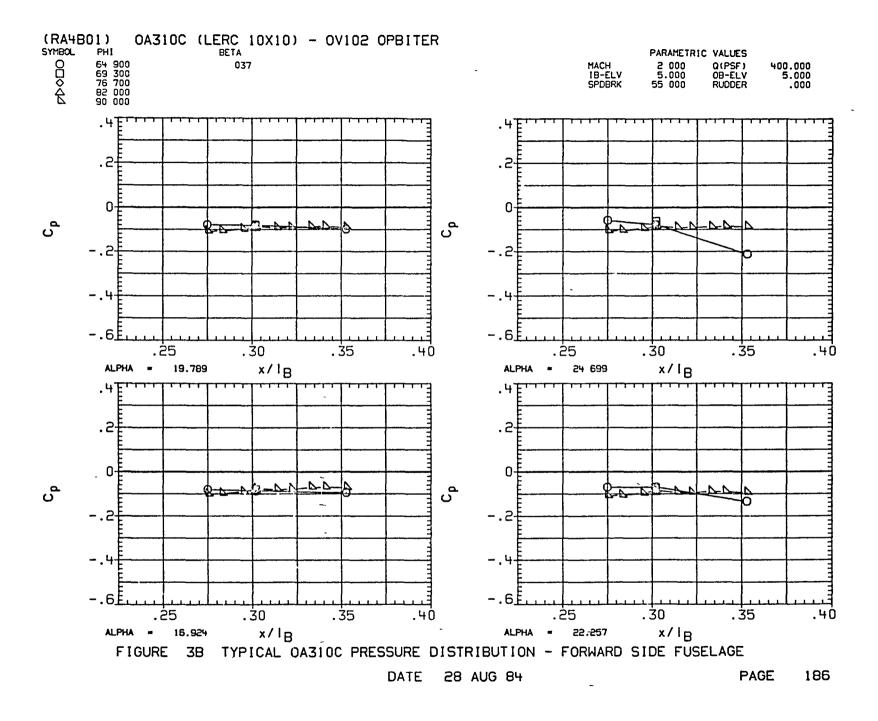


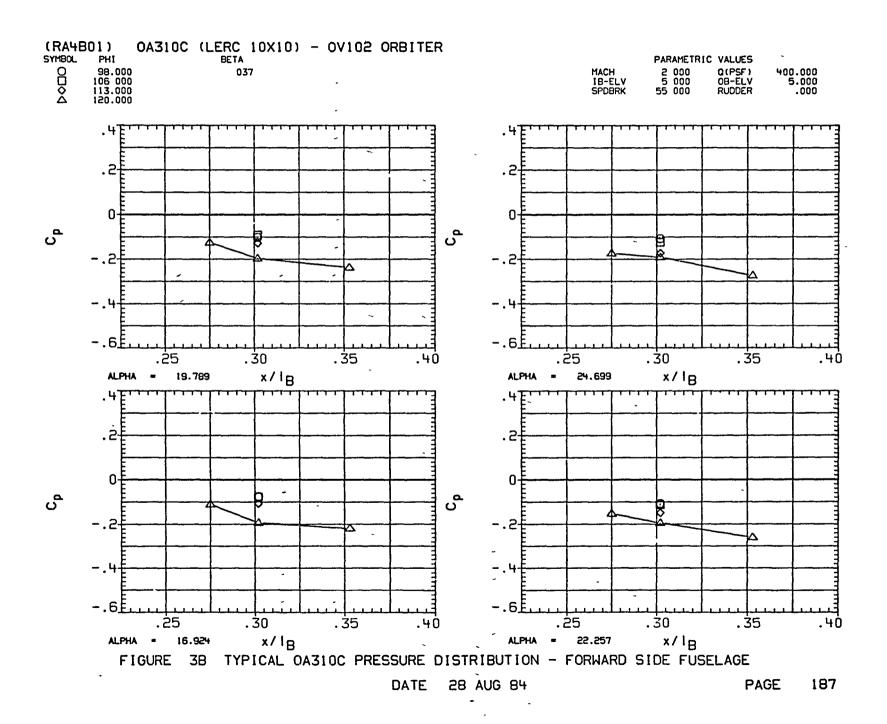


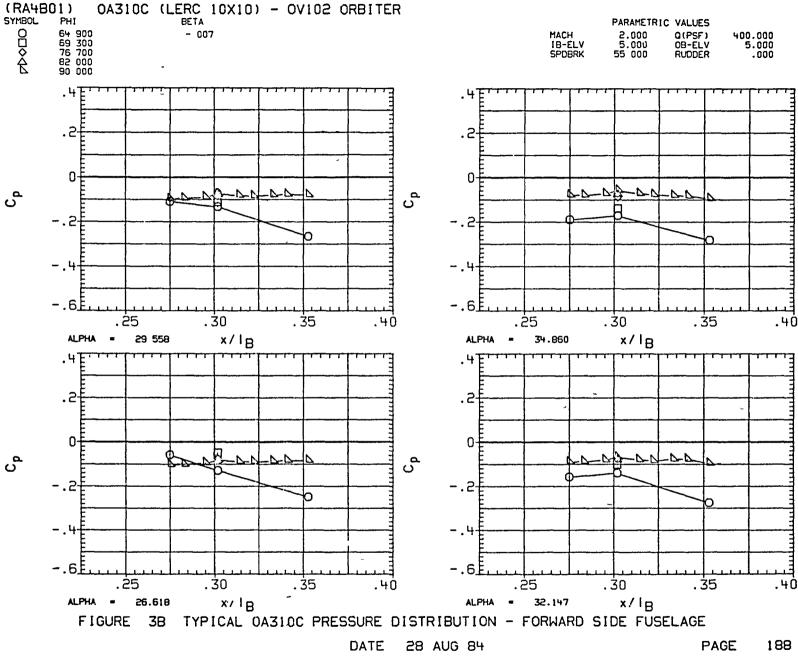


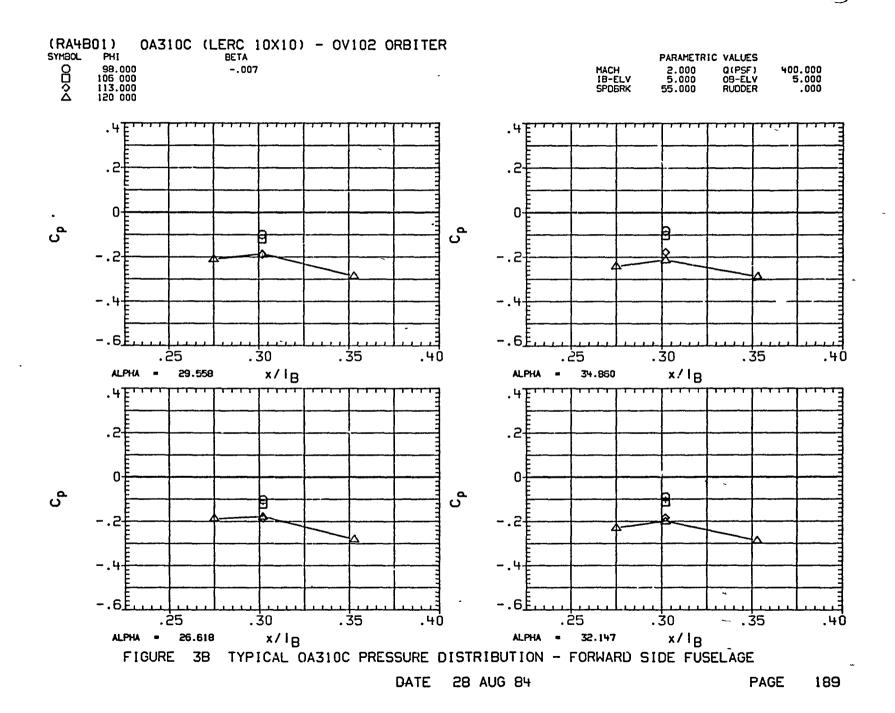


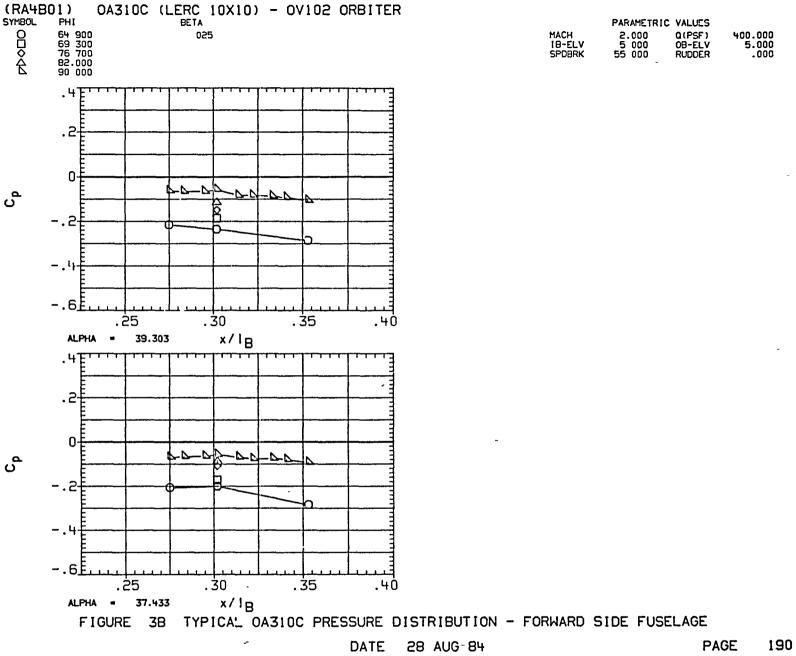


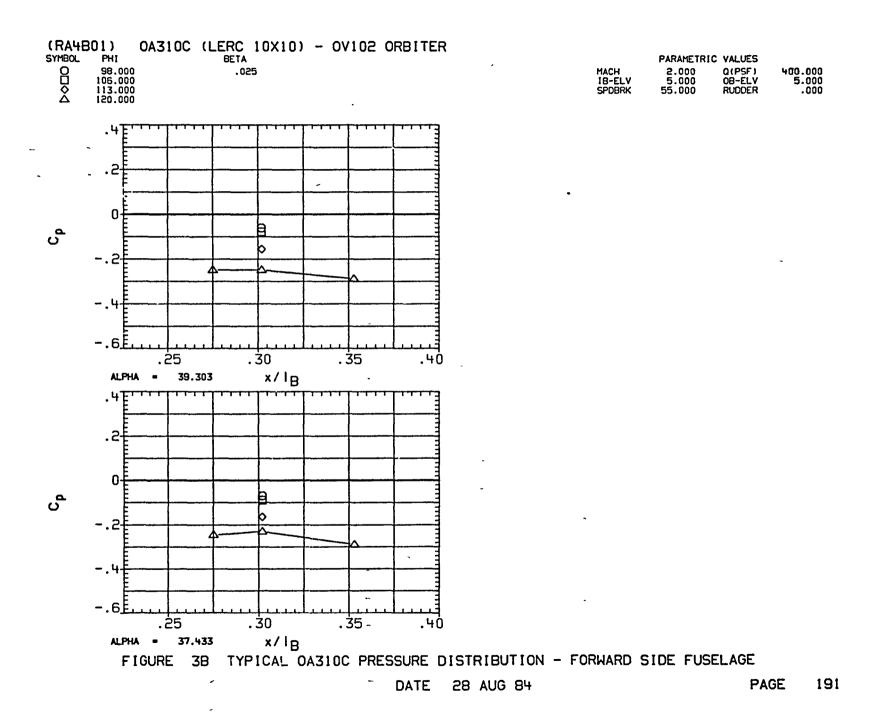


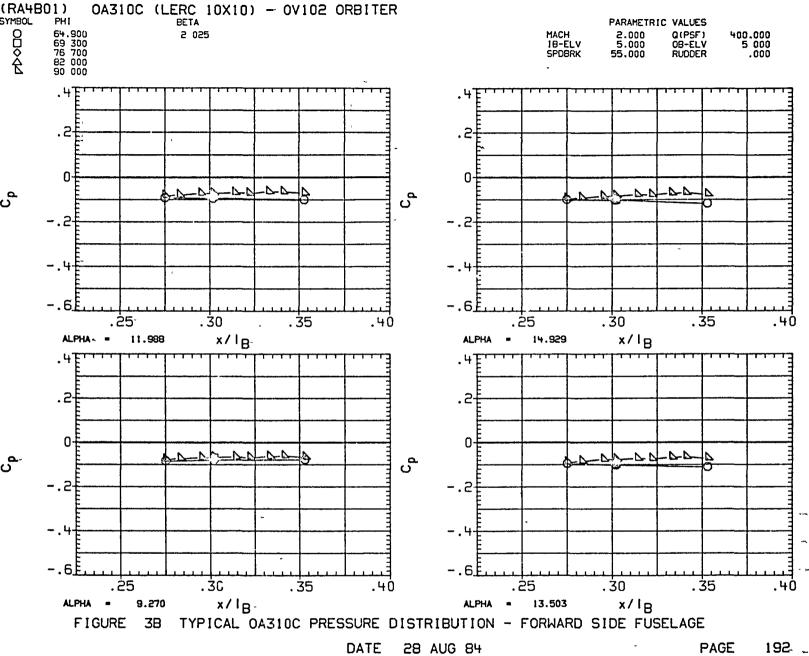


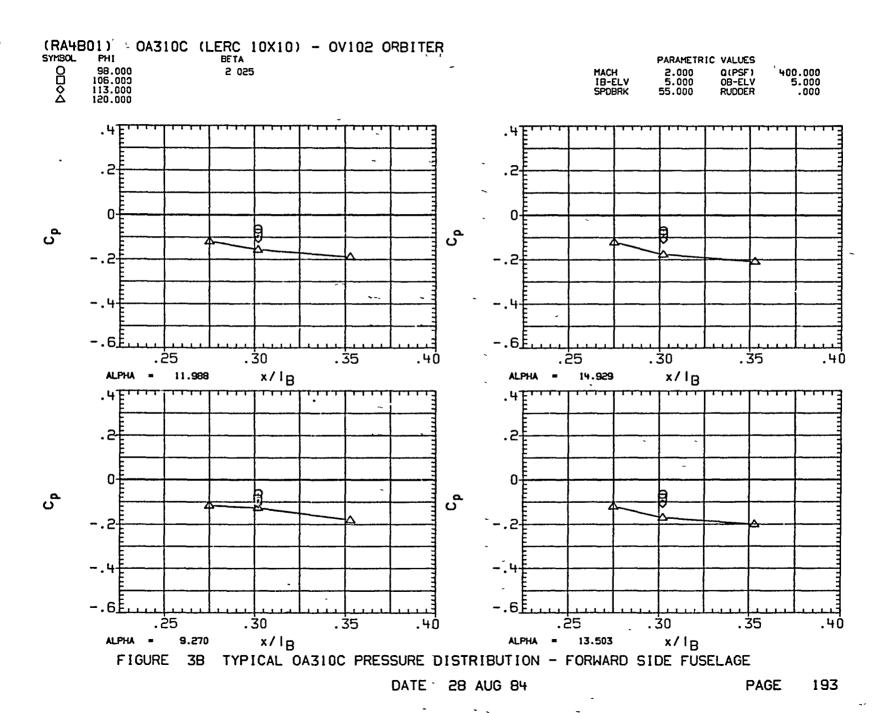


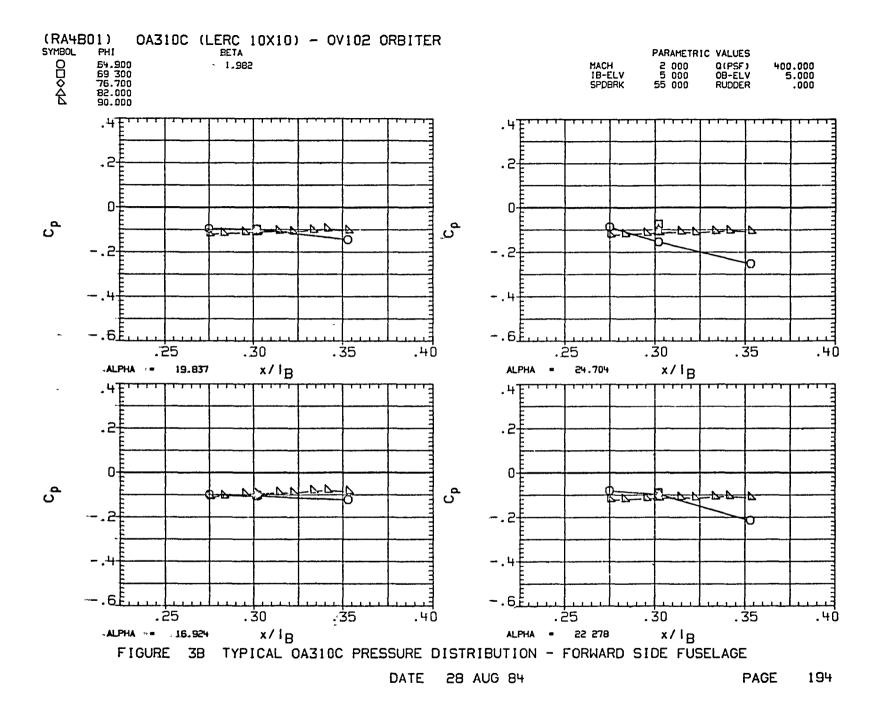


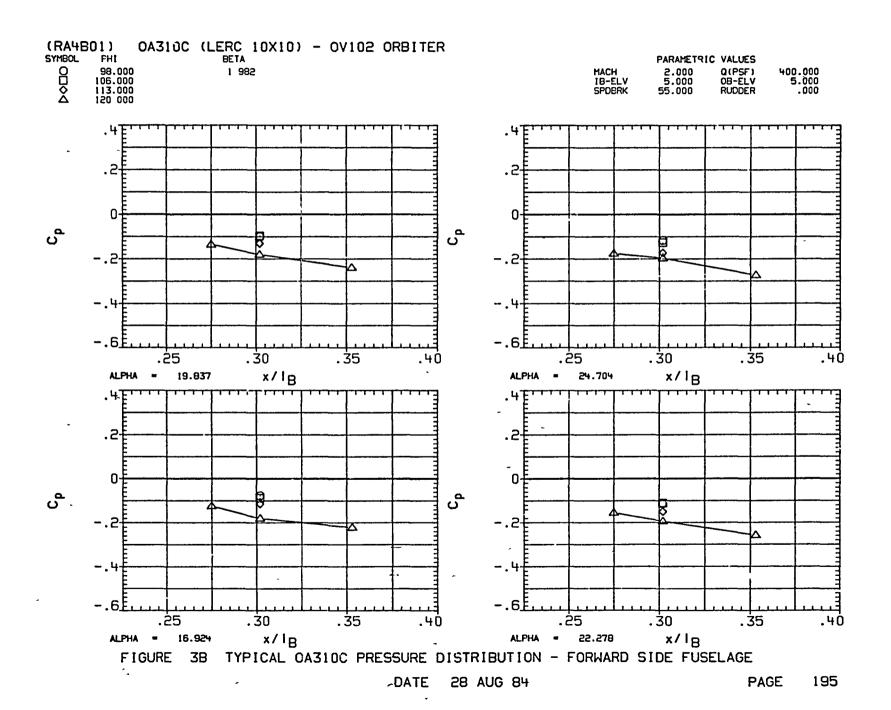


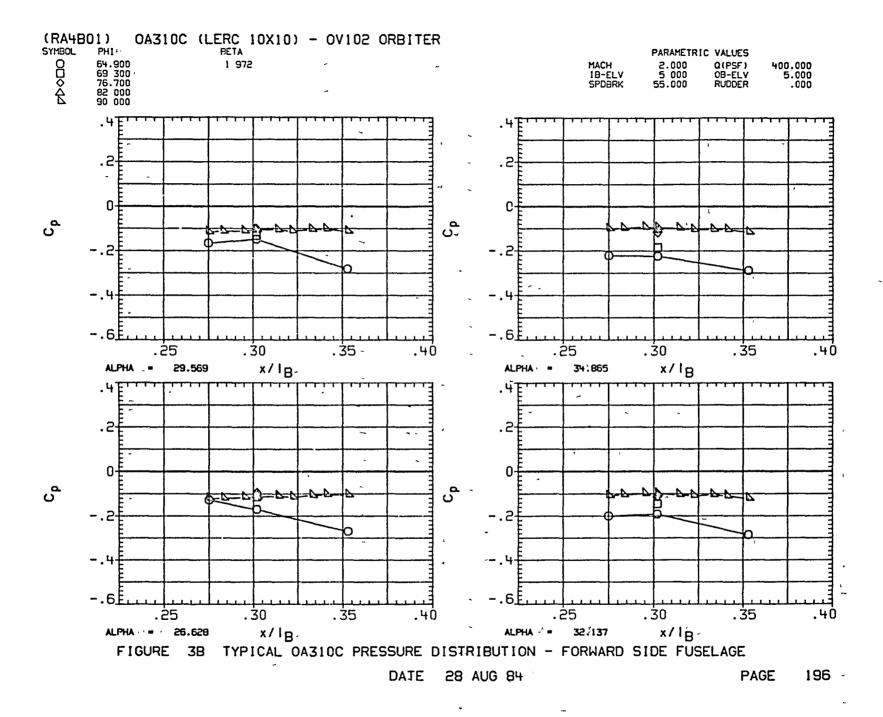


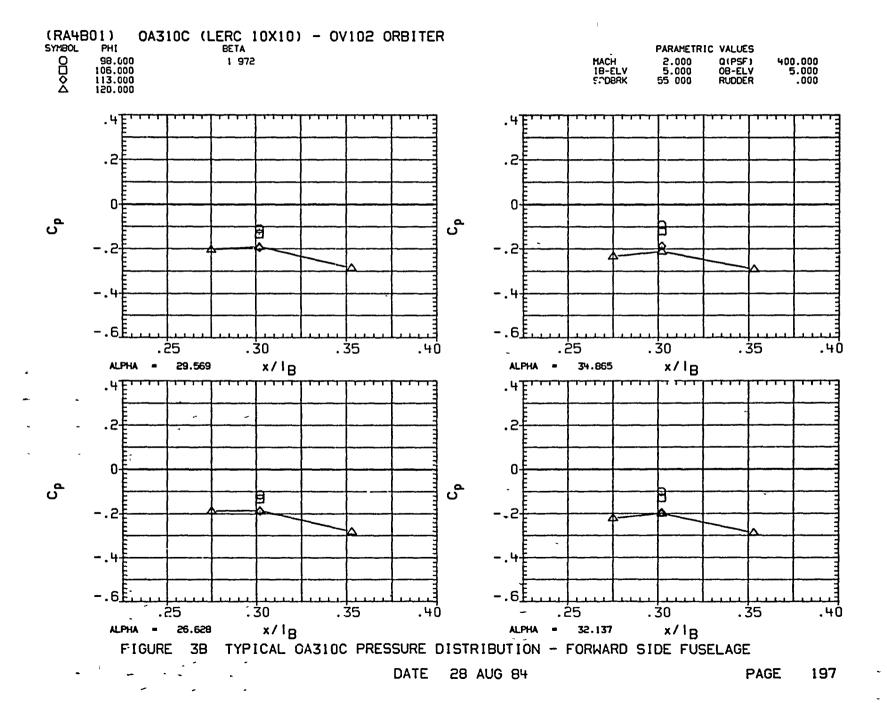


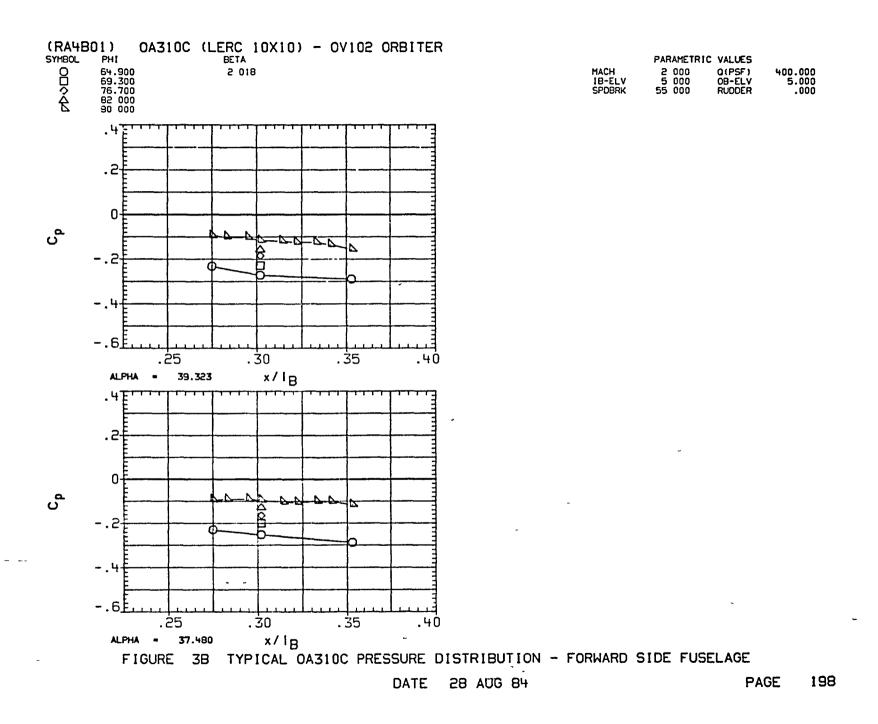


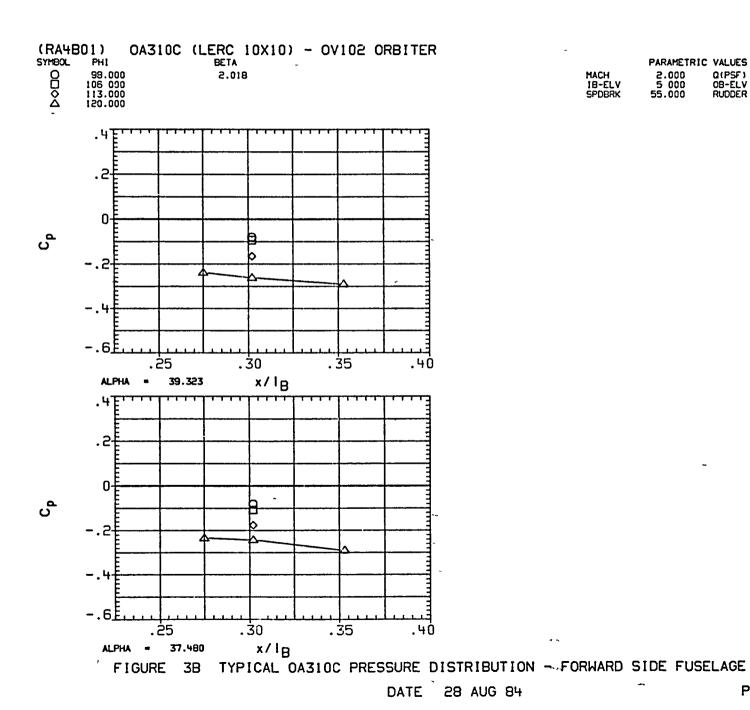








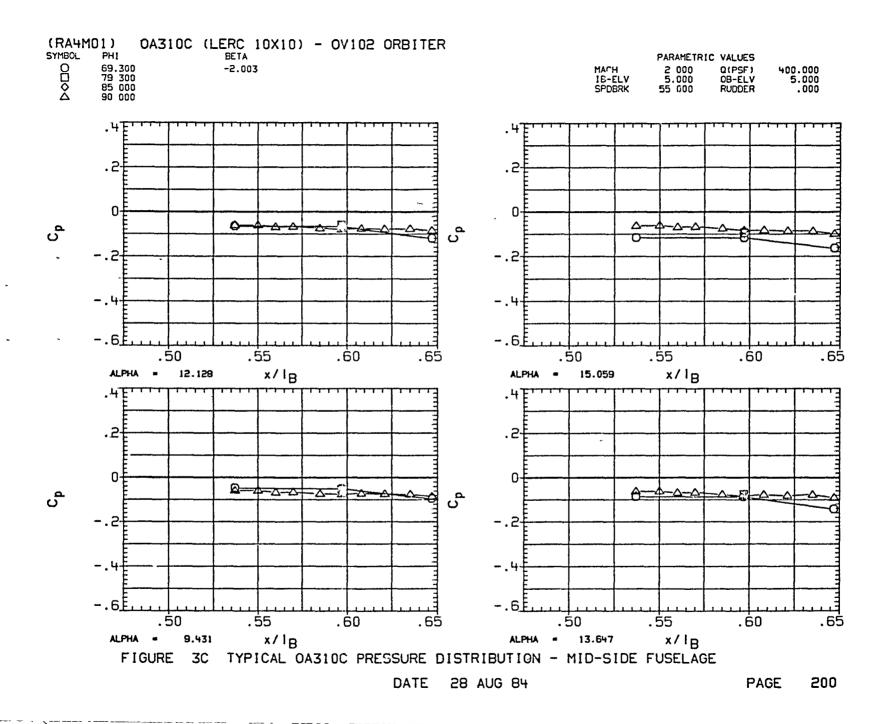


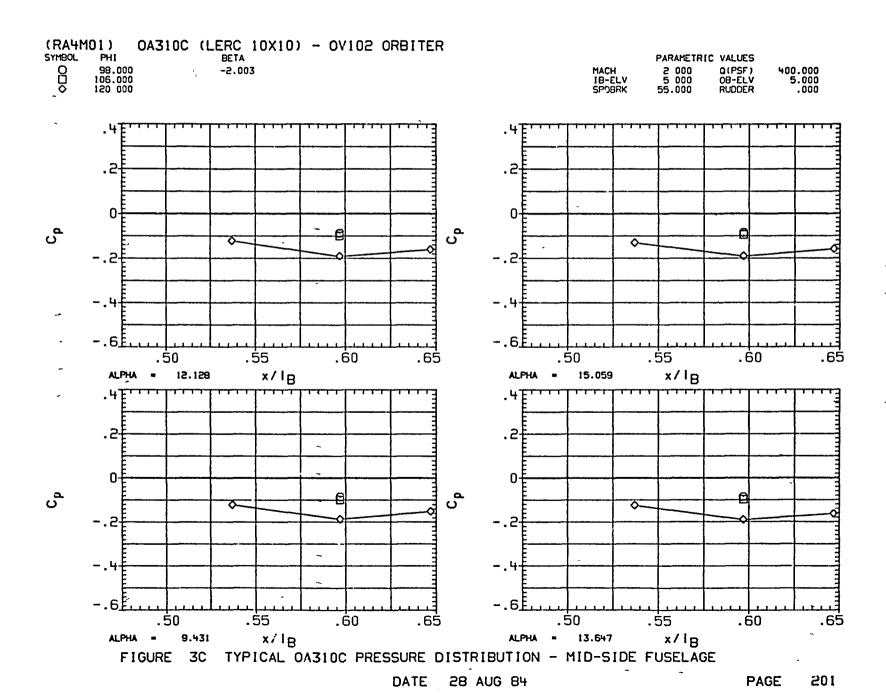


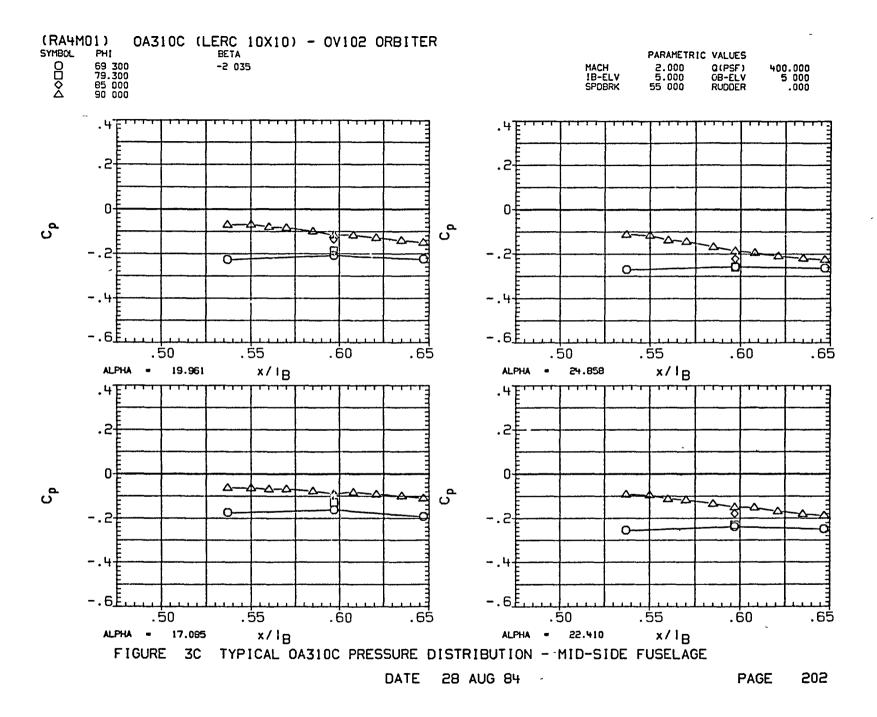
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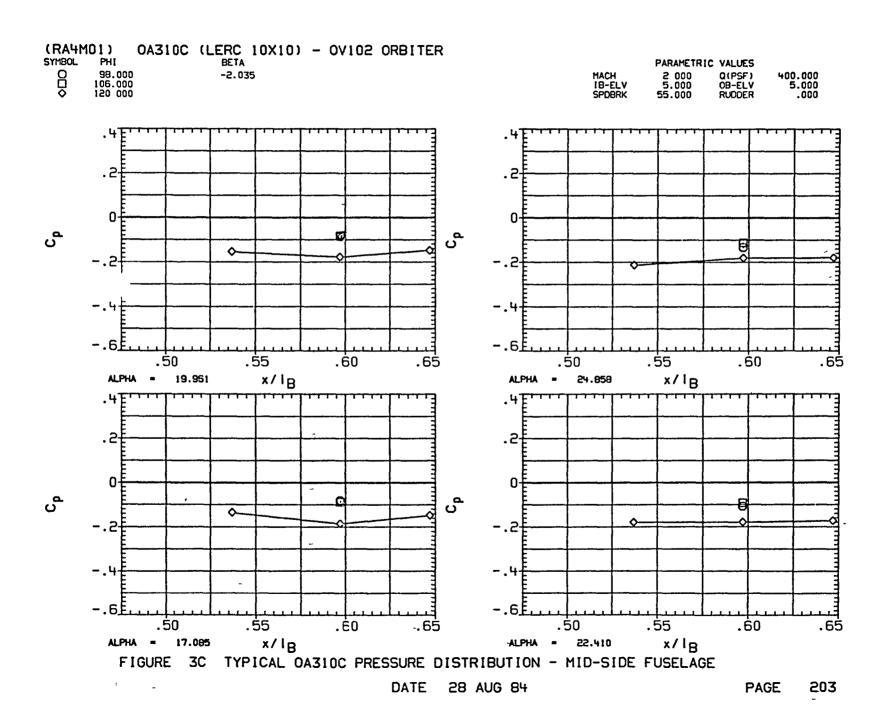
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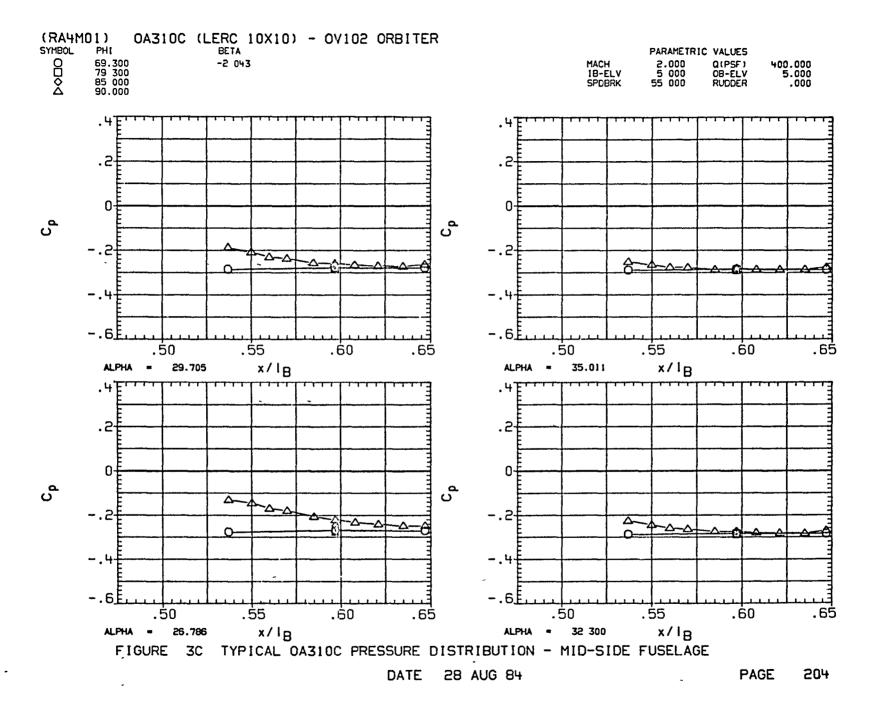
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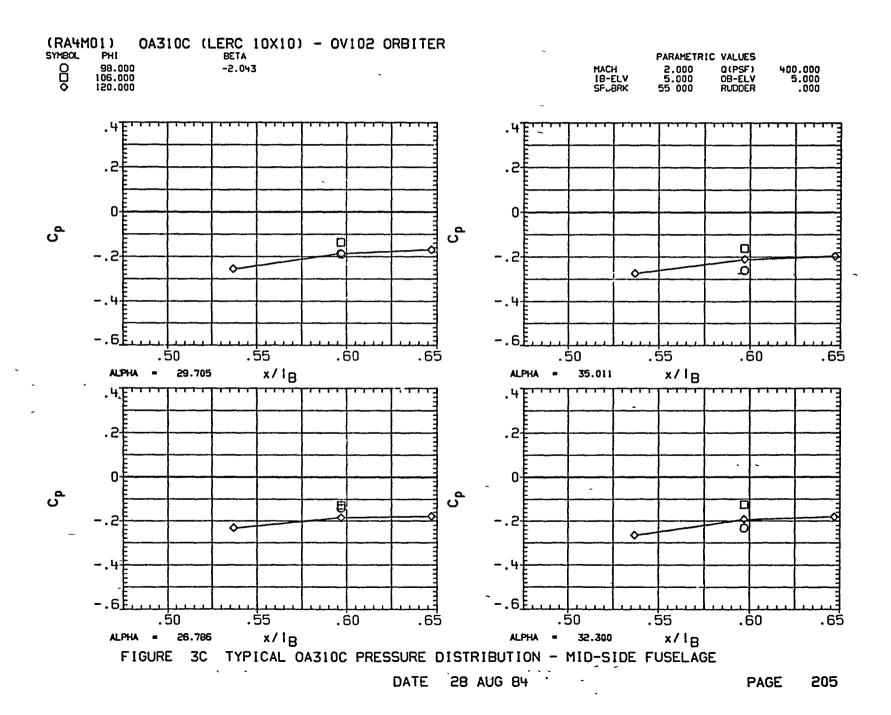


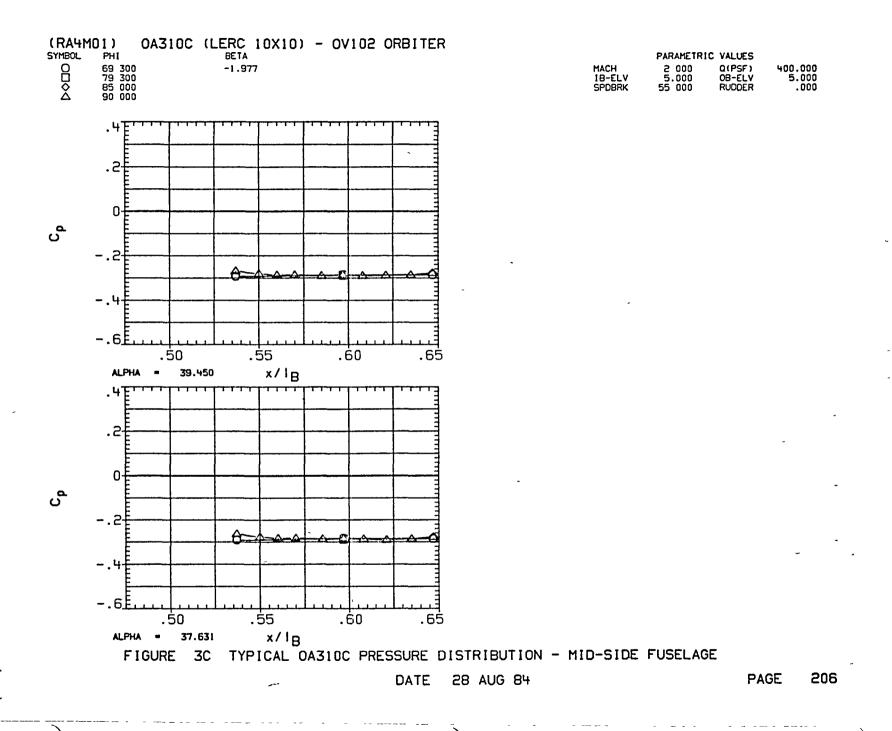


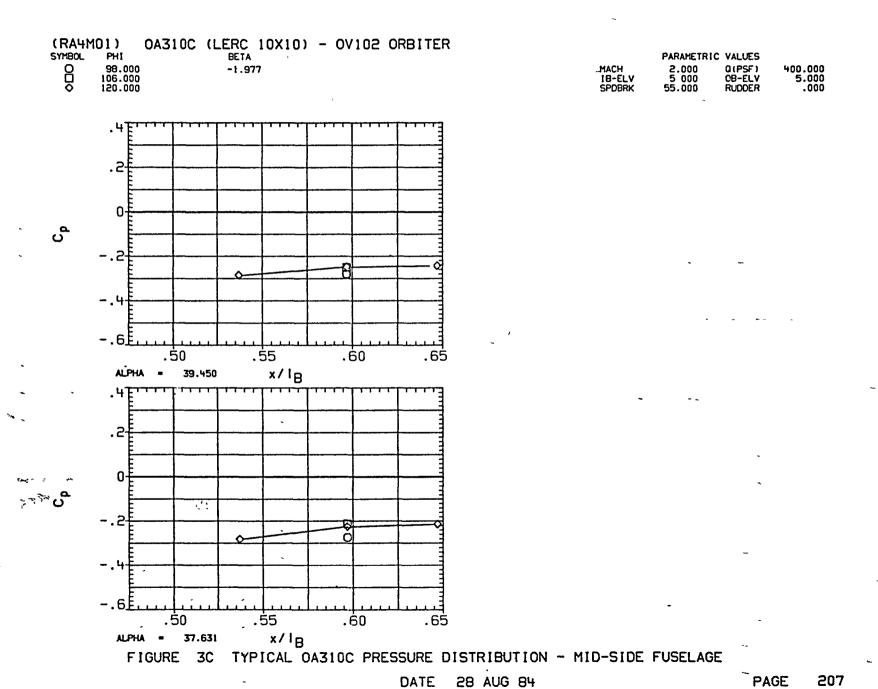


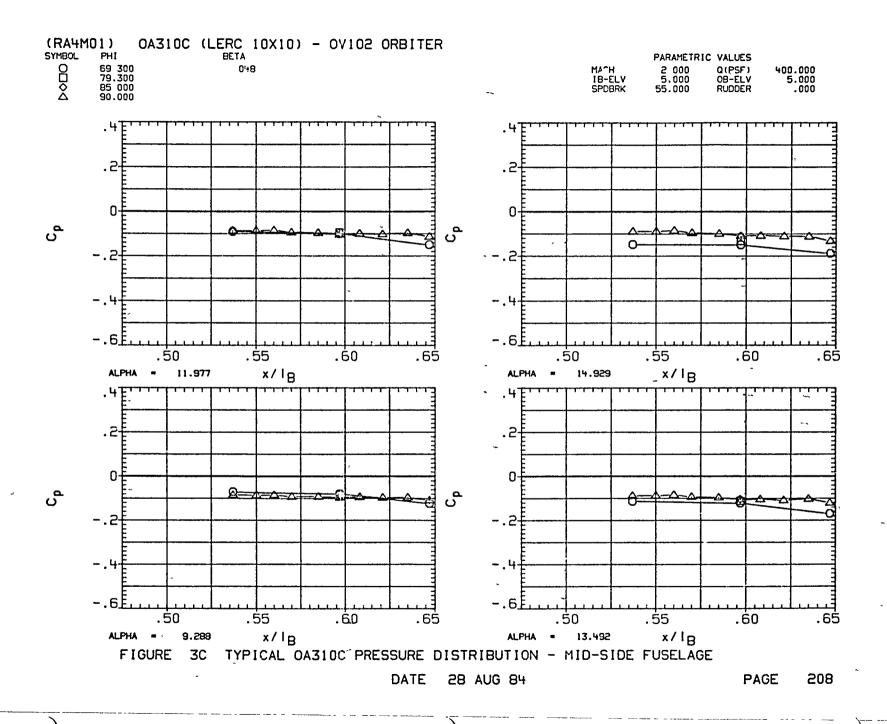


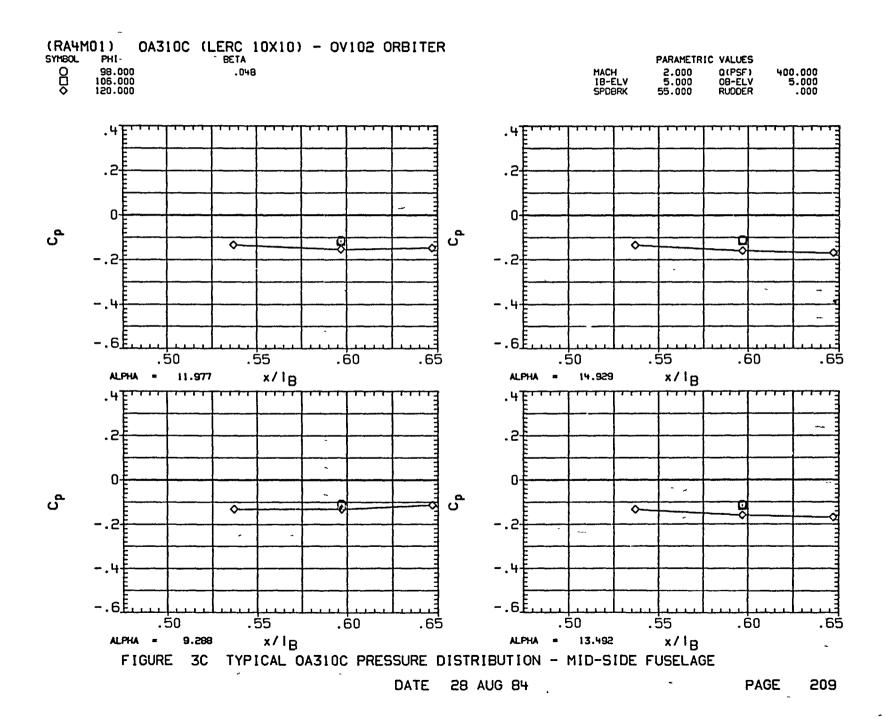


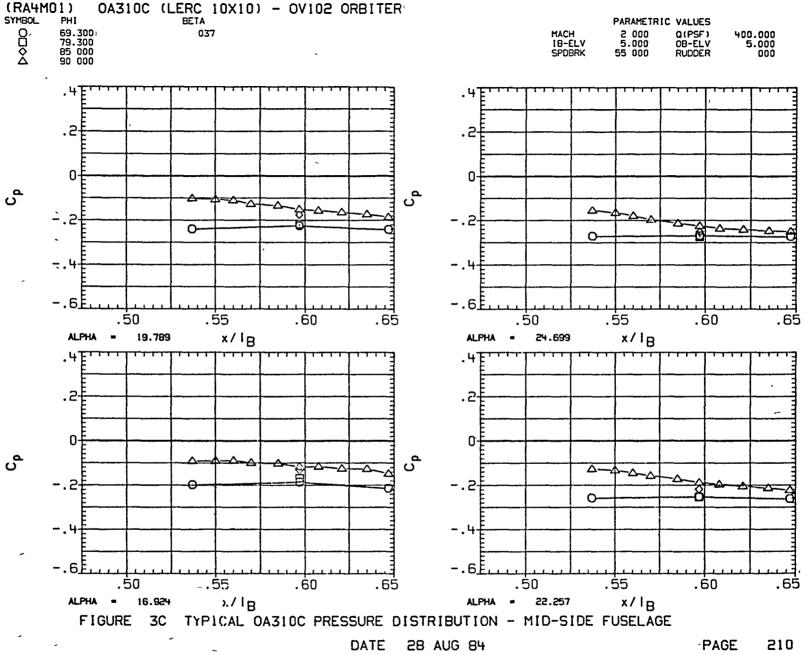


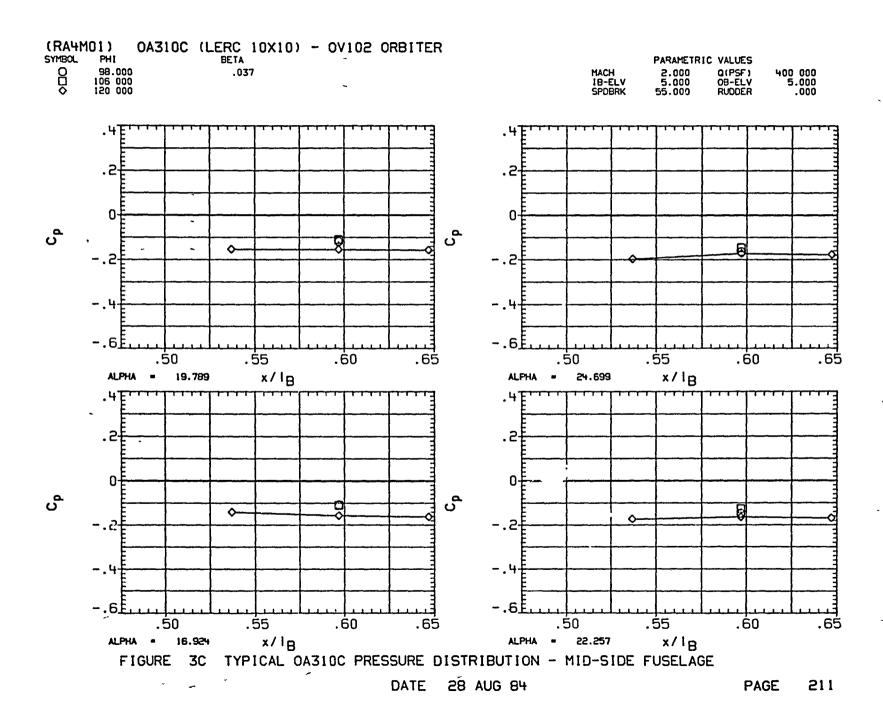


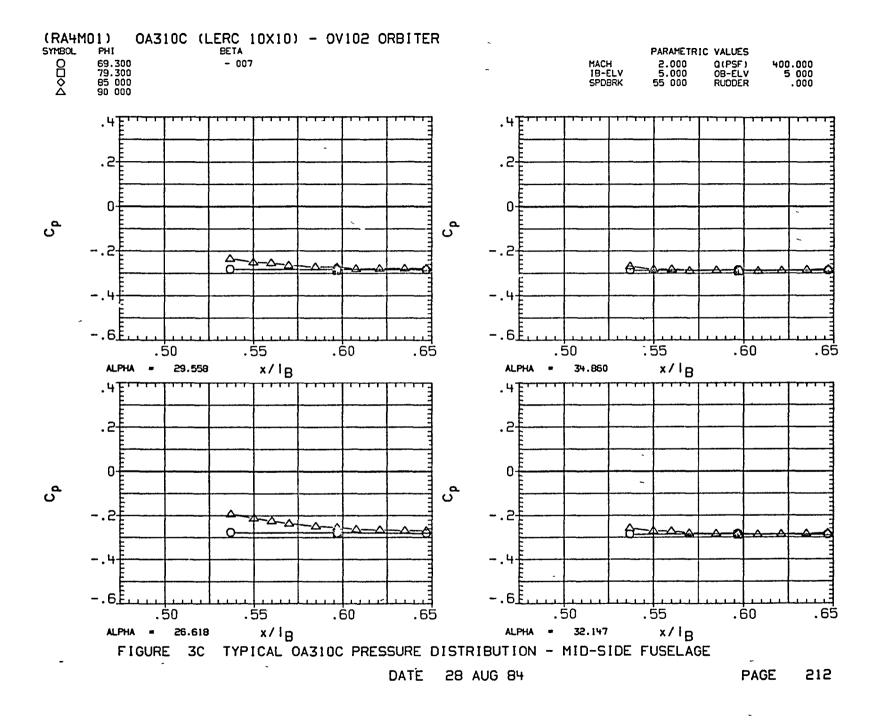


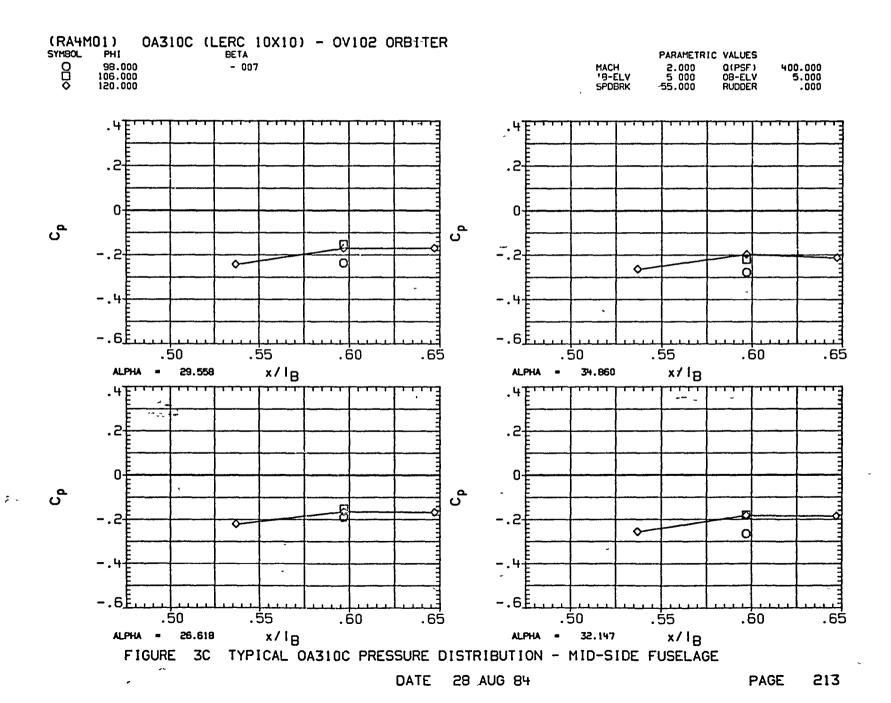


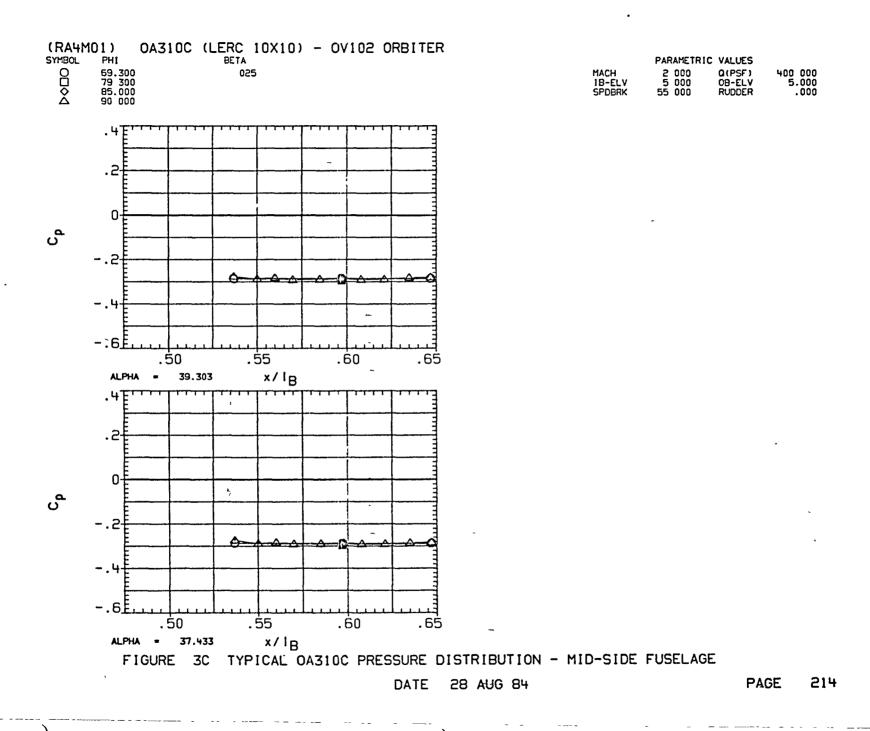


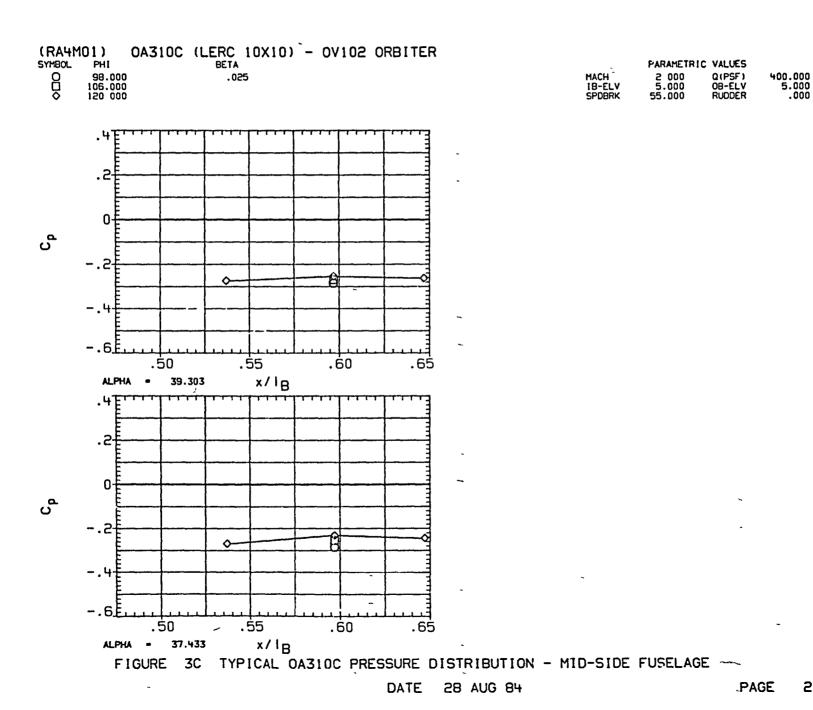


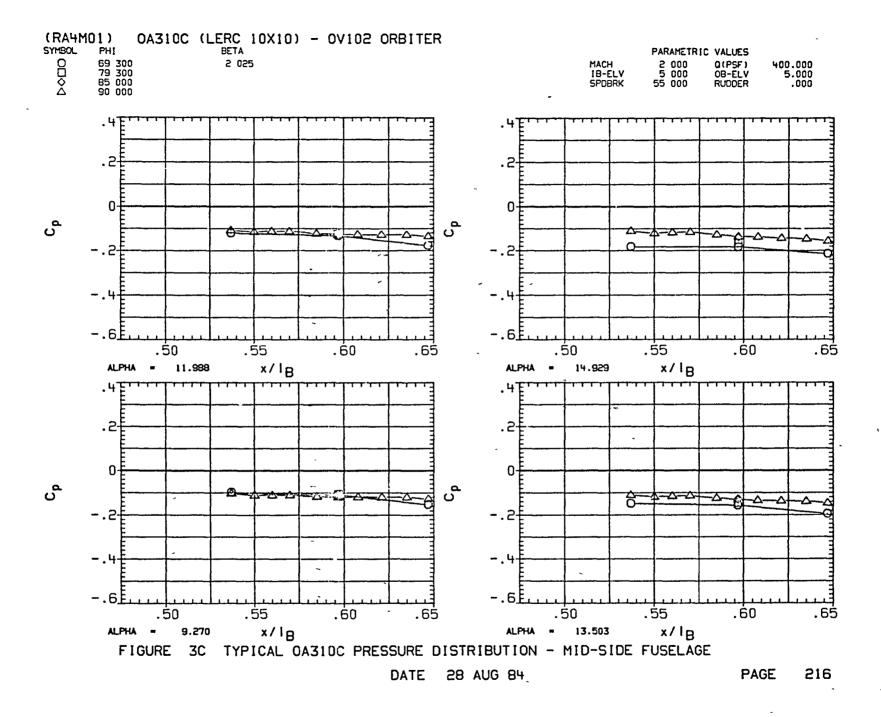


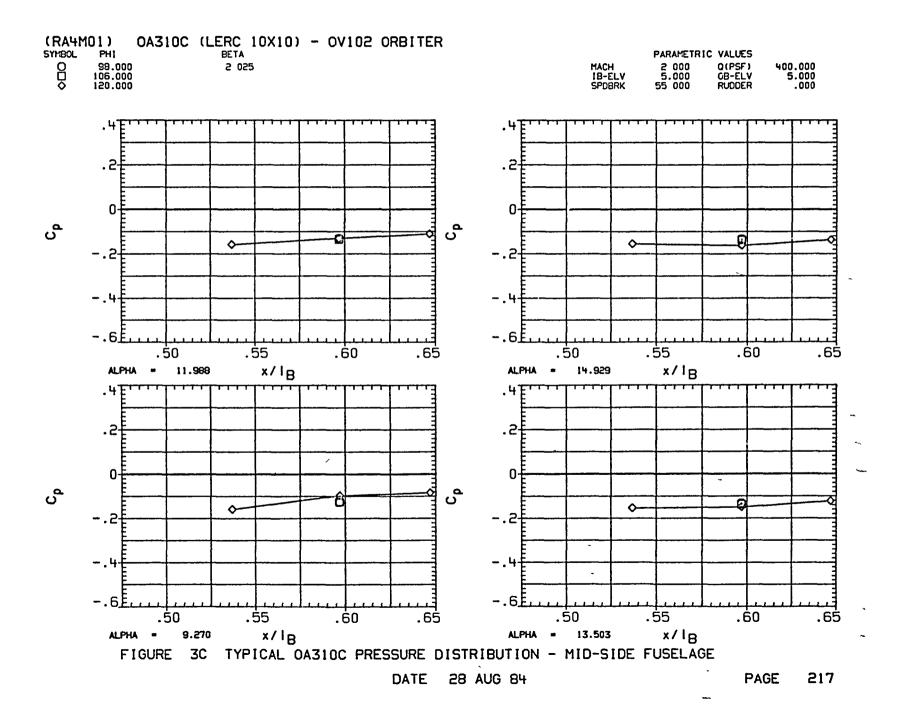


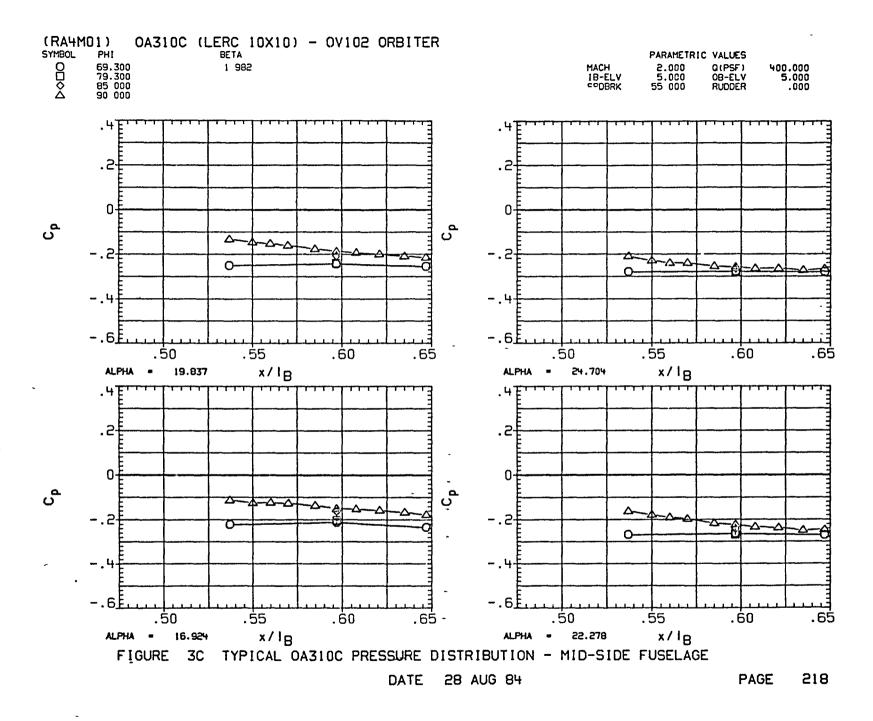


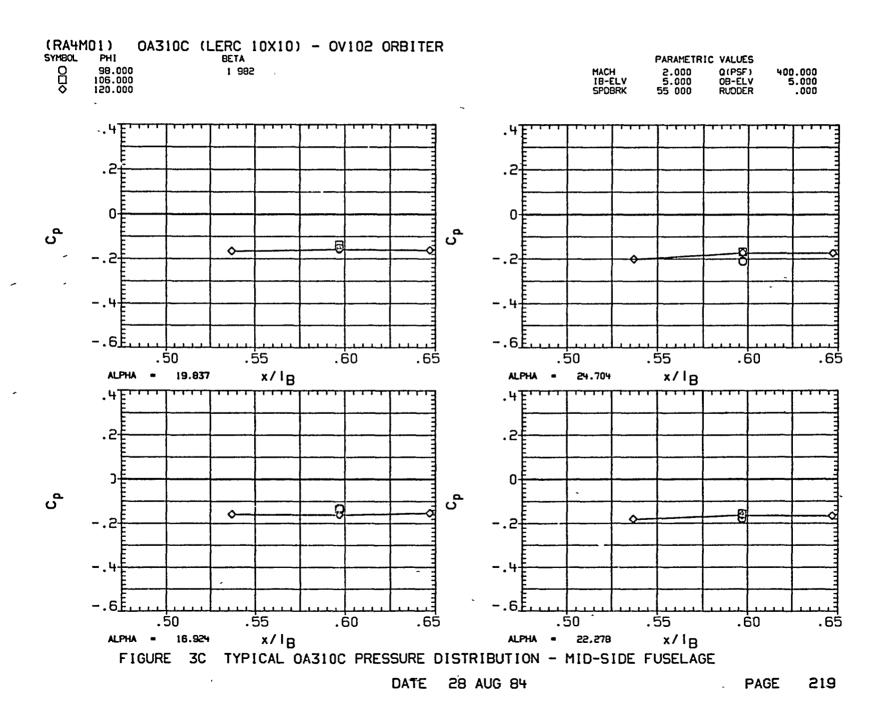


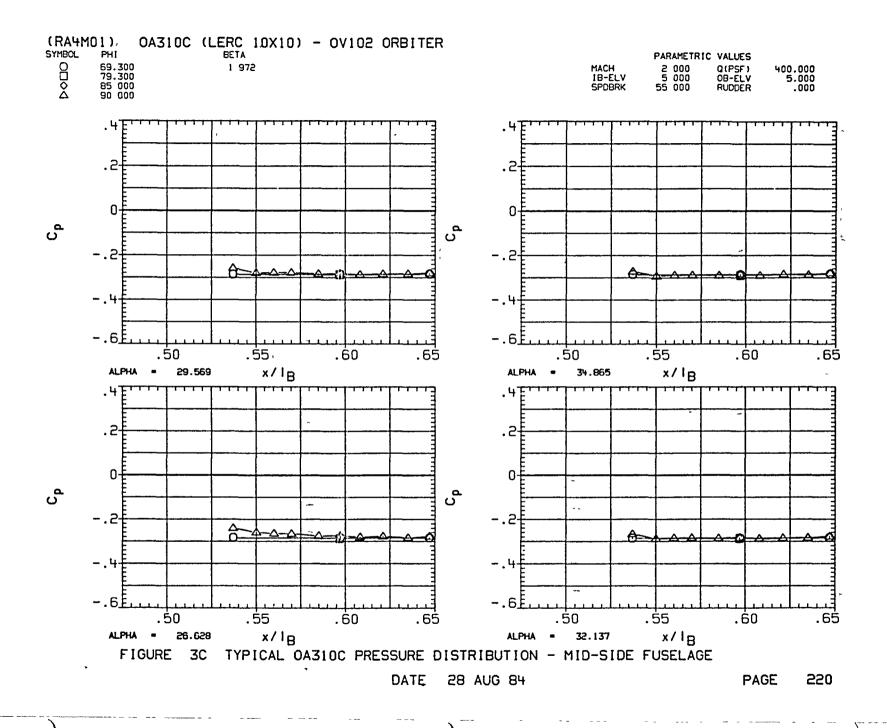


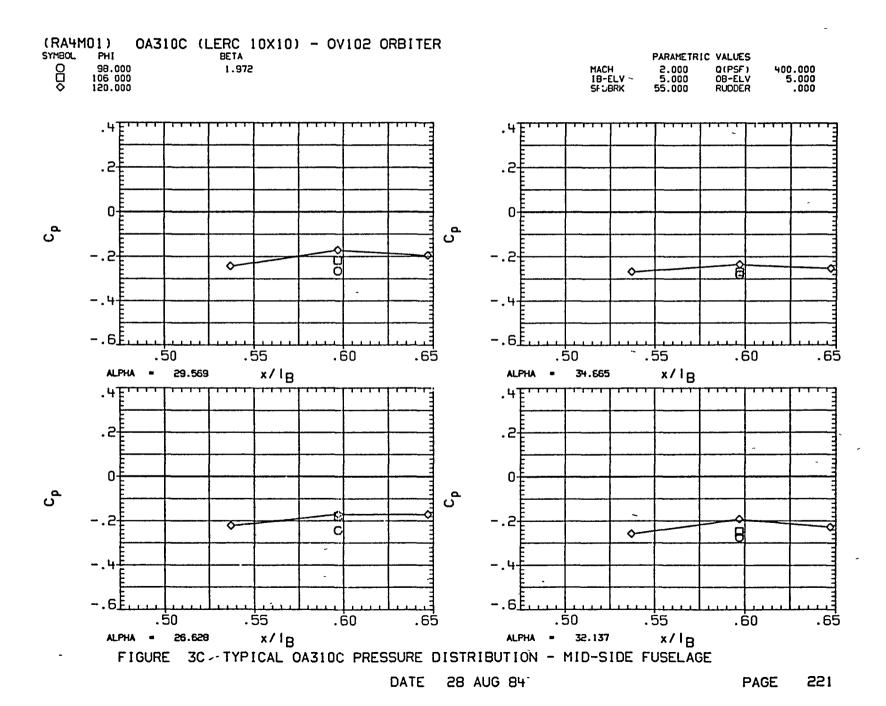


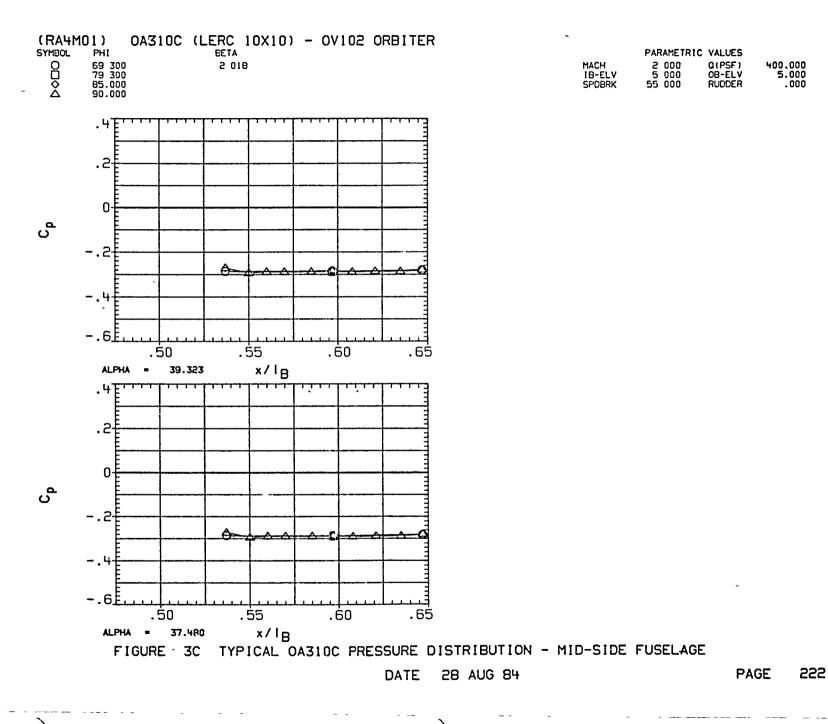


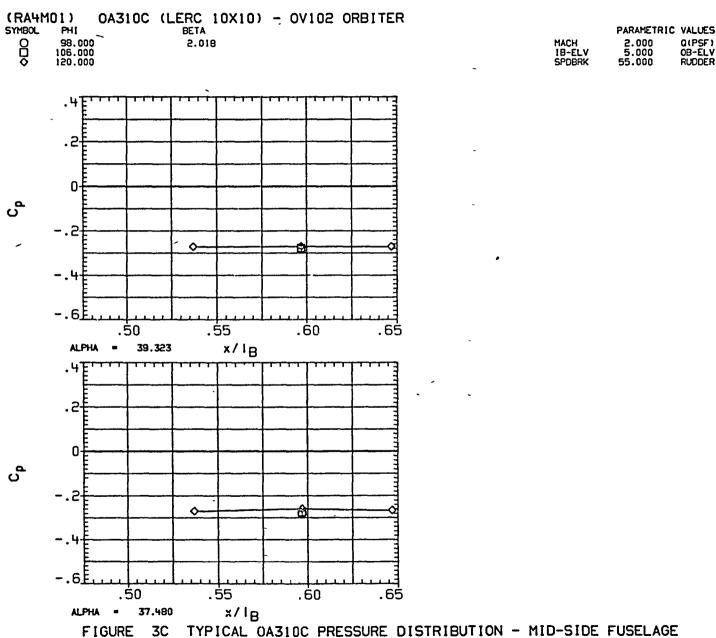








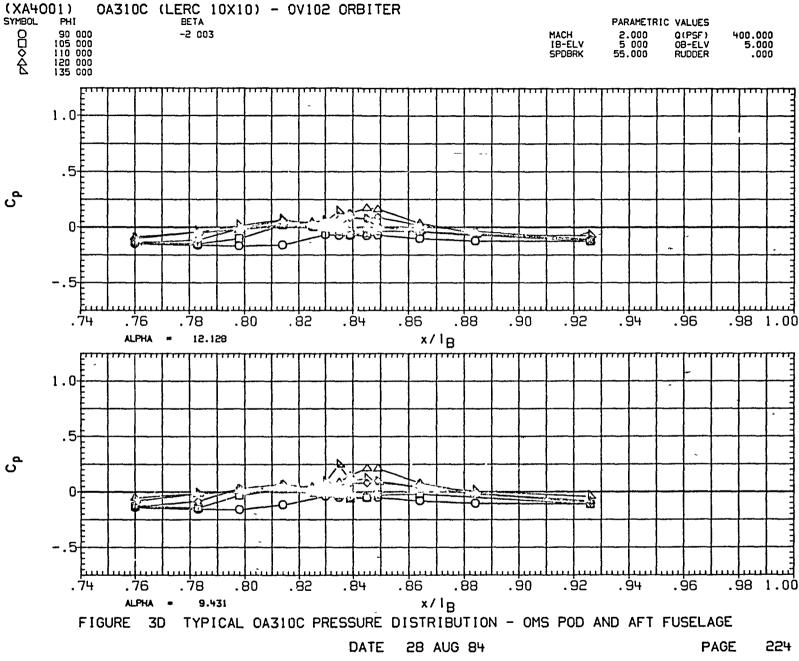


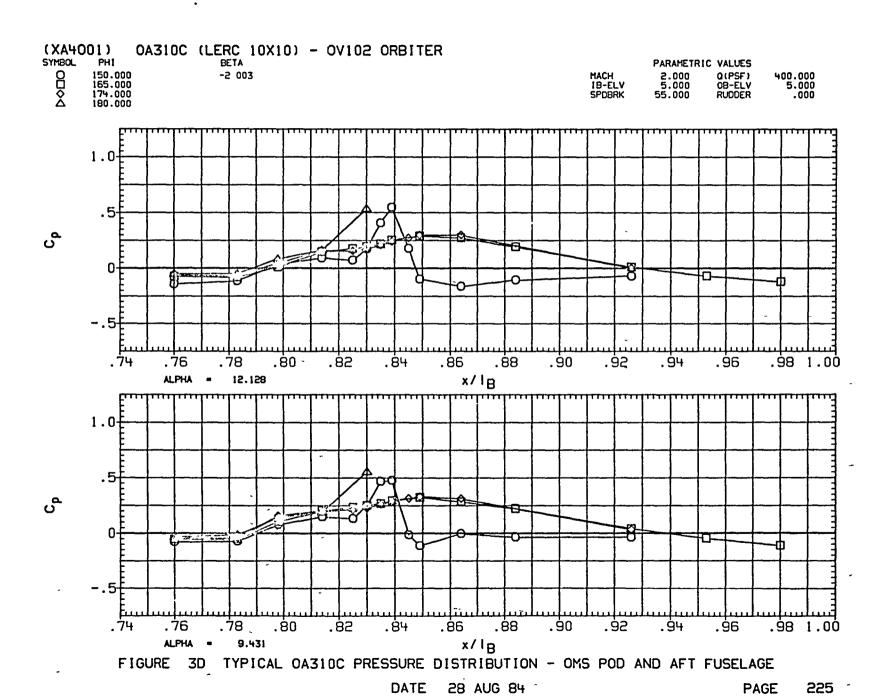


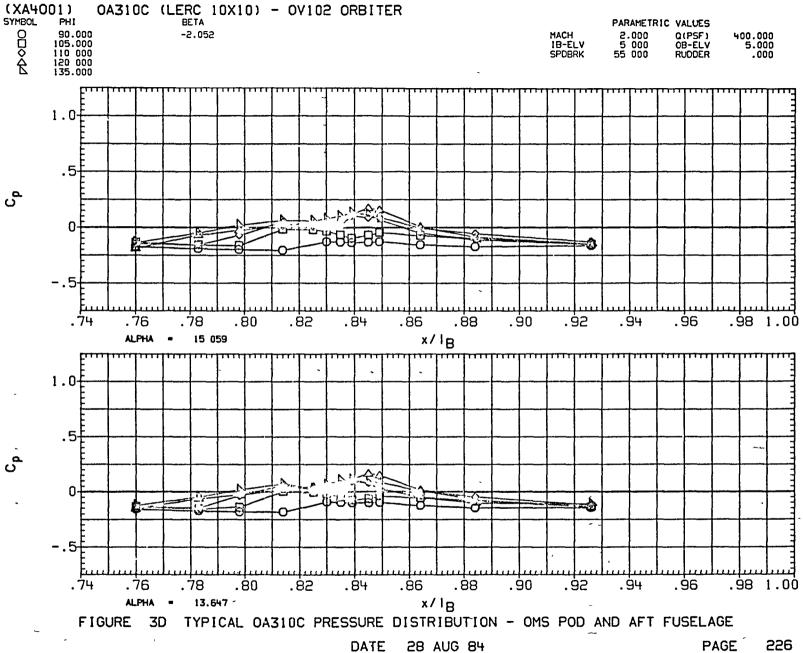
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Q(PSF) OB-ELV RUDDER

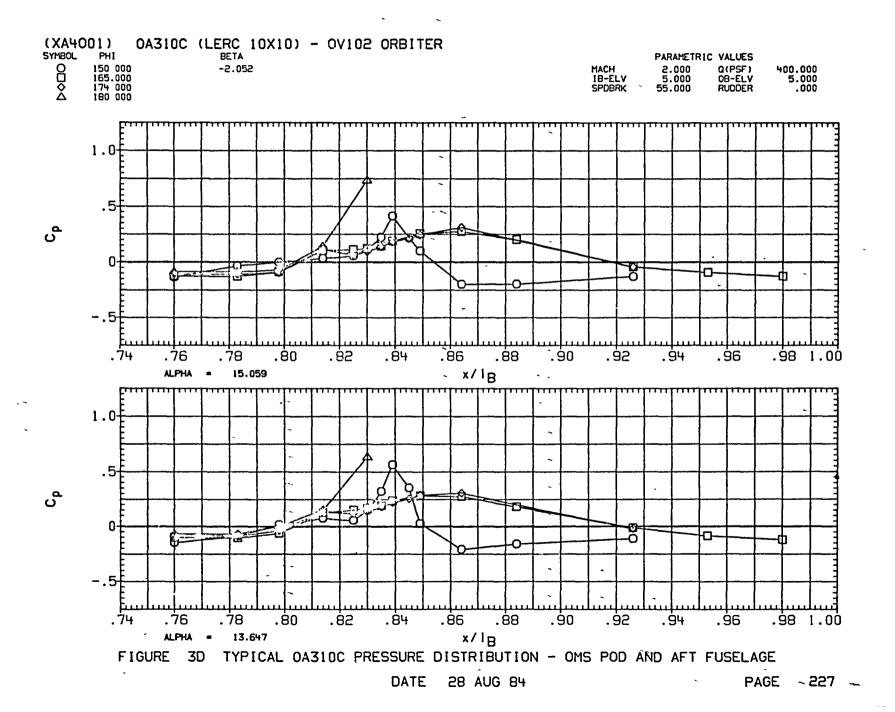
400.000 5.000 .000

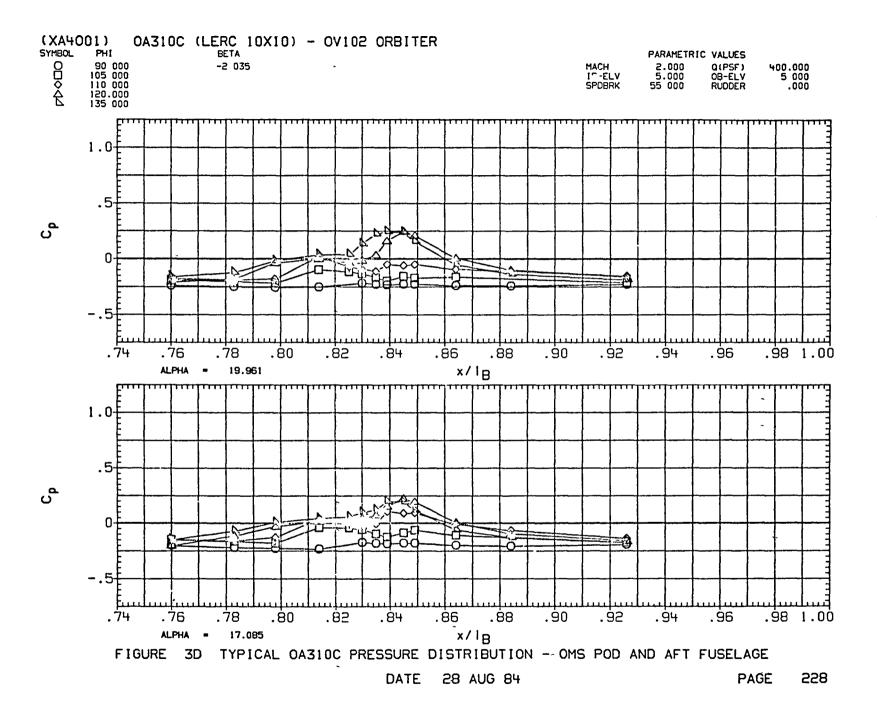


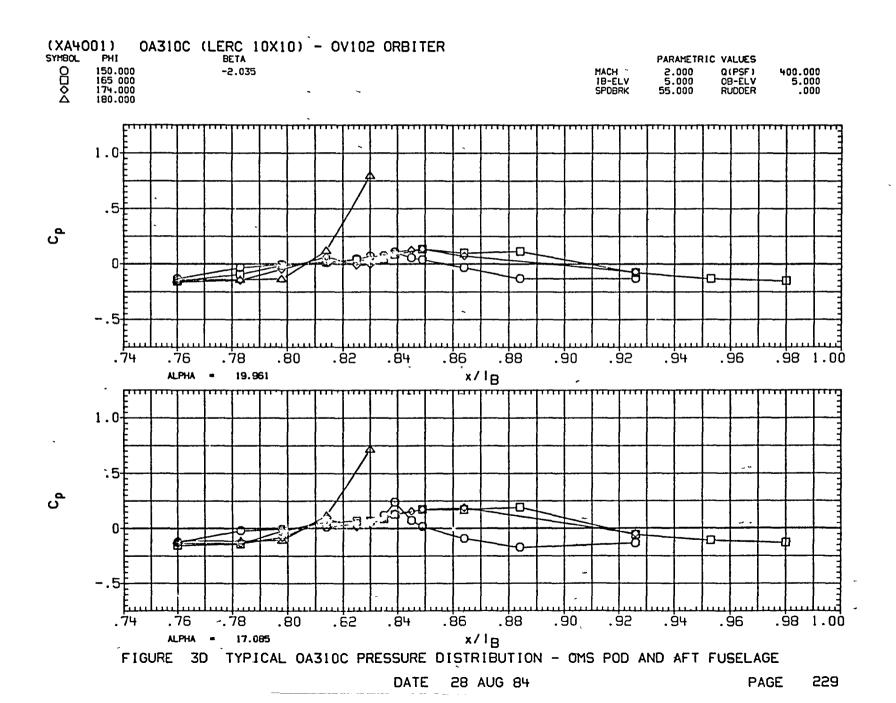


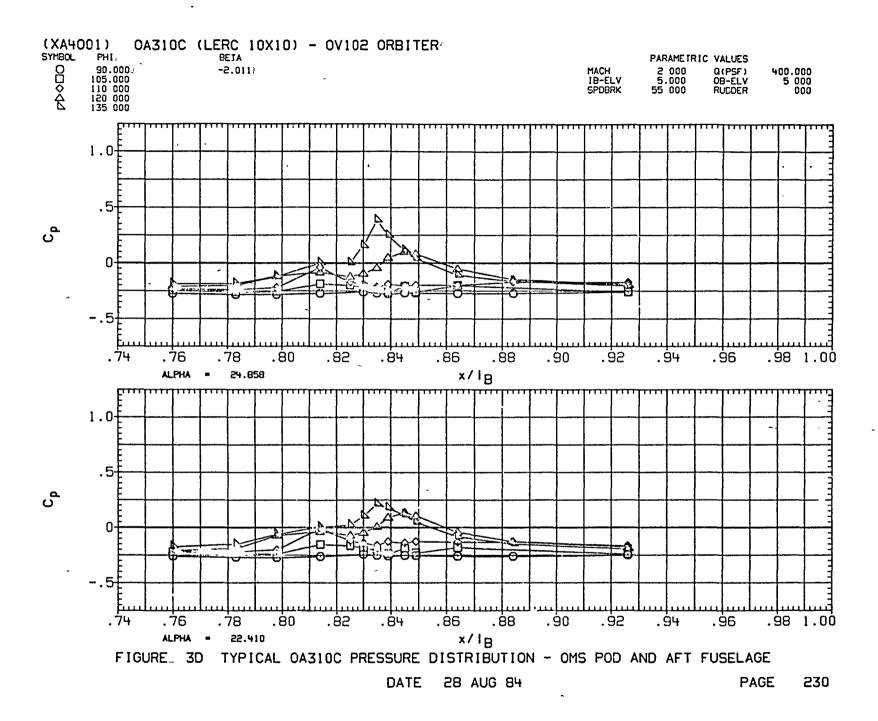


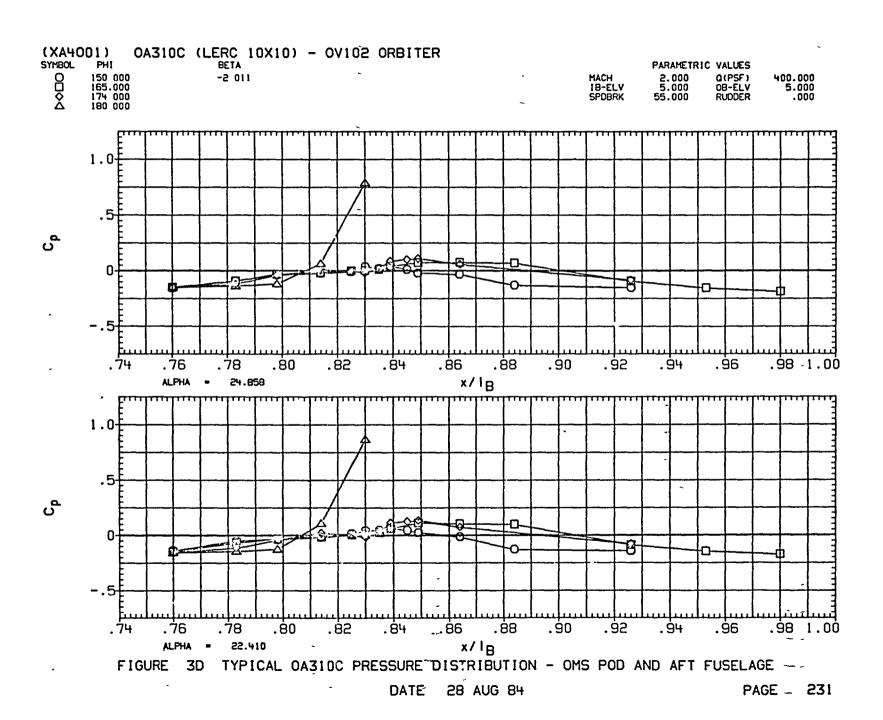
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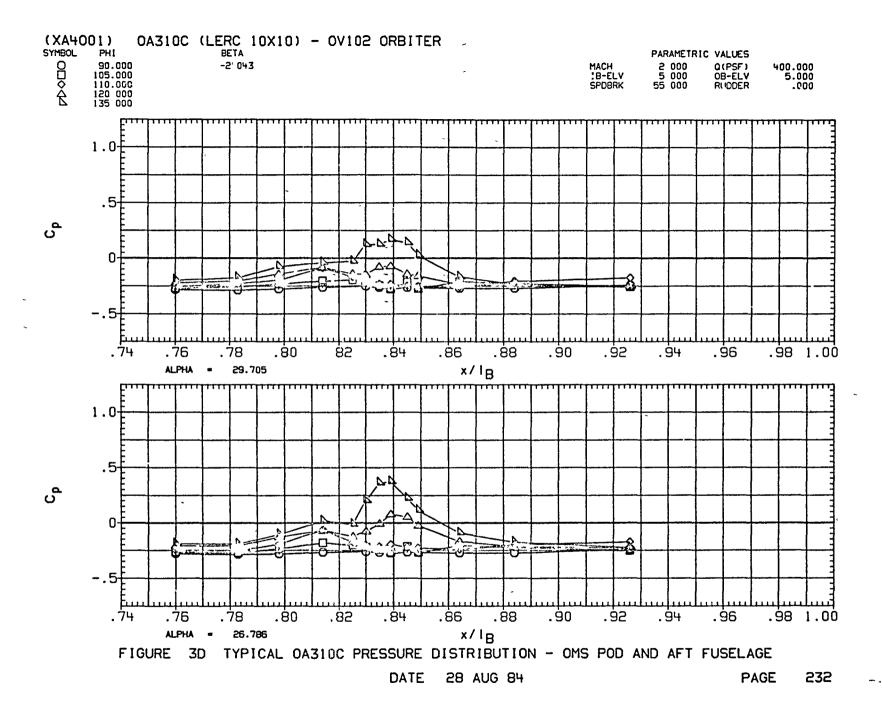


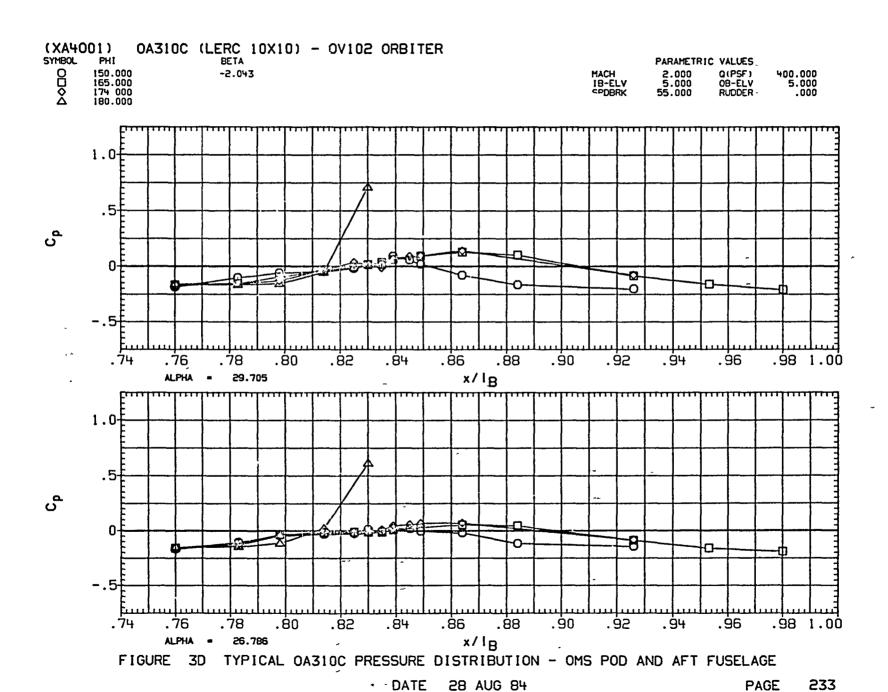


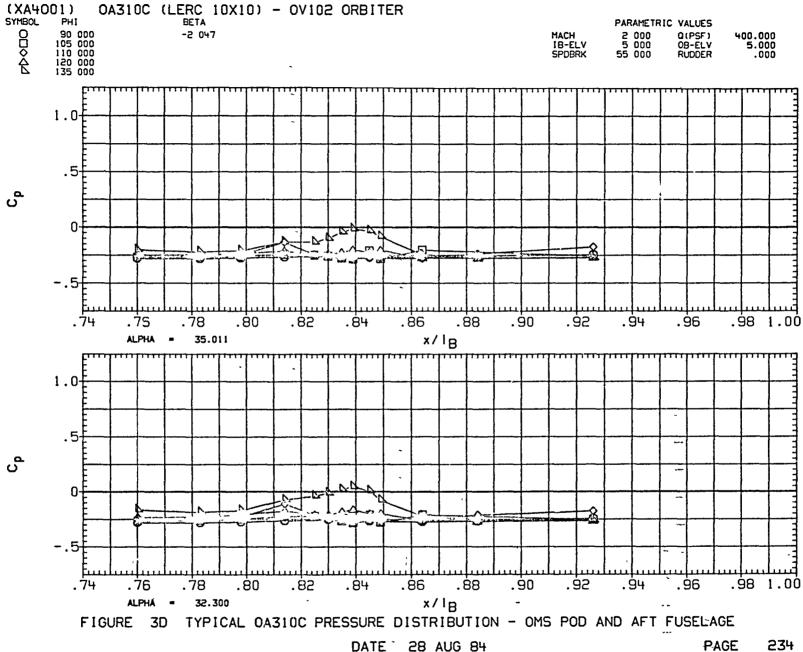


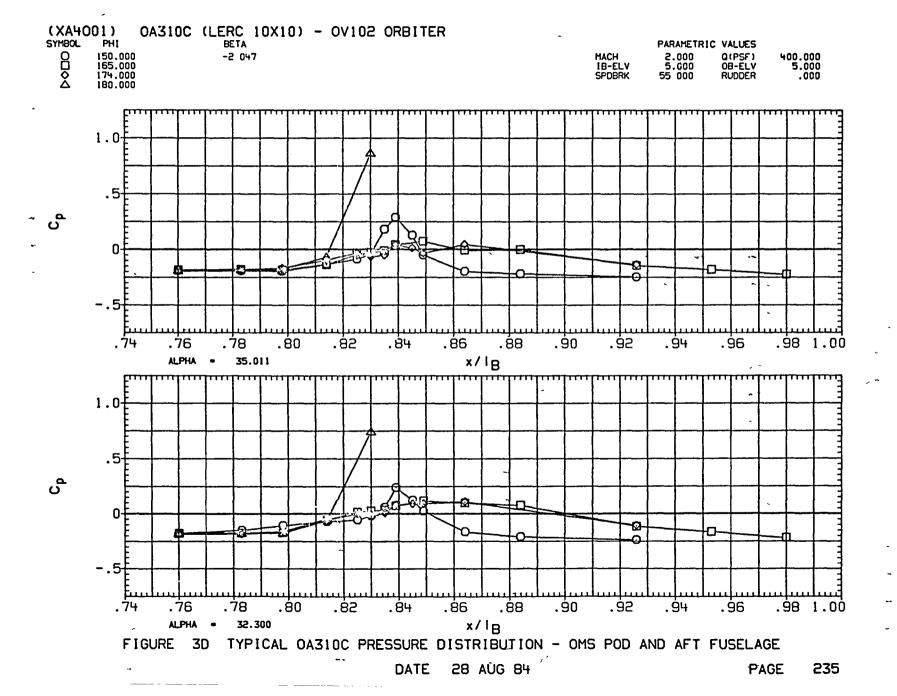


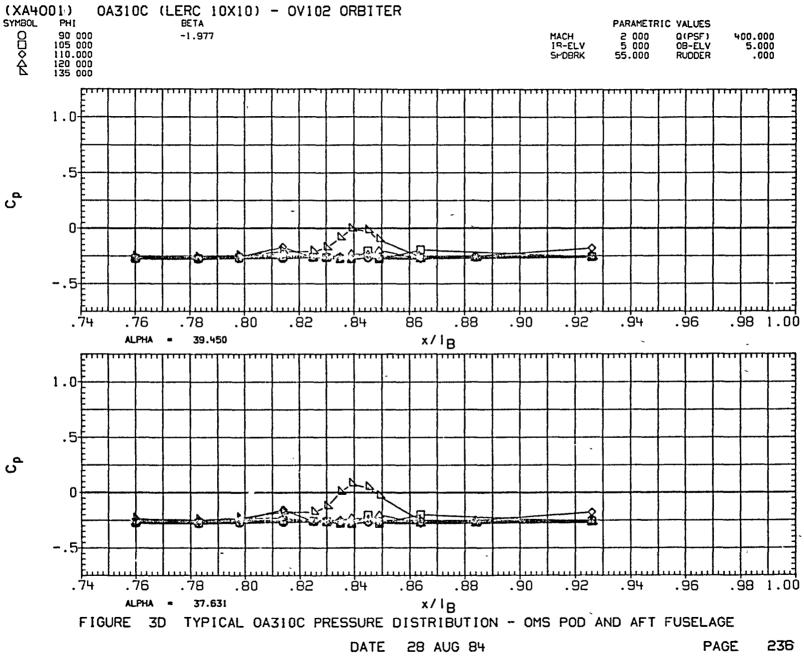


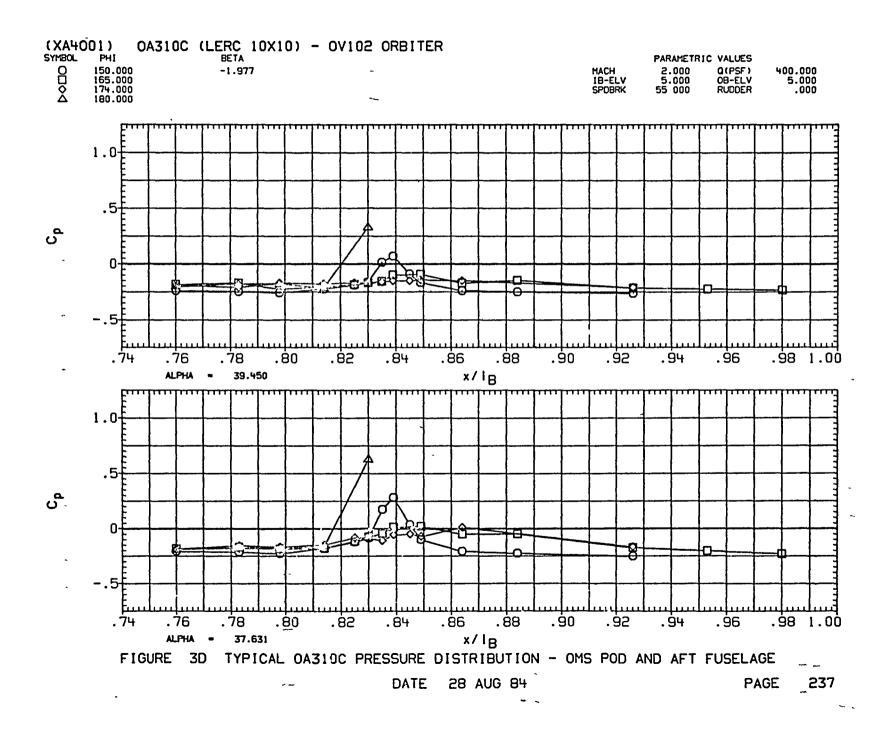


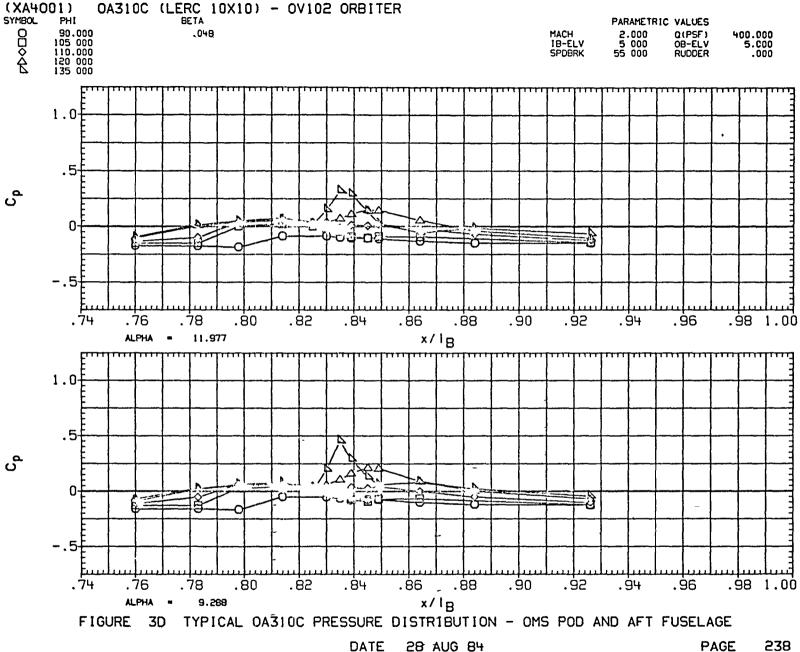


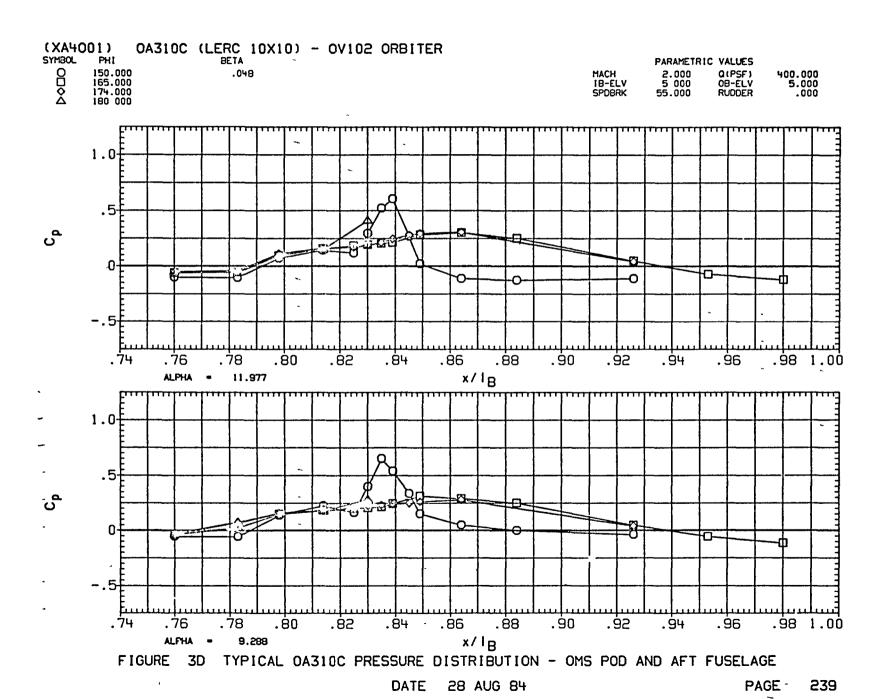


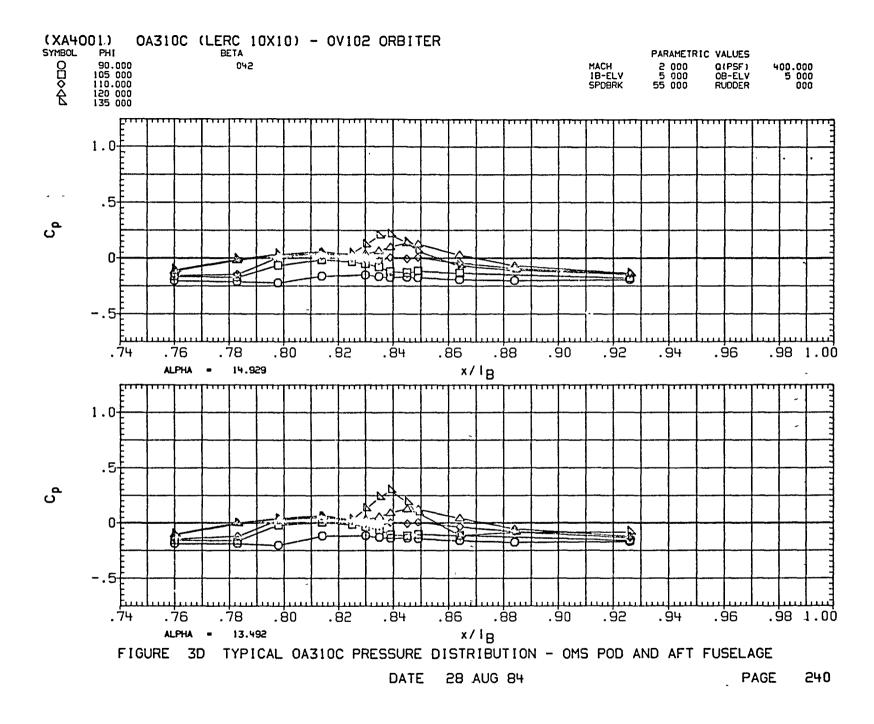




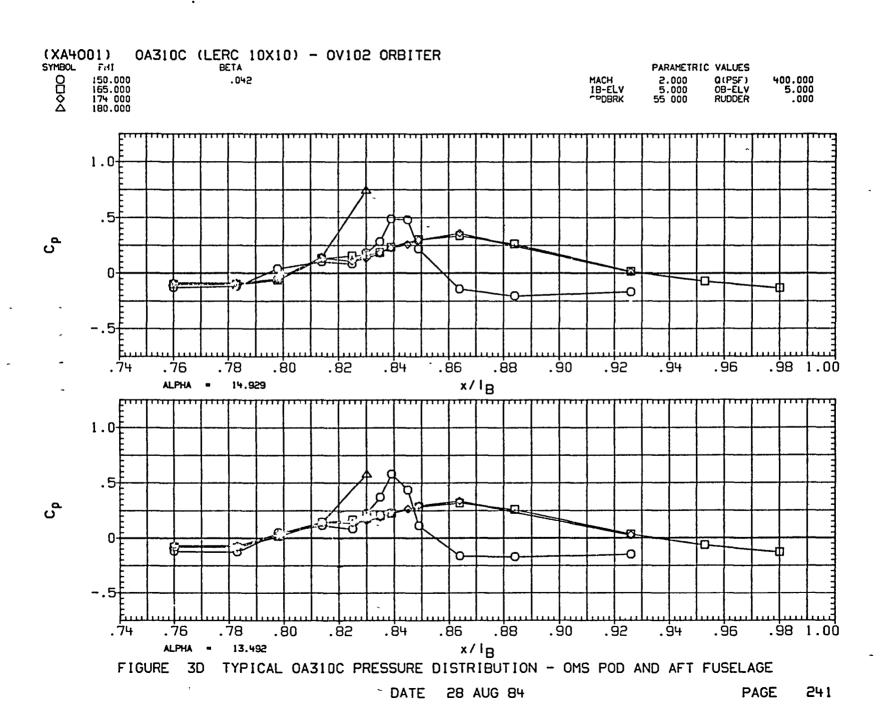


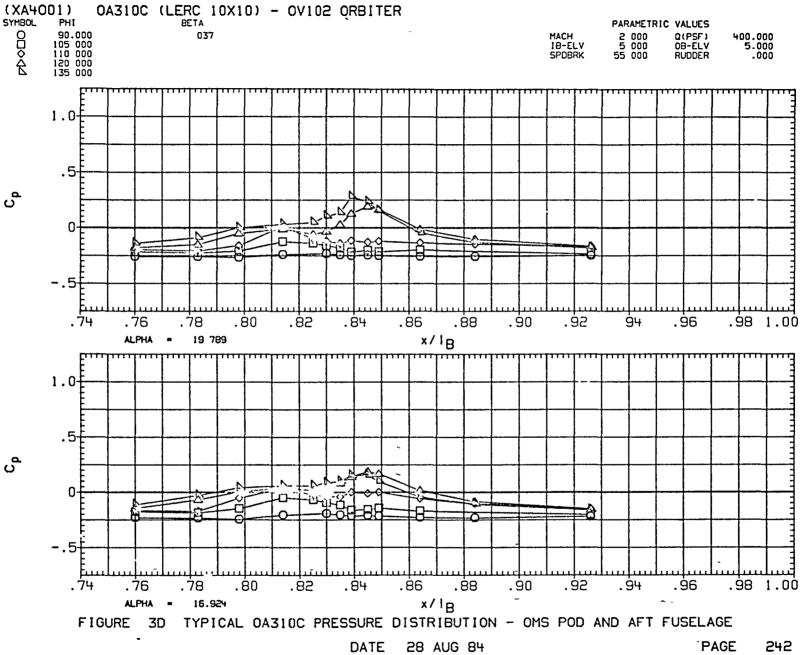


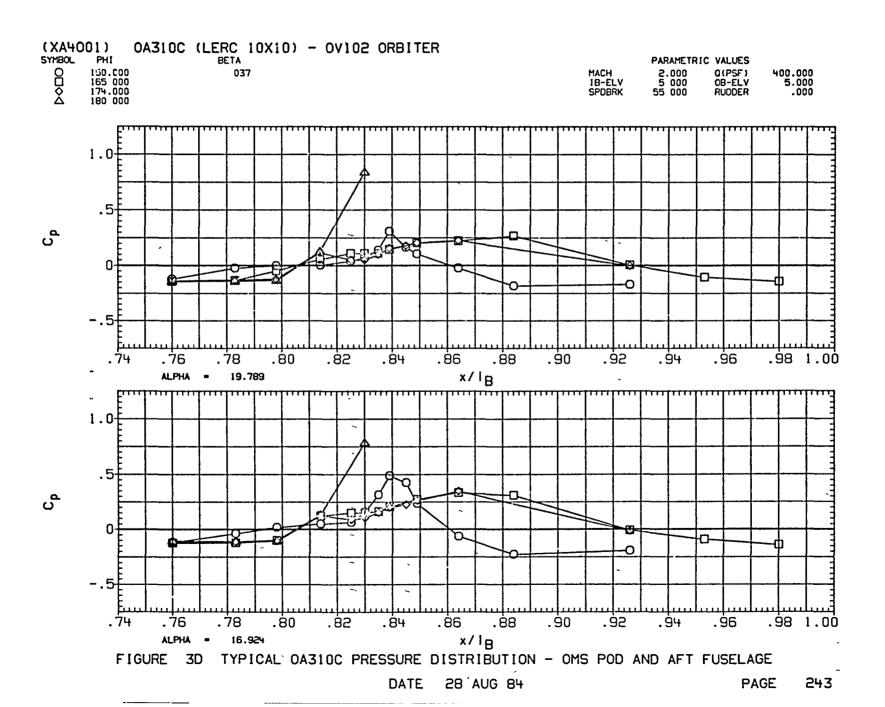


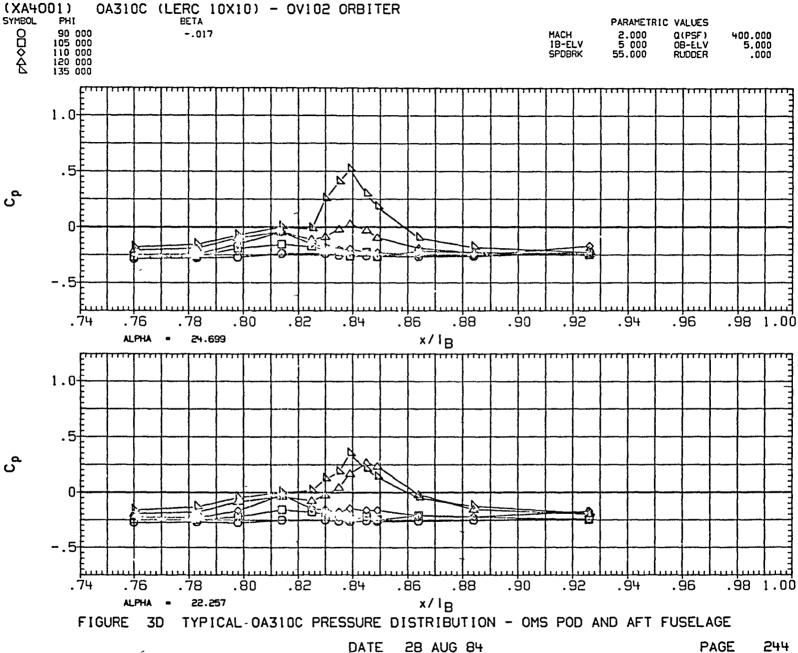


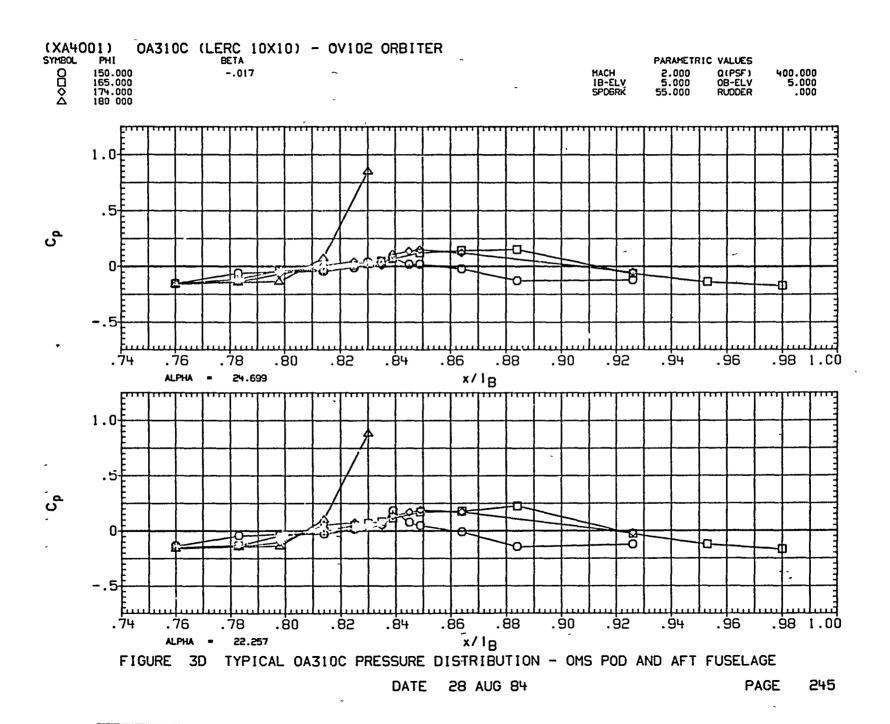
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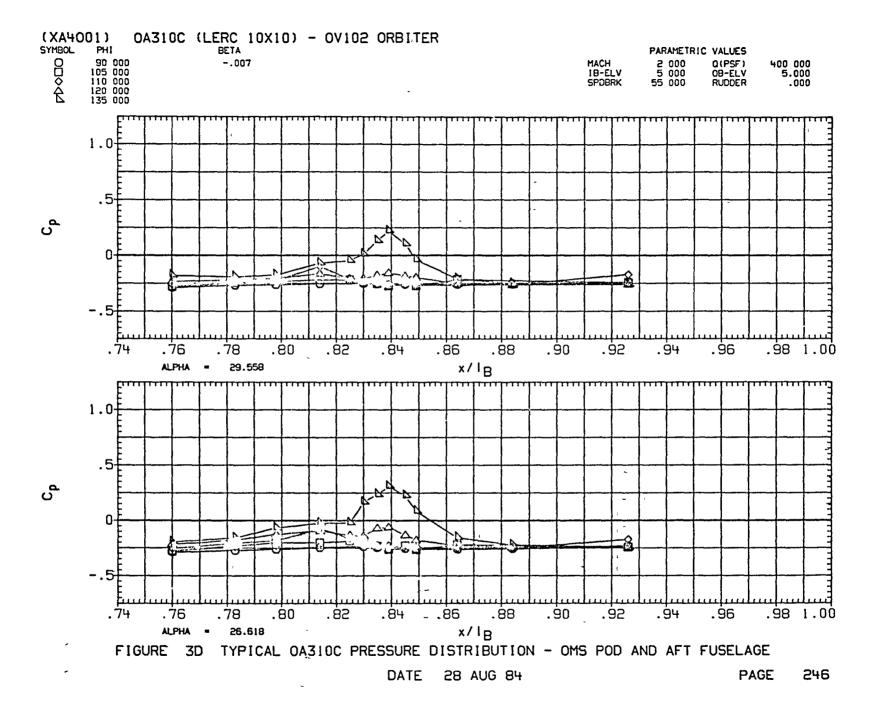




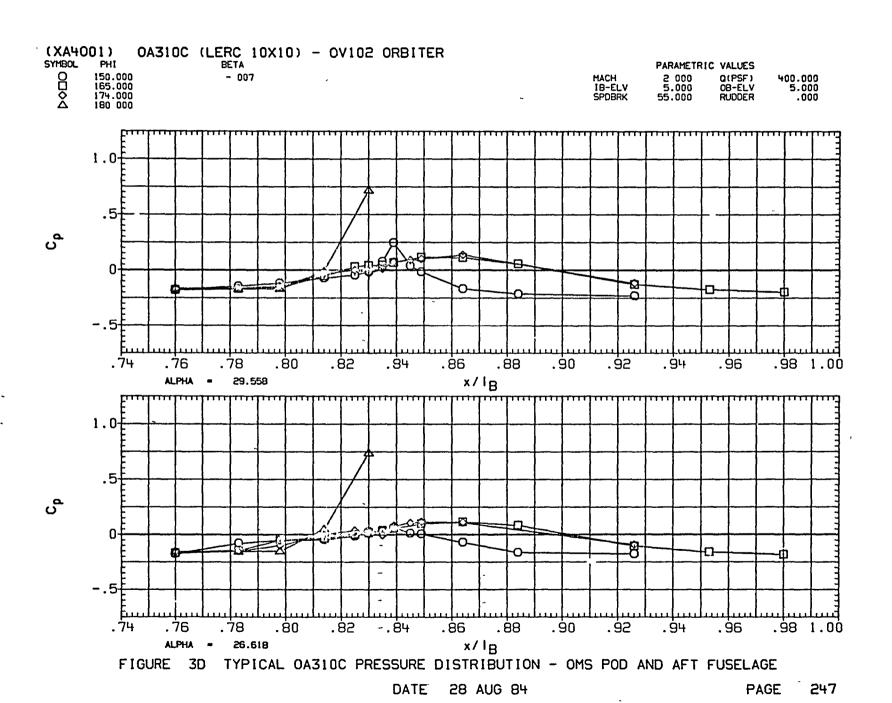


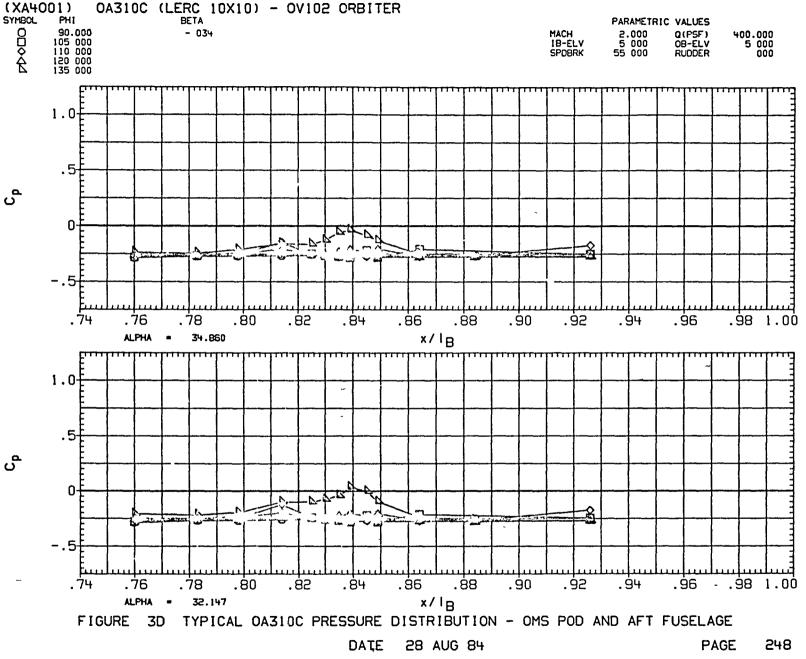


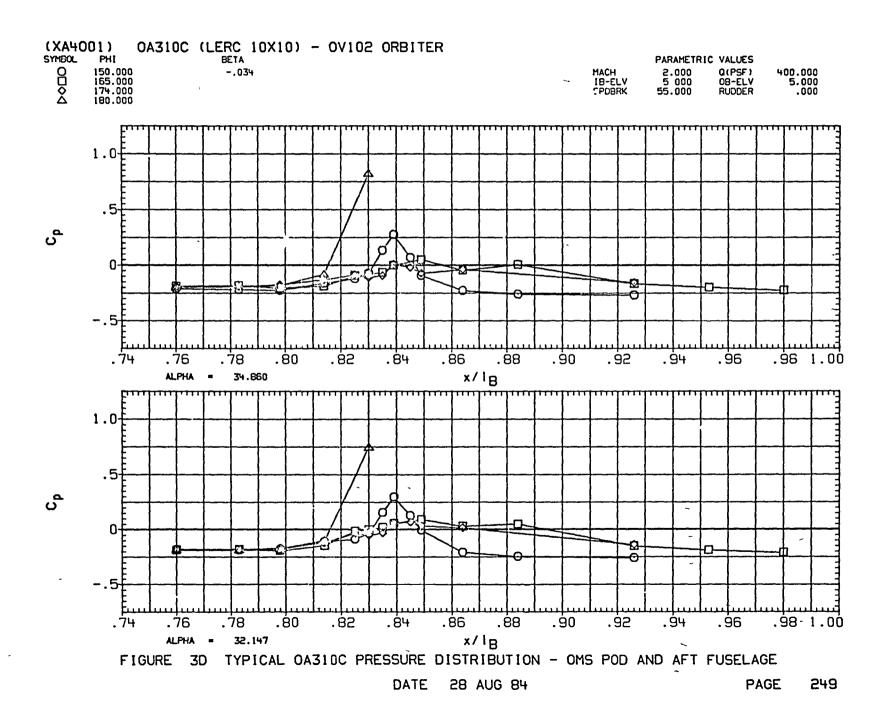


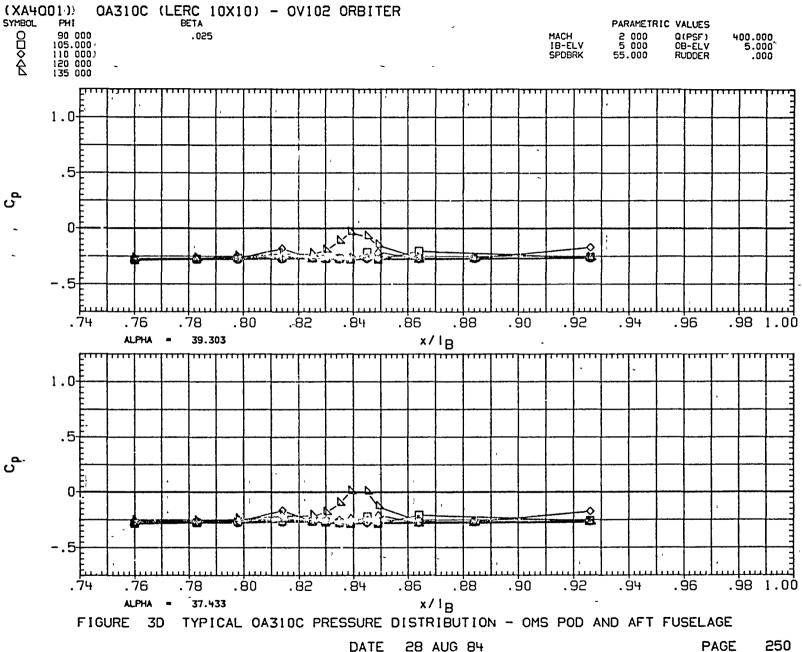


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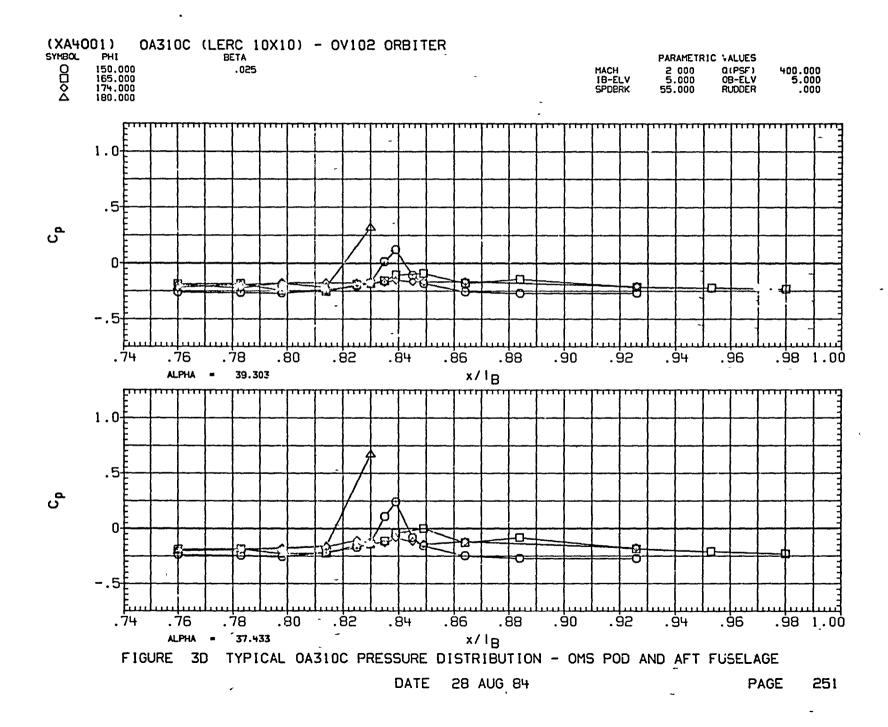


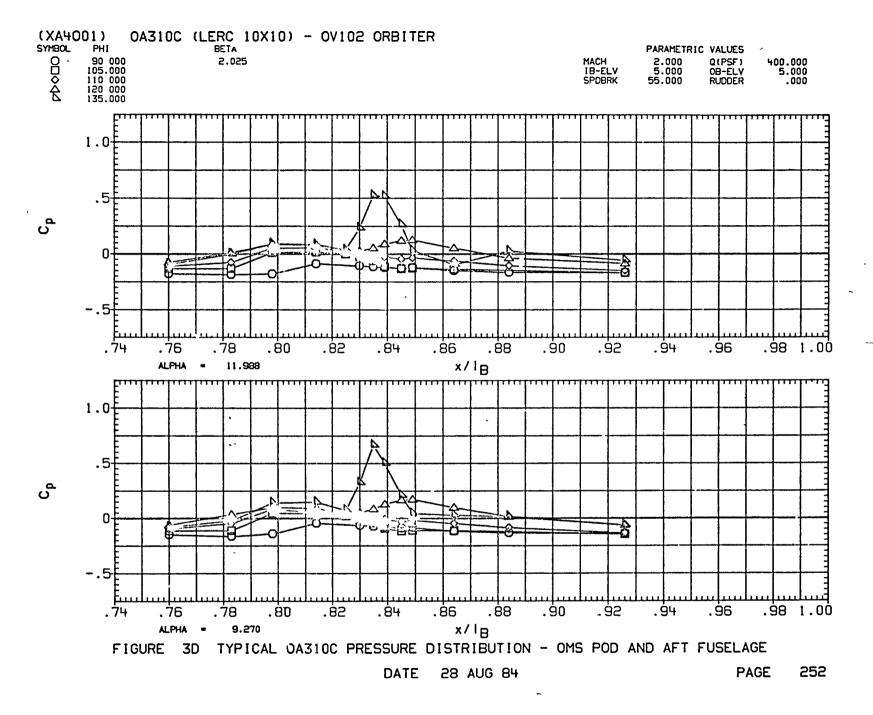






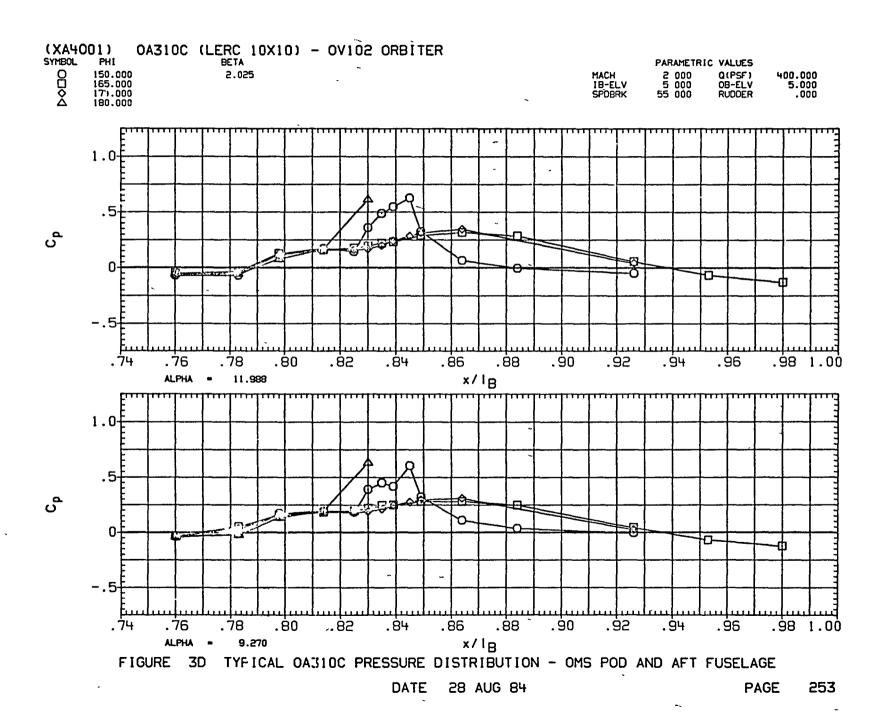
DATE 28 AUG 84 PAGE

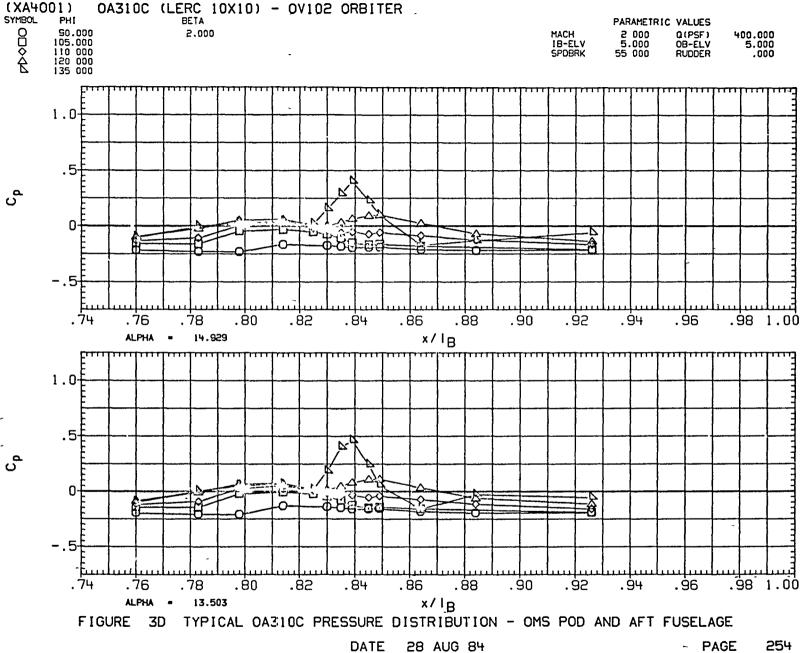


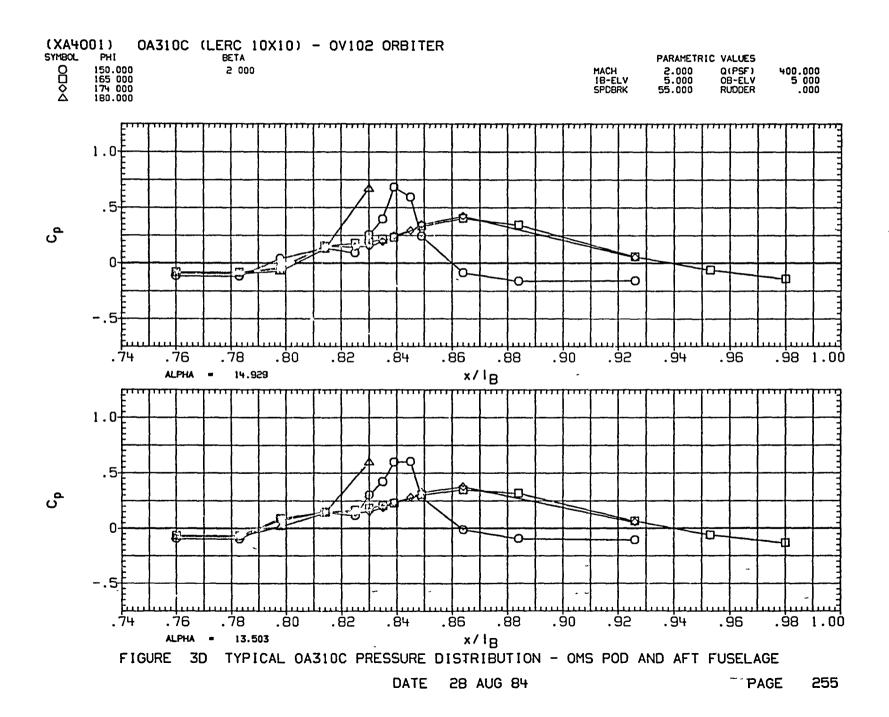


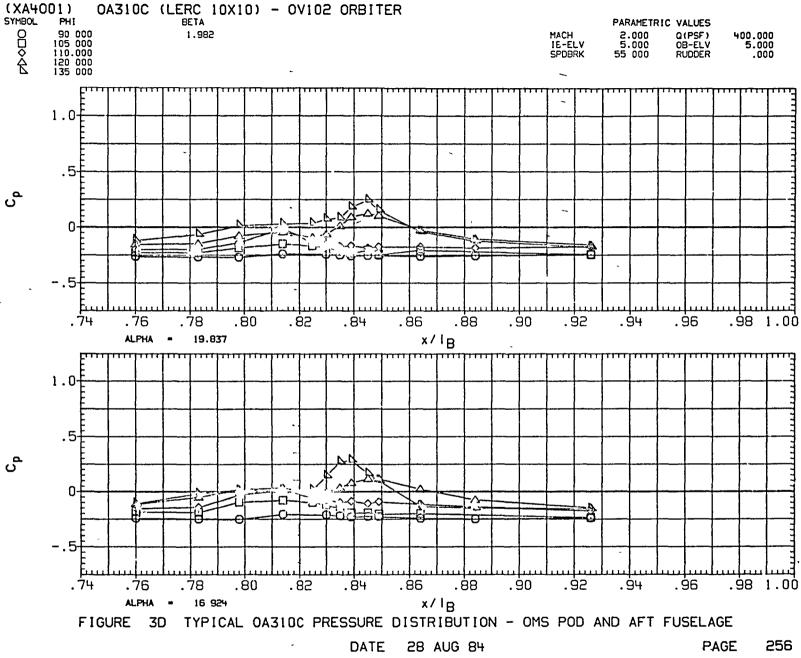
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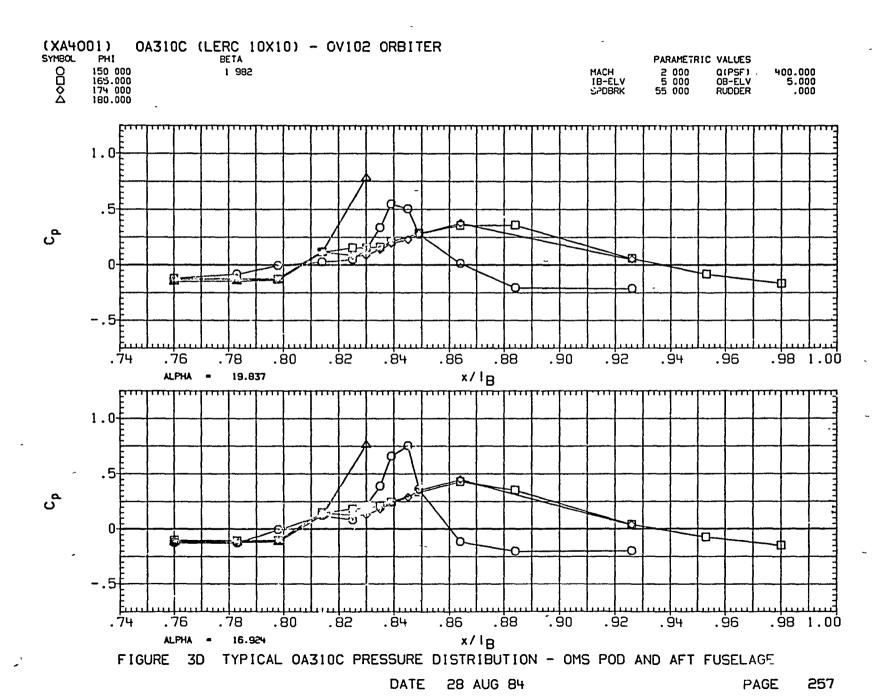
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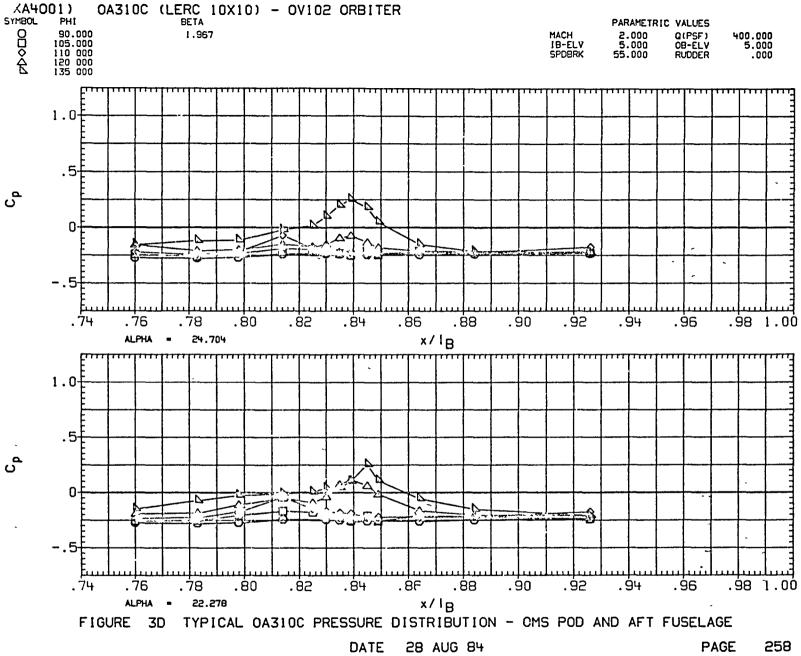








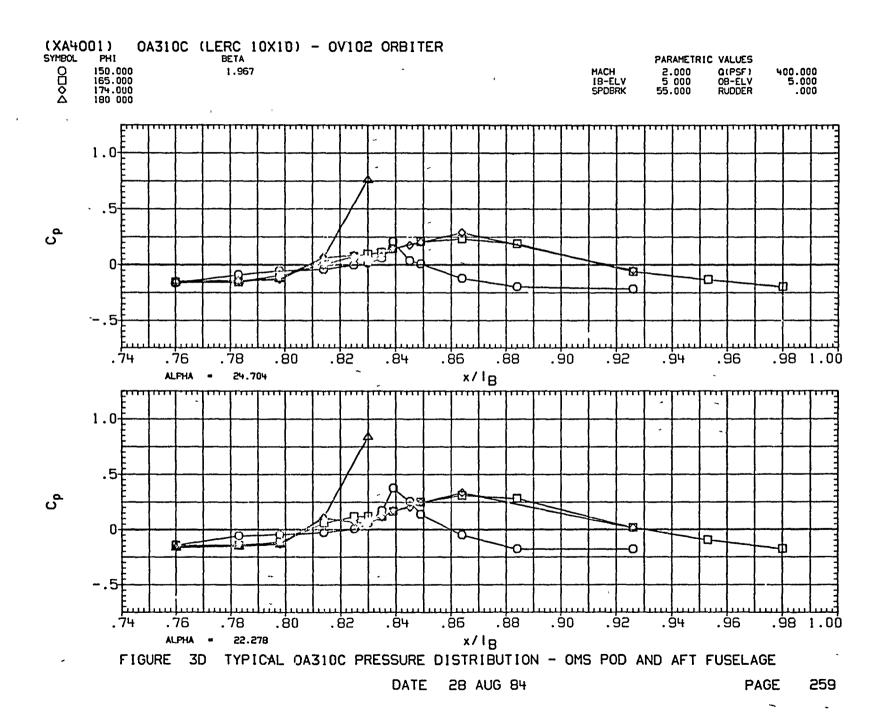


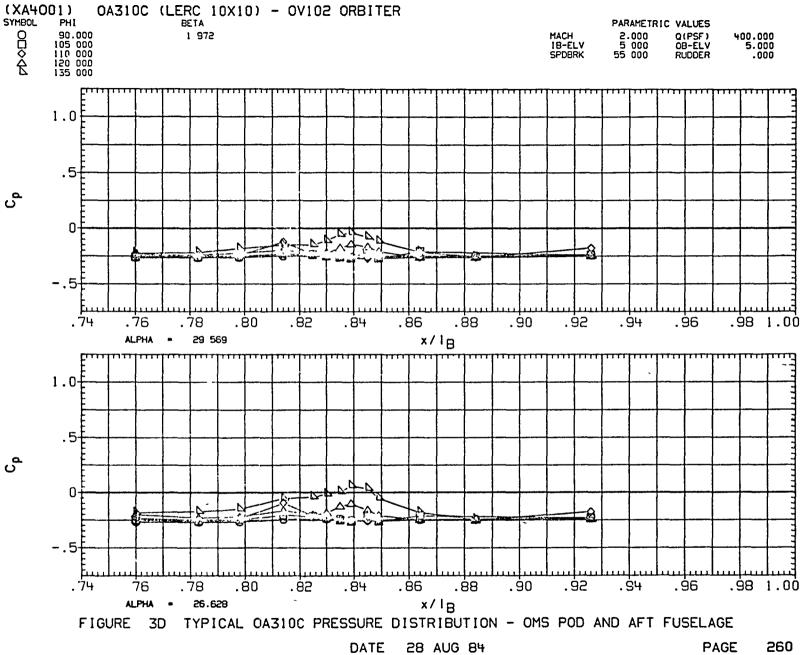


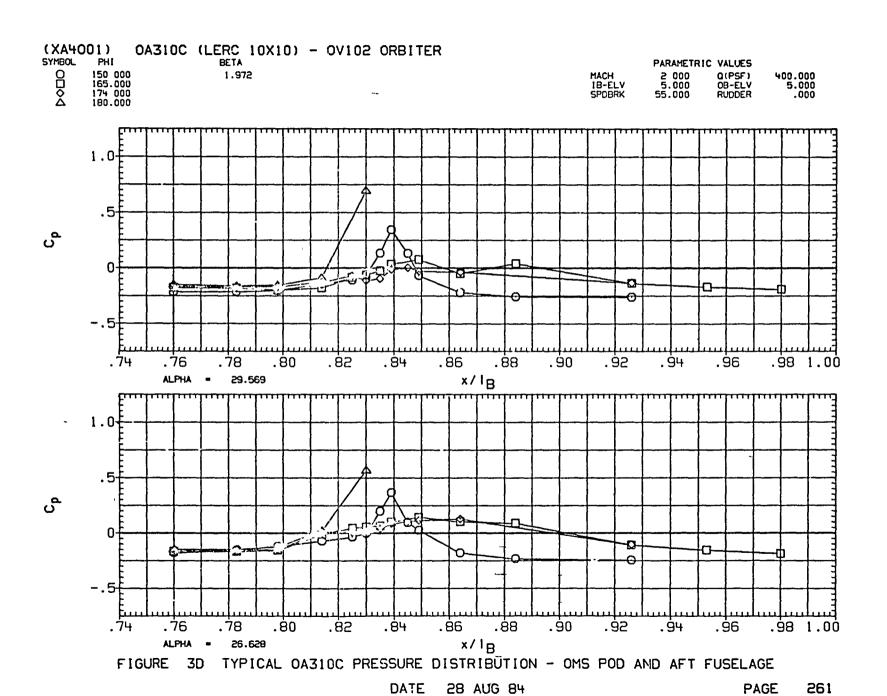
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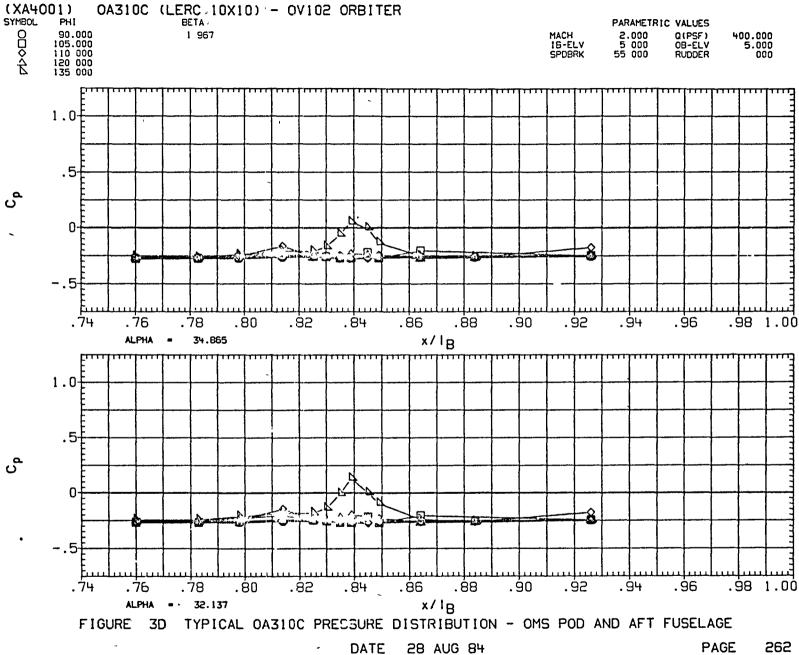
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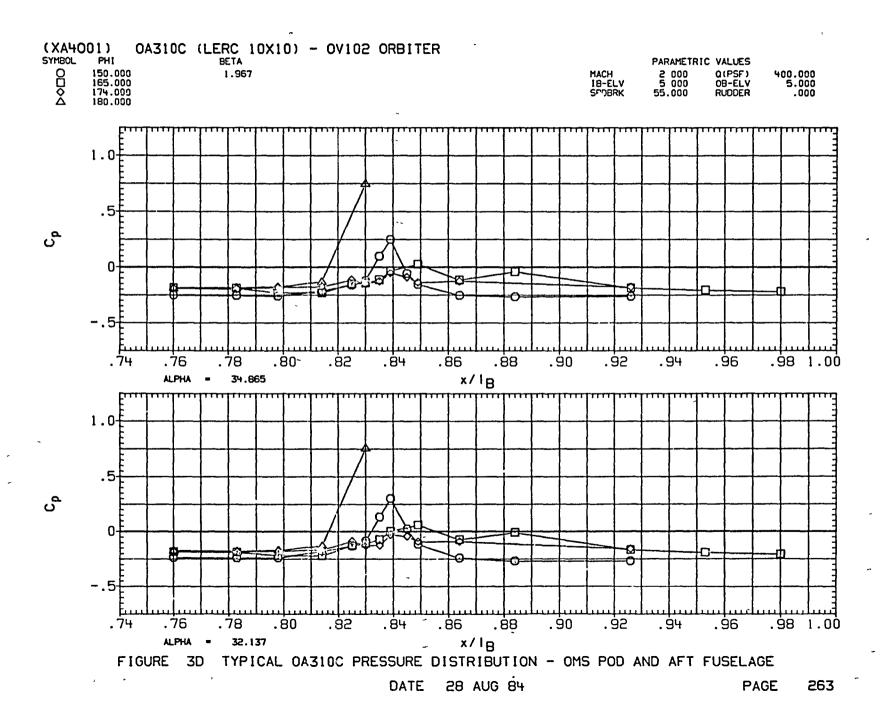
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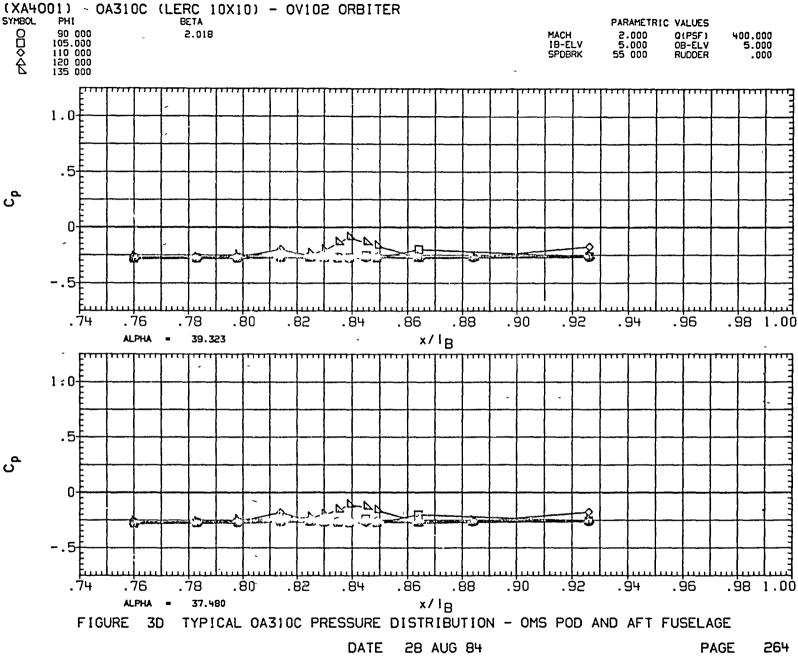


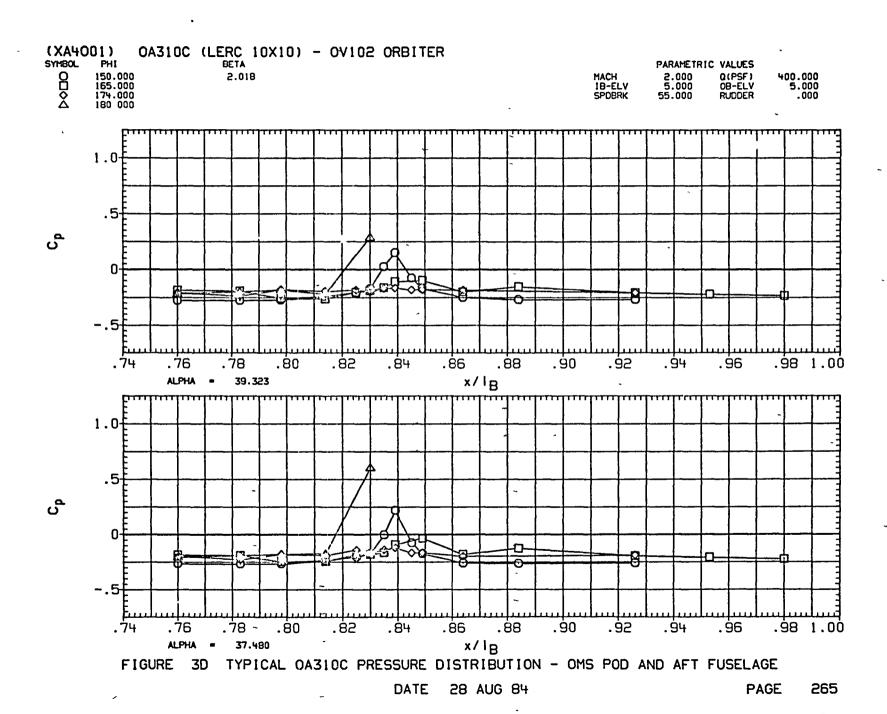


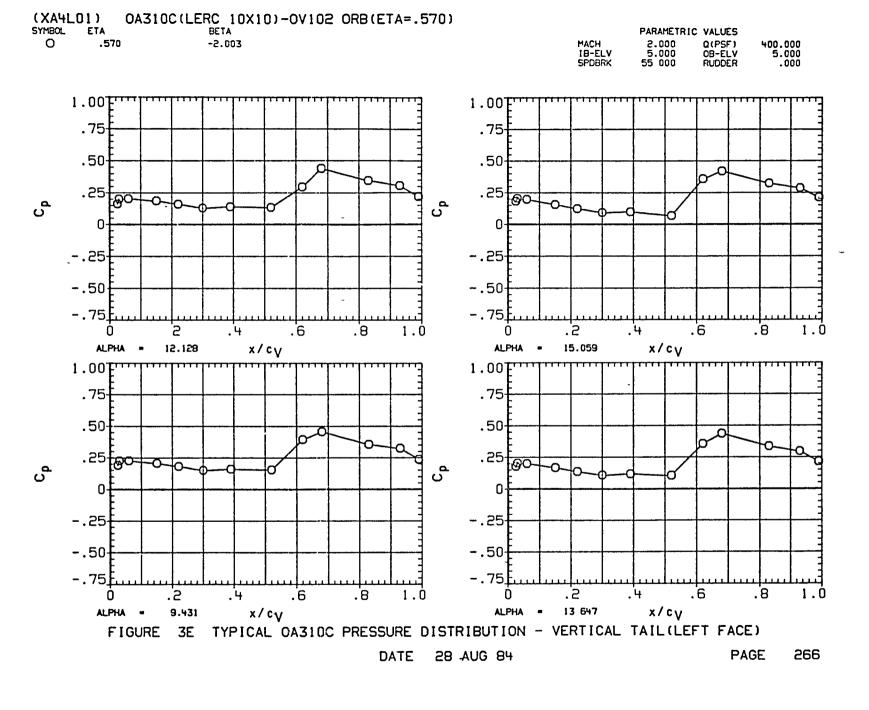


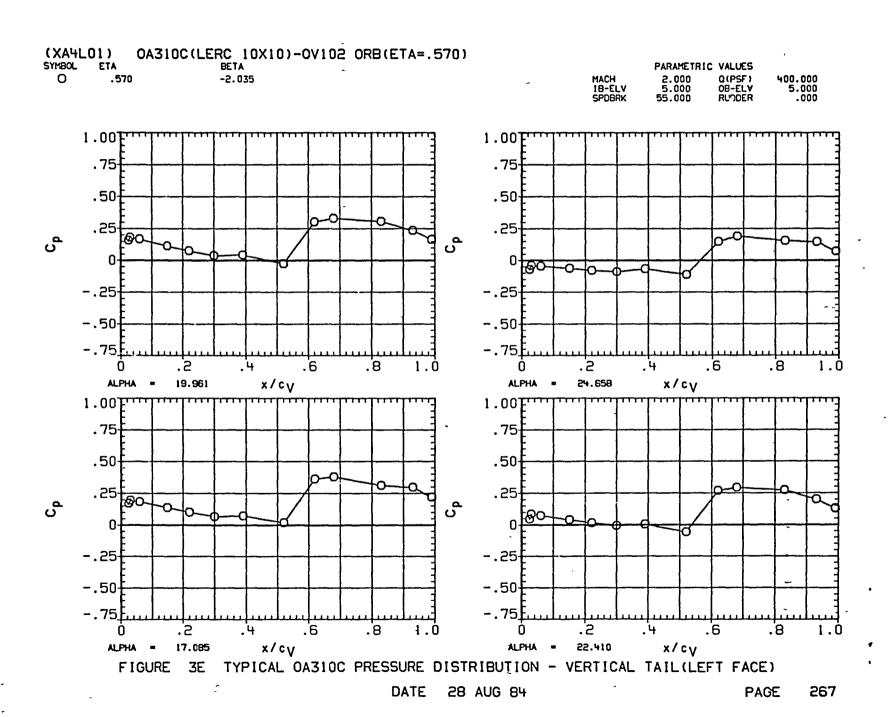


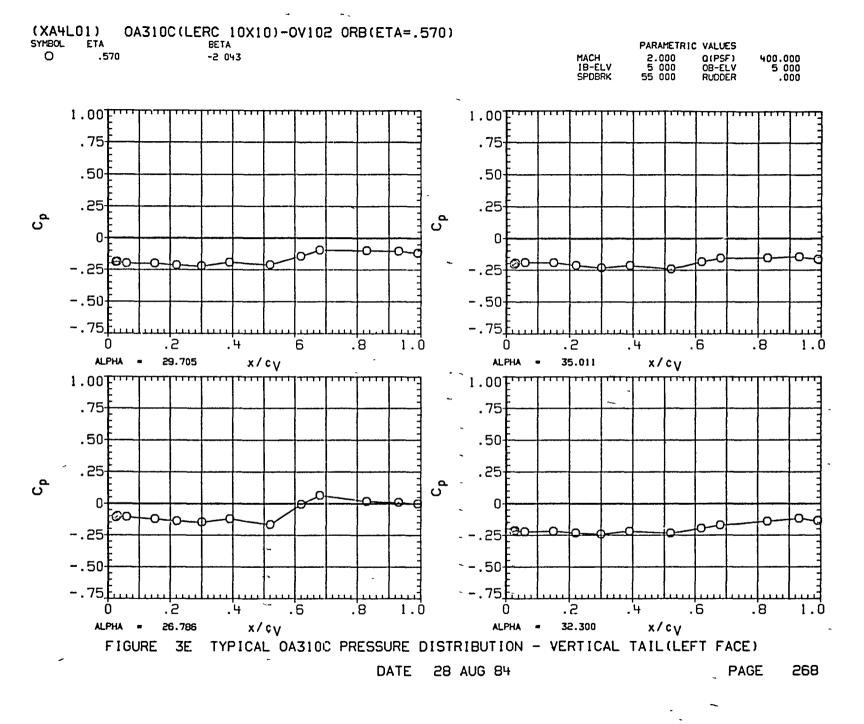












(XA4L01) OA310C(LERC 10X10)-OV102 ORB(ETA=.570)
SYMBOL ETA
O .570 -1.977

PARAMETRIC VALUES

MACH 2.000 Q(PSF) 400.000

1B-ELV 5.000 OB-ELV 5.000

SPDBRK 55.000 RUDDER .000

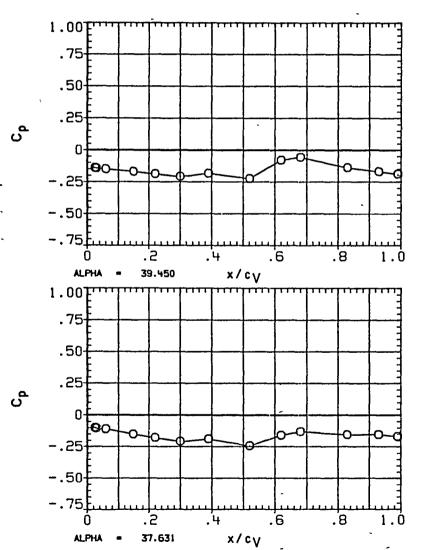
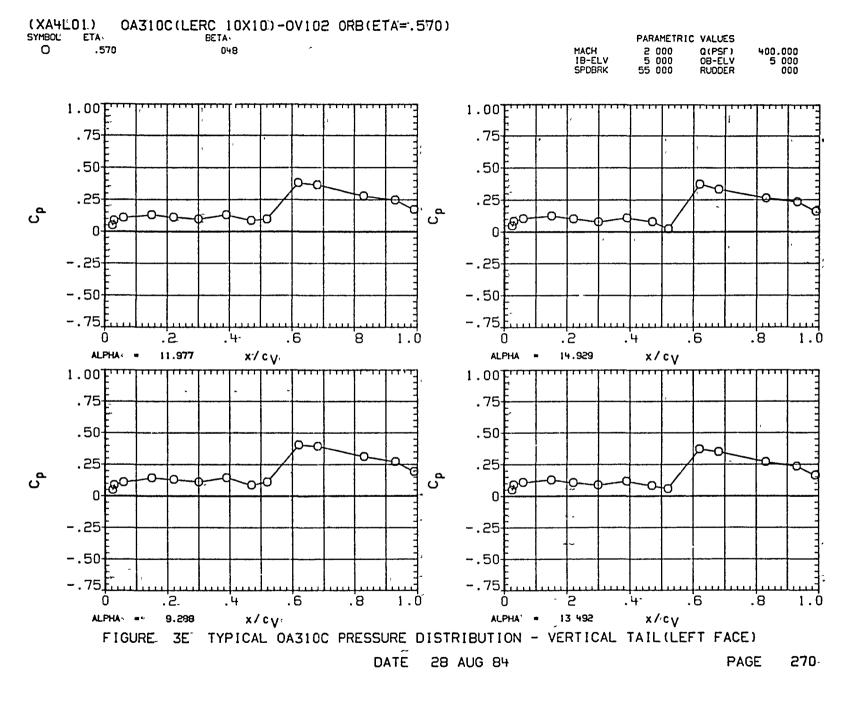
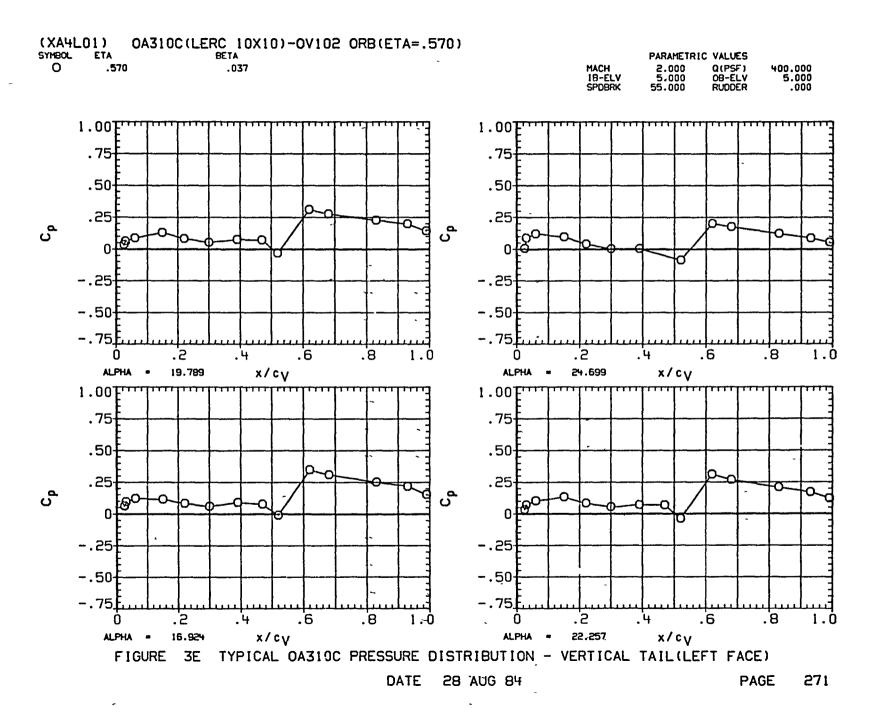
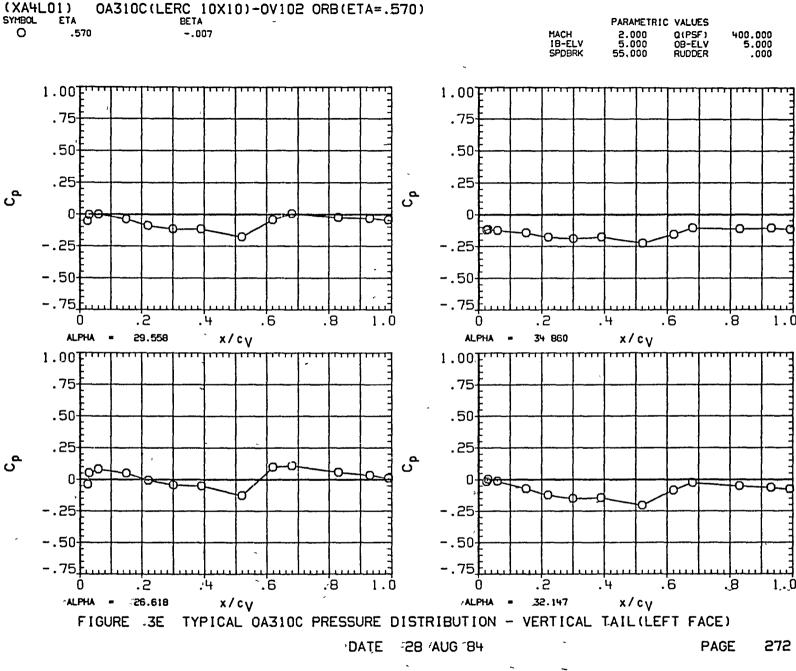


FIGURE 3E TYPICAL OA310C PRESSURÉ DISTRIBUTION - VERTICAL TAIL (LEFT FACE)







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(XA4L01) OA310C(LERC 10X10)-OV102 ORB(ETA=.570)
SYMBOL ETA
O .570 .025

PARAMETRIC VALUES

MACH 2.000 Q(PSF) 400.000

IB-ELV 5.000 08-ELV 5.000

SPDBRK 55.000 RUDDER .000

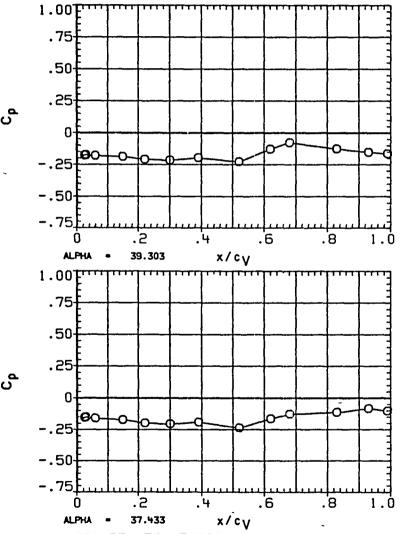
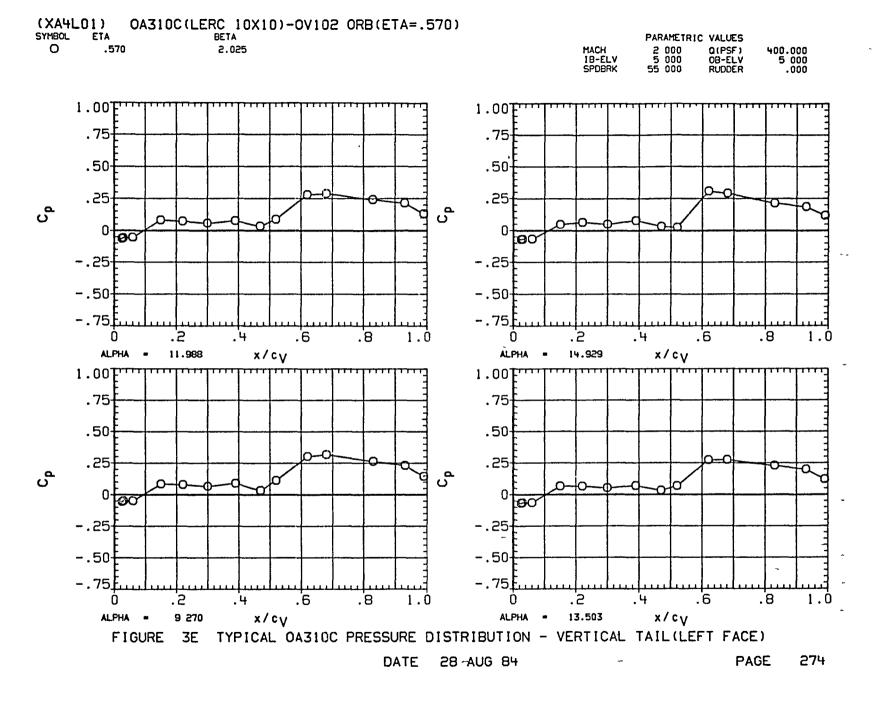
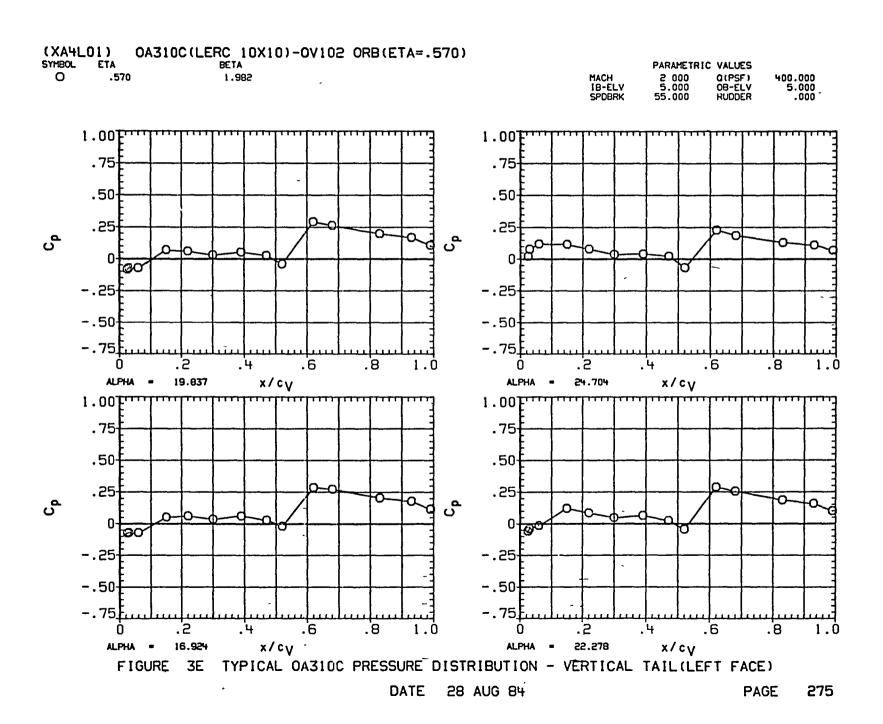
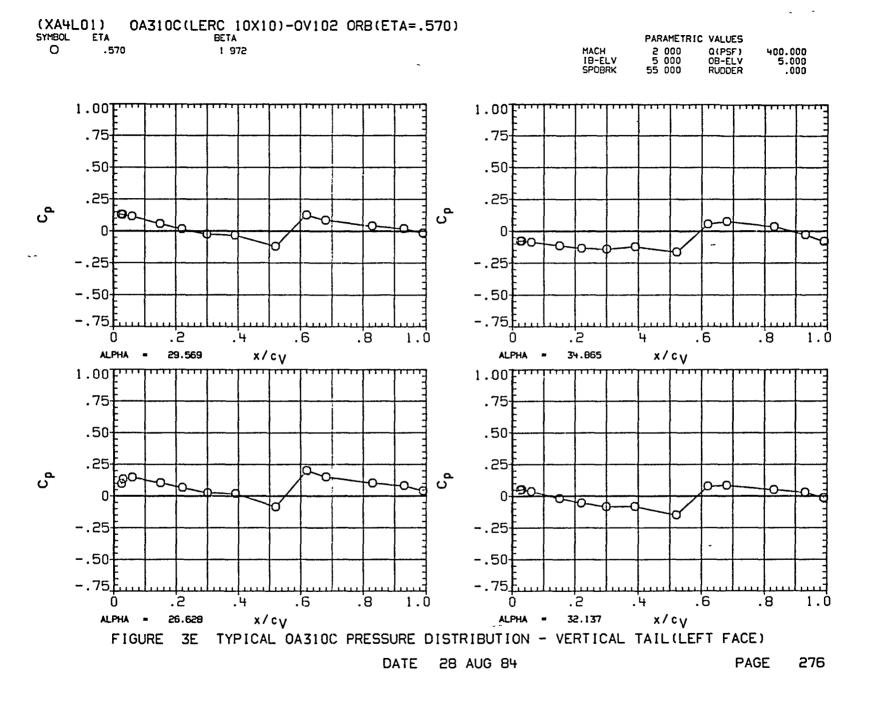


FIGURE 3E TYPICAL 0A310C PRESSURE DISTRIBUTION - VERTICAL TAIL(LEFT, FACE)







(XA4L01) OA310C(LERC 10X10)-OV102 ORB(ETA=.570)
SYMBOL ETA
O .570 BETA
2.018

PARAMETRIC VALUES

MACH 2.000 Q(PSF) 400.000

1B-ELV 5 000 0B-ELV 5.000

SPDBRK 55 000 RUDDER .000

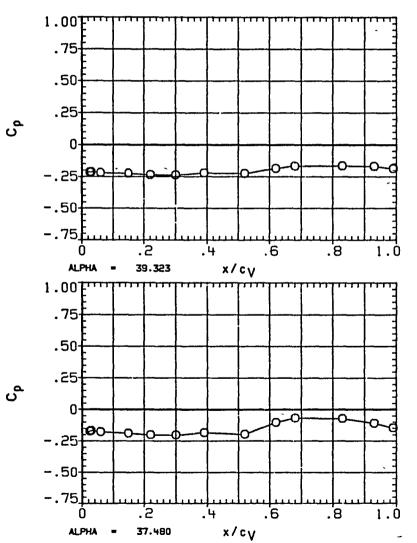
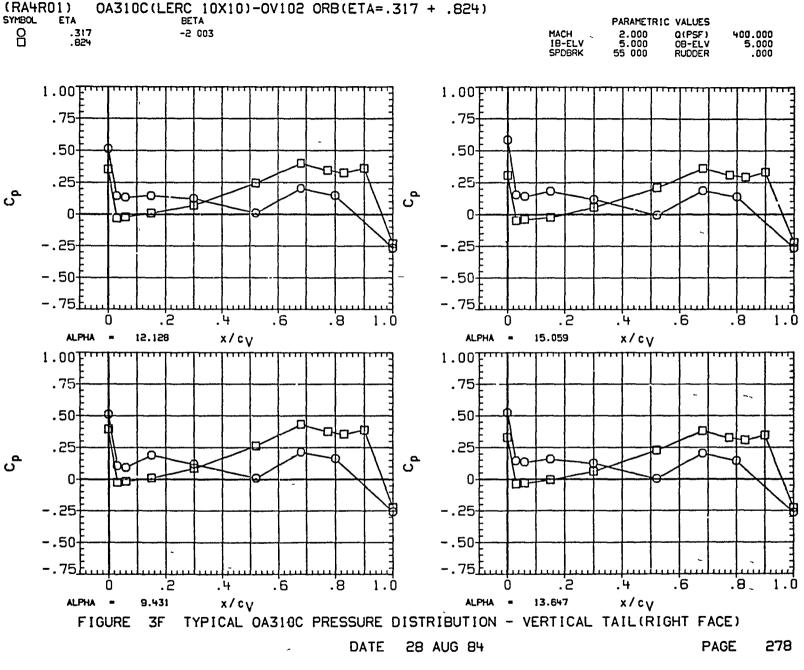
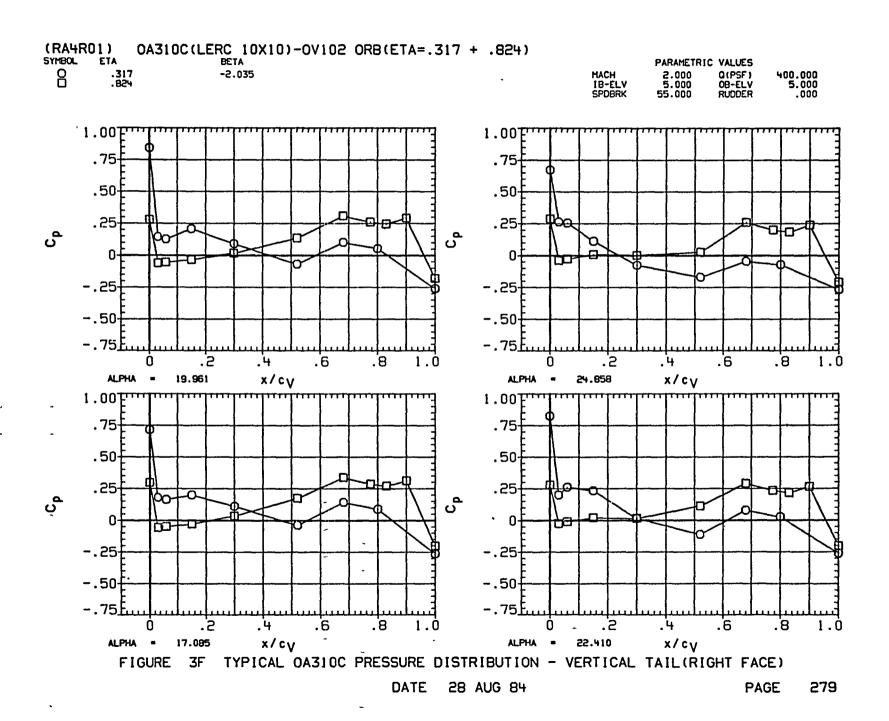
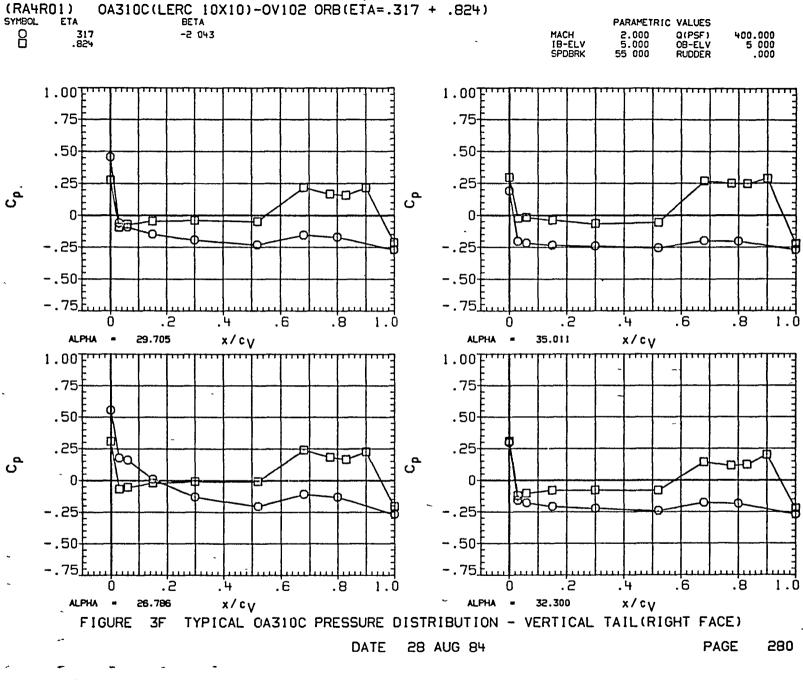


FIGURE 3E TYPICAL 0A310C PRESSURE DISTRIBUTION - VERFICAL TAIL (LEFT FACE)

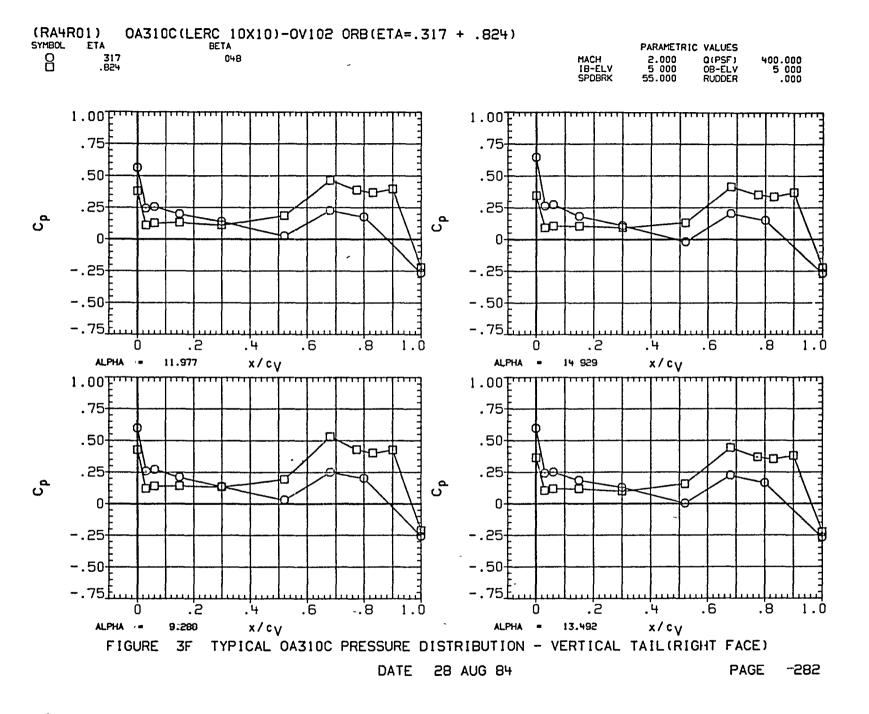


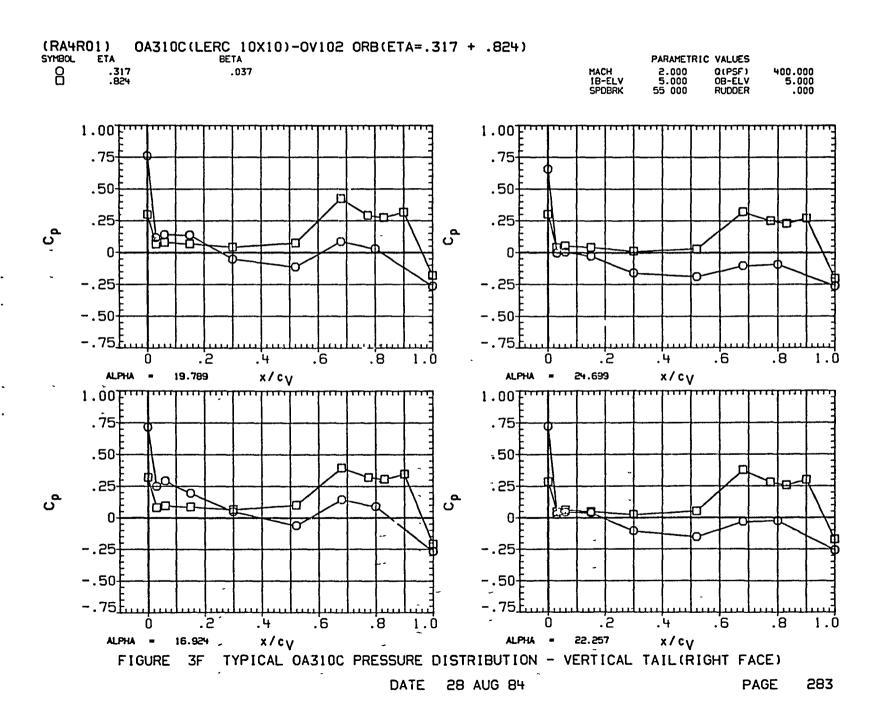


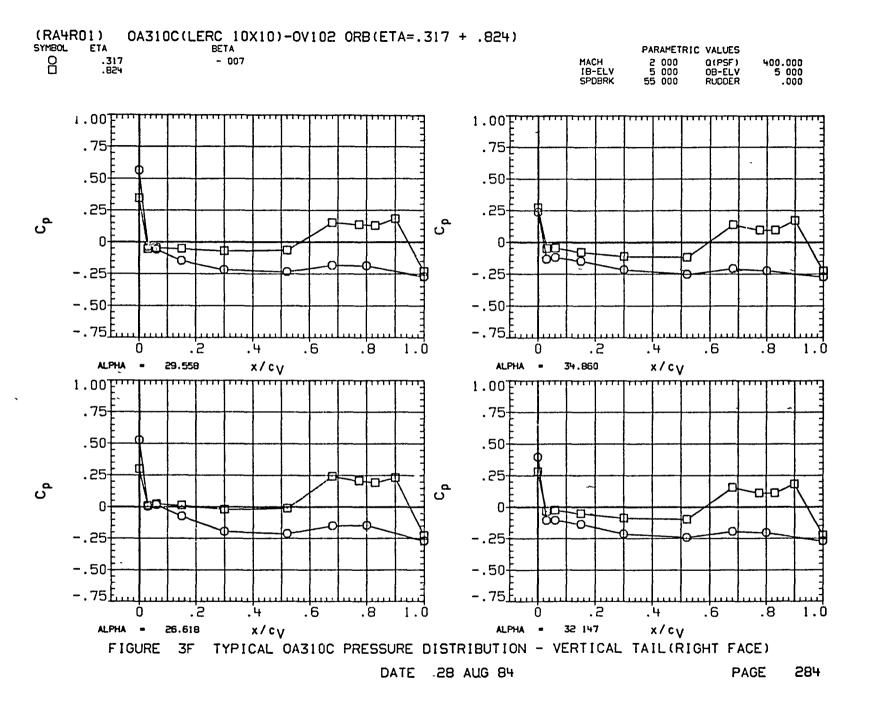


(RA4R01) OA310C(LERC 10X10)-OV102 ORB(ETA=.317 + .824) SYMBOL ETA PARAMETRIC VALUES 8 317 .824 Q(PSF) OB-ELV RUDOER -1.977 2.000 5.000 55.000 MACH 18-ELV SPDBRK 400.000 5.000 .000 1.007 .75 .50 .25 0--.50 -.75<u>F.</u> .'n .4 . 6 .'8 1.0 39.450 ALPHA = x/c_V 1.00FT .75 .50 0--.25 -.50 -.75<u>F</u> .2 .'6 .8 .'4 37.631 x/cy ALPHA = TYPICAL 04310C PRESSURE DISTRIBUTION - VERTICAL TAIL (RIGHT FACE) DATE 28 AUG 84 PAGE

281







400.000 5.000 .000

285

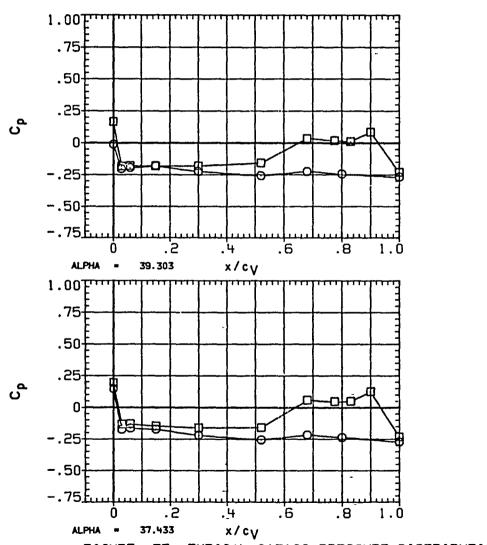
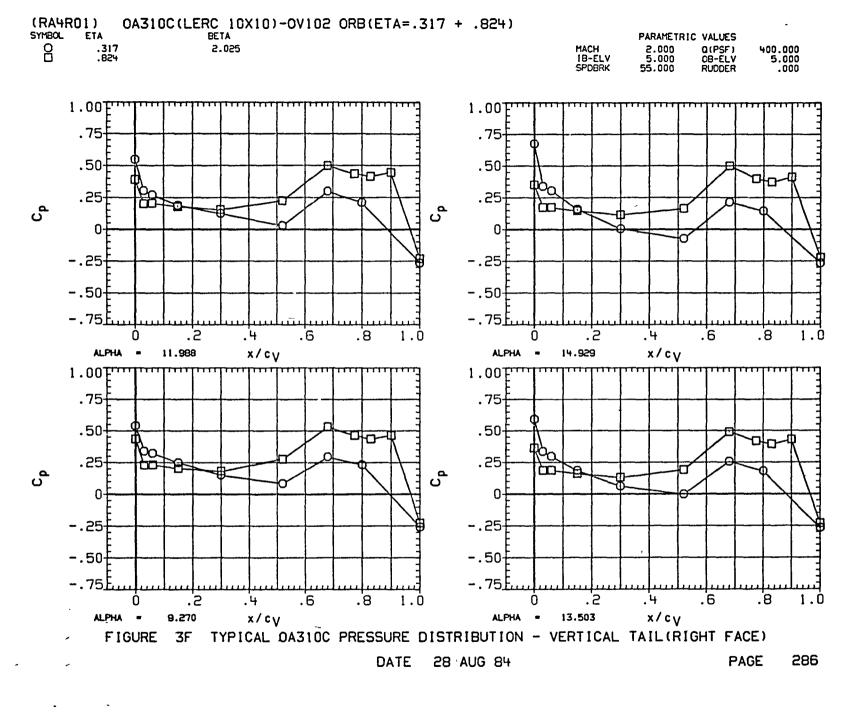
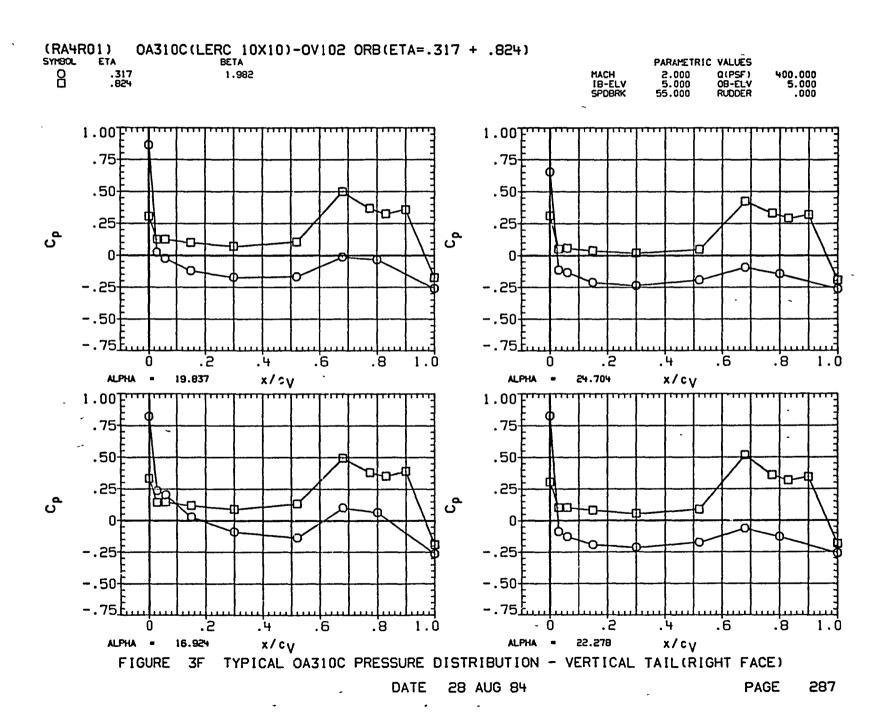
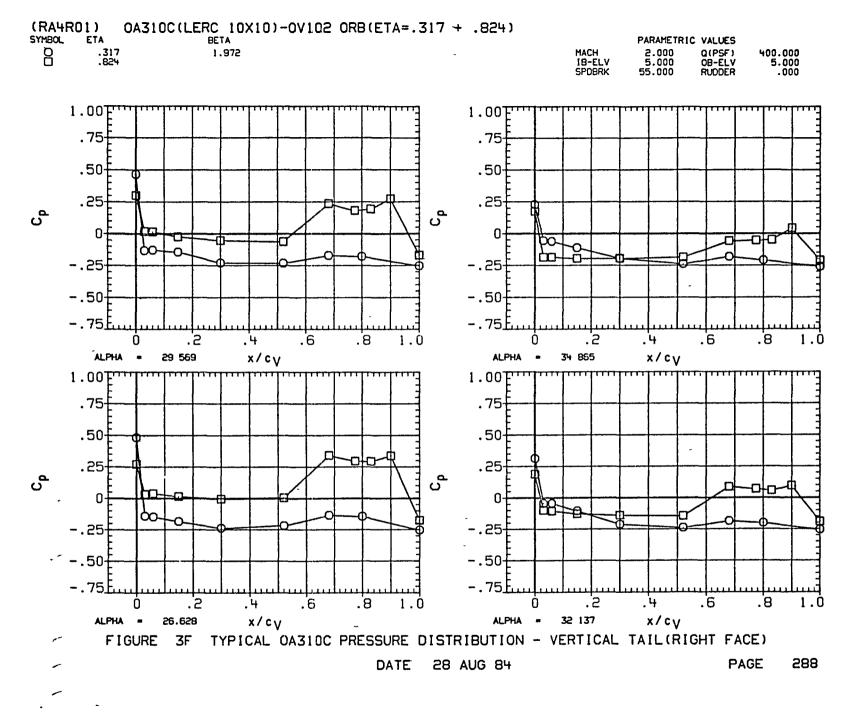


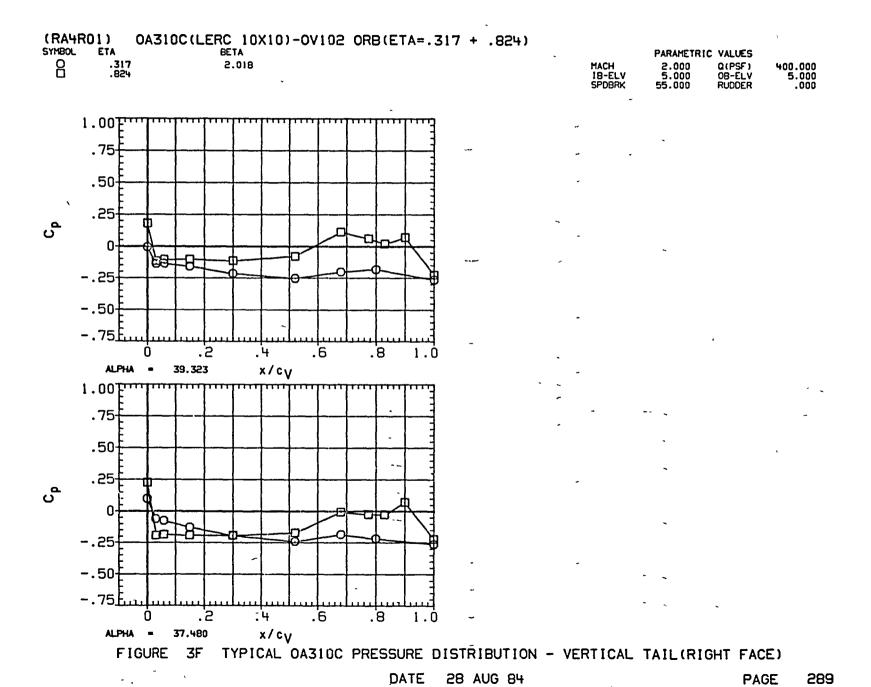
FIGURE 3F TYPICAL 0A310C PRESSURE DISTRIBUTION - VERTICAL TAIL(RIGHT FACE)

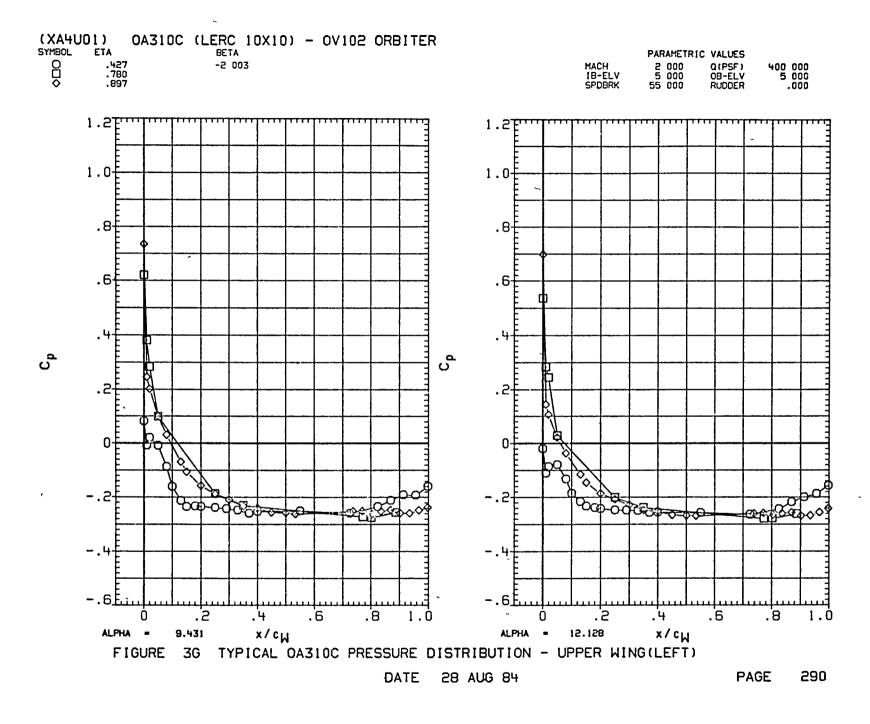
DATE 28 AUG 84 PAGE

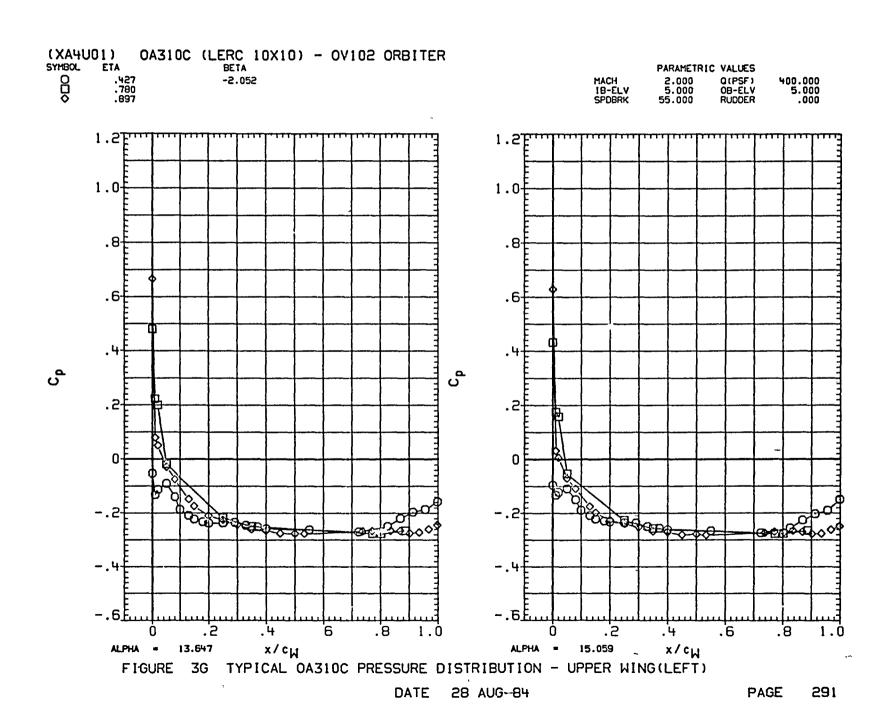


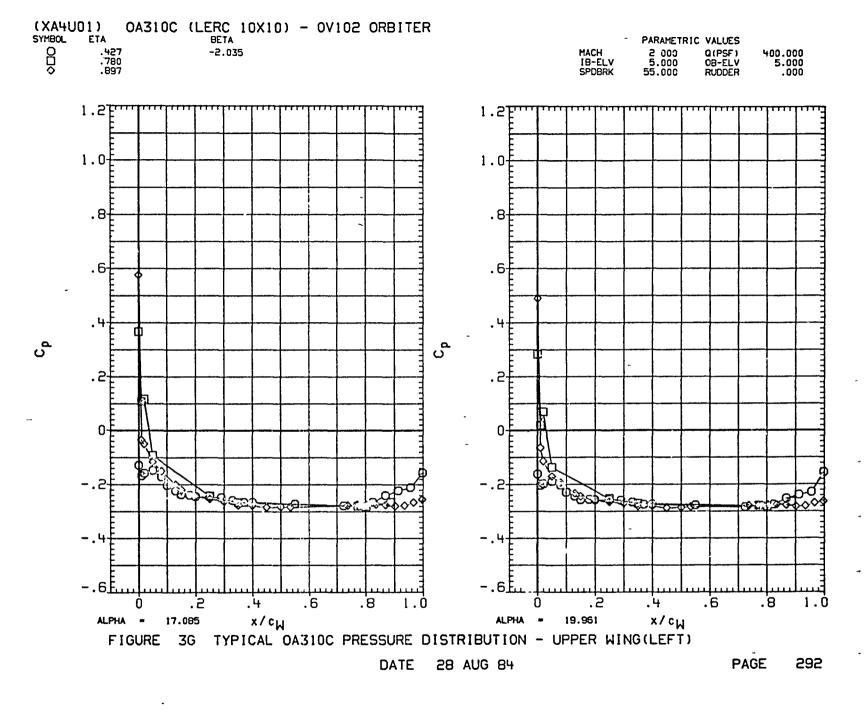




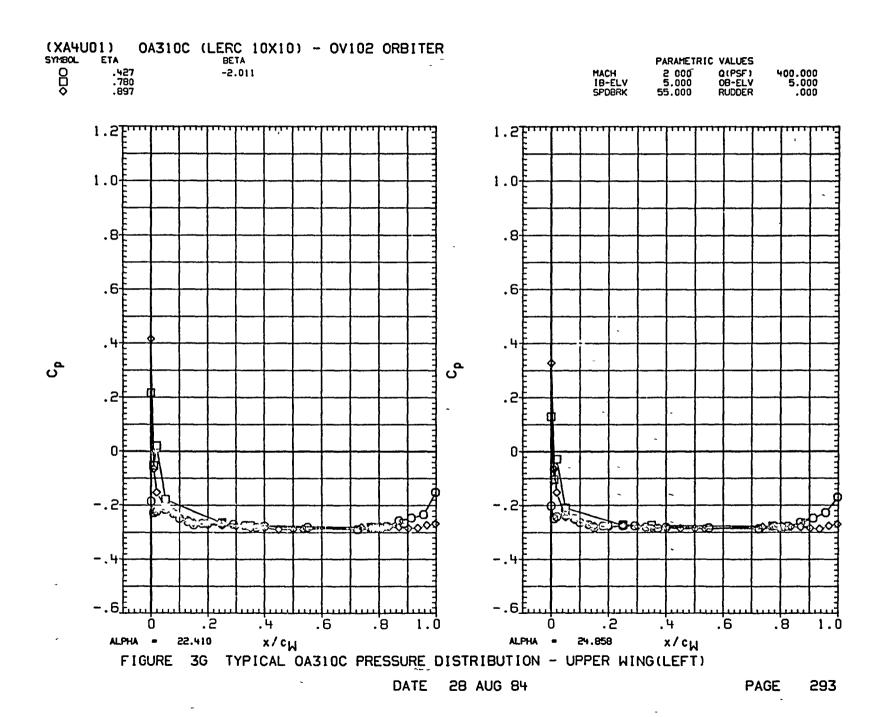


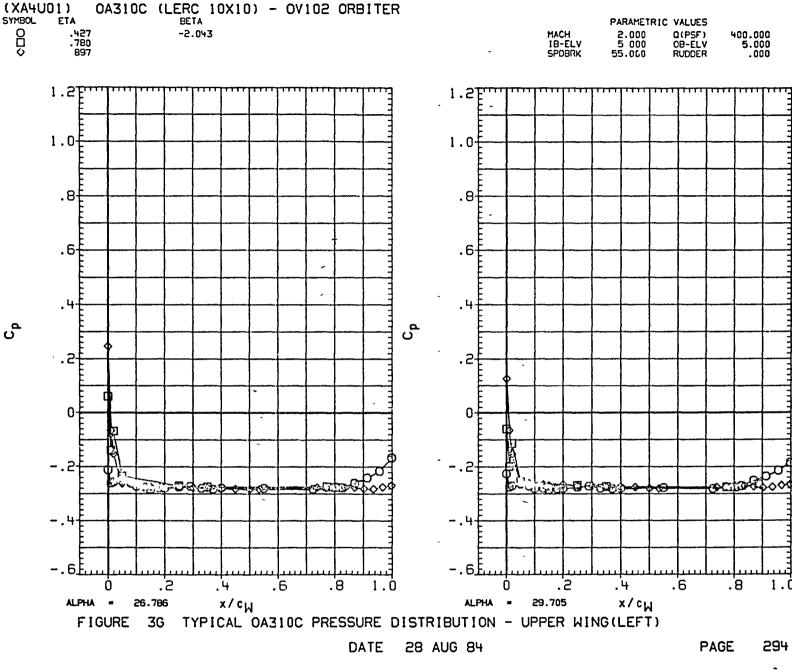


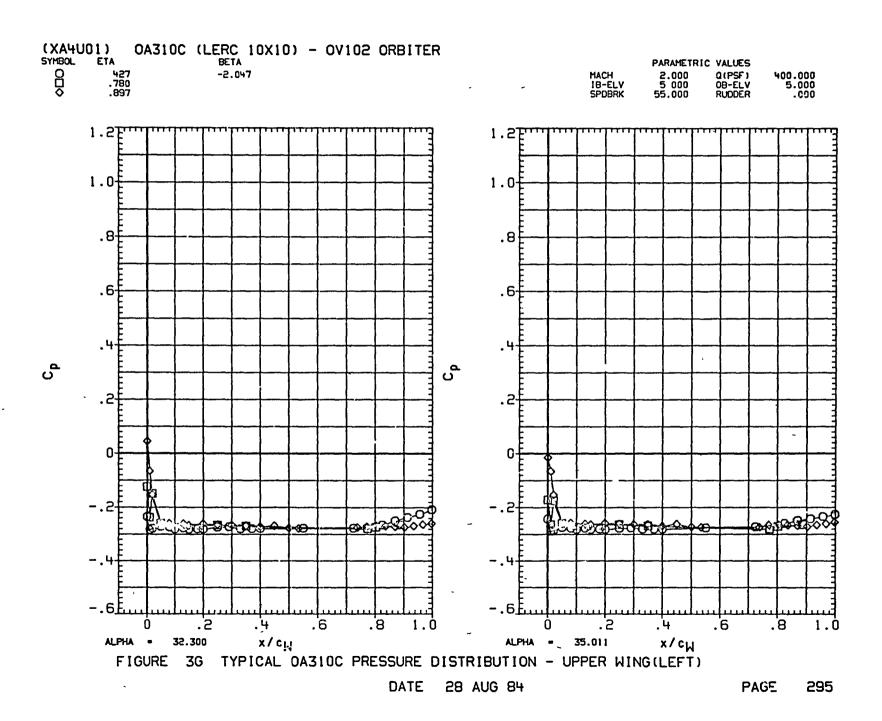


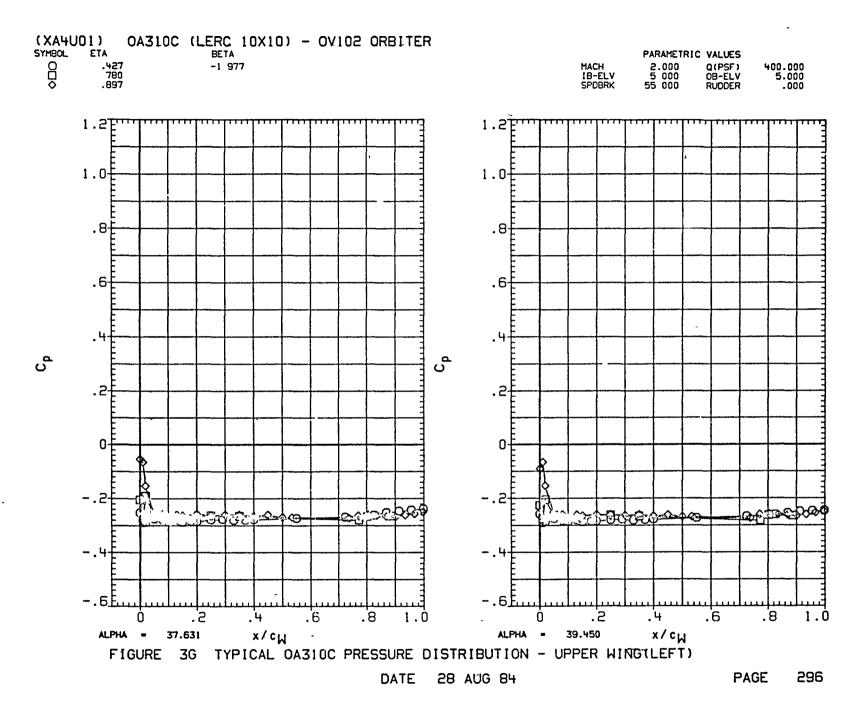


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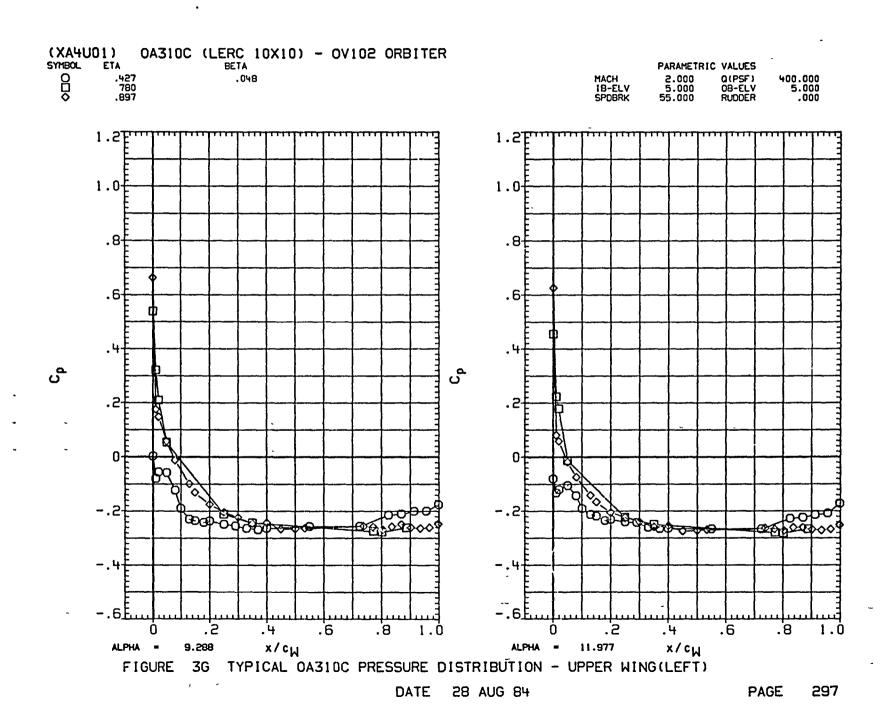


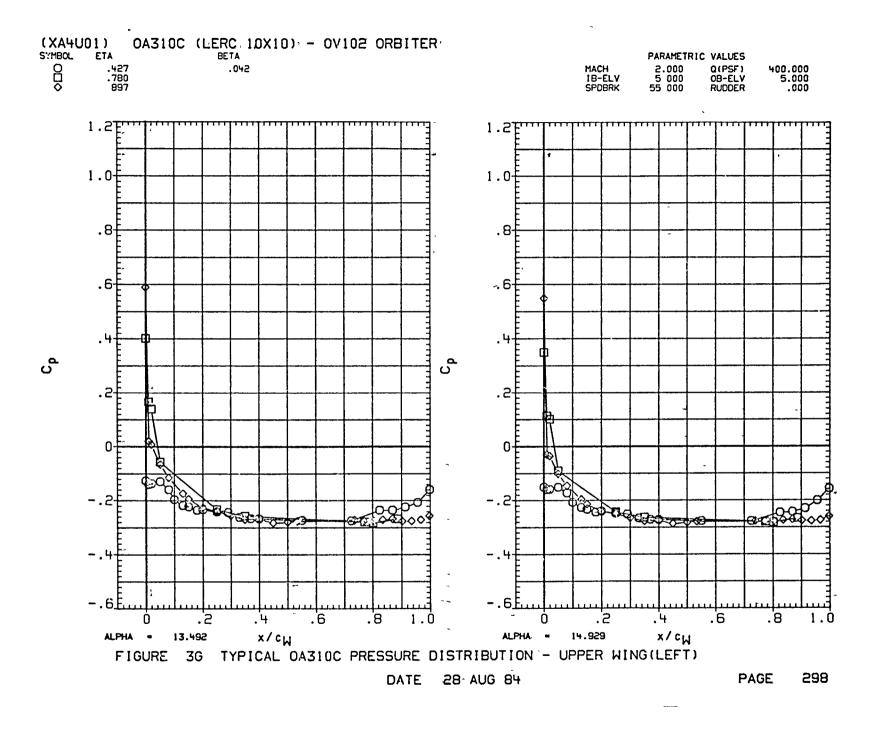


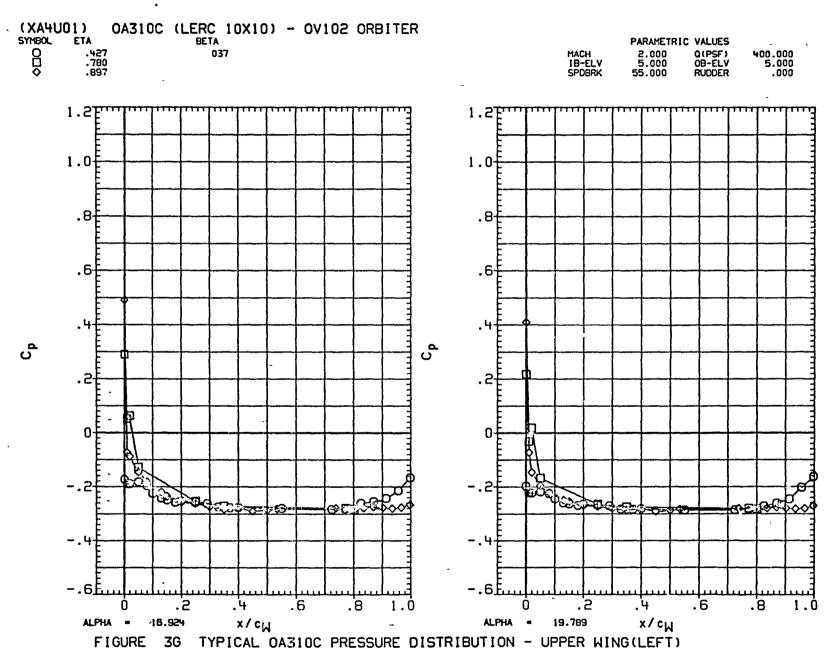




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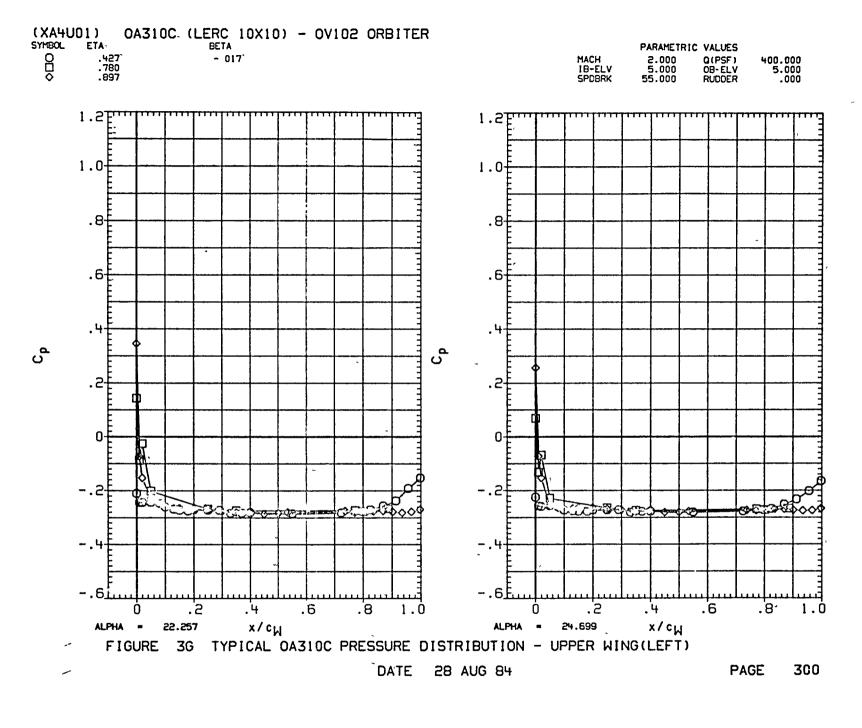


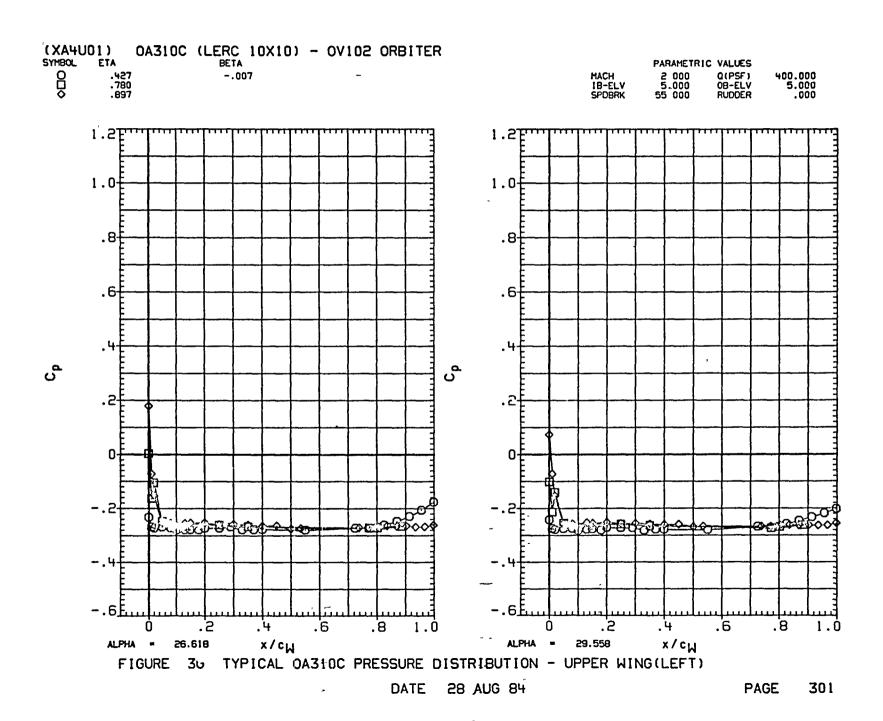


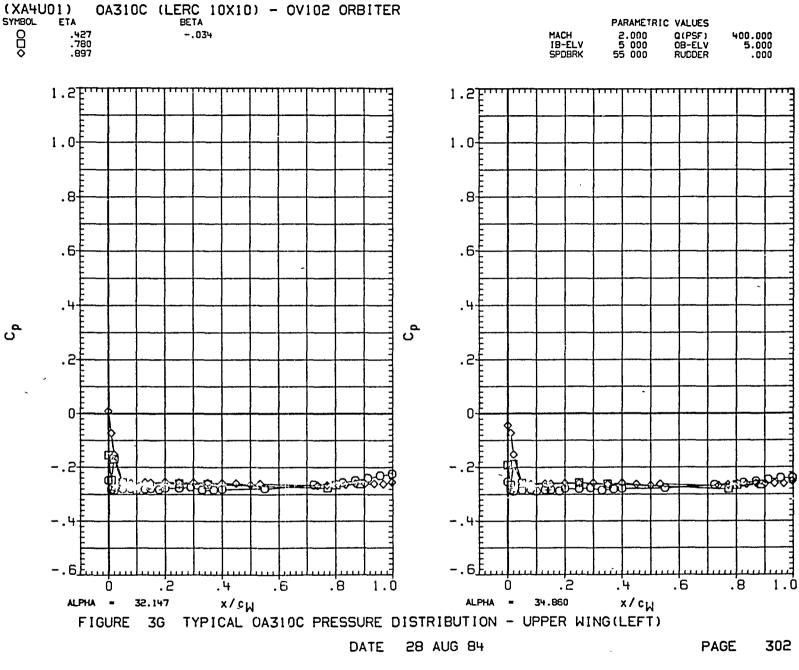


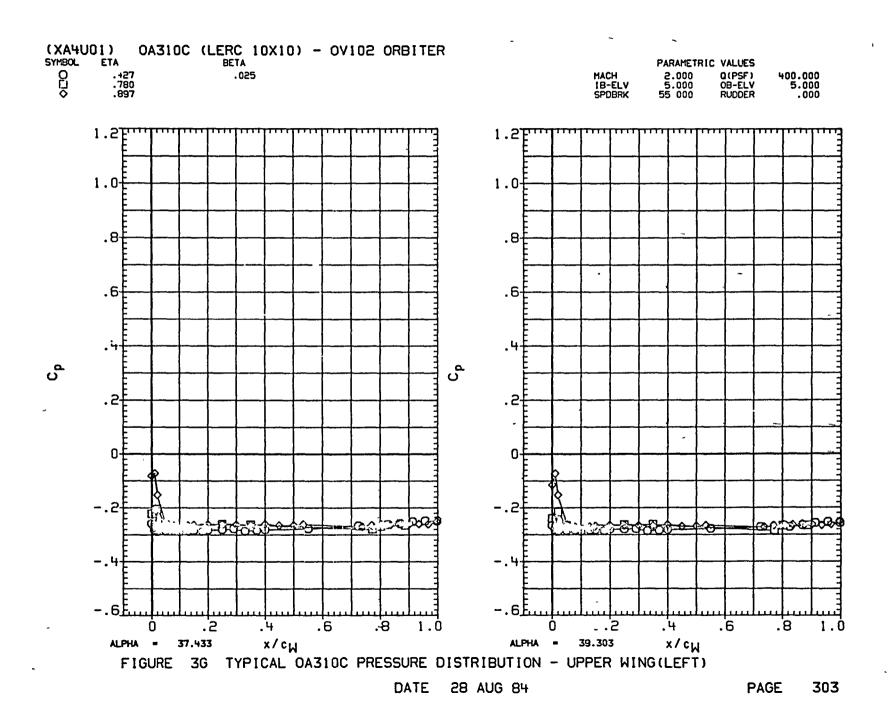
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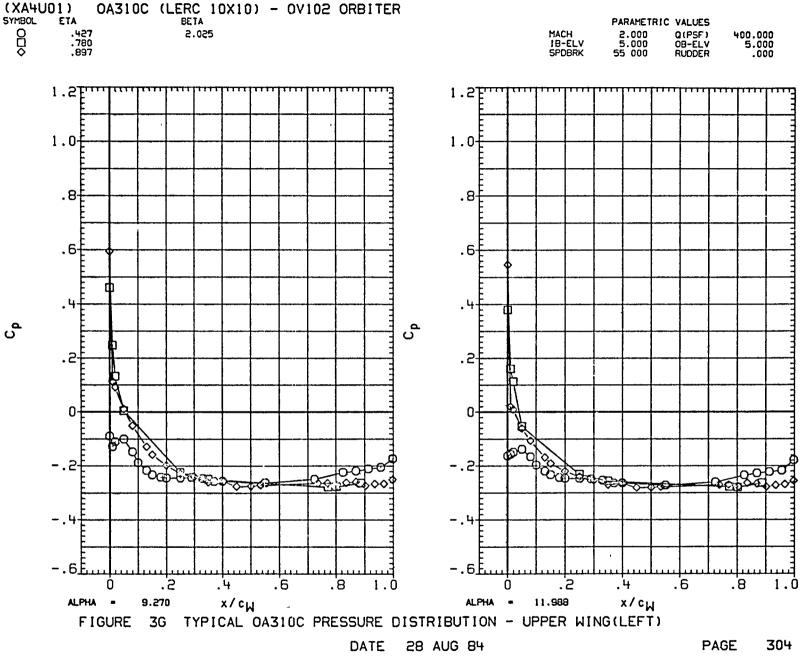
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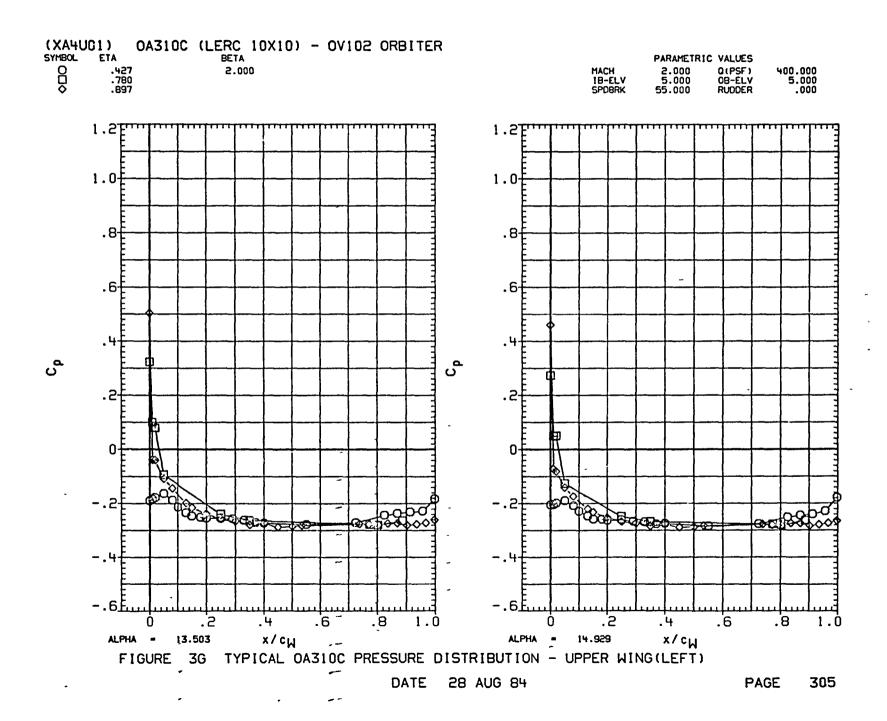


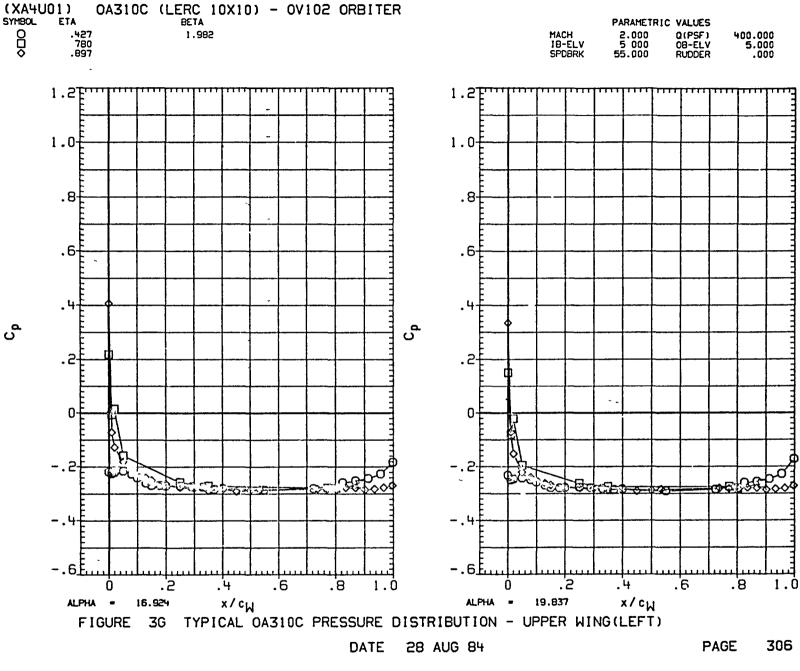


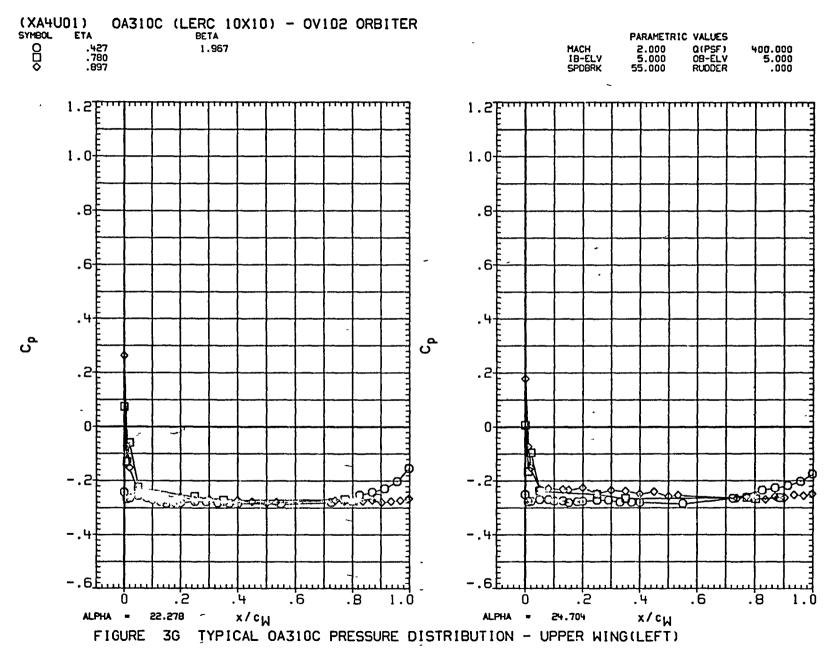








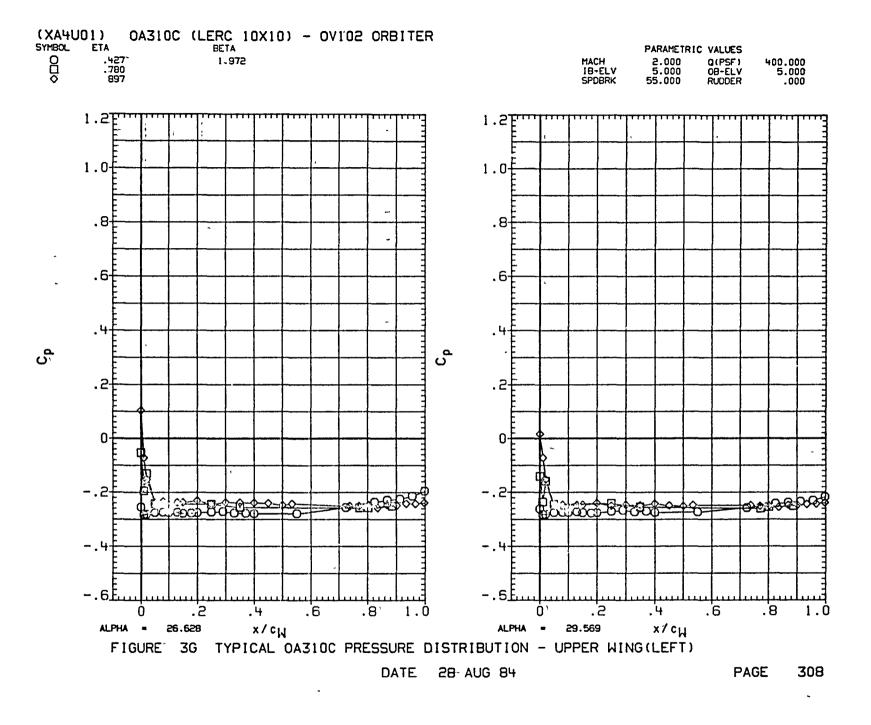


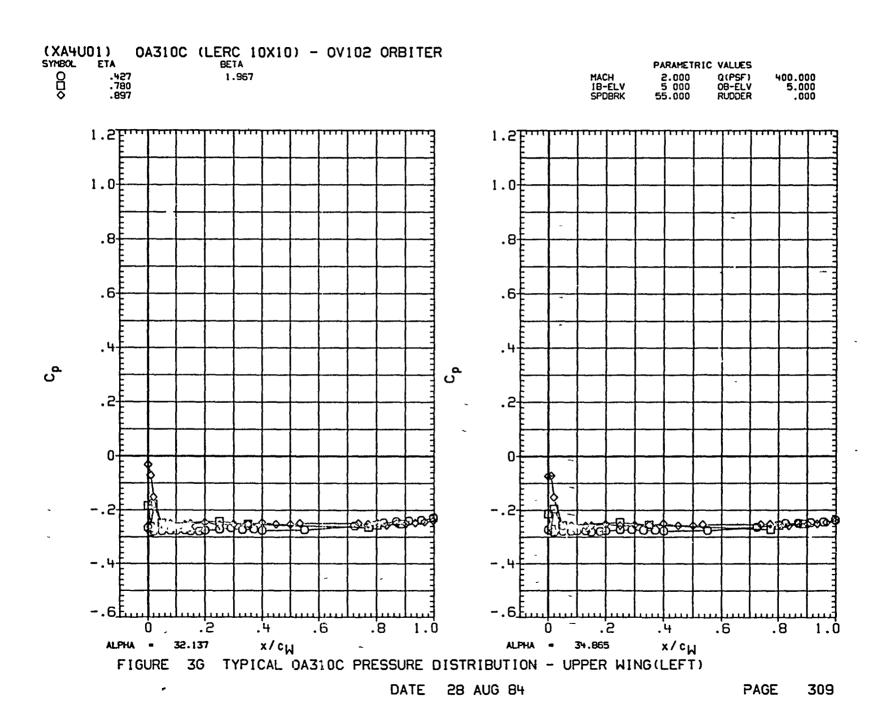


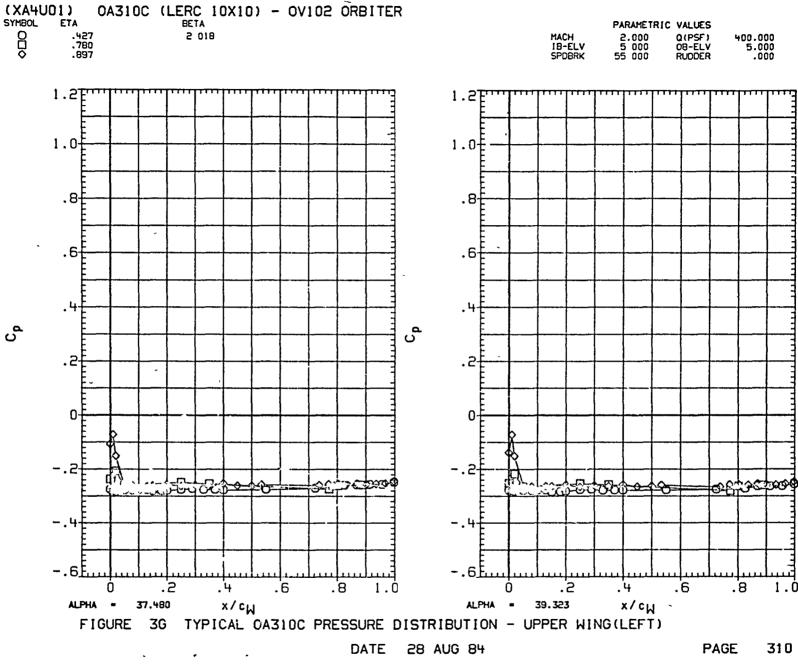
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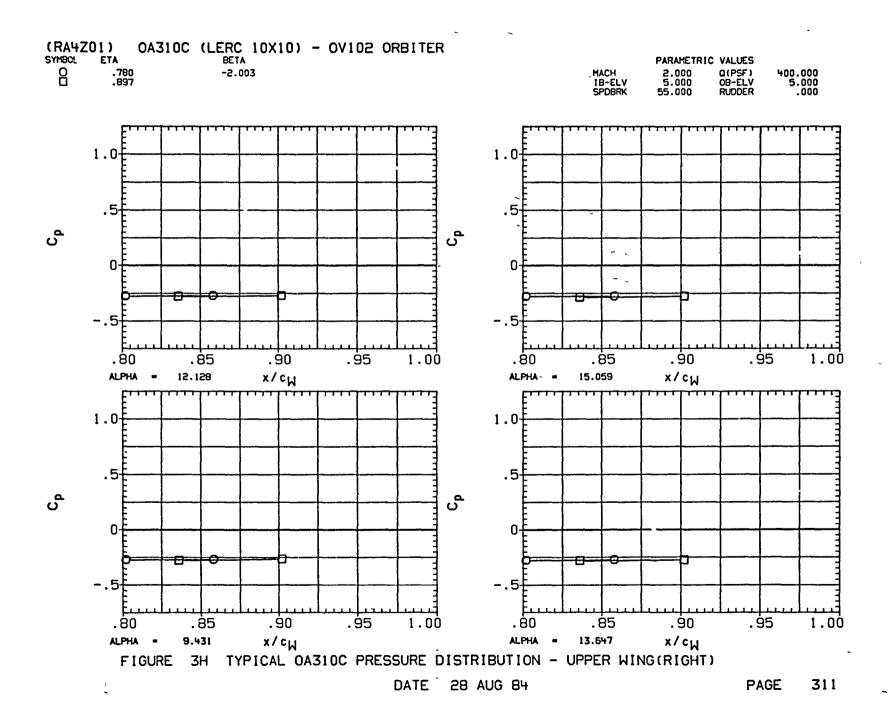
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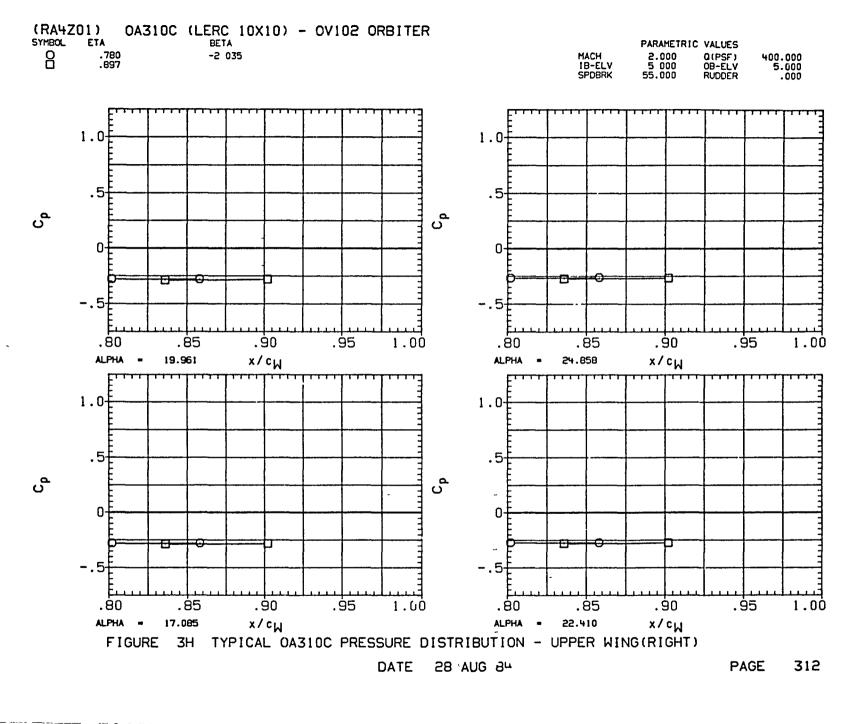
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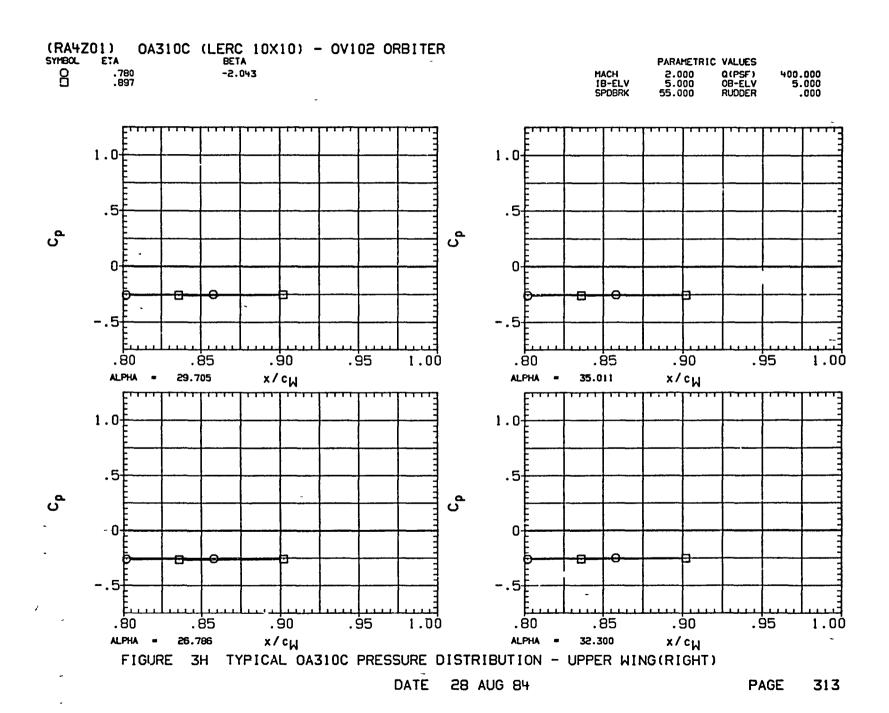


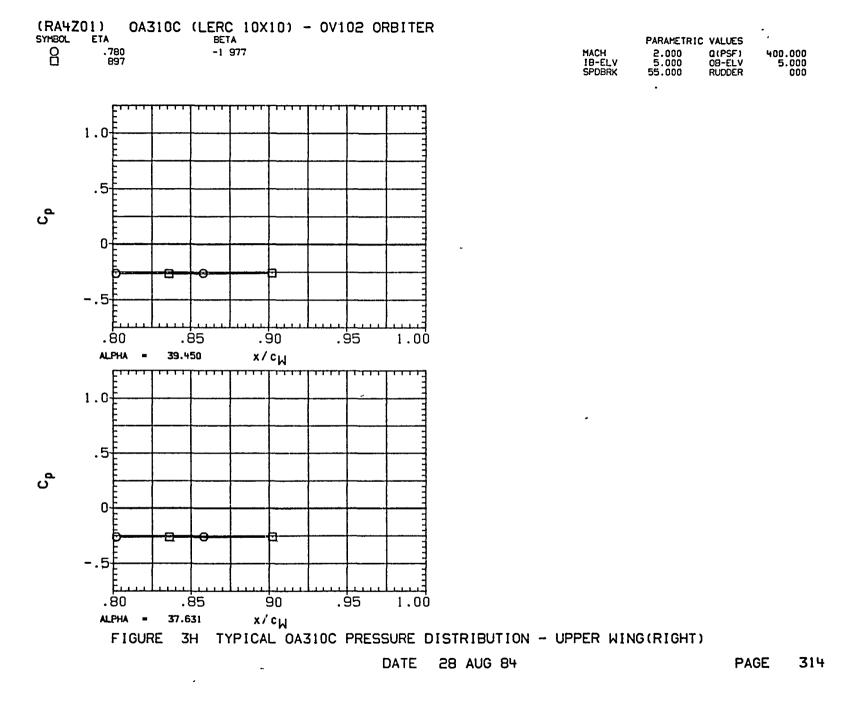


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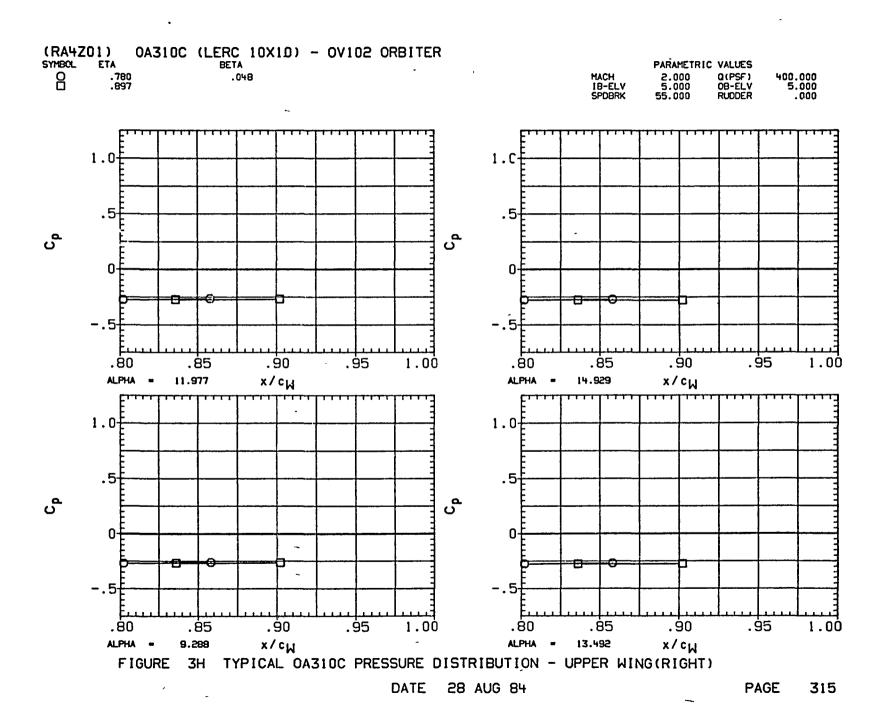
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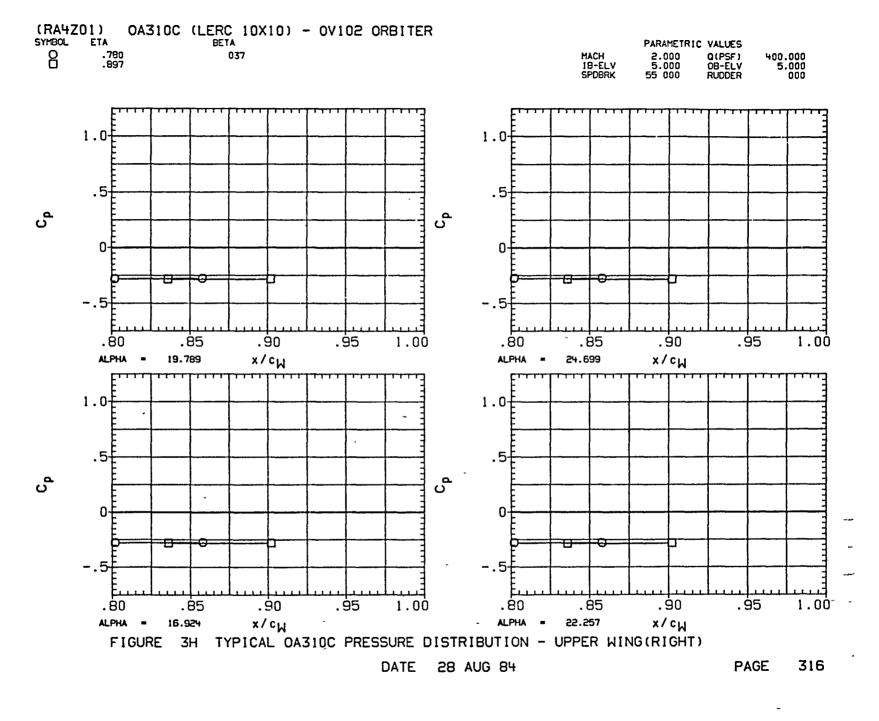


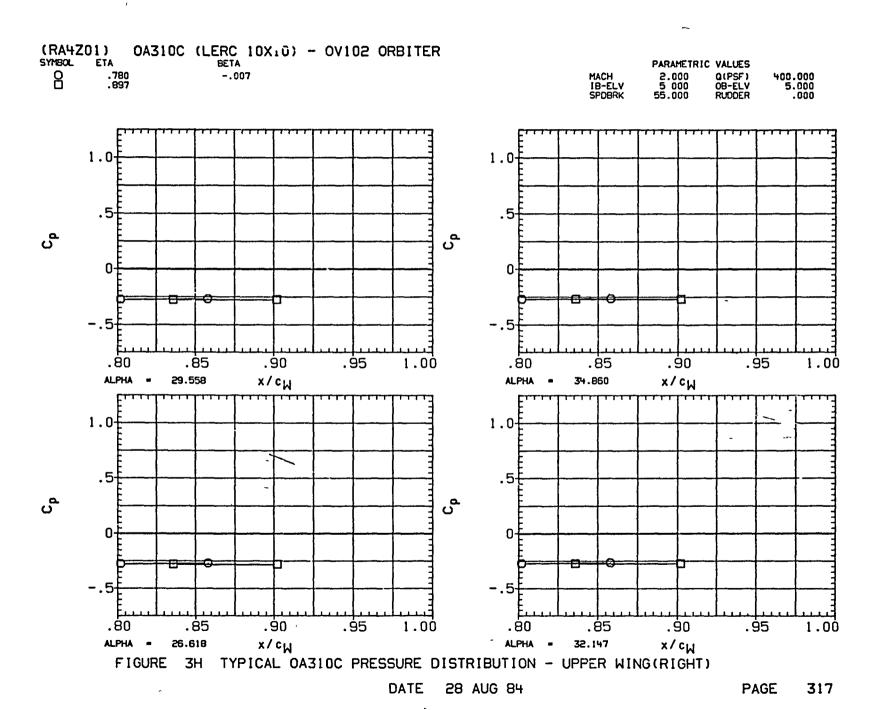


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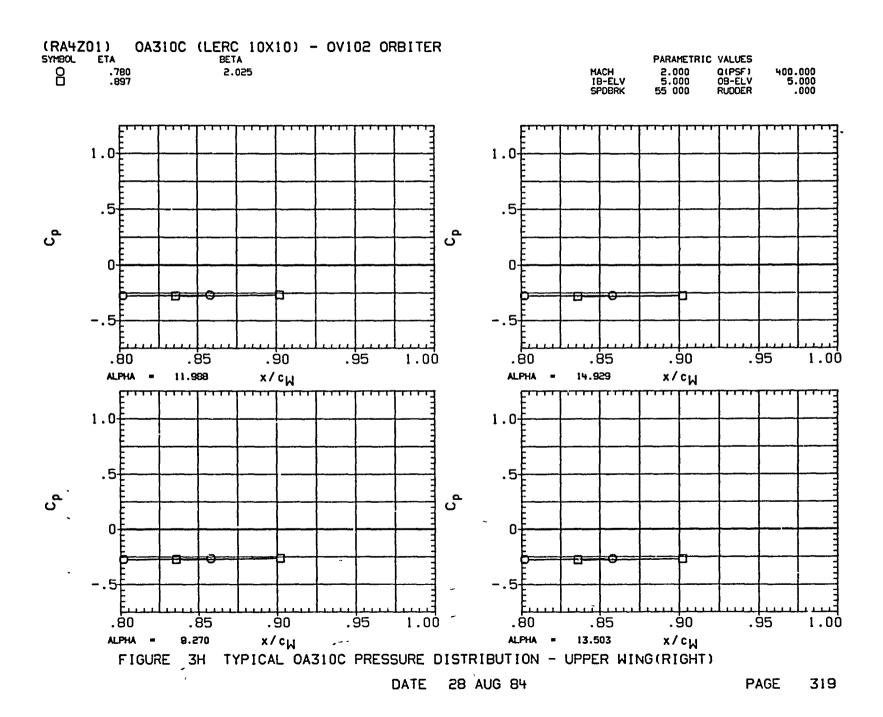
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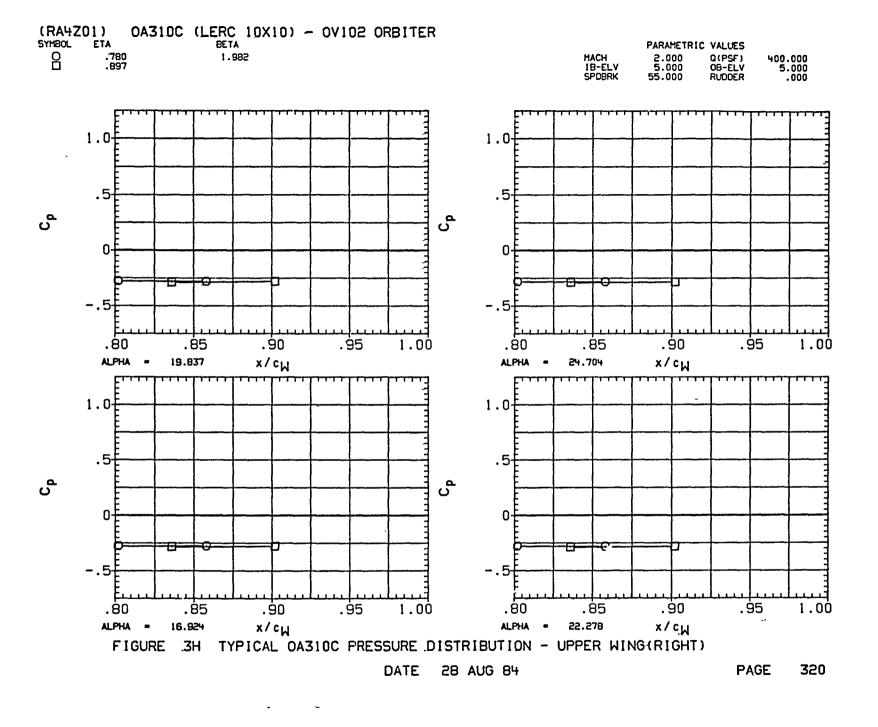


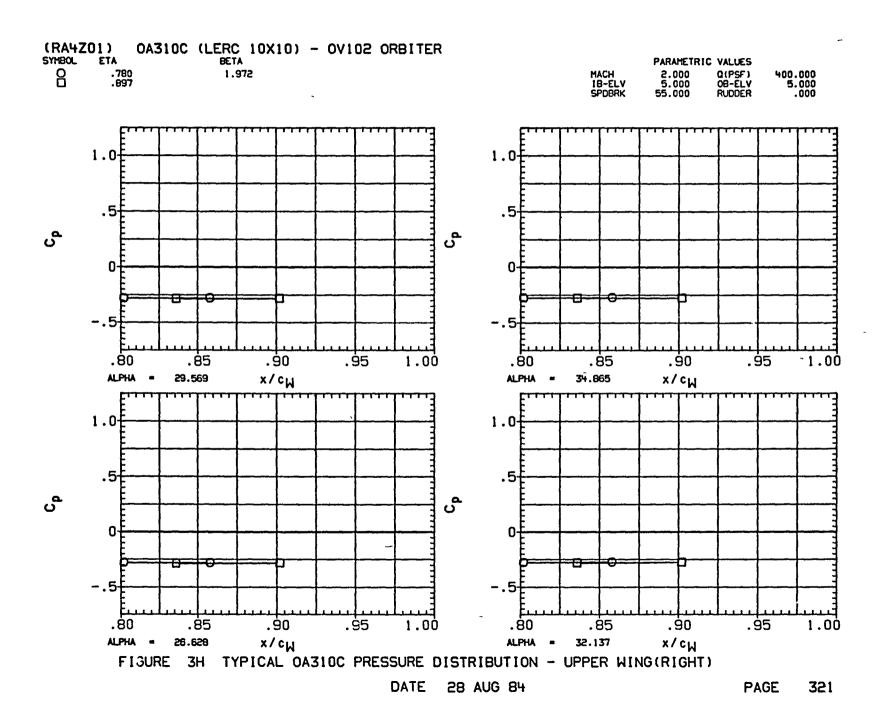


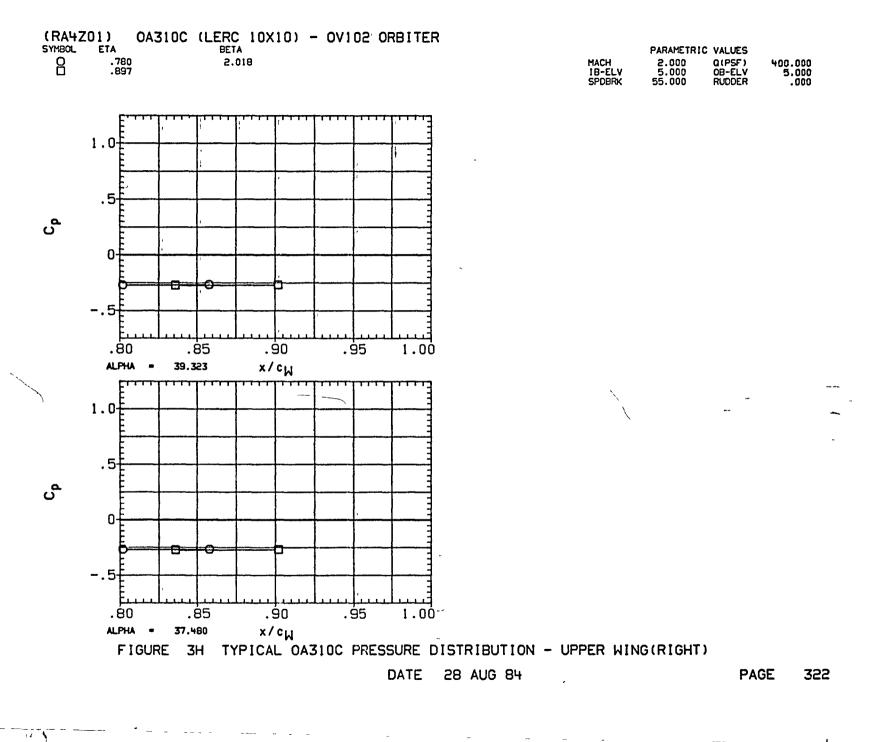
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